NASA Contractor Report 177982

AERODYNAMIC CHARACTERISTICS OF A PROPULSIVE WING/CANARD CONCEPT AT STOL SPEEDS

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V. R. Stewart

Rockwell International Corporation
Columbus, Ohio 43216

Contract NAS1-17171
November 1985

National Aeronautics and
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Langley Research Center
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ABSTRACT

A full span model of a wing/canard concept representing a fighter configuration has been tested at STOL conditions in the NASA Langley 4 by 7 Meter Tunnel. The results of this test are presented, and comparisons are made to previous data of the same configuration tested as a semispan model. The potential of the propulsive wing/canard to develop very high lift coefficients was investigated with several nozzle spans (nozzle aspect ratios). Although longitudinal trim was not accomplished with the blowing distributions and configurations tested, the propulsive wing/canard appears to offer an approach to managing the large negative pitching moments associated with trailing edge flap blowing. Also presented are data showing the effects of large flap deflections and relative wing/canard positions. Presented in the appendix to the report are limited lateral-directional and ground effects data, as well as wing downwash measurements.

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SYMBOLS

	A	~	Aspect Ratio ~ b^2/S
	AF	~	Aft Force - balance
	b	~	Span
	$b_{exposed}$	~	Exposed Span
	b_f	~	Flap Span
BN/B	b_j	~	Jet Span/Exposed Span
	BP	~	Butt Plane (B.P. = 0 at wing root)
	$\bar{c}$	~	Mean Aerodynamic Chord
CD	C_D	~	Drag Coefficient ~ $\frac{\text{Drag}}{qS}$
CDTR	$C_{D_{TR}}$	~	Thrust Removed Drag Coefficient
CL	C_L	~	Lift Coefficient ~ $\frac{\text{Lift}}{qS}$
CLTR	$C_{L_{TR}}$	~	Thrust Removed Lift Coefficient
CM	C_M	~	Pitching Moment Coefficient ~ $\frac{\text{Pitching Moment}}{qS\,\bar{c}}$
CMTR	$C_{M_{TR}}$	~	Thrust Removed Pitching Moment Coefficient
	C_p	~	Pressure Coefficient
CMU	C_μ	~	Blowing Coefficient ~ $\frac{\dot{m}\,V_j}{qS}$
CN	C_n	~	Yawing Moment Coefficient
CROLL	C_ℓ	~	Rolling Moment Coefficient
CY	C_Y	~	Side Force Coefficient
	FRL	~	Fuselage Reference Line
	H	~	Height of FRL above Ground
	K_b	~	Span Correlation Factor

SYMBOLS (Concluded)

	M	∿	Mach Number
	$\dot{m}$	∿	Nozzle Mass Flow
	NF	∿	Normal Force - Balance
	P_L	∿	Local Static Pressure
	P_∞	∿	Ambient Pressure
	q	∿	Dynamic Pressure ∿ $\frac{1}{2}\rho V^2$
	S	∿	Reference Area
V_∞	V	∿	Freestream Velocity
	V_j	∿	Jet Velocity
	Y	∿	Spanwise Measurement
	α	∿	Angle of Attack of Fuselage
	ρ	∿	Density of Air
DELC	δ_C	∿	Canard Deflection
DELF	δ_F	∿	Flap Deflection
	ϵ	∿	Downwash Angle

Subscripts

c	∿	Canard
N	∿	Nozzle
w	∿	Wing
TR	∿	Thrust Removed

1.0 INTRODUCTION

Attainment of very short ground roll distances (approximately 400 feet) for takeoff and landing requires that operation at flight speeds of approximately 50 knots must be attainable. In today's fighter design, high thrust to weight ratios are required for maneuvering and high speed flight. If a concept can be developed which can utilize these thrust levels to augment the aerodynamic lift, then it may be possible to operate at these speeds.

One such concept involves the use of all or a large percentage of the engine exhaust ducted over the flap of the lifting surfaces. The blown flap or propulsive wing has been recognized as a method of producing large circulation lift coefficients. However, these large induced circulation lift coefficients, as well as the deflected thrust vector, react well aft on the wing and produce sizable nose-down pitching moments. The addition of a propulsive canard offers a possible conceptual design relief for these sizable trim requirements. The displacement of a portion of the thrust forward, as well as the induced circulation lift on the canard, tend to balance the moment due to the aft loading on the wing.

An earlier program done on this propulsive wing/canard configuration was primarily a cruise and maneuvering investigation. During that study (ref. 1), it was demonstrated that the concept would provide an improved maneuvering lift/drag (L/D) at transonic speeds. Also, the earlier study indicated quite large circulation lift coefficients at relatively low flap deflections. The study was done with a semispan model and tests were conducted in the Ames 14 foot Transonic Tunnel and in the Rockwell 7 by 10 foot low speed tunnel. Flap deflections in these tests were limited to a maximum of 15 degrees. The major configuration variable investigated was the effect of wing/canard relative locations. Analysis of that data showed that the best position for the low flap deflection ($\delta_F \leq$ 15 degrees) and for all speeds (M = 0.1 to 0.9) was with the canard high relative to the wing. These test results were discussed in References 1 through 3.

The investigation of the propulsive wing/canard concept has continued with additional tests and analysis. The data have been extended to the higher flap deflections and into ground effects as required for STOL operation, see Reference 4.

This report covers this test phase of the propulsive wing/canard concept investigation in STOL conditions in the NASA Langley 4 by 7 Meter Tunnel. The major variables investigated were:

(1) blown flap span
(2) canard/wing relative location, and
(3) flap deflection with large blowing coefficients.

2.0 MODEL AND TEST PROCEDURE

2.1 Model Description

The model is a full span wing/canard configuration. Both wing and canard have blown trailing edge flaps which can be deflected from zero to 60 degrees. The nozzle slots are at the flap hinge line (the 80 percent local chord position) and are perpendicular to the fuselage centerline. Figures 1 and 2 are model sketches presenting dimensional data. Table 1 presents a tabulation of the model geometry. The canard can be placed in one of three positions on the fuselage, and the wing can be placed in one of two positions on the fuselage, as shown in Figure 1. Figure 1 also shows the location of the downwash probe mounted one mean aerodynamic chord behind the wing.

The span of the nozzle slot on the wing was also variable. Provisions were made for the nozzle to blow either full span, half span, or a quarter span of the flap while maintaining approximately the same nozzle exit area. The nozzle was also always on the inboard portion of the wing. The canard nozzle was similarly configured except that only full span or half span configurations could be tested. When flap deflections were tested, the flap was deflected as a full span flap regardless of the extent of nozzle span. Wing and canard airfoil coordinates are tabulated in Table 2.

Air for the blowing slots is introduced to the model through a pressure reducer valve to the main fuselage plenum (see Figure 3). From the main fuselage plenum, air is ducted to four smaller plenums in the fuselage, one each for two canards and two wings.

Figure 3 also shows the balance installation and the manner in which air is supplied across the balance. The wings and canards plug into the fuselage plenums, allowing the air to flow to the wing or canard high pressure plenums where the flow is stagnated and ducted through pressure drop supply ducts to the low pressure plenums which supply the nozzle slots. The air supply system provided nearly uniform jet exit velocity with span. Figure 4 is a sketch showing the model air supply from the LaRC supply pipe to the nozzle exits. Flow split is adjustable to each of the four blowing surfaces by use of valves located on the main fuselage plenum. High pressure air brought onto the model in this manner results in balance constants which are determined by calibration with the air pipe in place and by a pressure tare which is a function of model internal geometry. These calibrations and tares are a part of the standard, NASA provided, data reduction capability.

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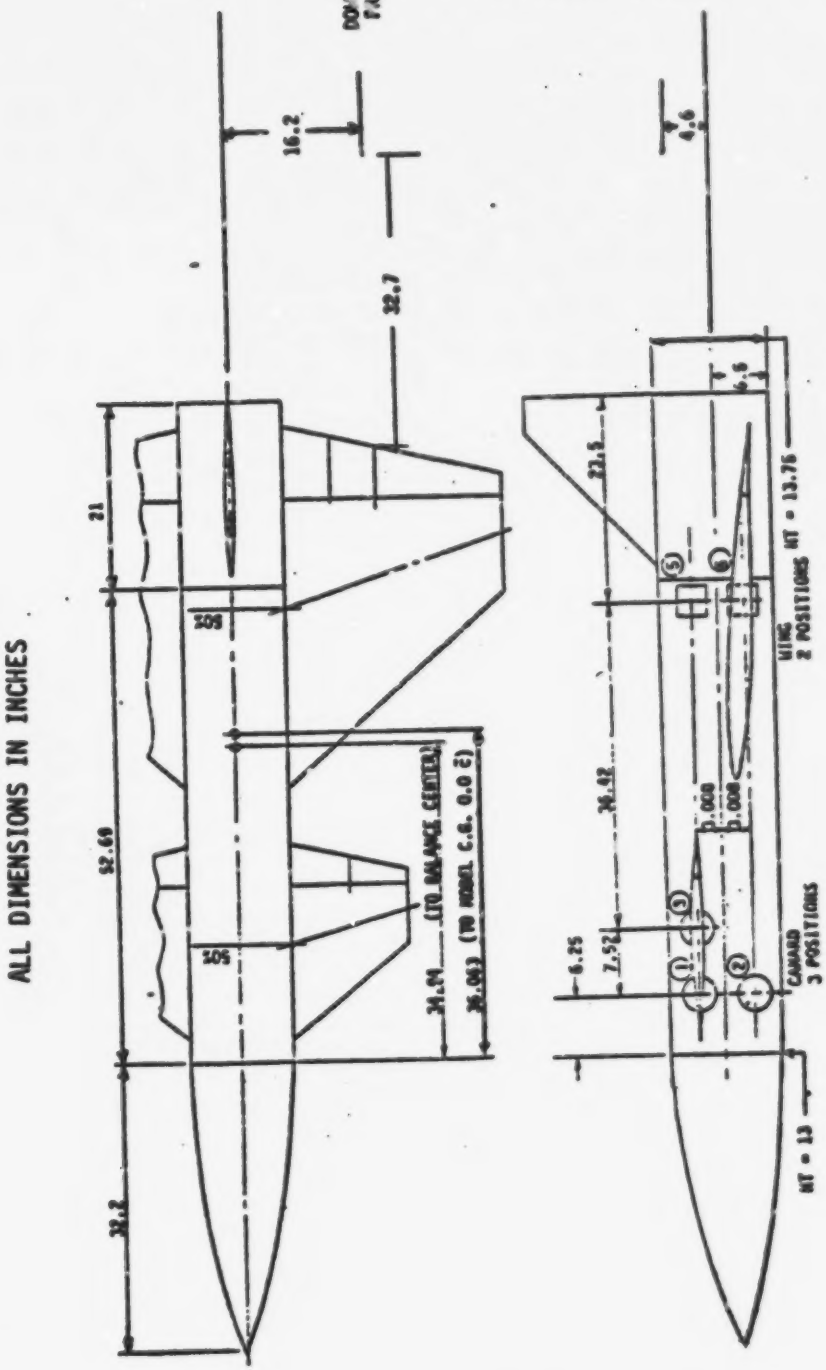

Figure 1. Model Sketch, Surface Locations

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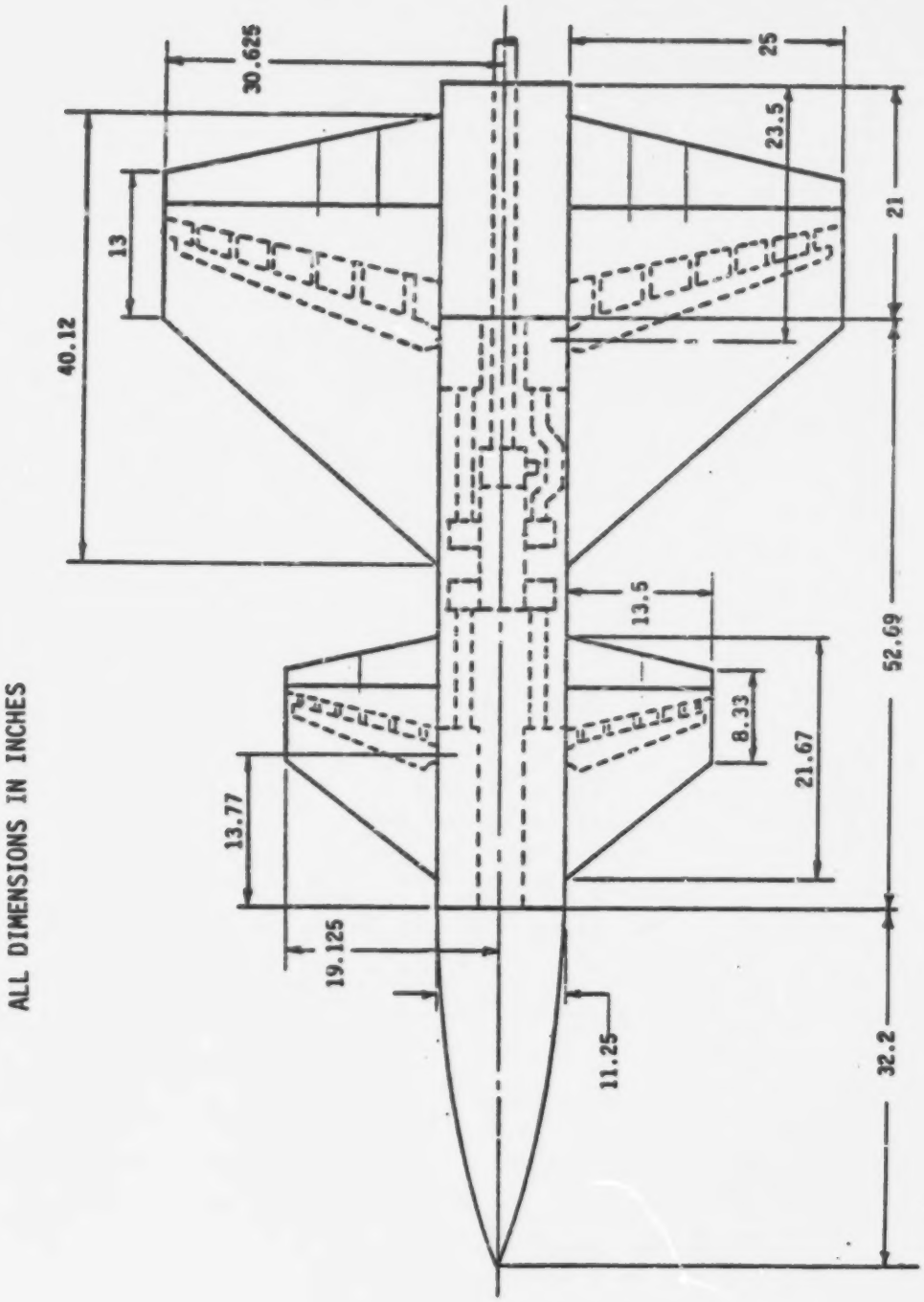


Figure 2. Model Sketch (Dimensions)

TABLE 1. - MODEL GEOMETRY

	WING	CANARD	BODY	TAIL
TIP CHORD	13 IN	8.33 IN	-	4.25 IN
ROOT CHORD EXPOSED	40.12 IN	21.67 IN	-	19.50 IN
ROOT CHORD TOTAL (BP 0)	46.21 IN	27.23 IN	-	-
TAIL HEIGHT	-	-	-	15.20 IN
SPAN TOTAL	61.25 IN	38.25 IN	-	-
AREA EXPOSED	9.222 FT^2	2.812 FT^2	-	1.25 FT^2
AREA TOTAL	12.591 FT^2	3.334 FT^2	-	-
ASPECT RATIO EXPOSED	-	1.80	-	2.56
ASPECT RATIO TOTAL	2.069	3.047	-	-
BODY LENGTH	-	-	105.89 IN	-
BODY WIDTH	-	-	11.25 IN	-
MINIMUM BODY HEIGHT	-	-	13.00 IN	-
MAXIMUM BODY HEIGHT	-	-	13.75 IN	-
MAC EXPOSED	28.87 IN	16.0 IN	-	-
MAC TOTAL	32.71 IN	19.45 IN	-	-
SWEEP	41 DEG	38.33 DEG	-	45 DEG

TABLE 2. - BASIC AIRFOIL COORDINATES (NO NOZZLE)

	WING				CANARD			
	NO DROOP		DROOP		NO DROOP		DROOP	
X/C	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER
0	0	0	-.035	-.035	0	0	-.025	-.025
.002	.0046	-.0044	-.0299	-.0395	.0046	-.0044	-.0195	-.0295
.005	.0068	-.00605	-.0263	-.04095	.0068	-.00605	-.0154	-.0311
.01	.0093	-.0077	-.0217	-.0411	.0093	-.0077	-.01035	-.0308
.02	.0126	-.0100	-.0150	-.0398	.0126	-.0100	-.003	-.029
.03	.0153	-.0120	-.0097	-.0383	.0153	-.0120	.0024	-.0278
.04	.0175	-.0133	-.0051	-.0370	.0175	-.0133	.007	-.0266
.06	.0212	-.0157	.0034	-.0346	.0212	-.0157	.0142	-.0249
.08	.0242	-.0176	.0107	-.0324	.0242	-.0176	.01955	-.0235
.10	.0264	-.0192	.0165	-.0303	.0264	-.0192	.0235	-.0226
.125	.0287	-.0208	.02235	-.0279	.0287	-.0208	.0273	-.0220
.15	.0305	-.0222	.0267	-.0258	.0305	-.0222	.030	-.0224
.20	.0329	-.0241	.0321	-.0246	.0329	-.0241	.0329	-.0241
.25	.0342	-.0254	.0342	-.0254	.0342	-.0254	.0342	-.0254
.30	.0350	-.0256	.0350	-.0256	.0350	-.0256	.0350	-.0256
.35	.03548	-.02545	.03548	-.02545	.0354	-.02545	.0354	-.02545
.40	.0357	-.0249	.0357	-.0249	.0357	-.0249	.0357	-.0249
.45	.03575	-.0241	.03575	-.0241	.0358	-.0241	.0358	-.0241
.50	.03565	-.0230	.03565	-.0230	.0358	-.0230	.0358	-.0230

TABLE 2. - BASIC AIRFOIL COORDINATES (NO NOZZLE) (Concluded)

		WING				CANARD			
		NO DROOP		DROOP		NO DROOP		DROOP	
	X/C	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER	Z/C UPPER	Z/C LOWER
	.55	.03535	-.02175	.03535	-.02175	.0358	-.02175	.0358	-.02175
	.060	.03488	-.01945	.03488	-.01945	.0356	-.01945	.0356	-.01945
	.65	.0342	-.0165	.0342	-.0165	.03535	-.0165	.03535	-.0165
	.70	.0332	-.0126	.0332	-.0126	.0348	-.0126	.0348	-.0126
	.75	.03165	-.0081	.03165	-.0081	.0340	-.0081	.0340	-.0081
	.80	.029	-.0028	.0290	-.0028	.0325	-.0028	.0325	-.0028
FLAP	.85	.02325	+.002	.02325	+.002	.02325	+.002	.02325	+.002
FLAP	.90	.0173	+.003	.0173	+.003	.0176	+.003	.0176	+.003
FLAP	.95	.00935	+.0008	.00935	+.0008	.0112	+.0008	.0112	+.0008
FLAP	1.00	0	-.004	0	-.004	.004	-.004	.004	-.004

L.E. RADIUS = .012

WING ROOT INCIDENCE +4.0°

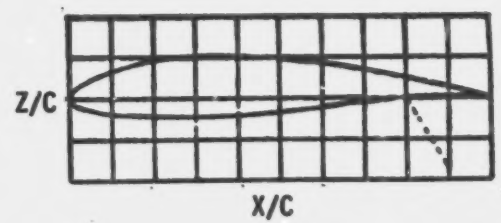
WING TIP INCIDENCE -1.0°

TWIST ALONG X/C = 0.80

Z/C

Z ~ VERTICAL DISTANCE

C ~ CHORD ~ IN.



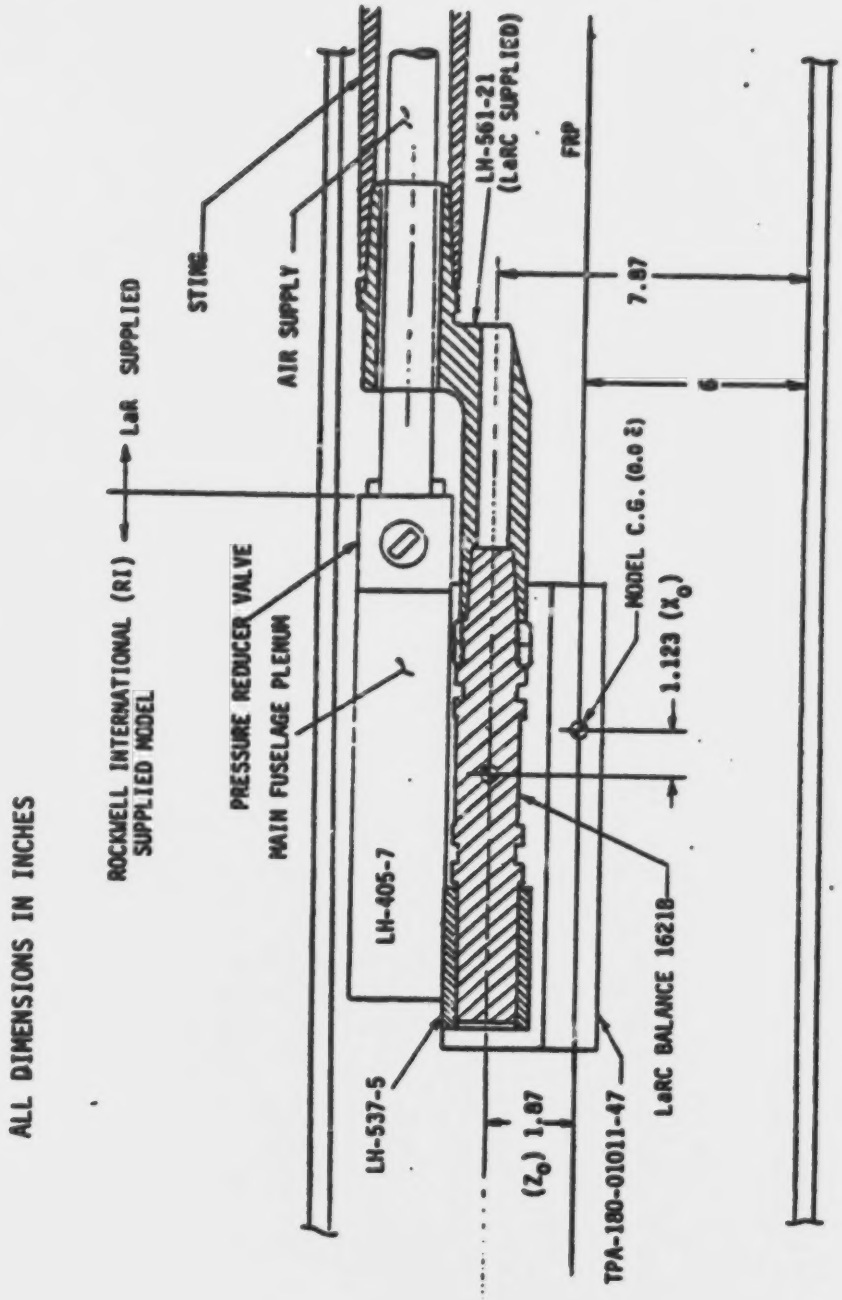


Figure 3. Schematic of Sting, Balance, and Air Supply

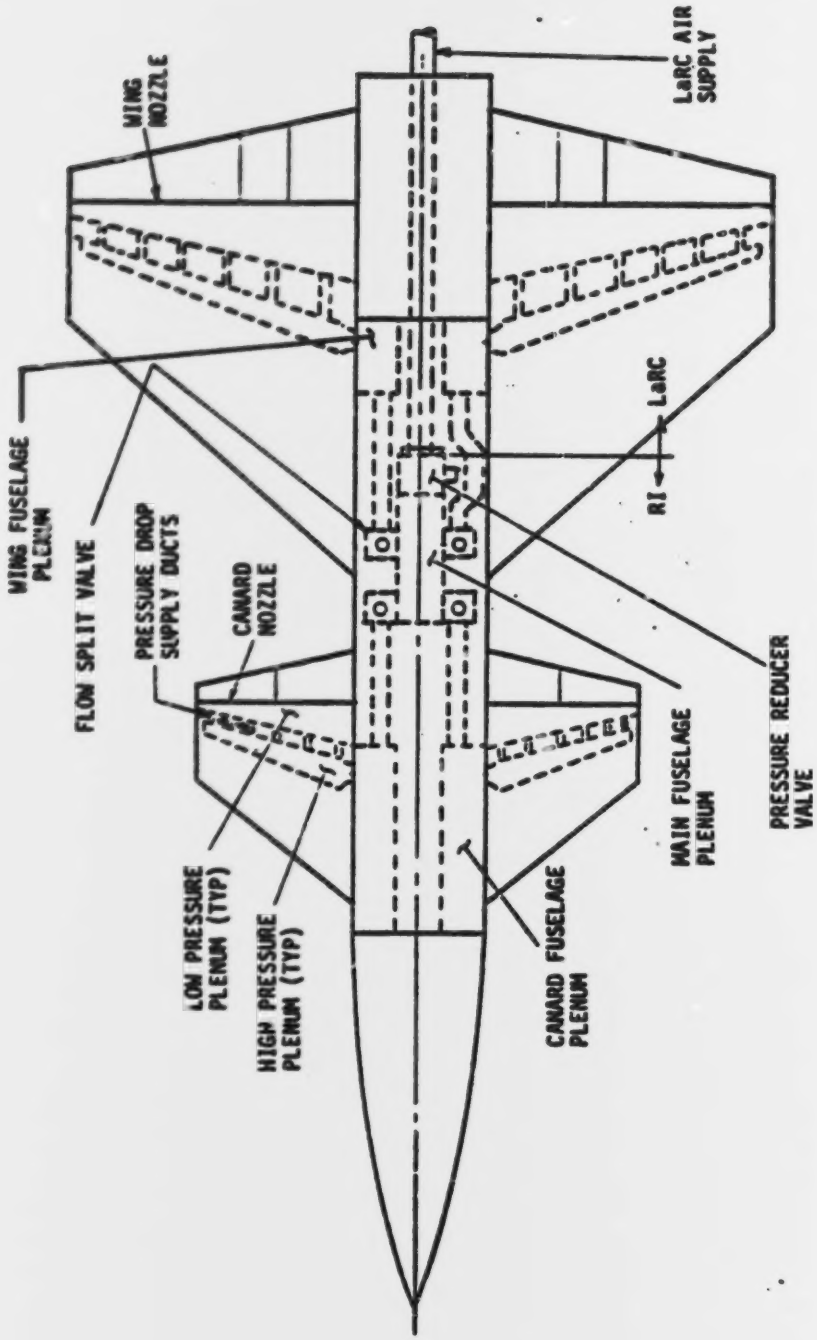


Figure 4. Model Air Supply Diagram

2.2 Model Instrumentation

The model and test instrumentation consisted of the following:

1. six-component internal balance
2. wing surface static pressure ports
3. canard surface static pressure ports
4. fuselage surface static pressure ports
5. internal flow pressure sensors
6. airflow measuring and calibrating instrumentation (NASA supplied)
7. calibrated downwash probe (NASA supplied)

Surface pressure instrumentation consisted of a total of 229 surface taps with 151 located on the left hand wing, 56 located on the left hand canard, and 22 located at 3 inches left of the plane of symmetry on the fuselage upper and lower surfaces. These were all monitored with a standard scani-valve system. Figure 5 shows the location of the wing and canard pressure taps by butt plane and local chord. The locations of the fuselage pressures along the fuselage length are presented with the tabulated pressure data in Appendix A.

The downwash probe was a five-hole directional probe mounted approximately one chord length aft of wing trailing edge (see Figure 1). The calibration of the probe provided local angle of attack relative to model FRL. These angles of attack have been converted to downwash angle and are presented in Appendix A.

2.3 Model - Data Reduction

The force data have been reduced to standard six component force and moment coefficient about the stability axis by standard data reduction equations. The blowing coefficient C_μ is obtained by expanding the measured mass flow to the fully expanded, isentropic, velocity and normalizing on freestream dynamic pressure and the total wing area.

$$C_\mu = \frac{\dot{m}\, V_j}{qS}$$

The thrust removed coefficients are obtained by adjusting the balance raw data output by the static thrust tares and then applying normal corrections to the adjusted data, i.e.:

$$NF_{TR} = NF_{uncorrected} - NF_{static\ thrust}$$

$$AF_{TR} = AF_{uncorrected} - AF_{static\ thrust}$$

etc.

This procedure required that a thrust tare be taken each time a configuration change involving the thrust nozzle system was made.

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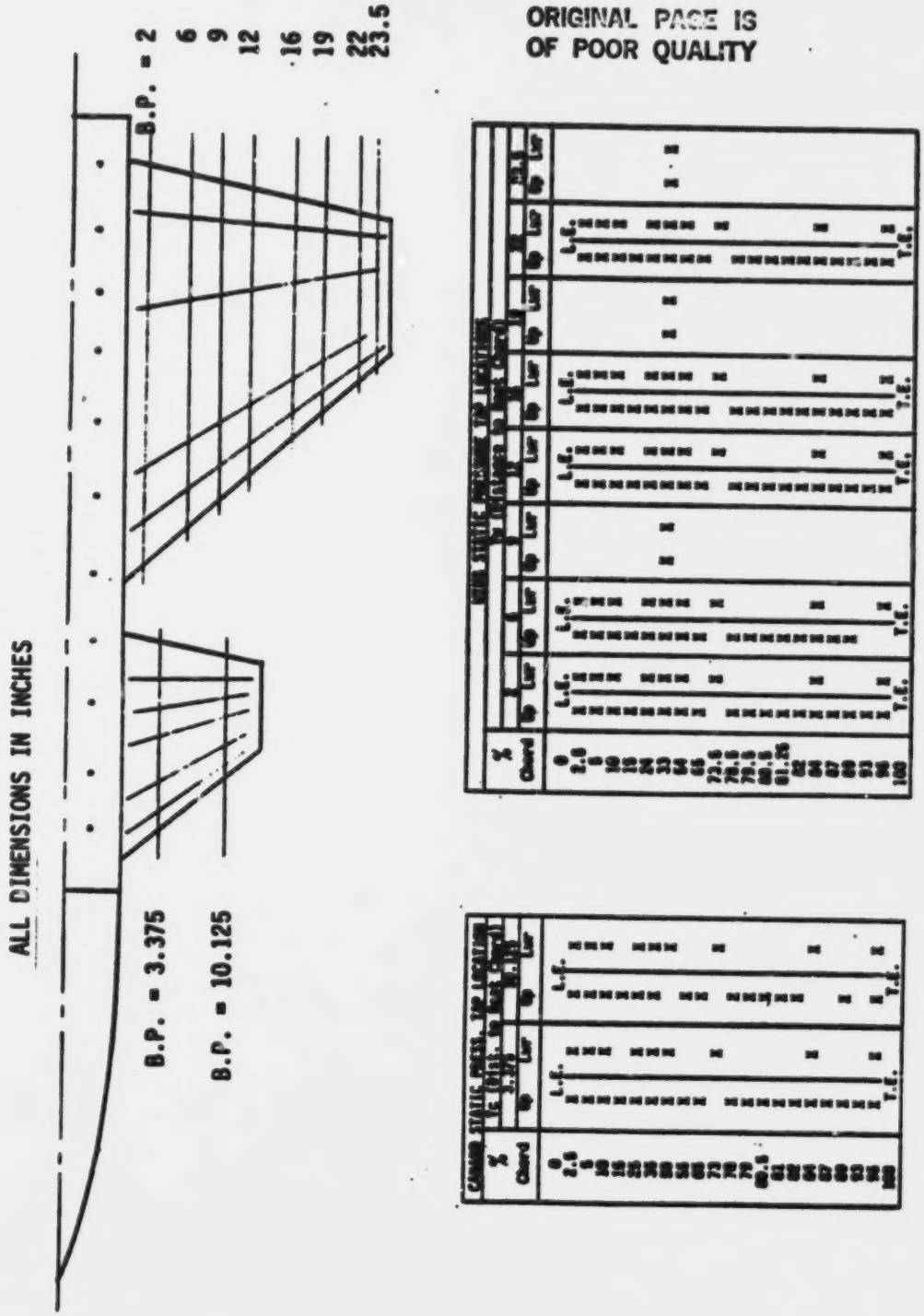

Figure 5. Surface Pressure Tap Locations

Surface pressure distributions were reduced to pressure coefficient by the normal data reduction equations.

$$C_p = \frac{\Delta P}{q} = \frac{P_L - P_\infty}{q}$$

Test data are presented in Appendix A. The force data are plotted, and selected pressure data and downwash angle measurements are tabulated. Table 3 defines the configurations for which data are presented.

TABLE 3. CONFIGURATION DEFINITION

SYMBOL	DEFINITION	CANARD POSITION	WING POSITION
B OR BV	BODY, VERTICAL	OFF	OFF
BC1V	BODY, CANARD, VERTICAL	1	OFF
BC1W6V	BODY, CANARD, WING, VERTICAL	1	6
BC2W6V	BODY, CANARD, WING, VERTICAL	2	6
BC2W5V	BODY, CANARD, WING, VERTICAL	2	5
BC3W6V	BODY, CANARD, WING, VERTICAL	3	6
BW6V	BODY, WING, VERTICAL	OFF	6

2.4 Installation and Test Procedure

The model is shown installed in the test section of the NASA-Langley 4 x 7 meter tunnel in Figure 6. High pressure air for jet simulation was supplied to the model through an air line in the model support sting, and supplied to each nozzle as described in Section 2.1. The nozzle pressure ratio was maintained at 1.7 or greater, and blowing coefficient was controlled by varying tunnel dynamic pressure. The nozzles were calibrated with the tunnel off, and this static thrust was utilized to obtain the thrust removed coefficients.

The configurations which were tested included: (1) flap deflections of 0, 15, and 45 degrees, (2) blowing spans of 1/4, 1/2, and full span on the wing and 1/2 and full span on the canard, (3) relative wing/canard placement. Blowing coefficients were varied from 0 to 4.0. These configurations were tested both in and out of ground effect. Basic data were obtained through a pitch range of -4 to 20 degrees angle of attack. Selected configurations were tested through a range of yaw angles of +8 to -8 degrees. Where ground effects were investigated, the model was tested at heights from approximately h/c = 0.4 to free air. The sideslip and

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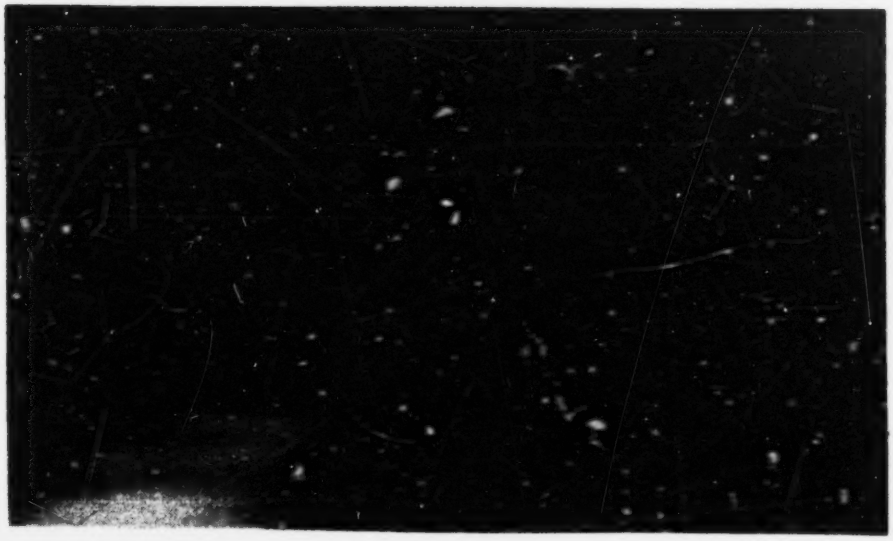

(a) Front View

(b) 3/4 Side View

Figure 6. Full Span Model in LaRC 4 x 7 Meter Wind Tunnel Facility

height data were obtained at an angle of attack near zero degrees. Downwash angle was measured behind the wing for most of the configurations. Figure 1 shows the relative wing canard locations of the model, and Figure 5 presents the wing and canard pressure tap locations.

3.0 SUMMARY OF TEST RESULTS

A brief analysis of the effect of the blowing nozzle span and wing/canard positioning effects on longitudinal aerodynamic characteristics is presented. The lateral-directional characteristics of the propulsive wing/canard obtained during the test are presented in Appendix A with the complete data. The effect of ground proximity is also presented in Appendix A, A177 to A-209. No unexpected ground effects were noted.

3.1 Effect of Nozzle Span

Nozzle span was varied from approximately 1/4 span to full span with a full span flap deflection of 45 degrees. The effects of nozzle span and blowing coefficient on lift coefficient and thrust removed lift coefficient are presented in Figures 7 through 10. Total C_L at angles of attack of zero and eight degrees are presented in Figures 7 and 8 with the corresponding thrust removed lift, $C_{L_{TR}}$, presented in Figures 9 and 10.

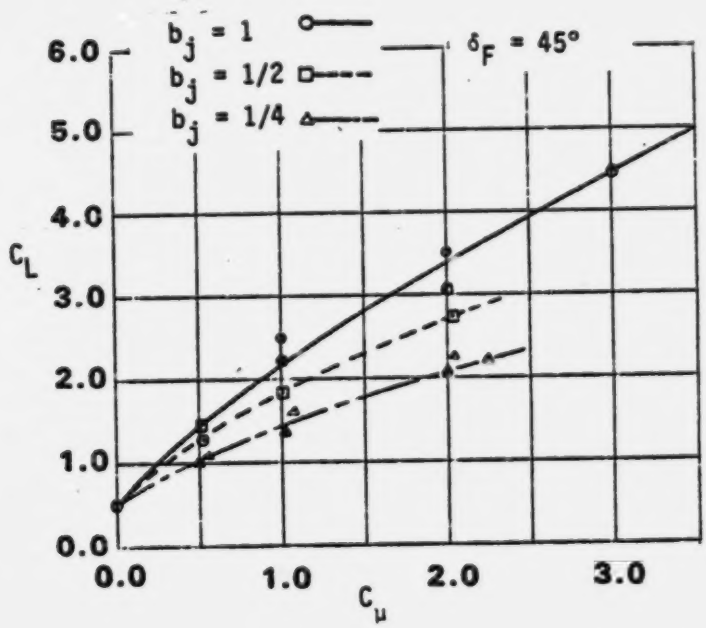

Figure 7. Effect of Blowing Span on Total Lift Coefficient, Canard Off, α = 0 Degrees

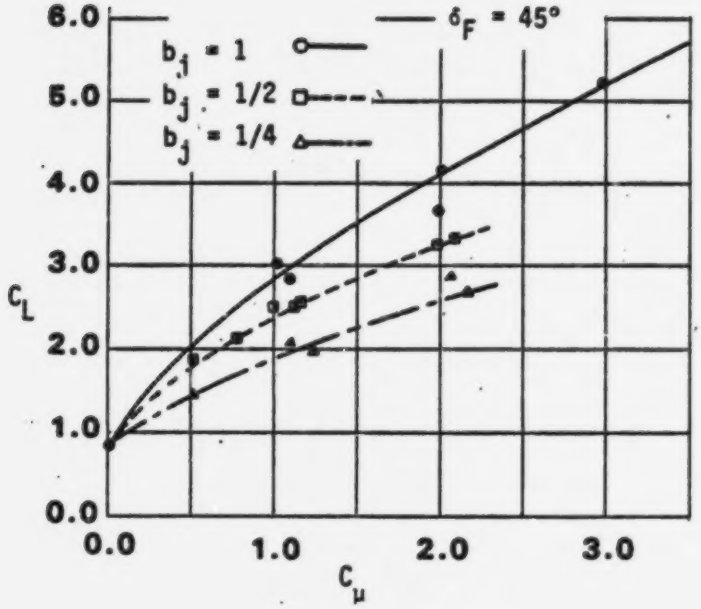

Figure 8. Effect of Blowing Span on Total Lift Coefficient, Canard Off, α = 8 Degrees

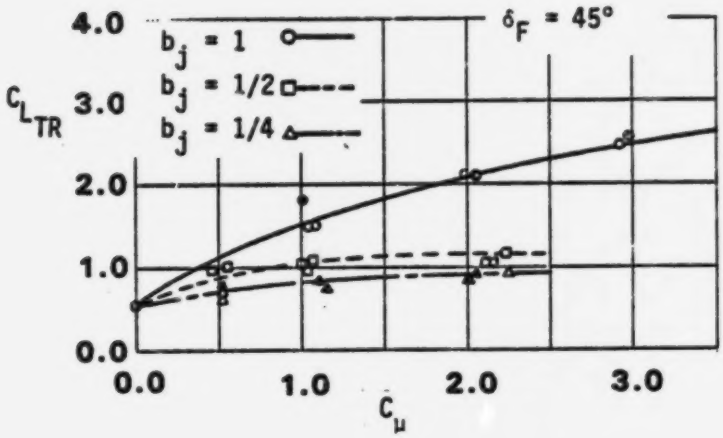

Figure 9. Effect of Blowing Span on the Thrust Removed Lift Coefficient, Canard Off, α = 0 Degrees

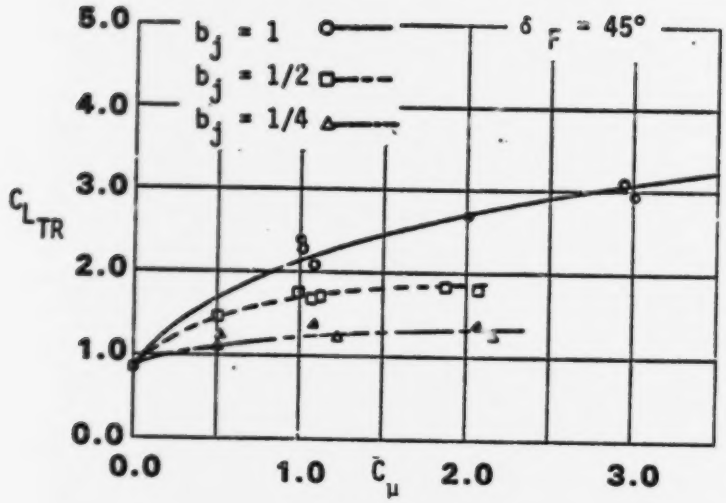

Figure 10. Effect of Blowing Span on the Thrust Removed Lift Coefficient, Canard Off, α = 8 Degrees

These thrust removed lift coefficients are developed by removing the static thrust components from the total force measurements. The results of Figures 9 and 10 show that the full span flap is much superior in developing circulation lift. The full span configuration continues to show an increase in circulation to levels of C_μ greater than three while the shorter spans appear to have attained their maximum level of circulation lift at a much lower C_μ. These characteristics are indicative of the greater local blowing coefficient of the shorter nozzles compared to the longer, full span nozzle. The circulation lift increase with blowing is controlled by the local or section blowing coefficient. The jet flap theory accounts for the local C_μ by the use of the affected wing area ratio, Reference 5.

The drag of the propulsive wing is presented in Figure 11 for the thrust included case and in Figure 12 with the thrust removed. The drag variation demonstrates the same characteristics as the lift. Large induced drag is indicated for the full span nozzle and less induced drag with the shorter span nozzles. This result is to be expected in view of the variation of lift coefficient discussed earlier.

Wing/body pitching moment coefficients presented in Figures 13 and 14 also show the same trend. The moments are about the leading edge of the mean aerodynamic chord; therefore, the vectored thrust inputs a significant negative moment. The thrust removed moment indicates that the aerodynamic load is acting at about the 70 percent chord of the wing. This is not unexpected for a blown flap with a chord of 20 percent of the wing chord.

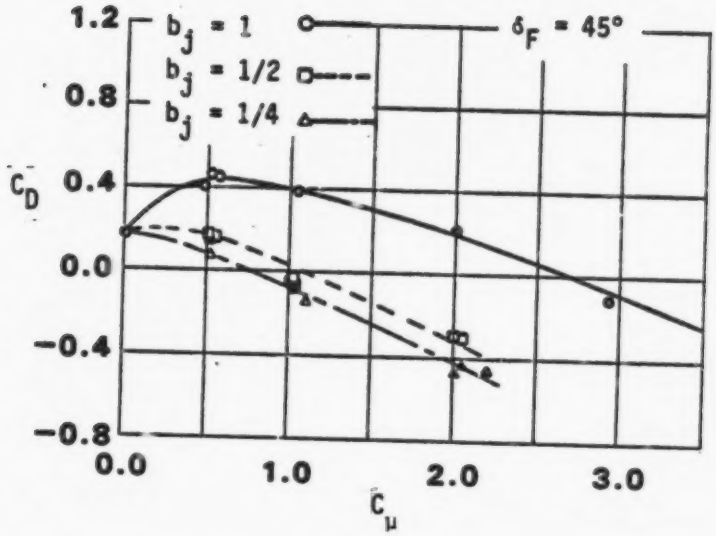

Figure 11. Effect of Blowing Span on the Total Drag Coefficient, Canard Off, α = 0 Degrees

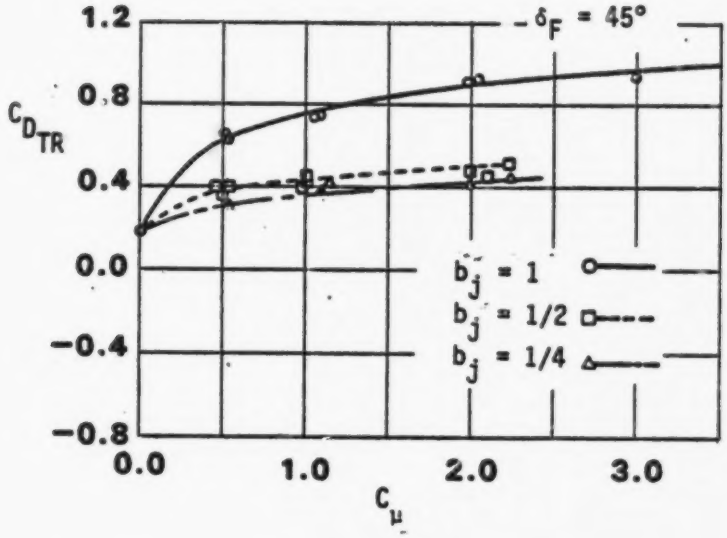

Figure 12. Effect of Blowing Span on the Thrust Removed Drag Coefficient, Canard Off, α = 0 Degrees

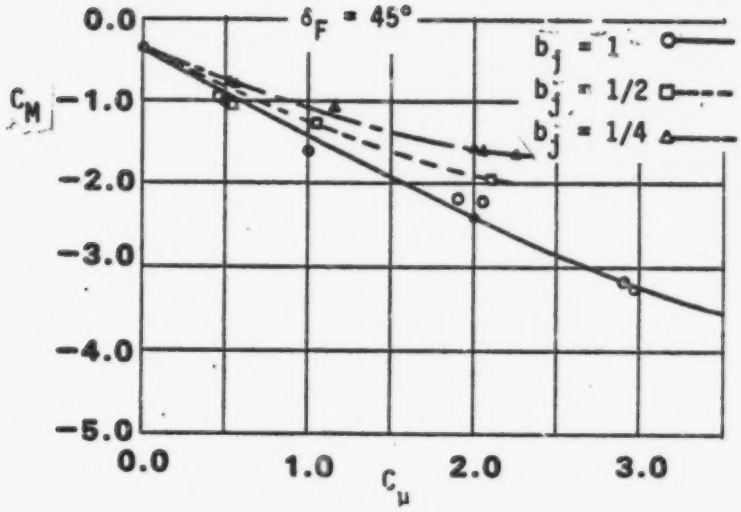

Figure 13. Effect of Blowing Span on the Total Pitching Moment Coefficient, Canard Off, α = 0 Degrees

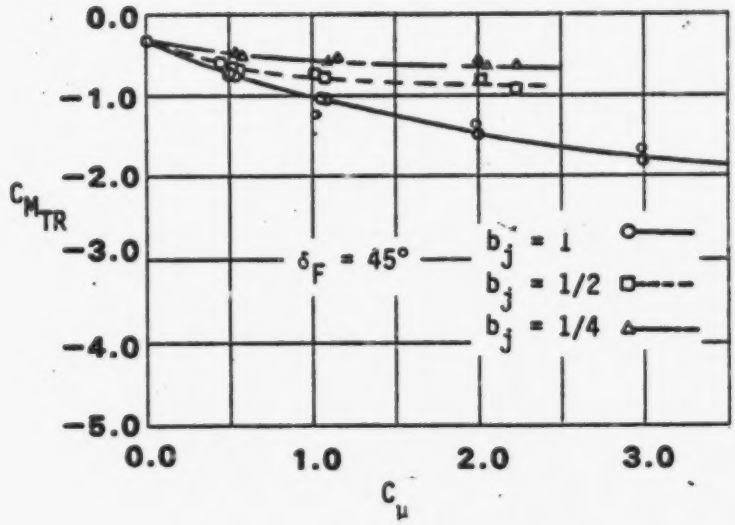

Figure 14. Effect of Blowing Span on the Thrust Removed Pitching Moment Coefficient, Canard Off, α = 0 Degrees

3.2 Canard Effects

The negative pitching moments of the propulsive wing require a large force well forward of the center of gravity of the configuration to obtain trim. One means of accomplishing this is using a blown canard. Figure 15 presents the pitching moment coefficient at zero angle of attack with the canard installed in the high forward position and a blowing coefficient of 1/2 that of the wing. Figure 16 presents similar data at an angle of attack of eight degrees. It can be seen that a significant pitch-up moment is gained from the canard, even though it is not sufficient to trim at this center of gravity (CG) location. Increased canard flap deflection, canard incidence angle, or increased canard blowing rates are means of increasing the trim power of the canard. These variables will be addressed in future studies.

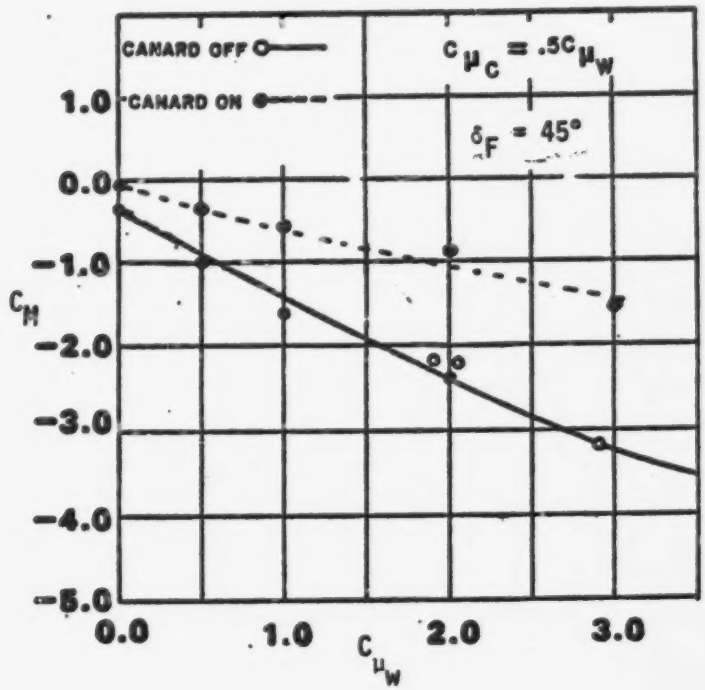

Figure 15. Effect of the Canard on the Pitching Moment Coefficient, $\alpha = 0$ Degrees

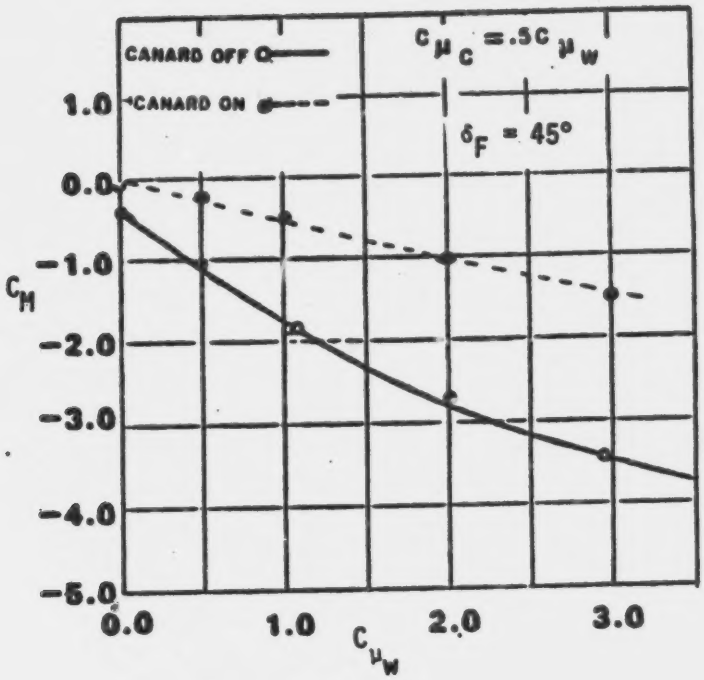

Figure 16. Effect of the Canard on the Pitching Moment Coefficient, α = 8 Degrees

The relative position of the canard and the wing is also a very important consideration in the propulsive wing/canard concept. As is true in most design variables, there is no single position which is the so-called "best" for all conditions. Previous studies (Refs. 1, 2, and 3) showed that for deflections of 15 degrees and less, the high canard, low wing had the better characteristics. This held for blowing as well as for the non-blowing conditions. Early, unpublished, studies on a similar configuration indicated that at nozzle deflections of 60 degrees, the canard should be low relative to the wing. The results of a positioning variation on the current wing/canard model are presented in Figure 17. The flap deflection for this test was 45 degrees; however, with the upper surface contour aft of the nozzle included, the total jet turning angle was 60 degrees. The data show that for this configuration, the low canard, high wing, position 5,2, has the better lift characteristics for the blown, large deflection conditions.

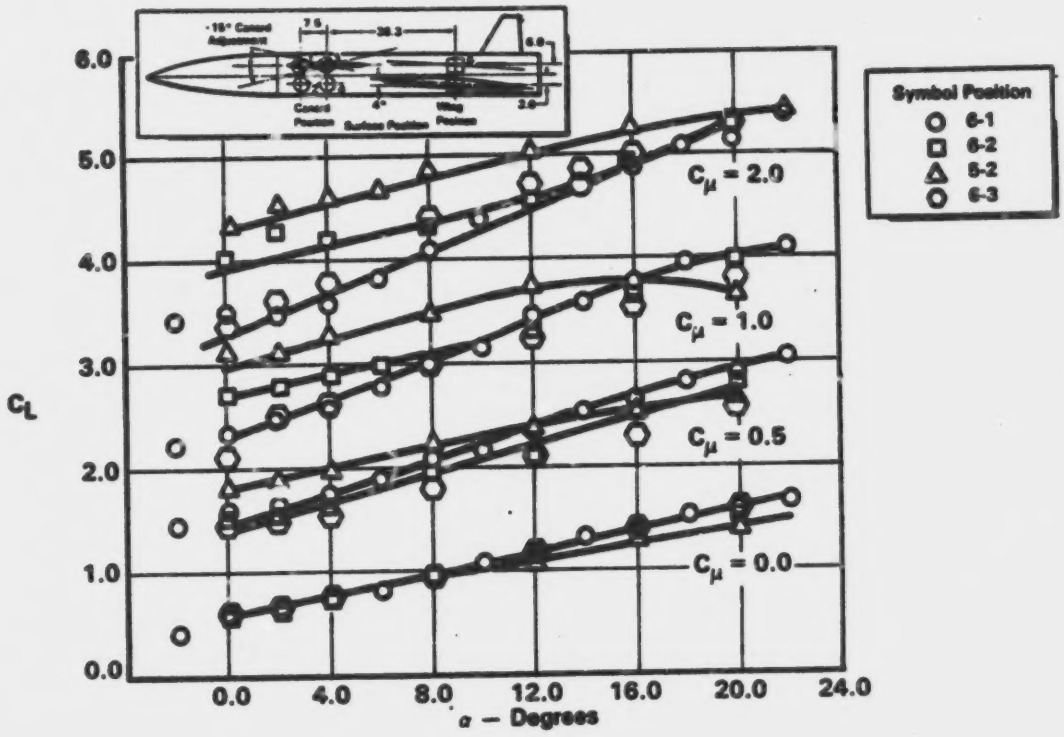

Figure 17. Effect of Canard Position on Total Lift Coefficient

The flight requirements of the STOL wing/canard airplane would dictate the compromises which would finally place the wing and canard in their positions. It is expected that the high speed requirements would prevail, and a propulsive wing/canard configuration would have a high canard with a low wing. Most operations would be at low flap deflections, 15 degrees or less, and the larger deflections would be used only for very short field operations, less than 400 feet.

3.3 Semi/Full Span Comparison

The propulsive wing/canard model has been tested in a semispan, as well as a full span configuration. However, the blowing distribution between the wing and canard was changed between the semi and full span configurations, so a comparison of the two configurations powered cannot be made. However, comparisons can be made for the wing alone at blowing coefficients of 0, 1.0, and 2.0 and for the wing/canard configuration with a blowing coefficient of zero.

The comparison of the full and semispan test results for the wing/canard configuration are presented in Figure 18 for the zero blowing case. Very good agreement is seen in most of the parameters. A difference is seen in the lift curve slope, but almost perfect agreement is seen in the drag coefficient. The pitching moment indicates a slightly greater stability for the semispan model. These small differences may have been the result of a change on the canard downwash on the wing.

The comparison of the wing/body is presented in Figure 19 for blowing coefficients of 0, 1.0, and 2.0 Near perfect agreement between the two models is shown in lift and in drag. Again, the semispan model indicates slightly greater stability.

4.0 CONCLUSIONS

The experimental STOL investigation of the propulsive wing/canard has been partially completed. The study, thus far, has shown these results.

The full span nozzle is more effective than the partial span nozzles in producing circulation lift.

The low canard, high wing positioning is preferred for large flap deflection and blowing; however, at other conditions, such as takeoff flaps, cruise, and transonic flight, the high canard is preferred. It is expected that these conditions will prevail, yielding a high canard, low wing configuration.

Very good agreement between full span and semispan model test results were obtained where exact conditions were duplicated.

An additional test of this concept will be conducted with a contoured fuselage. The major objectives with the revised fuselage lines will be to determine the lateral-directional characteristics, to investigate means to trim the wing pitching moments, and to investigate the effect of the moving ground board. Flow field measurements behind the canard will be made for comparison with and as an extension of the semispan model data bank.

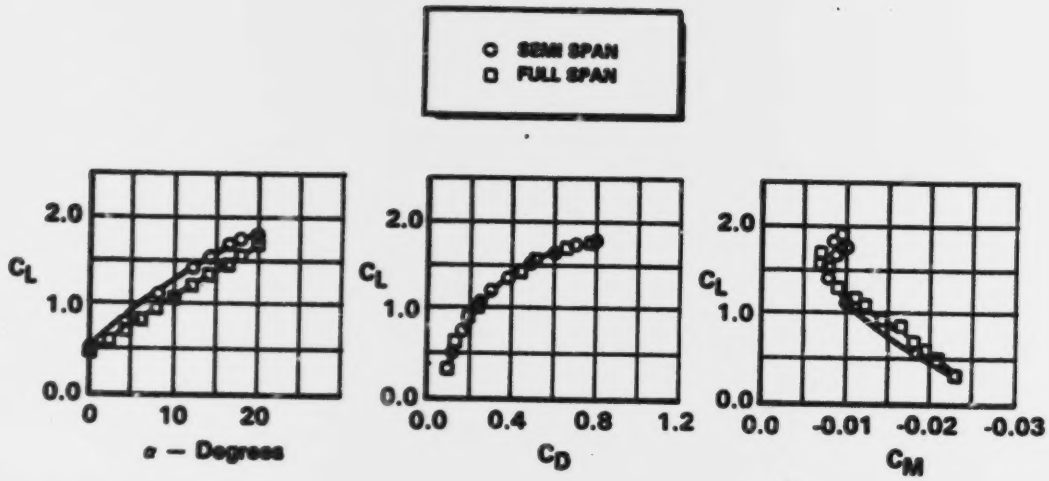

Figure 18. Comparison of Semispan and Full Span Test Results, Wing/Canard/Body, $C_\mu = 0$, $\delta_F = 0$

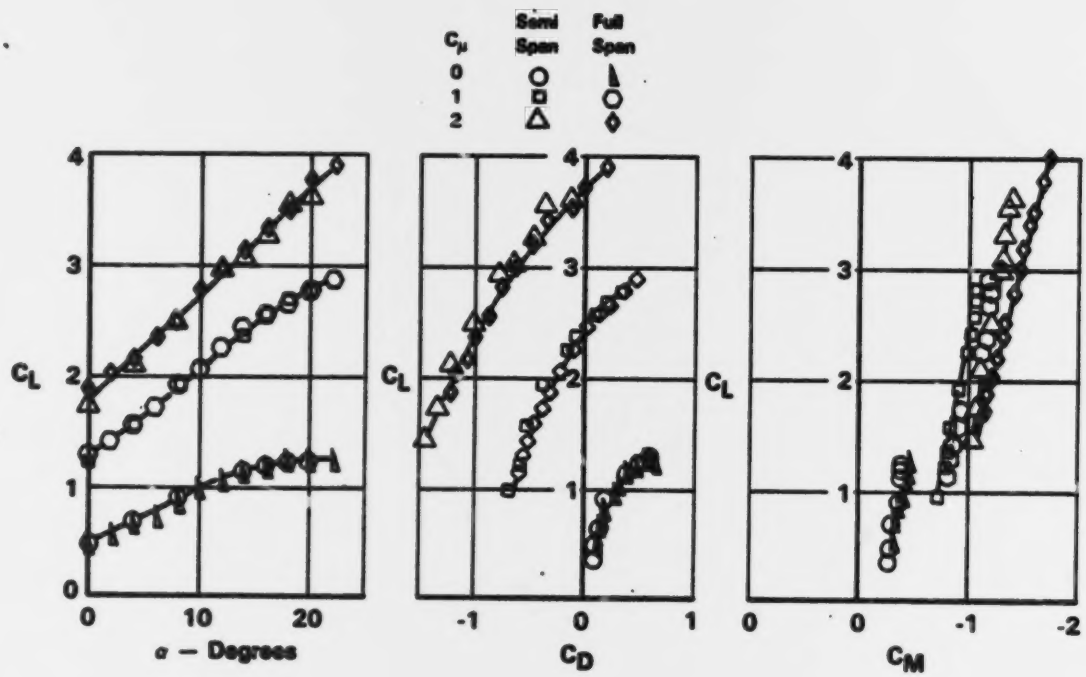

Figure 19. Comparison of Semispan and Full Span Test Results Wing/Body, $\delta_F = 15$ Degrees

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APPENDIX A

PRESENTATION OF BASIC TEST DATA

APPENDIX A

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A1.0 INTRODUCTION

The test data are presented in Appendix A. Some mechanical and internal airflow fouling difficulties were experienced during the test period. The air pipe fouling was, apparently, the result of a gradual failure of the air coil internal to the support sting. The air coil eventually ruptured (Run 247), and the sting was replaced with Air Sting #2 ("bent") for the remainder of the test. The mechanical foul appeared to be a combination of several circumstances including: (1) large nose-down moments with the wing alone configuration, (2) an excessively long sting-to-balance adapter coupled with (3) the normal balance flexibility. It is not known if the internal coil failure entered into the mechanical foul, although the initial signs of coil failure did occur prior to the mechanical foul. In any event, those force data runs where a foul can be detected are not presented. In certain cases, surface pressure data and downwash data runs are presented without force data results. These runs represent those for which repeat force runs were made without the pressure instrumentation operating. The surface pressure data and downwash data presented in Appendix A are correct in that the foul experienced did not affect these data. The corresponding force data in some runs may be in error due to the fouling. In these cases, the correct force data run to be used in analysis is noted on the tabulated data.

A2.0 PLOTTED FORCE DATA

The force and moment data have been plotted and are presented in Section A2.0. The data are presented in three categories. Figures A1 through A24 present the force and moment coefficients for the basic angle of attack variations. Figures A25 through A28 present these coefficients for sideslip variations. And, Figures A29 through A39 show the results of ground height variations. Data are presented for major variables of canard/wing positioning, flap deflection, and flap span. Each major variable is presented as a function of blowing coefficient.

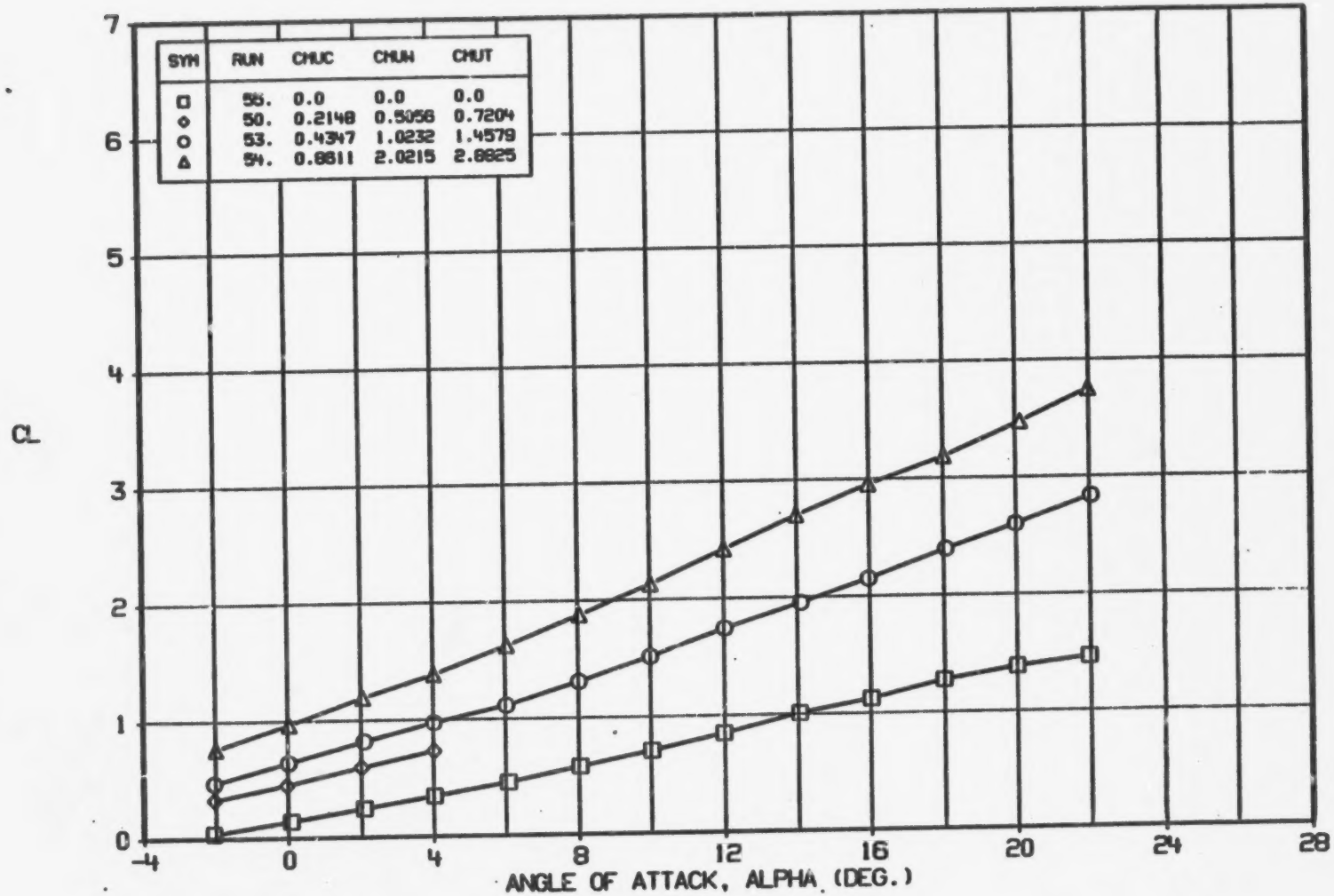

FIGURE A1a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=0, BN/B=1

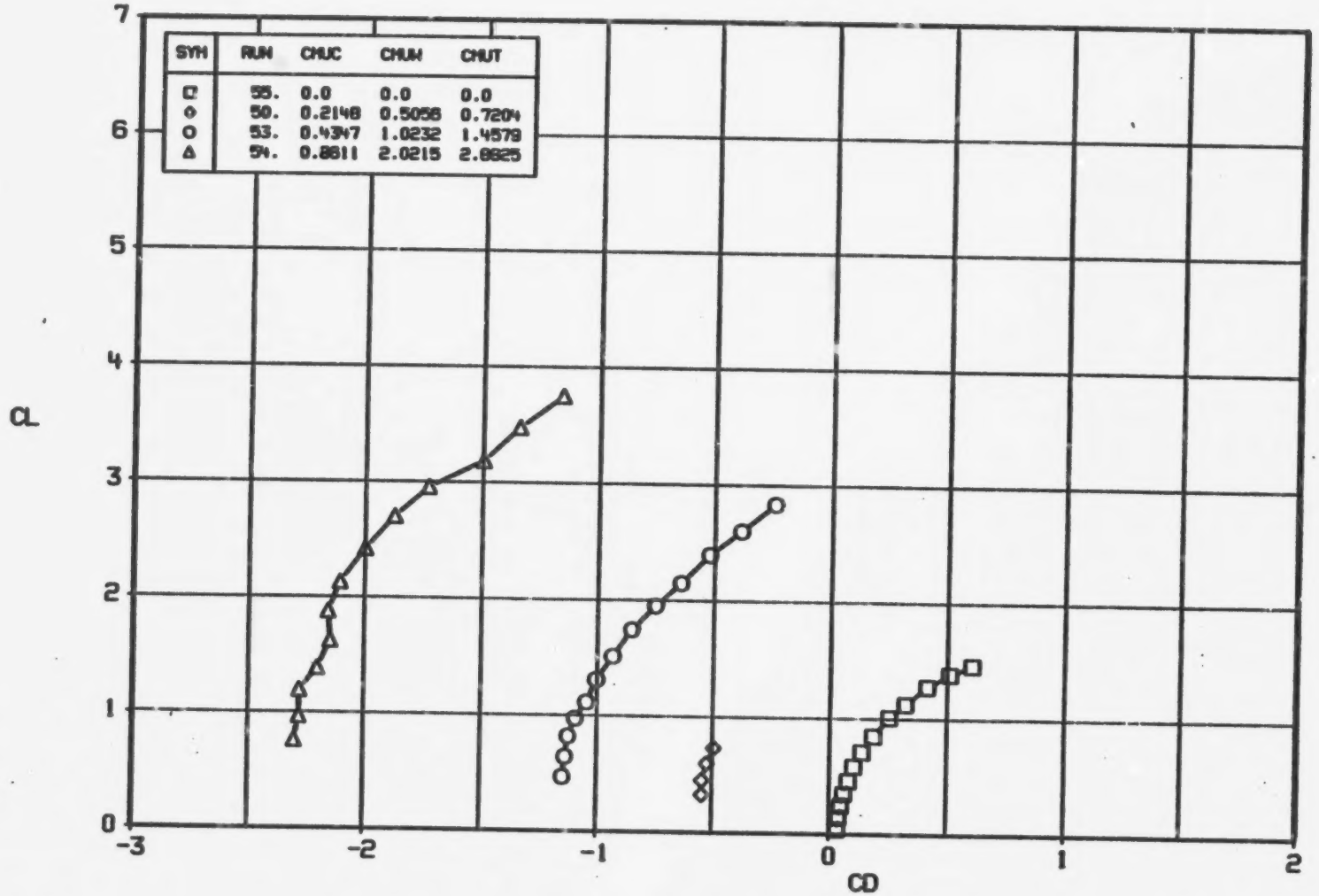

FIGURE A1b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=0, BN/B=1

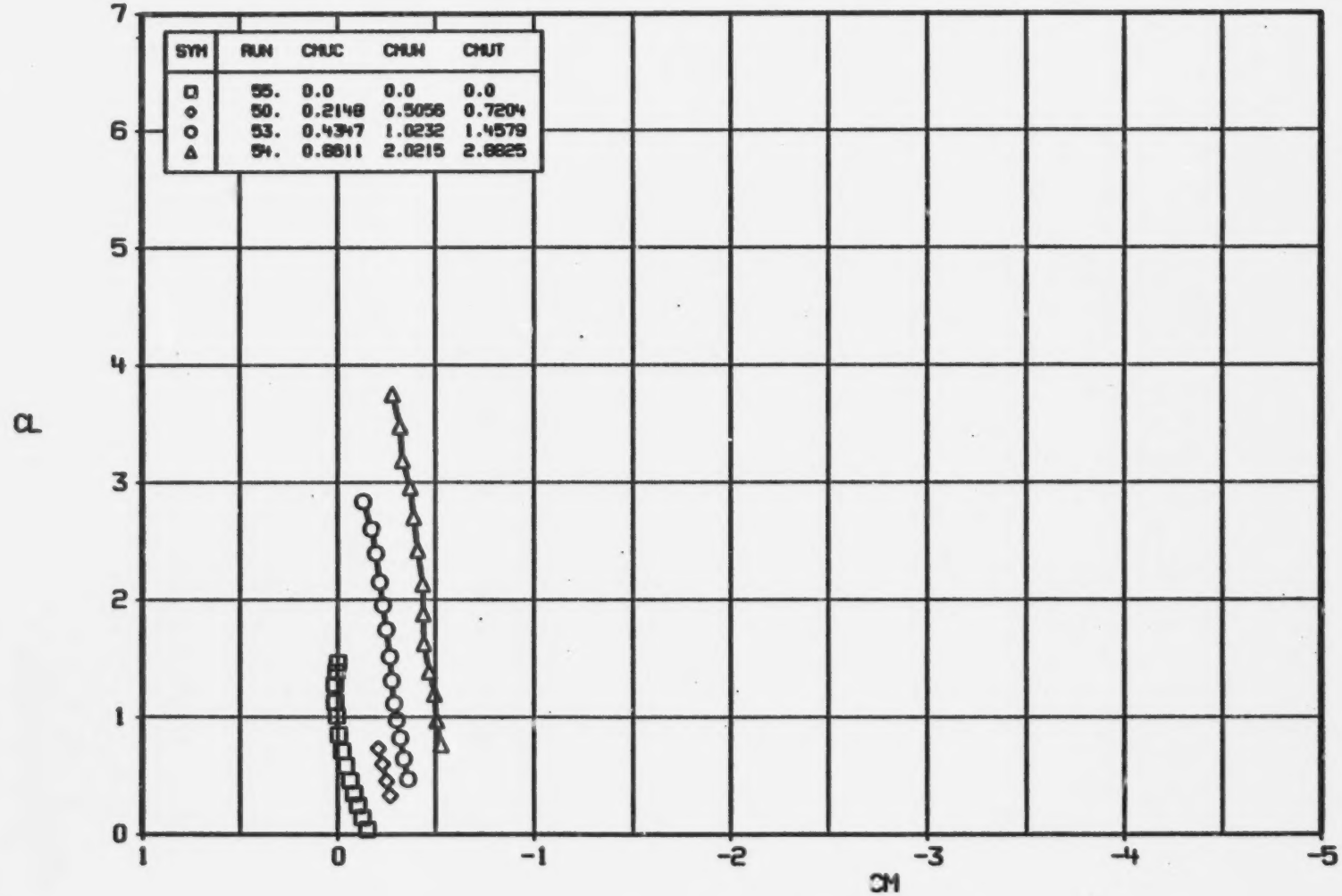

FIGURE A1c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=0, BN/B=1

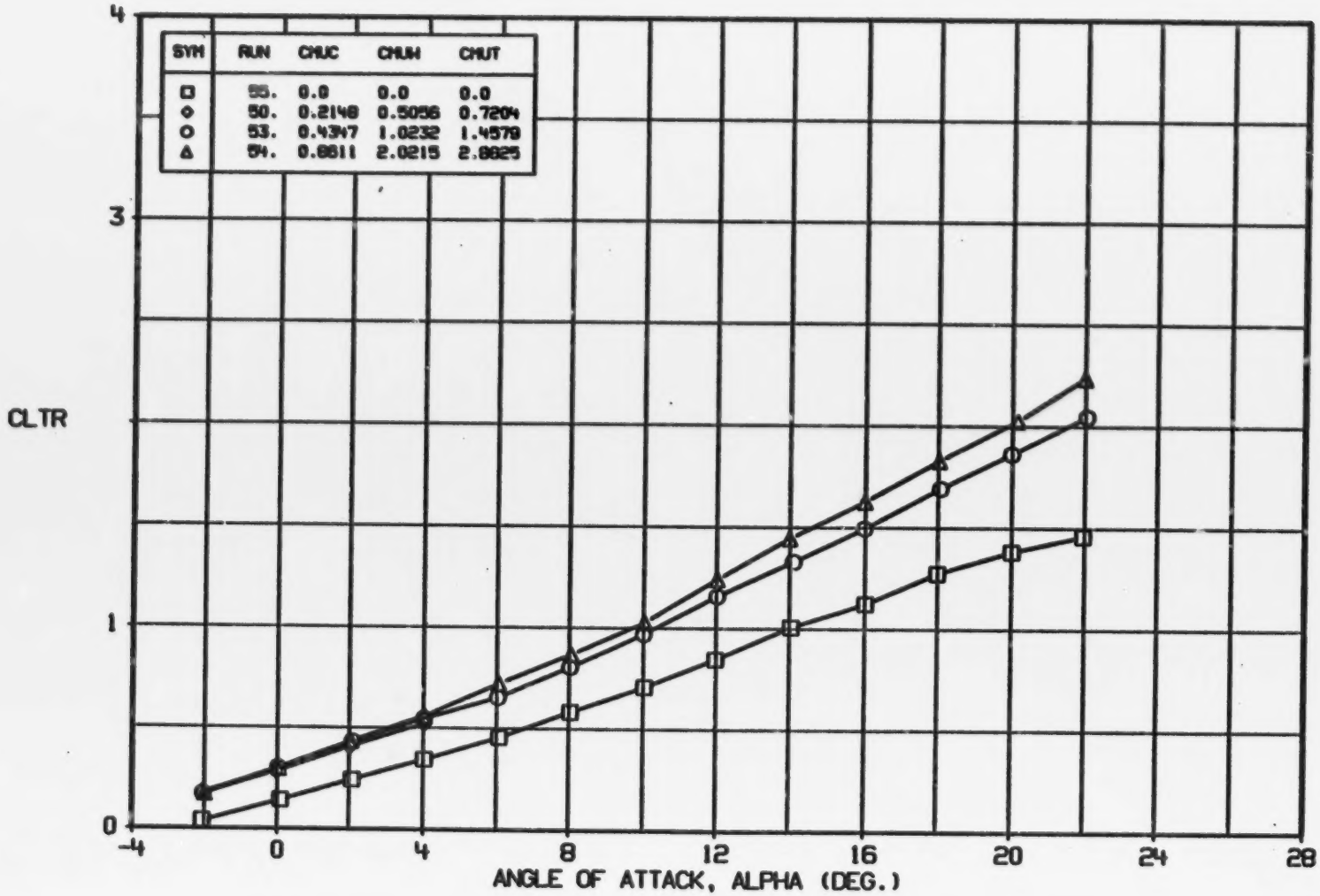

FIGURE A1d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=0, BN/B=1

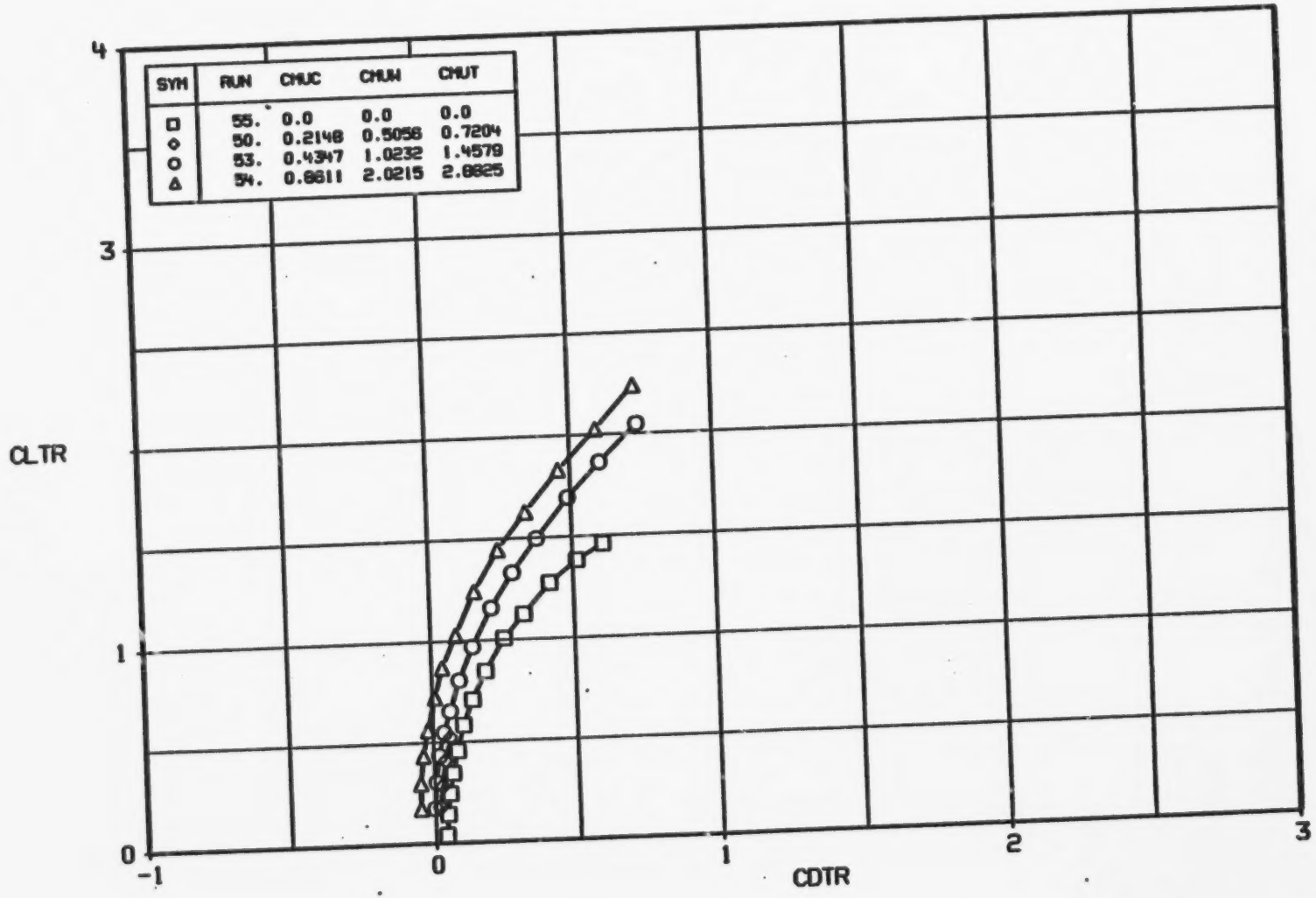

FIGURE A1e BASIC DATA EFFECT OF CMU
CONFIGURATION BC!W6V, DELF=0, BN/B=1

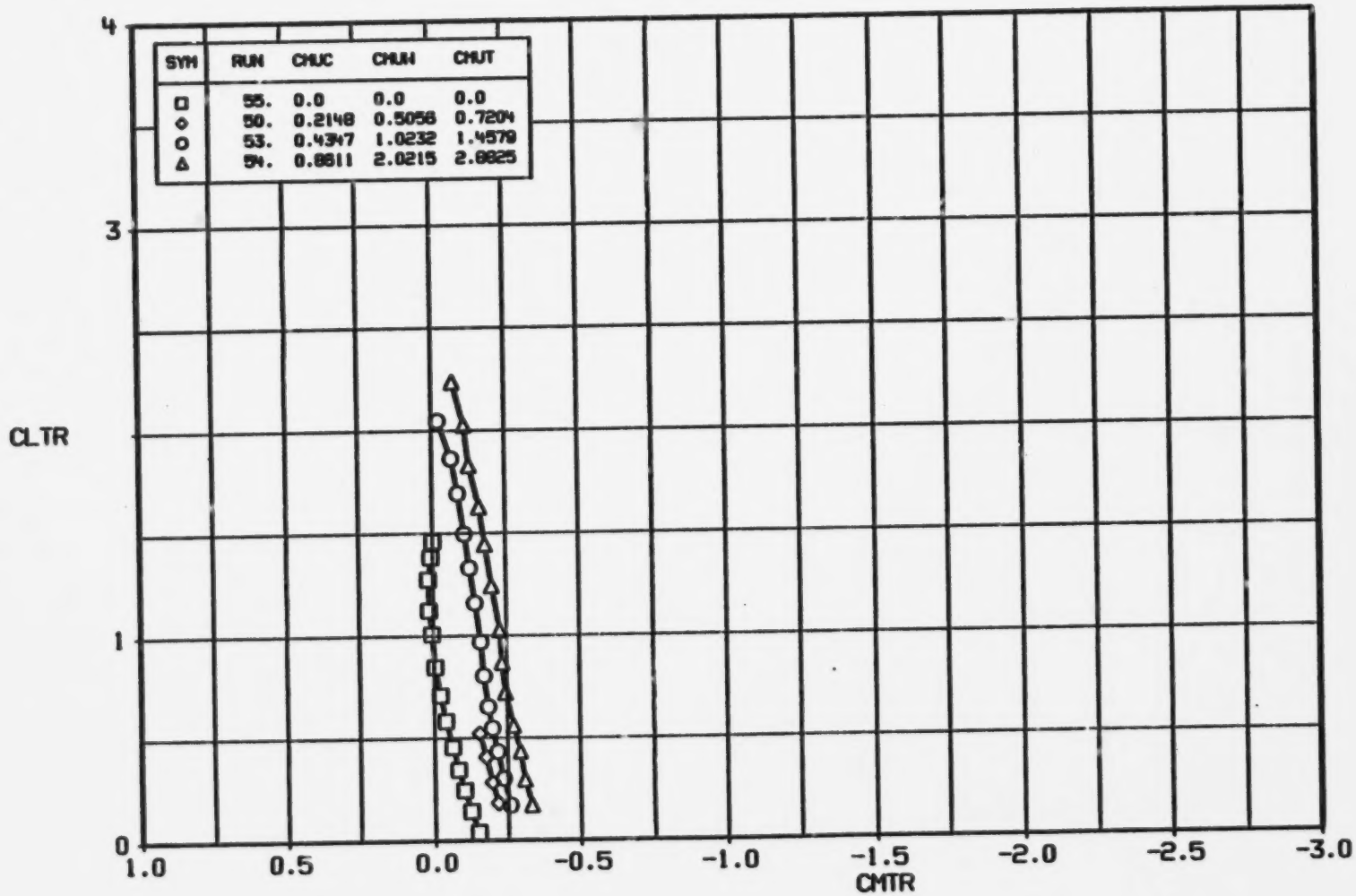

FIGURE A1f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=0, BN/B=1

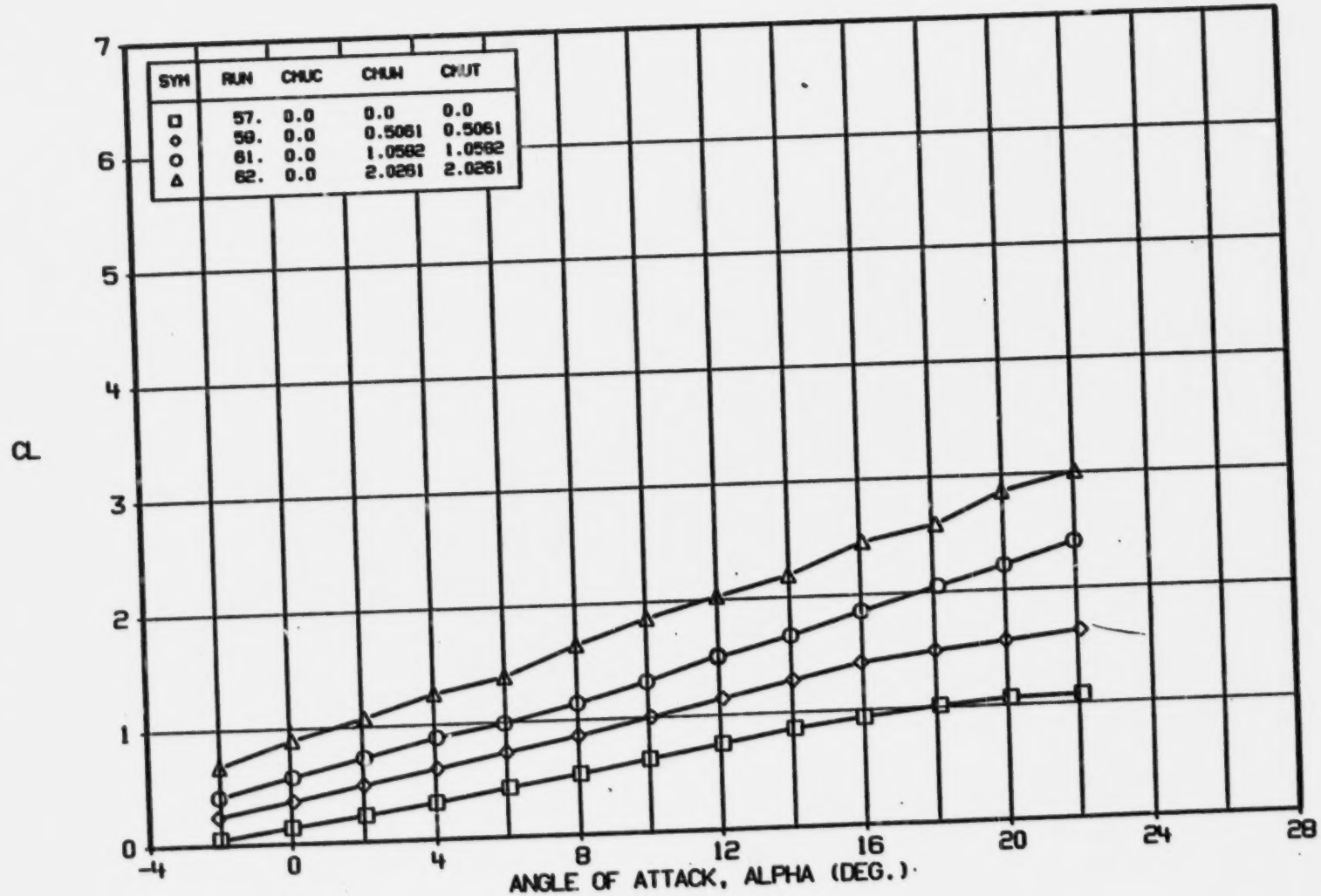

FIGURE A2a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

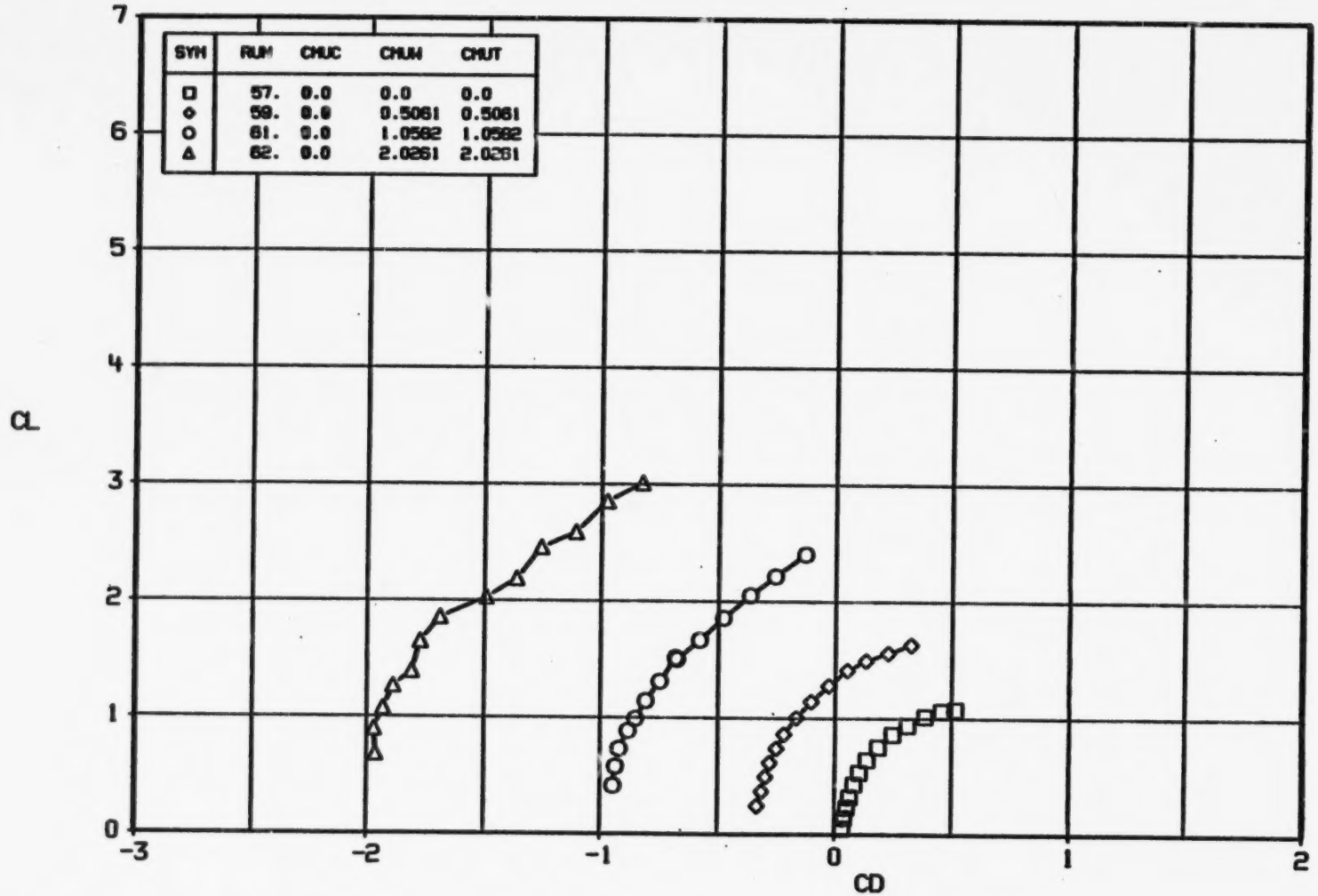

FIGURE A2b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

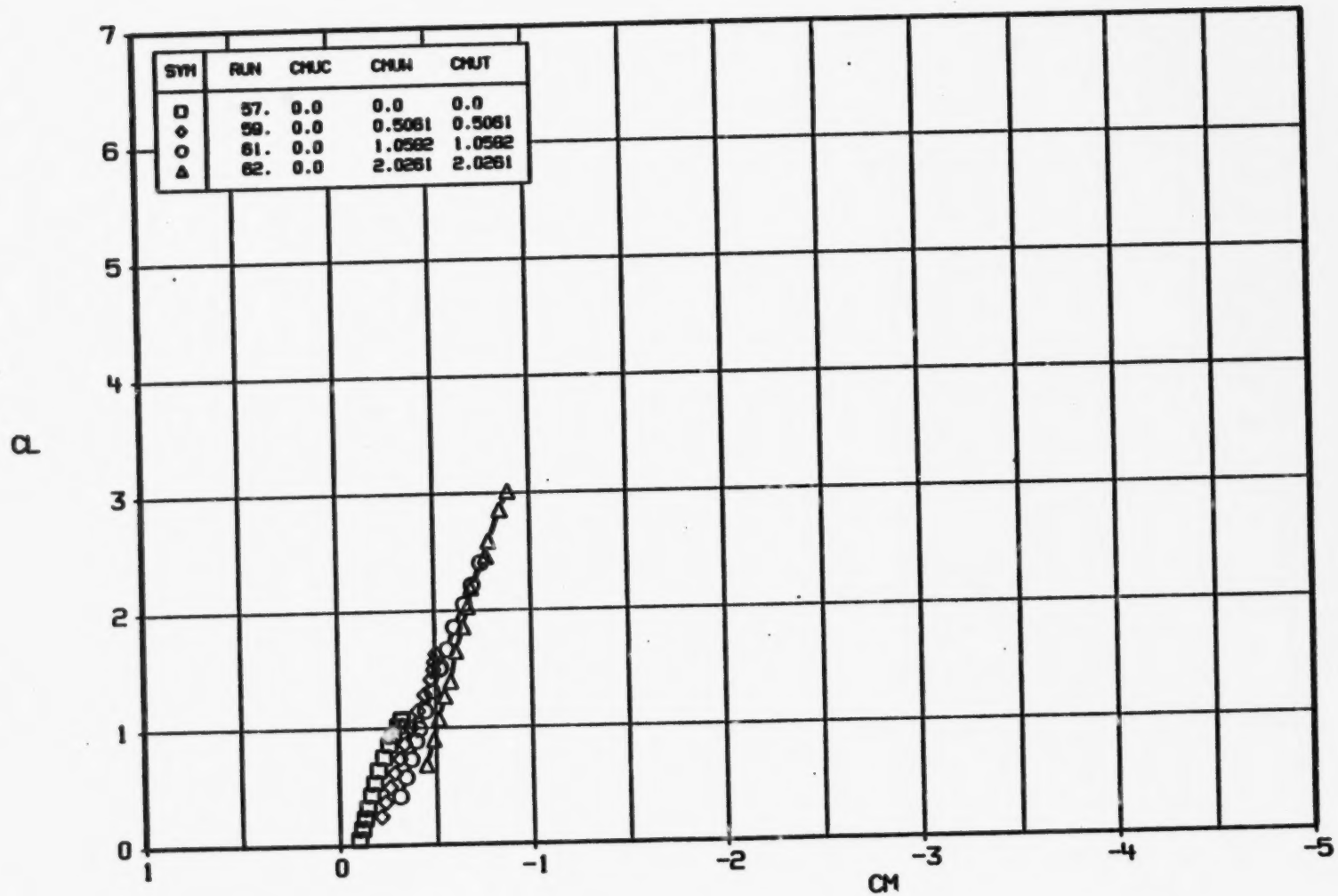

FIGURE A2c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

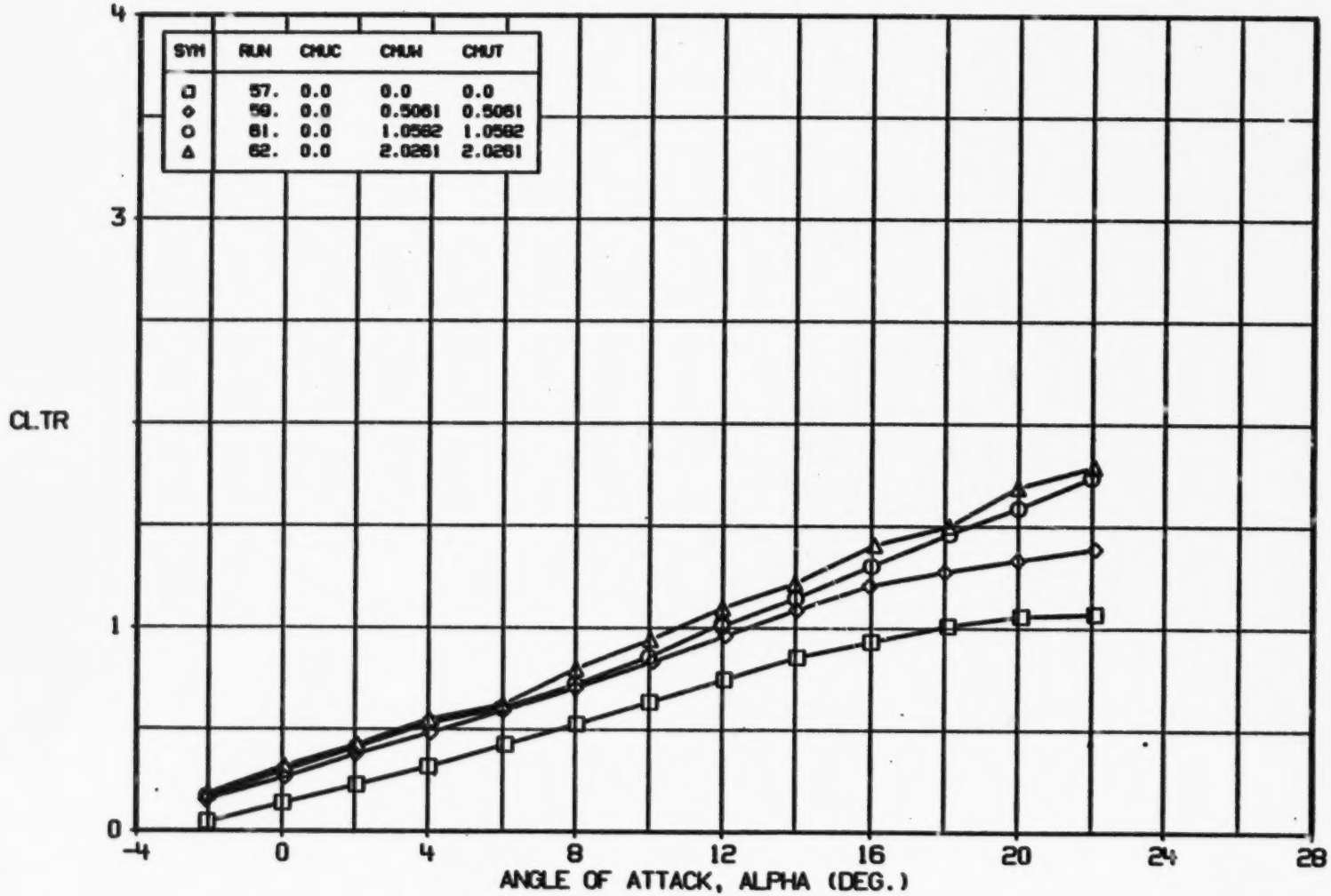

FIGURE A2d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

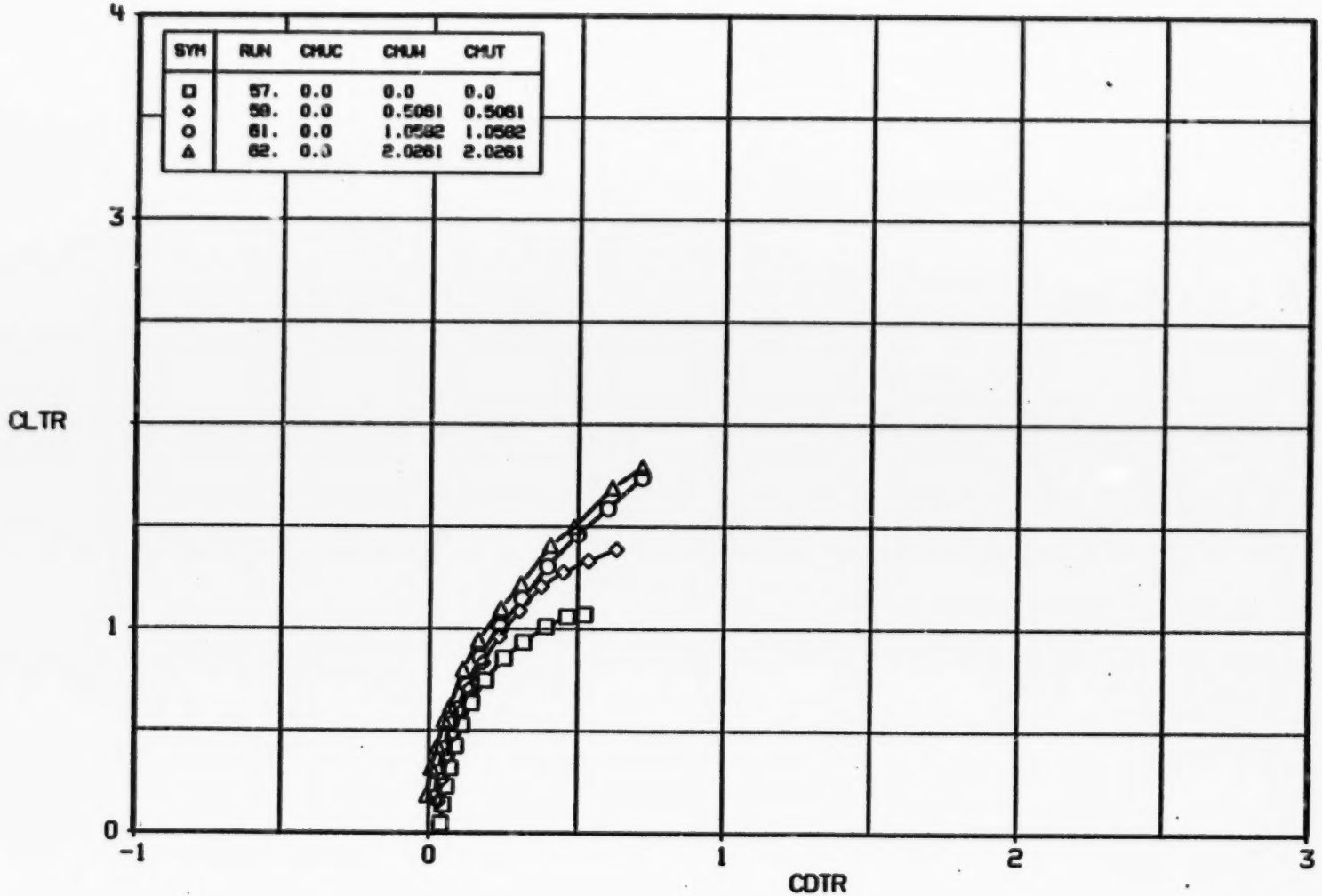

FIGURE A2e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

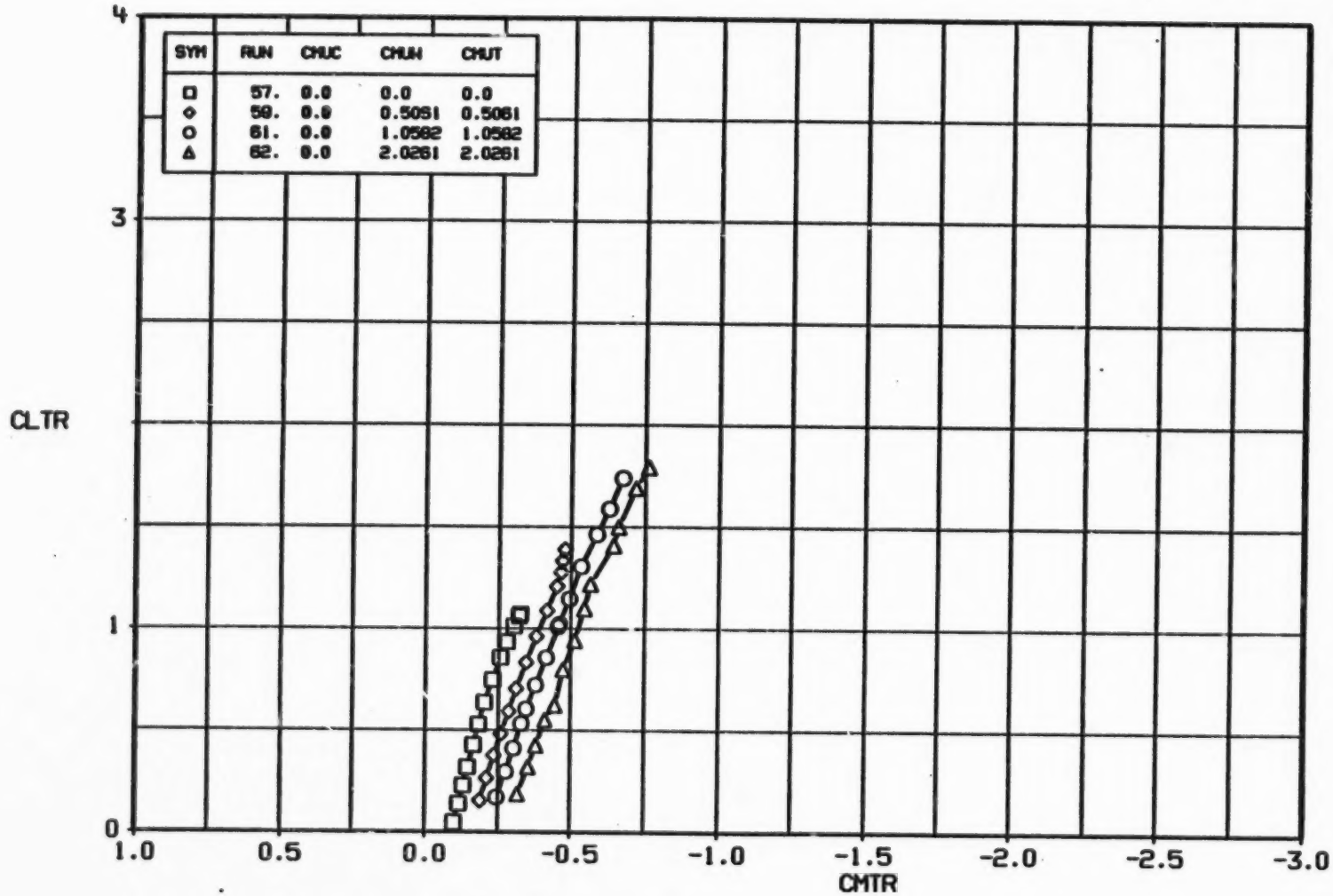

FIGURE A2f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=0, BN/B=1

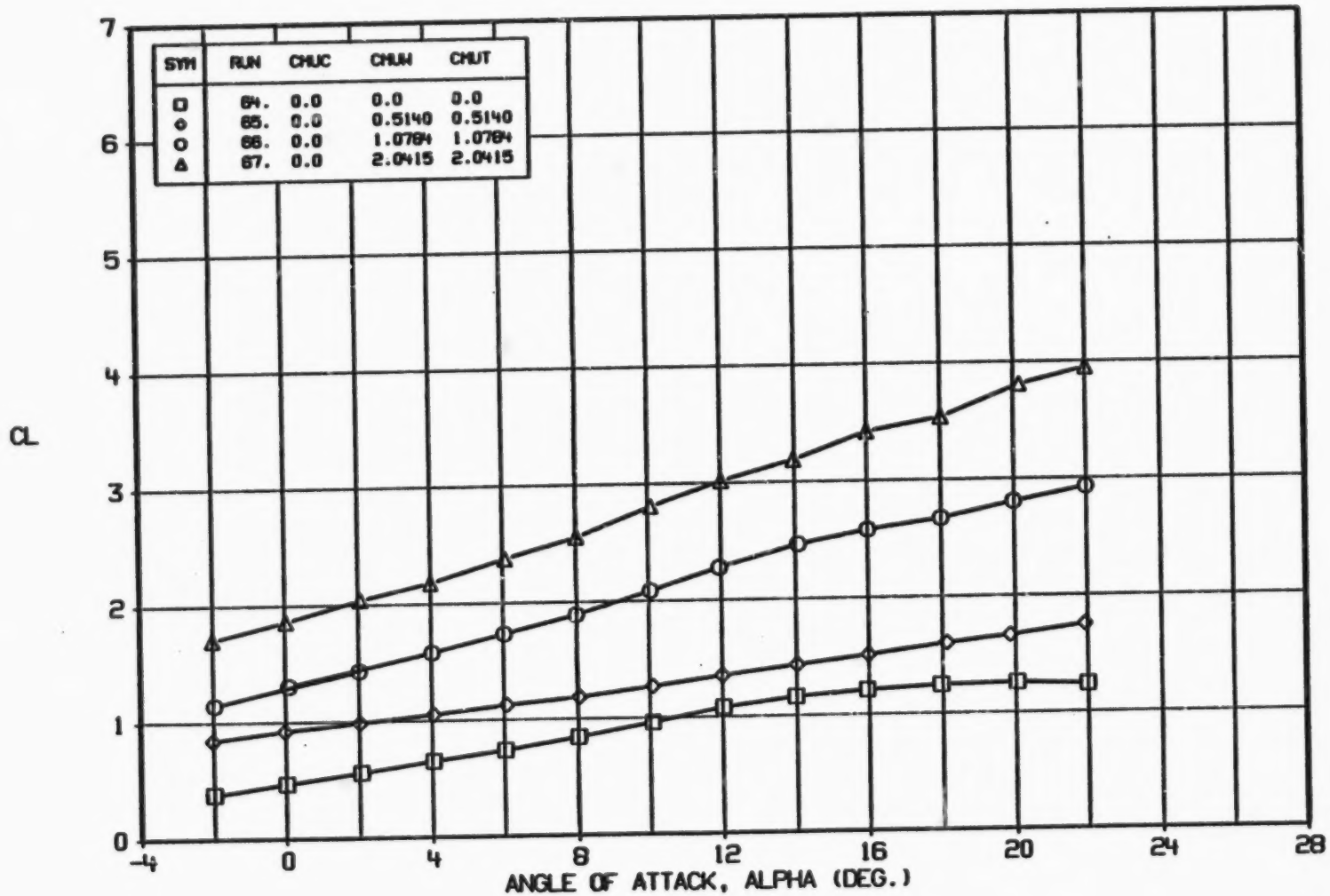

FIGURE A3a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=15, BN/B=1

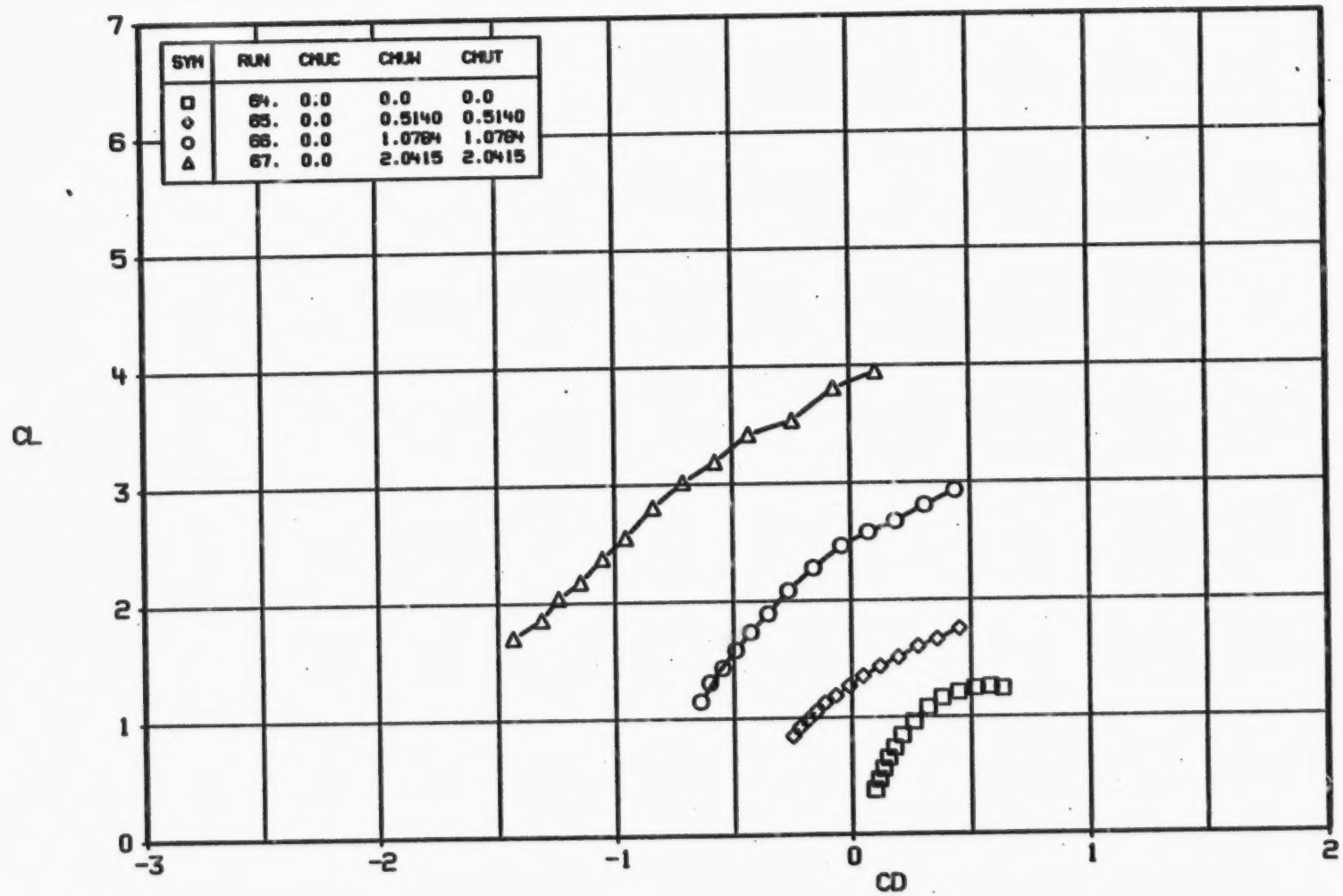

FIGURE A3b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=15, BN/B=1

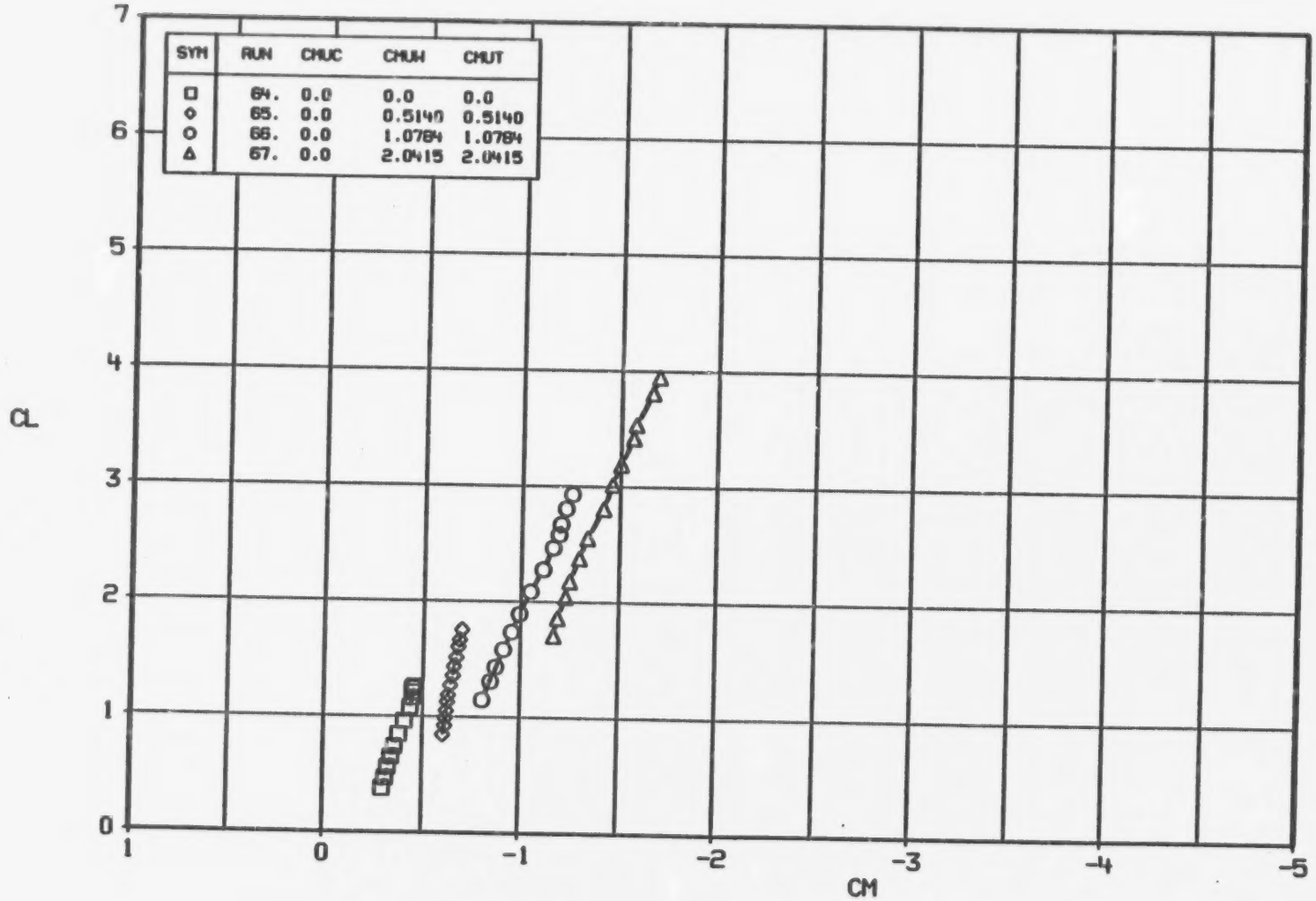

FIGURE A3c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=15, BN/B=1

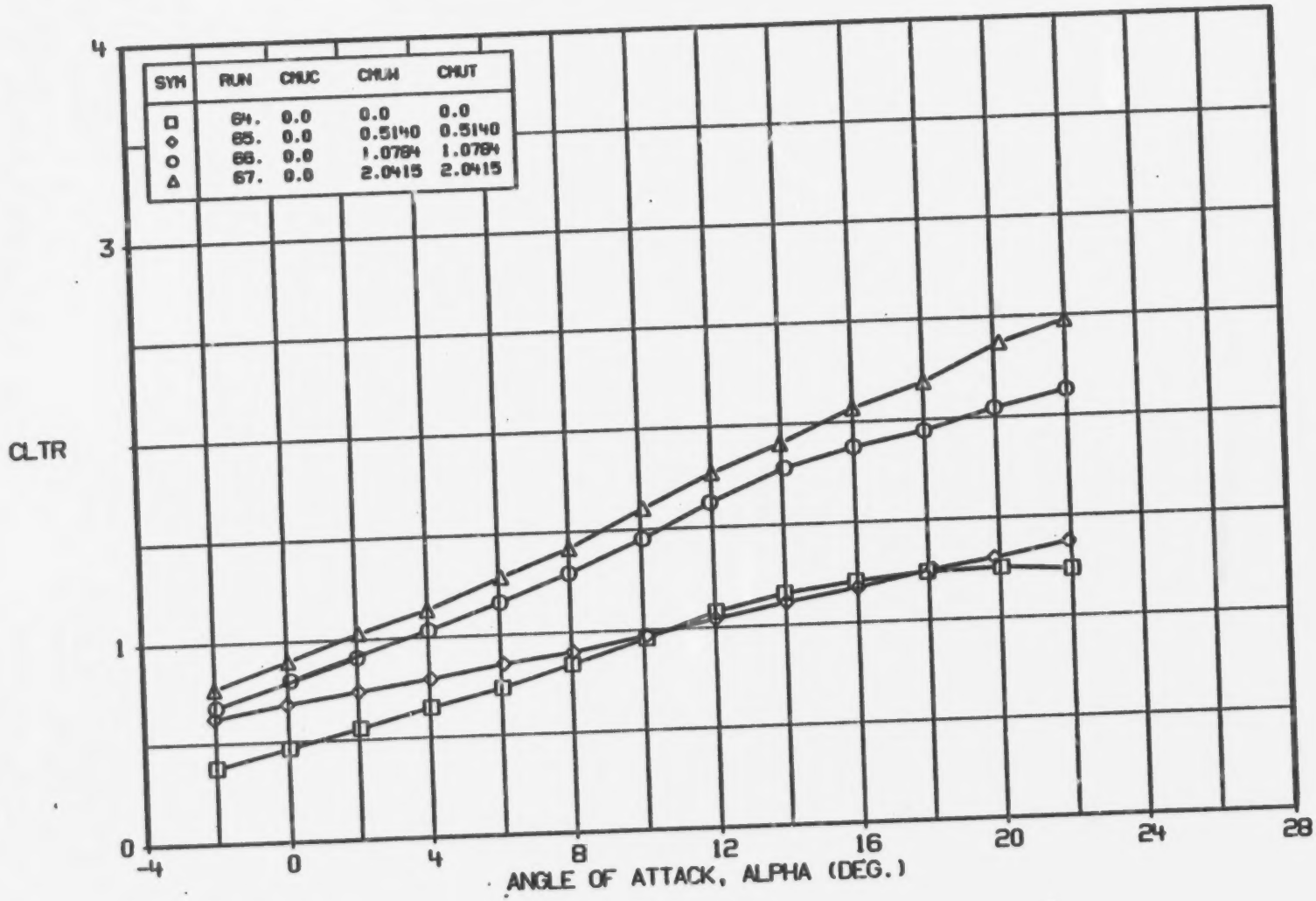

FIGURE A3d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=15, BN/B=1

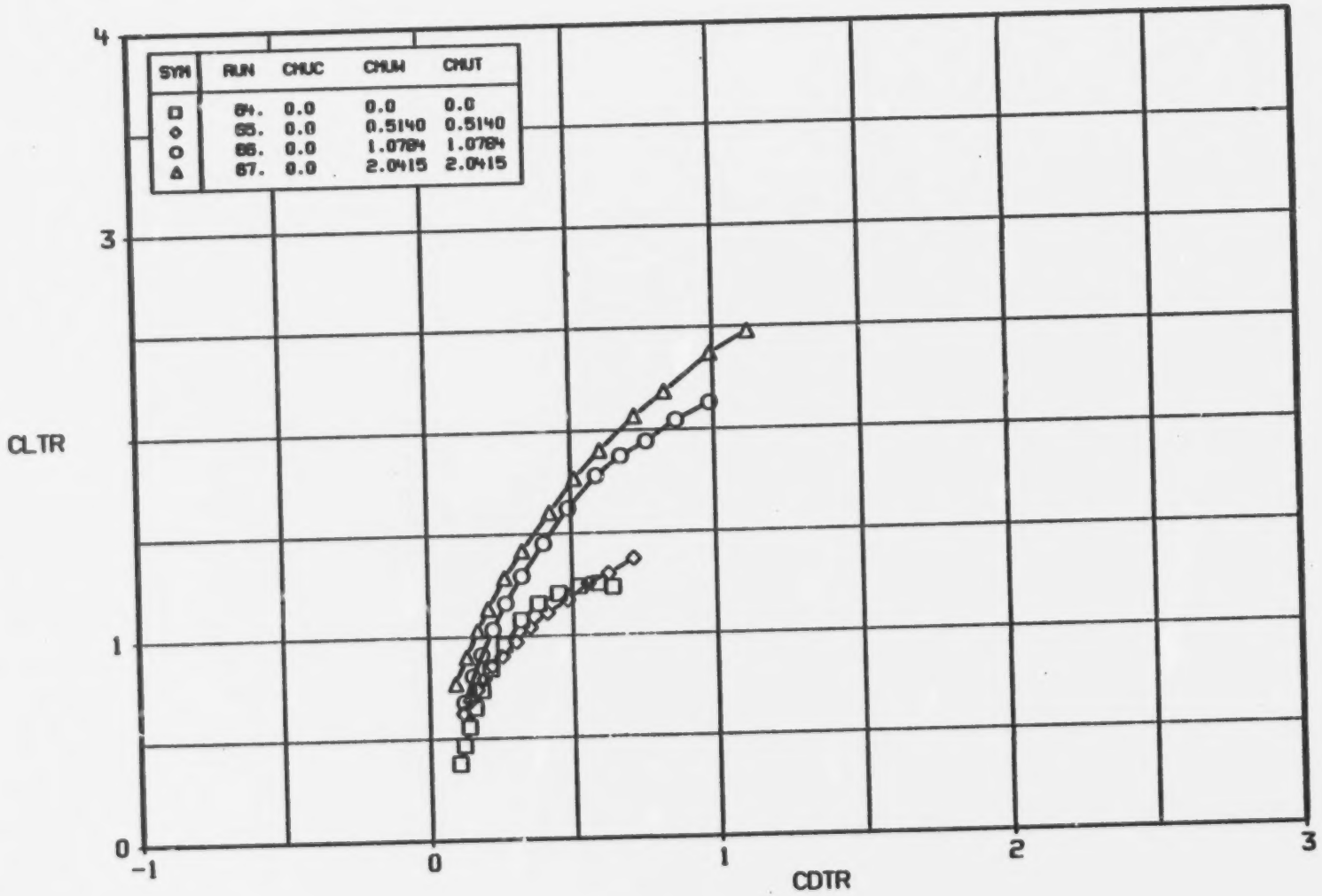

FIGURE A3e BASIC DATA EFFECT OF CMU
CONFIGURATION EW6V, DELF=15, BN/B=1

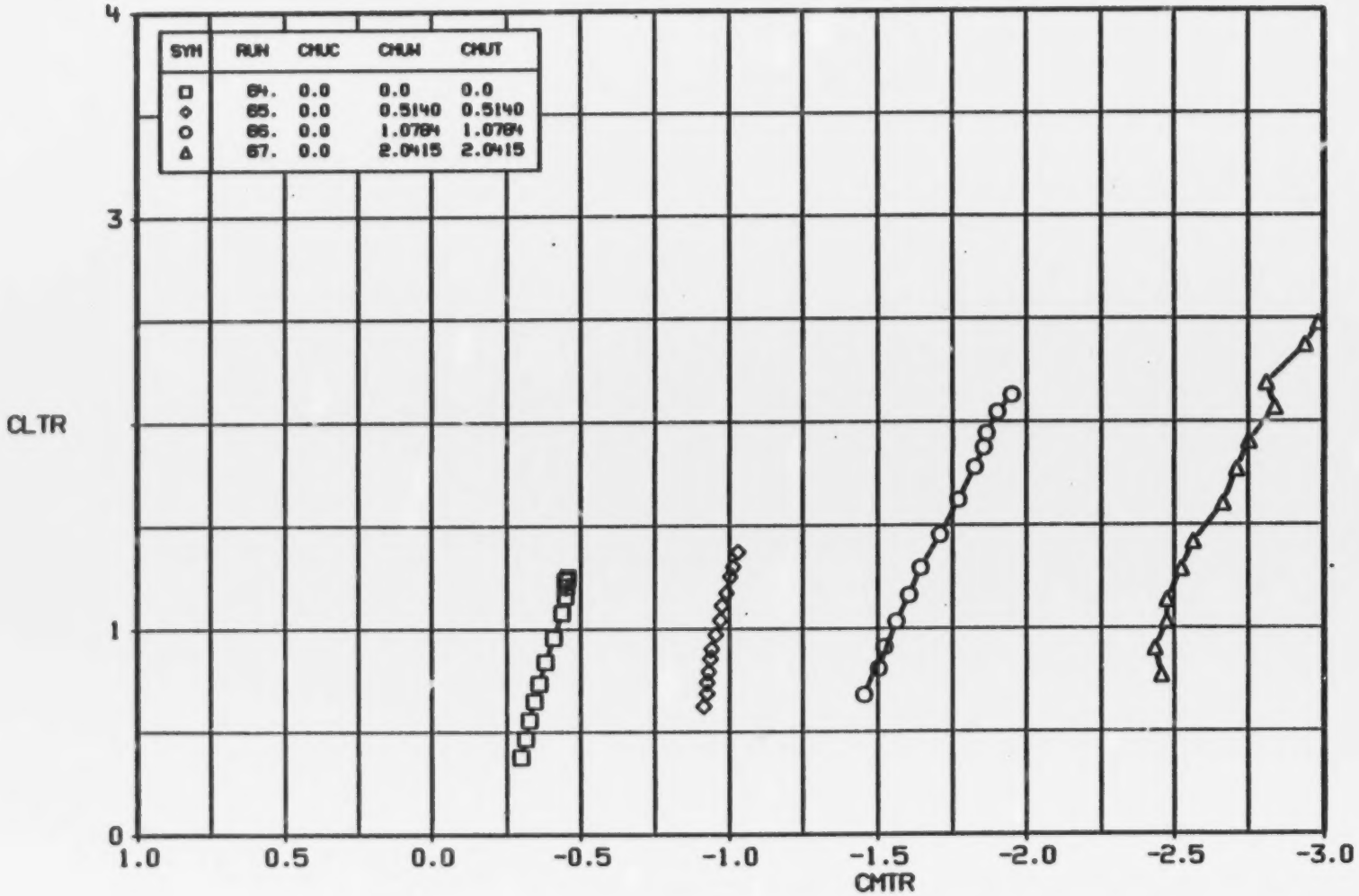

FIGURE A3f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=15, BN/B=1

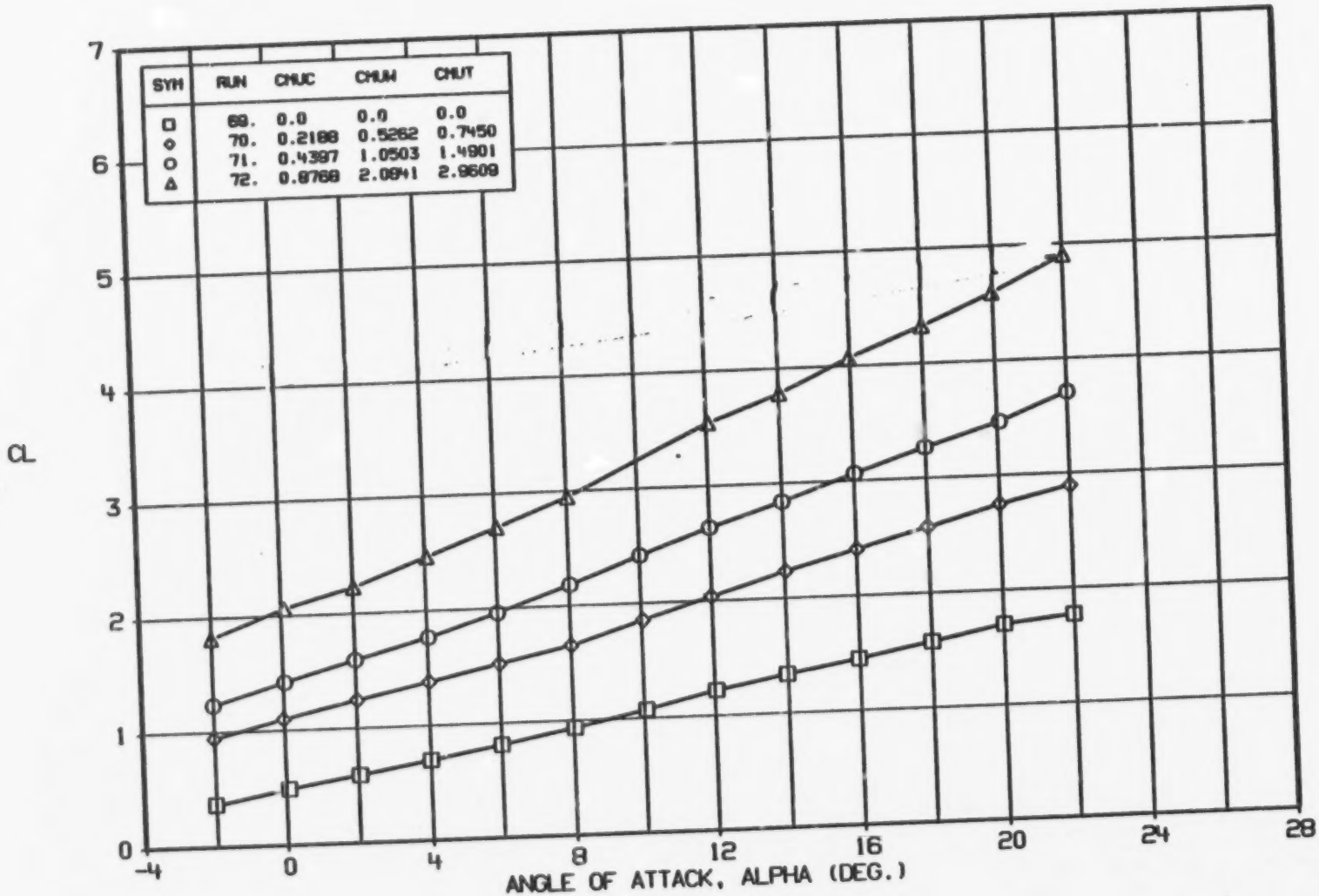

FIGURE A4a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, DELF=15, BN/B=1

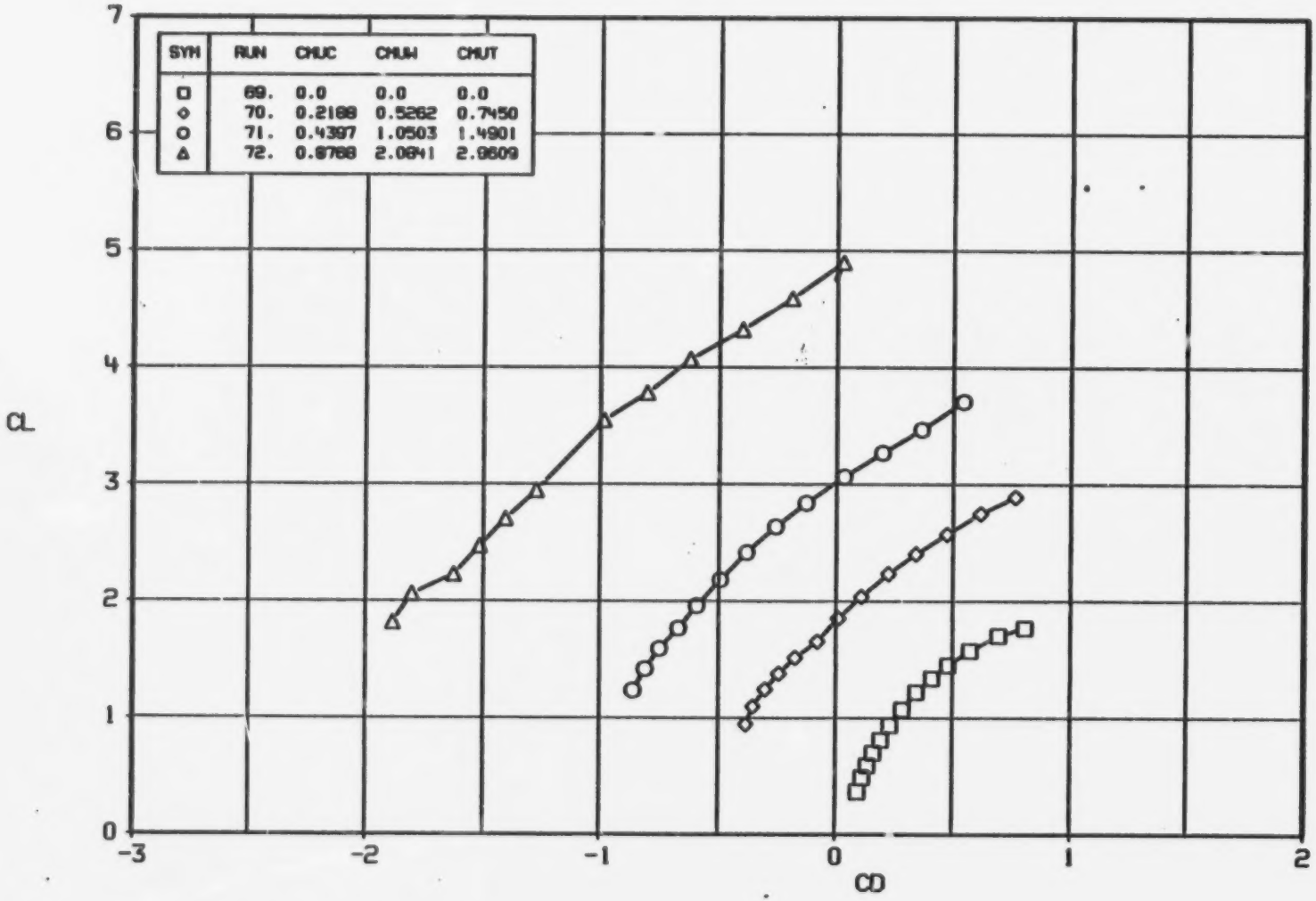

FIGURE A4b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=15, BN/B=1

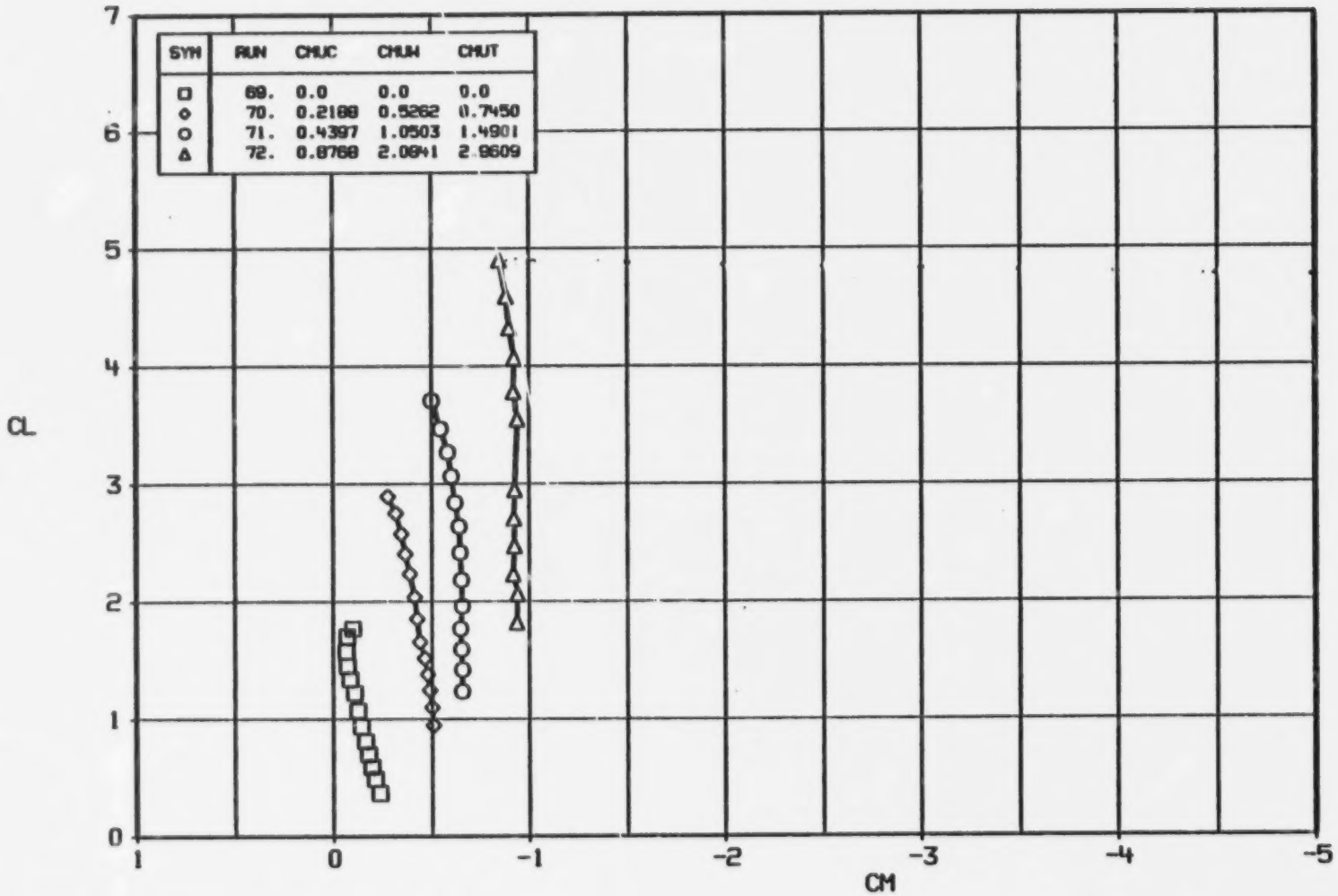

FIGURE A4c BASIC DATA EFFECT OF CMU
CONFIGURATION BCIW6V, DELF=15, BN/B=1

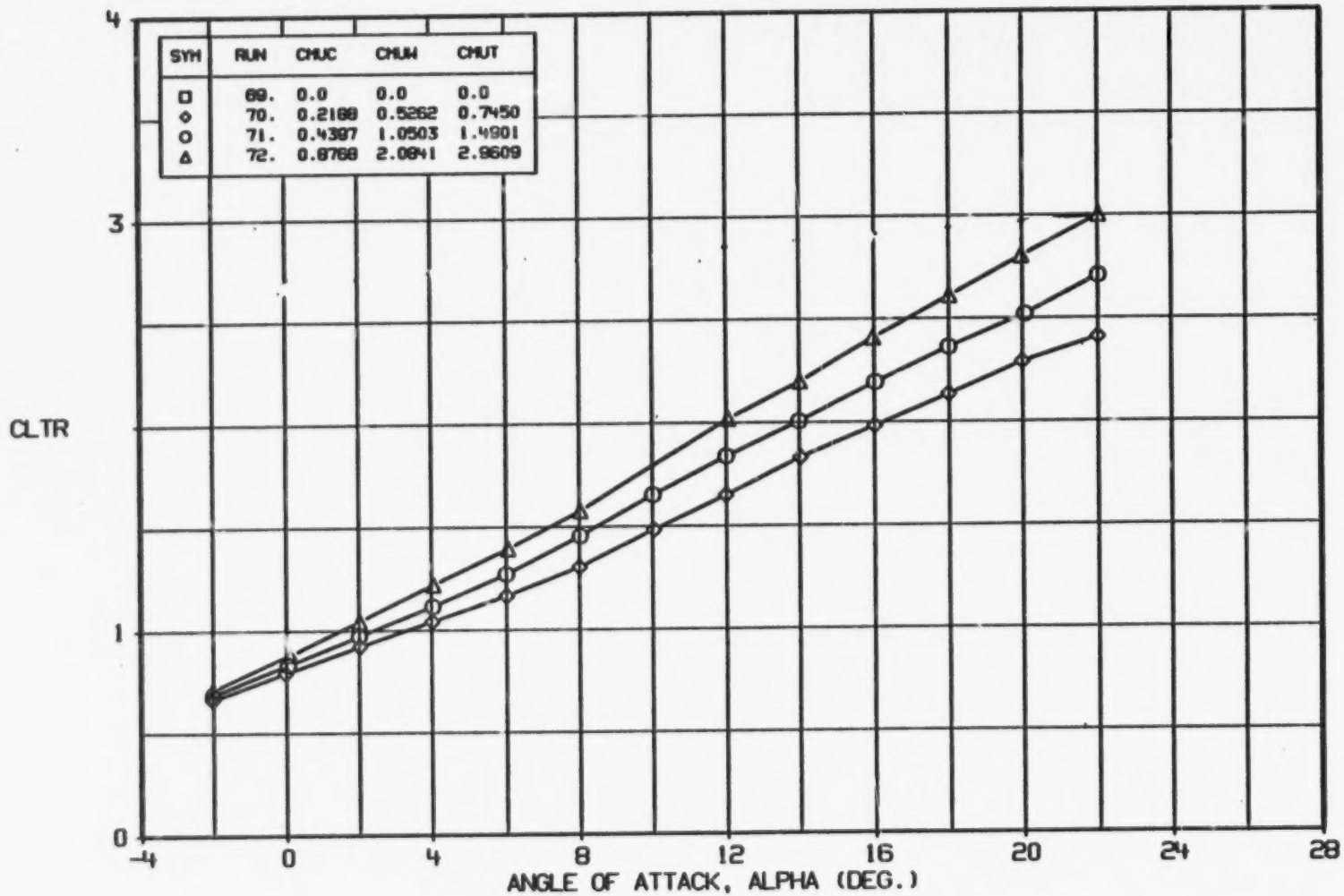

FIGURE A4d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=15, BN/B=1

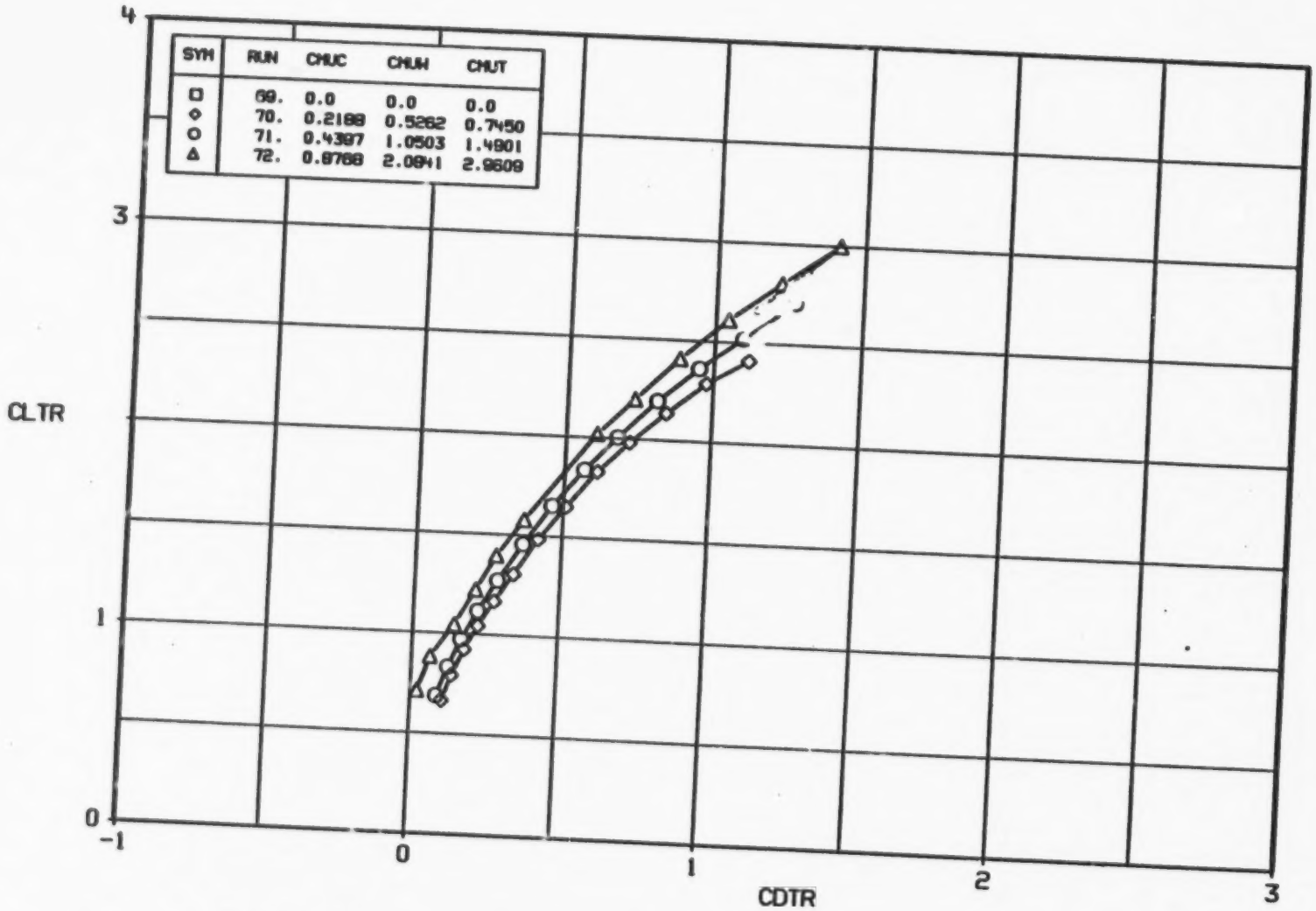

FIGURE A4e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=15, BN/B=1

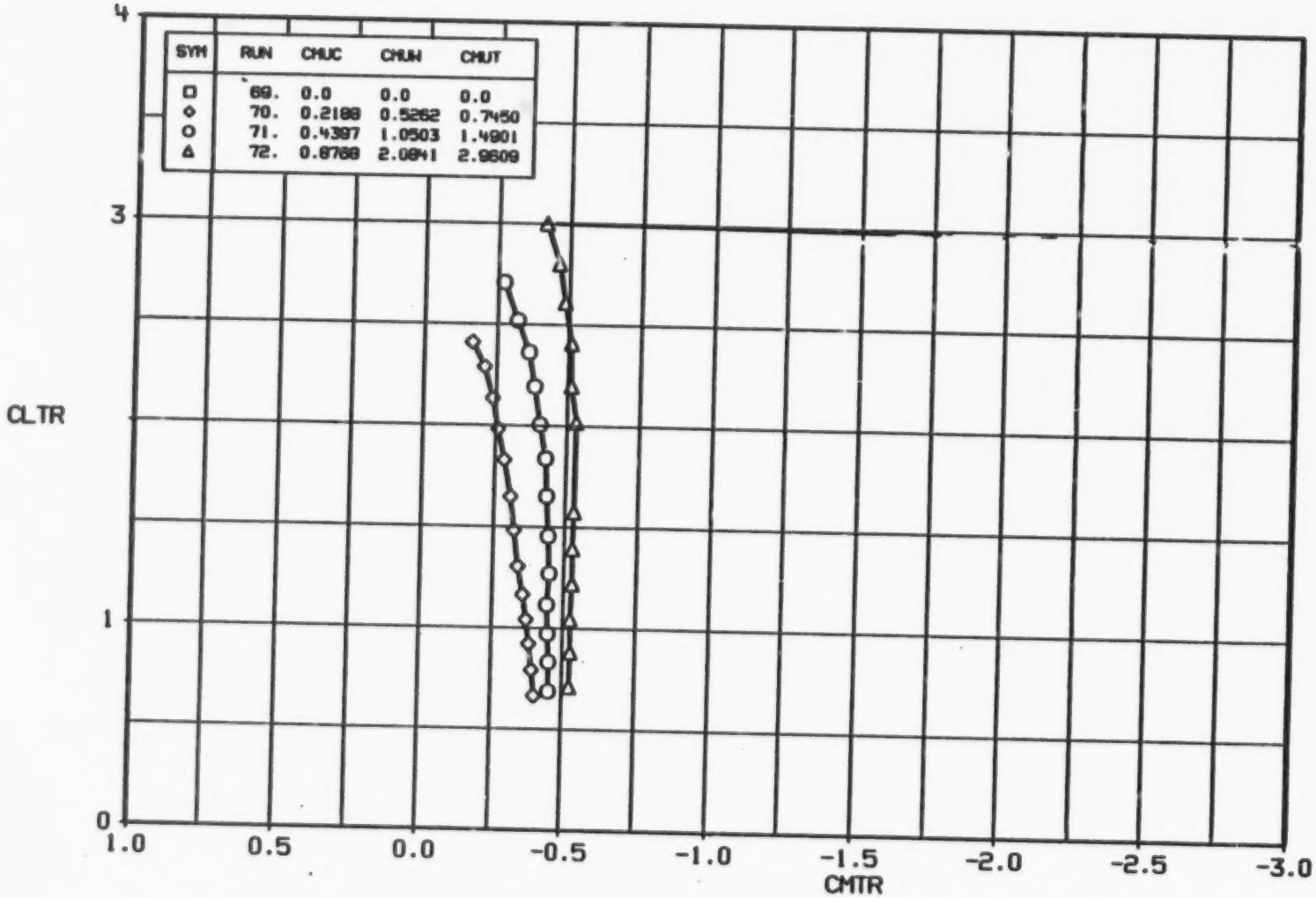

FIGURE A4f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=15, BN/B=1

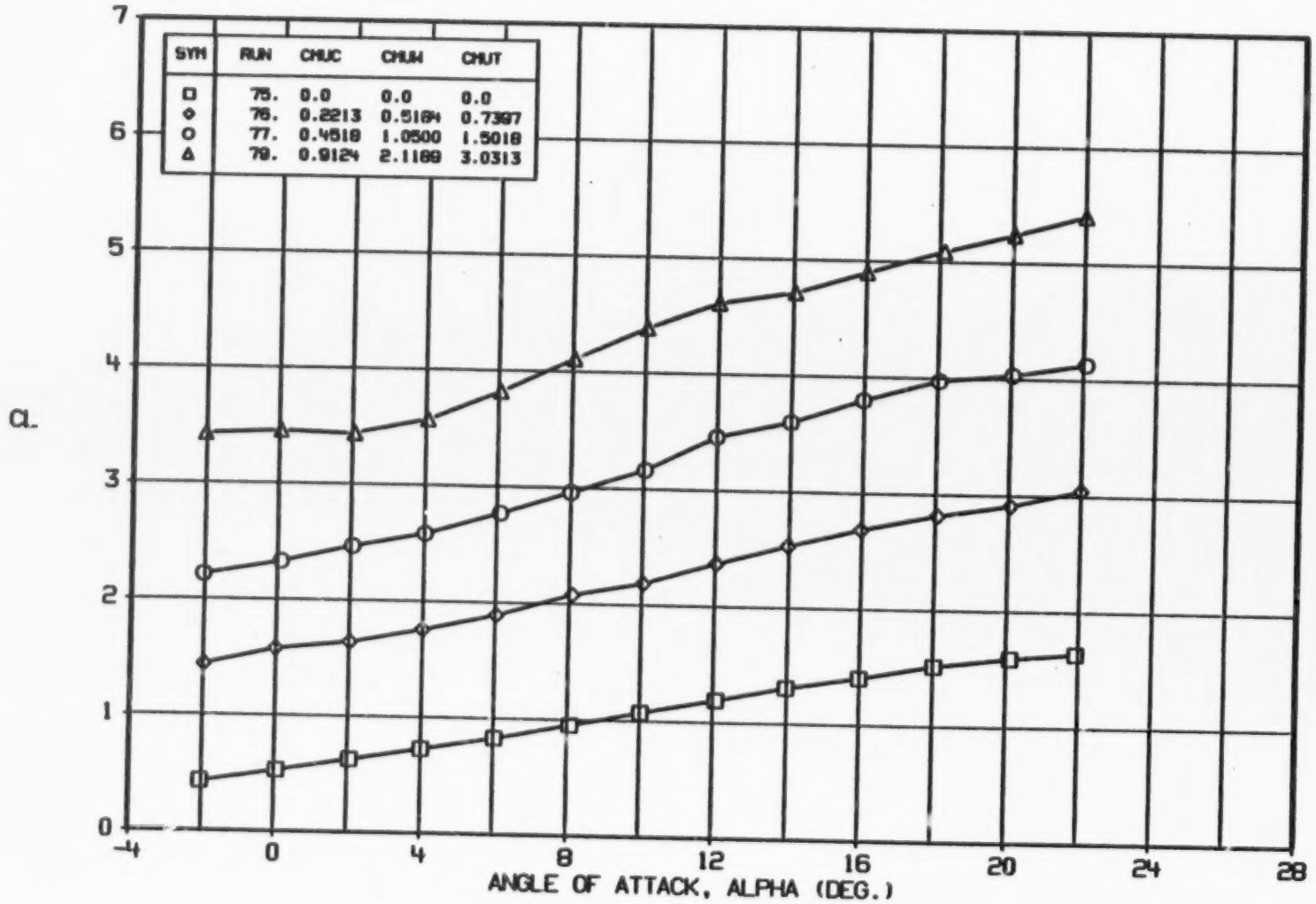

FIGURE A5a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

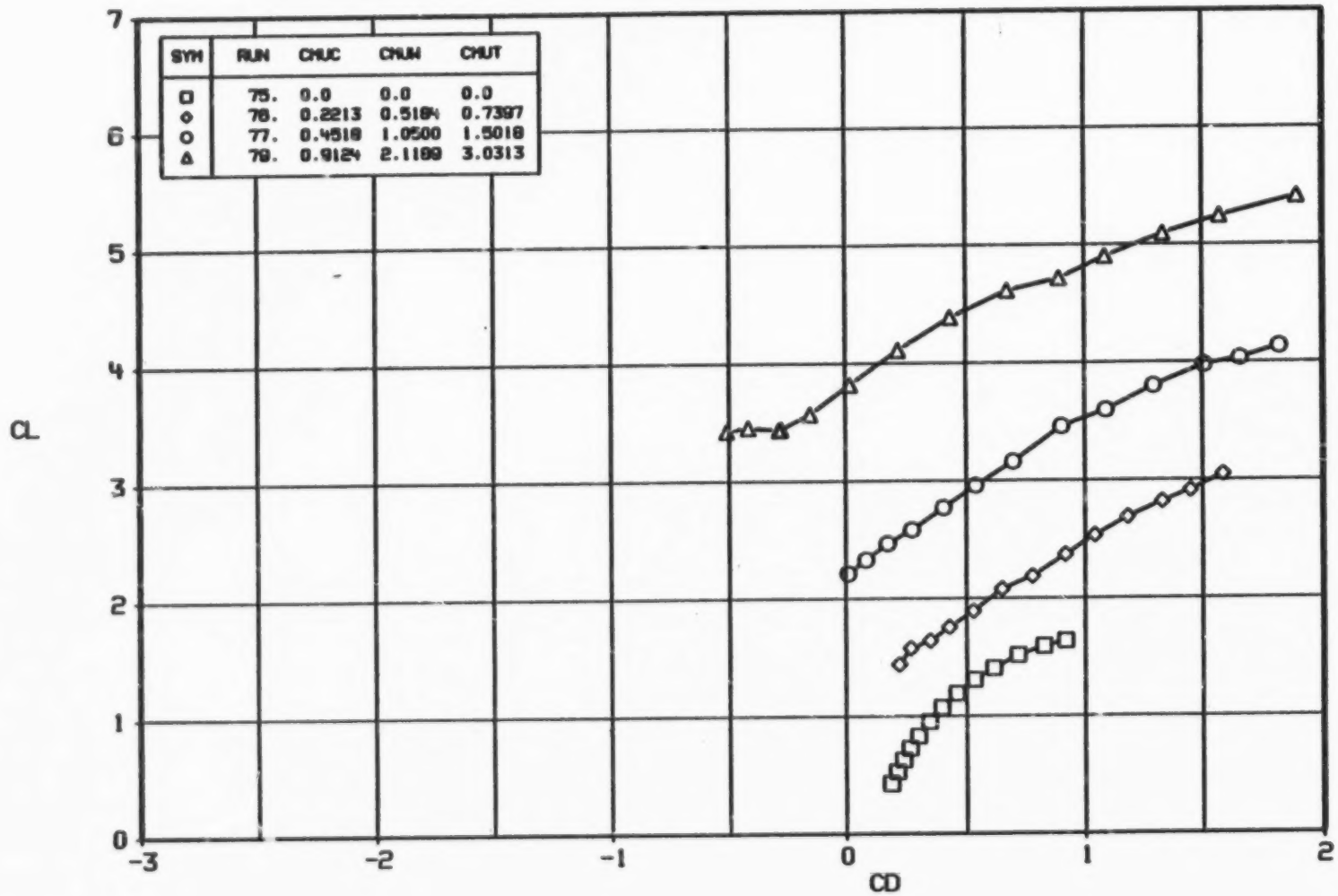

FIGURE A5b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, DELF=45, BN/B=1

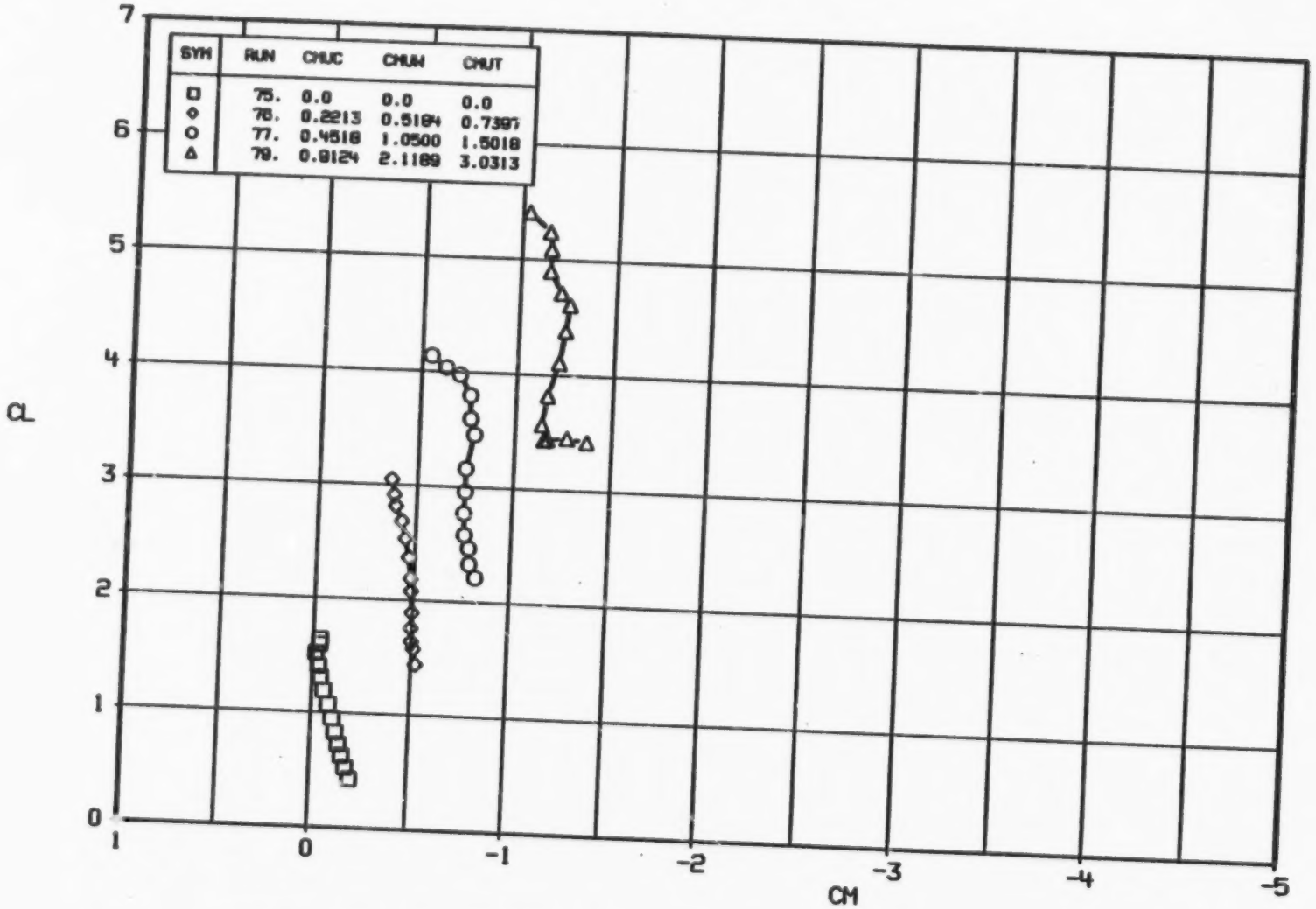

FIGURE A5c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

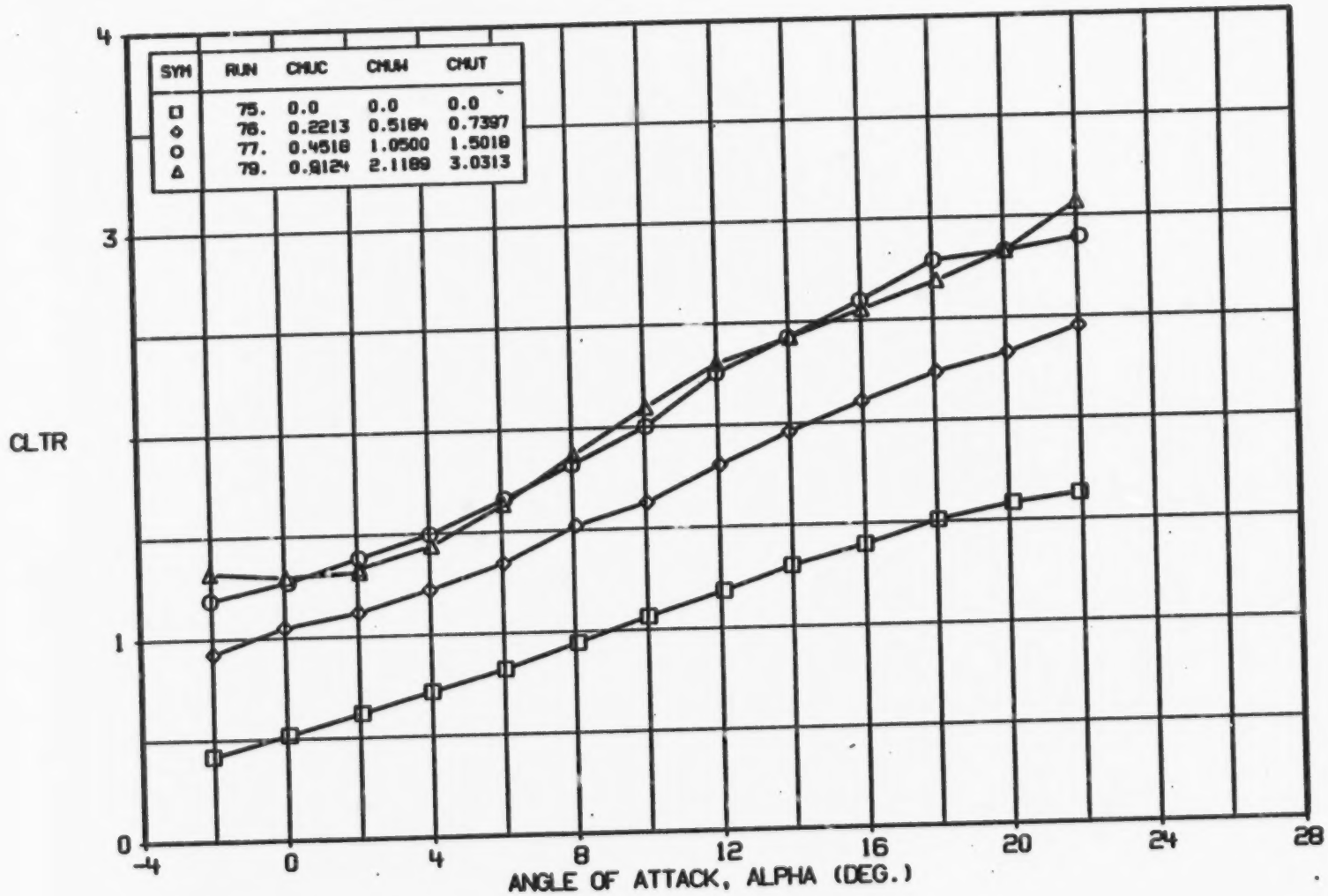

FIGURE A5d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

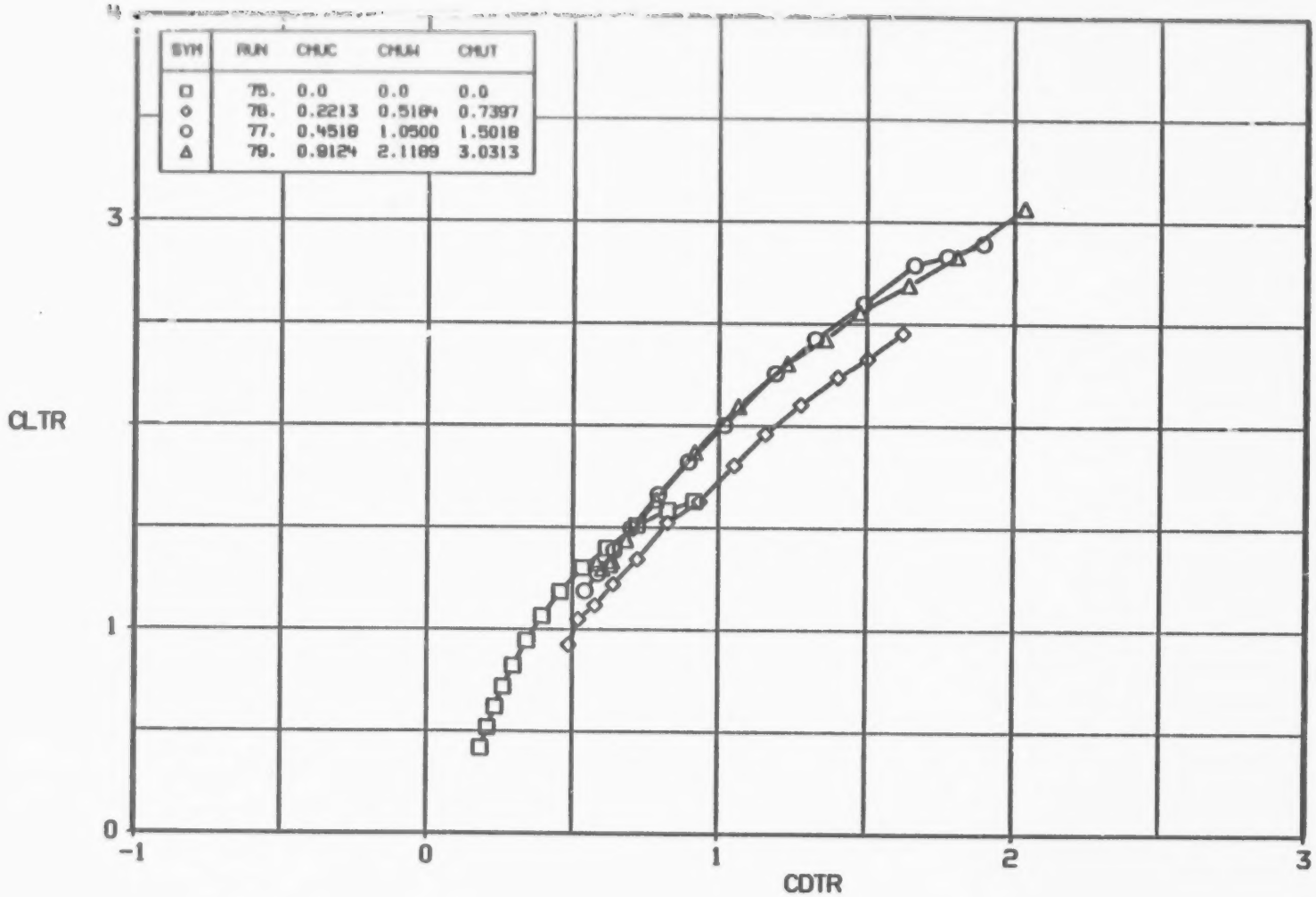

FIGURE A5e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

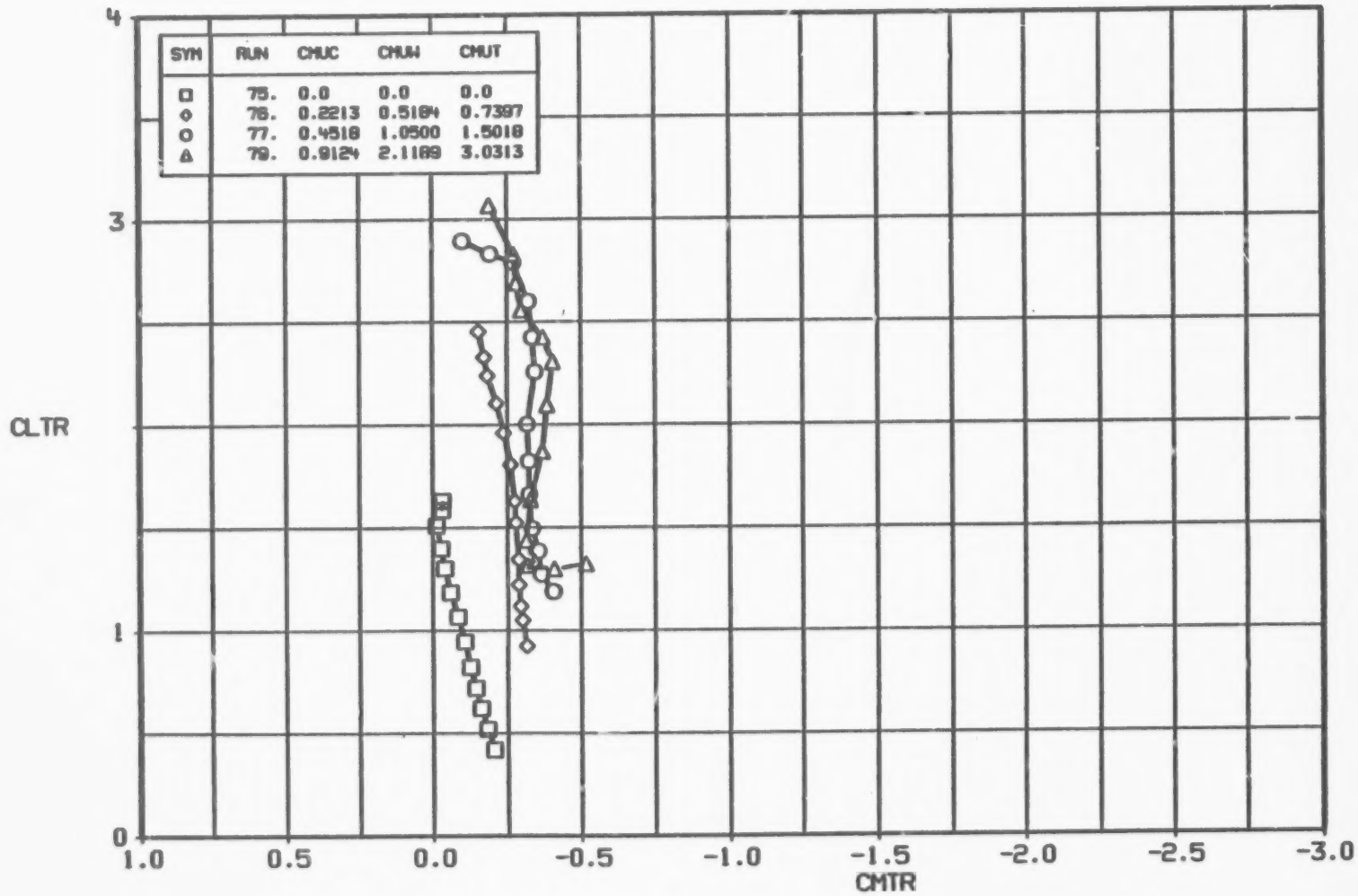

FIGURE A5f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

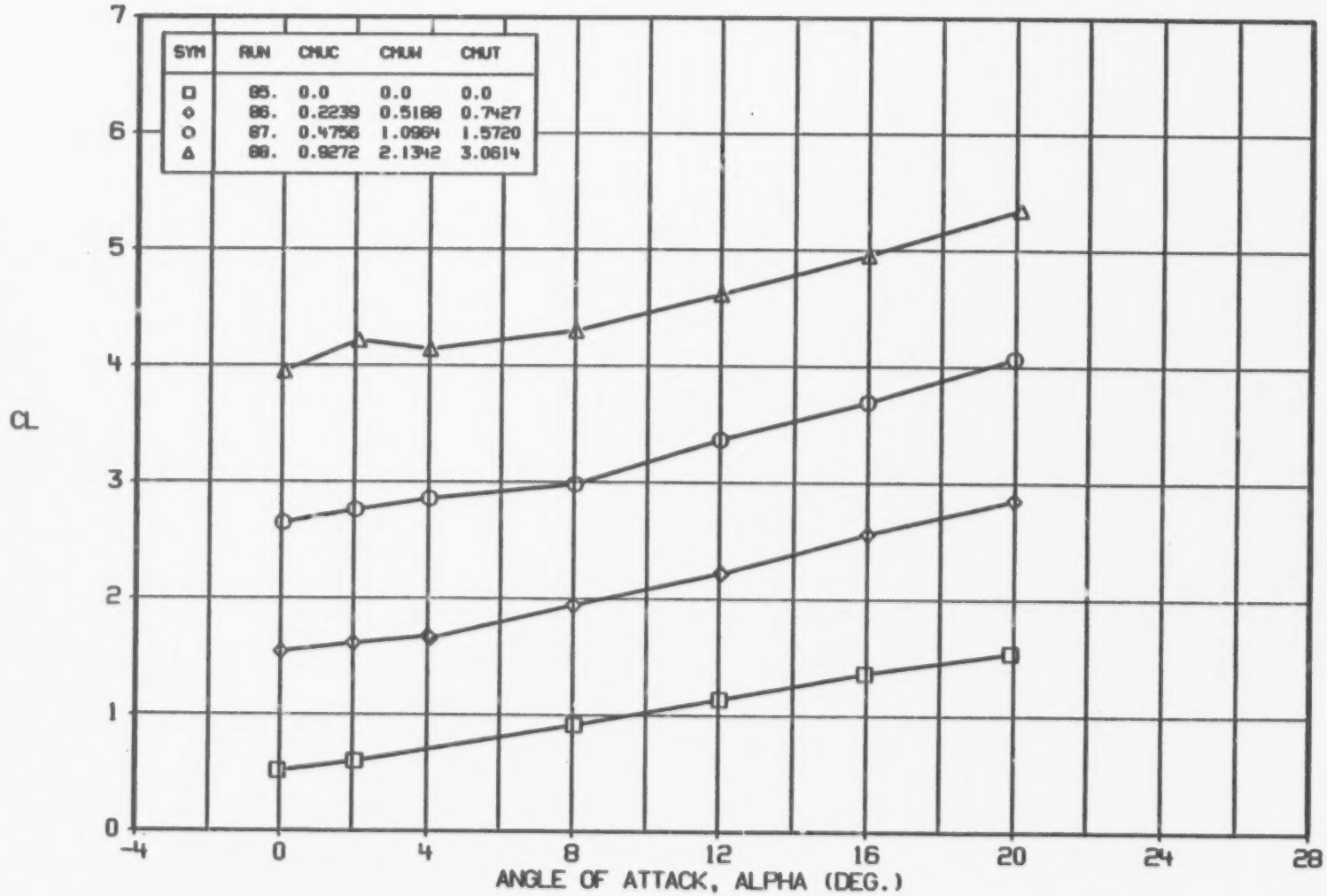

FIGURE A6a BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

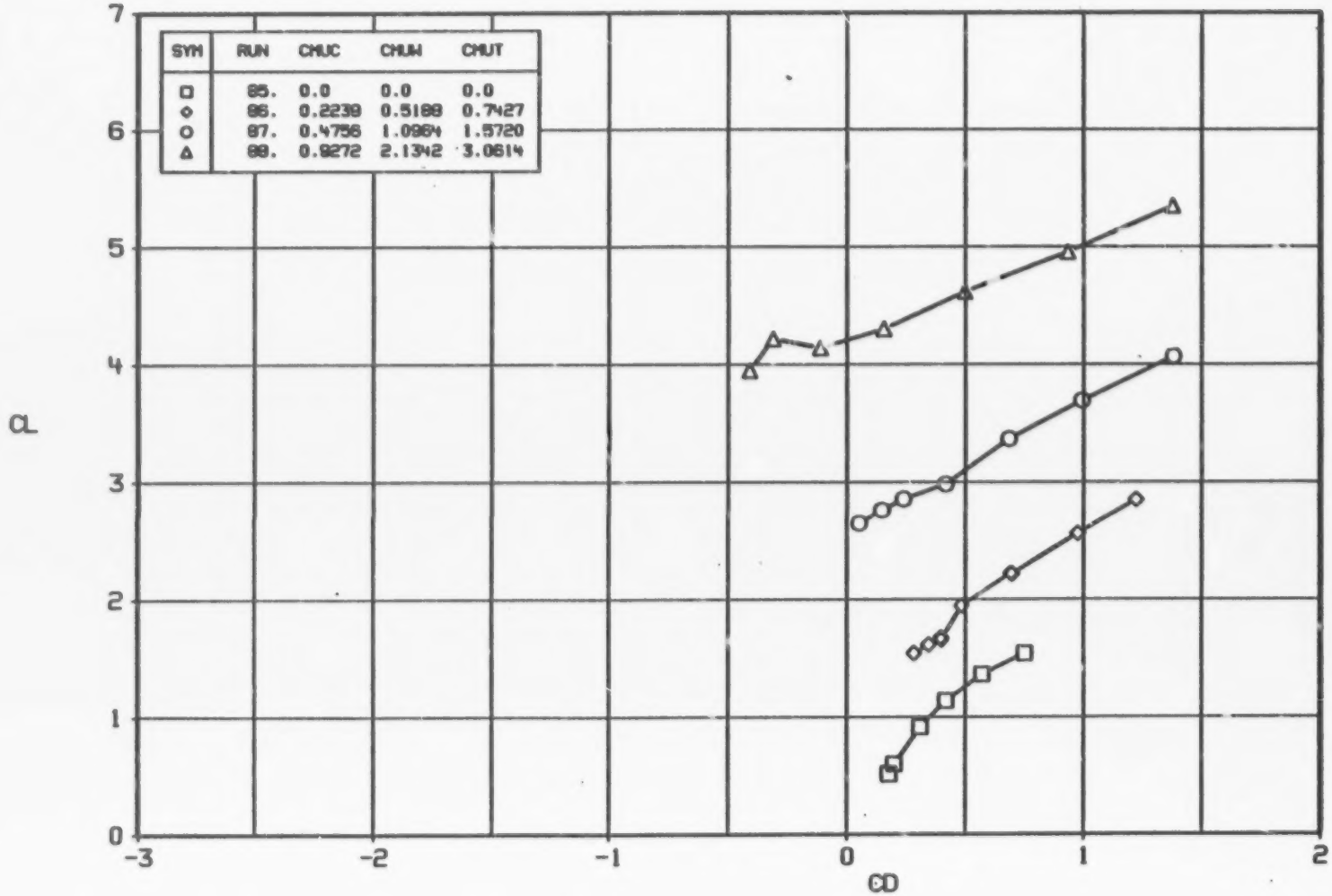

FIGURE A6b BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

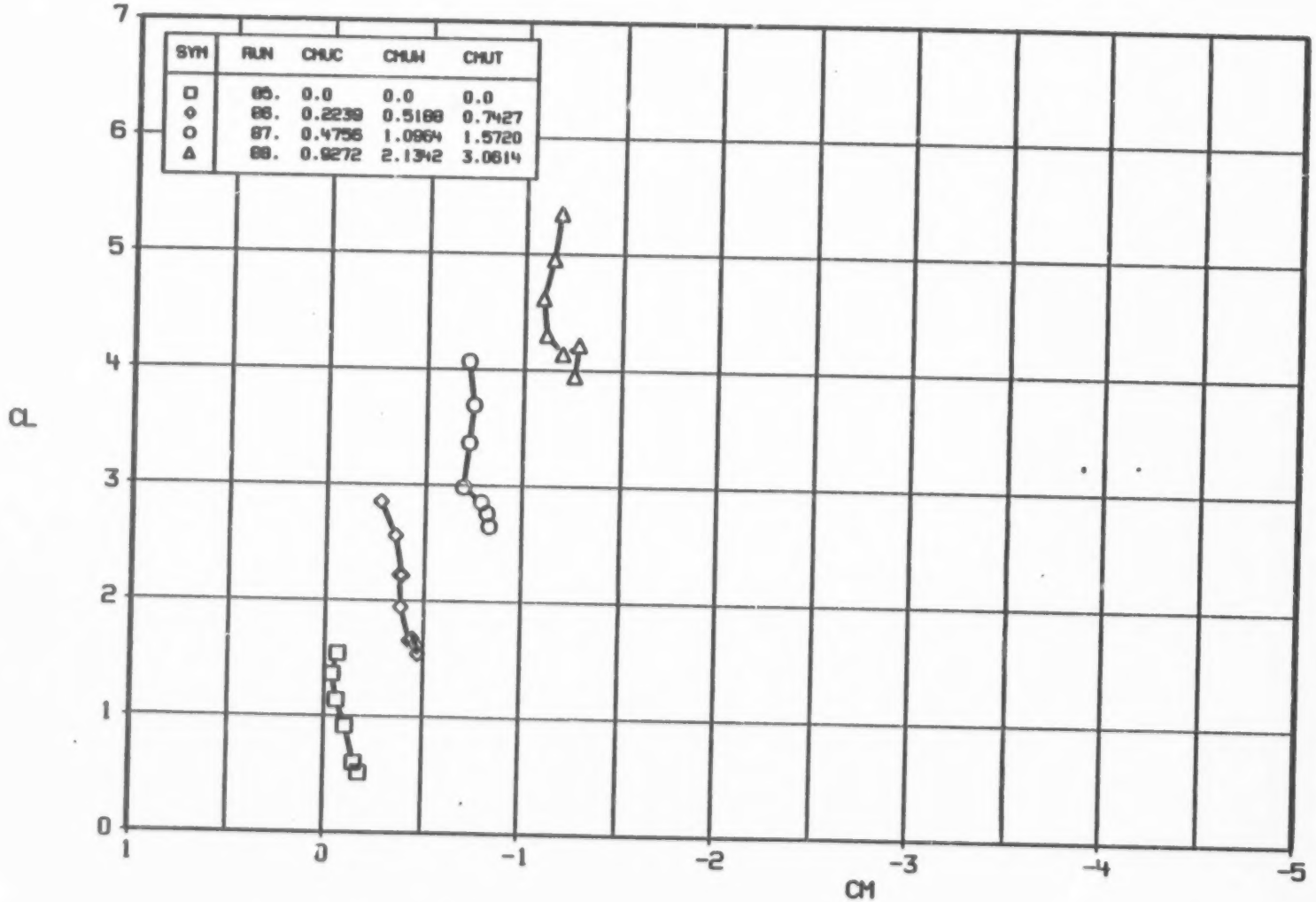

FIGURE A6c BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

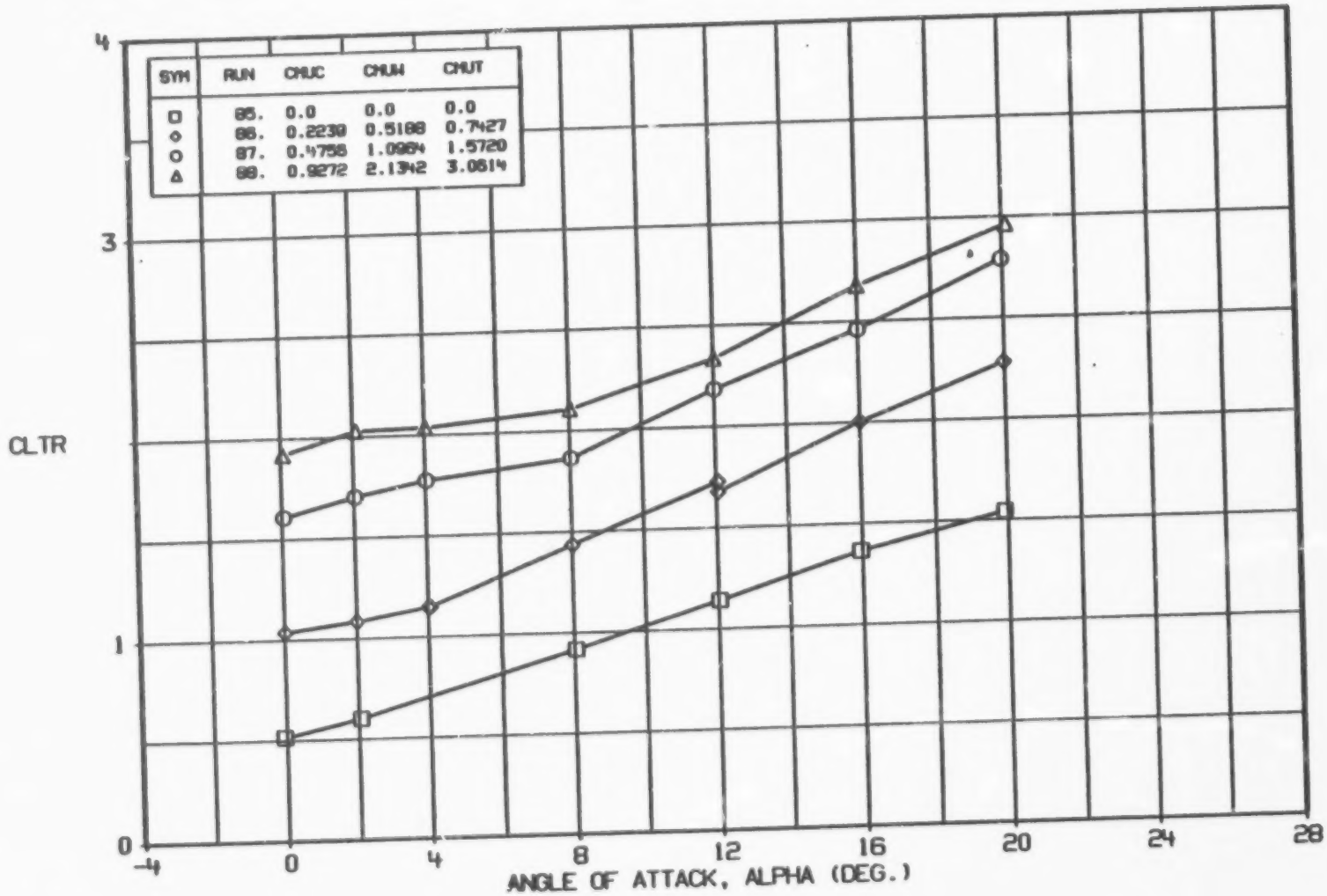

FIGURE A6d BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

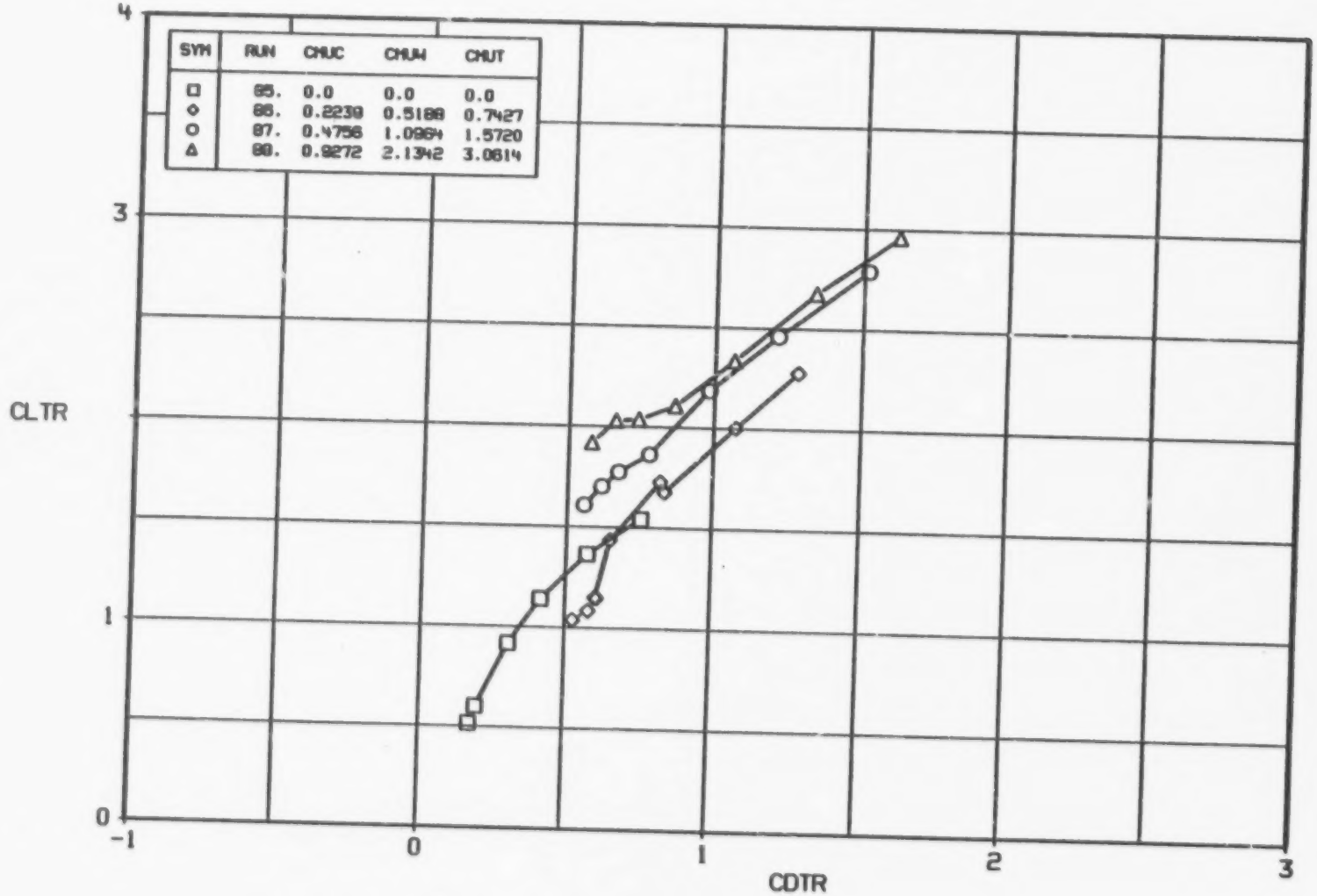

FIGURE A6e BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

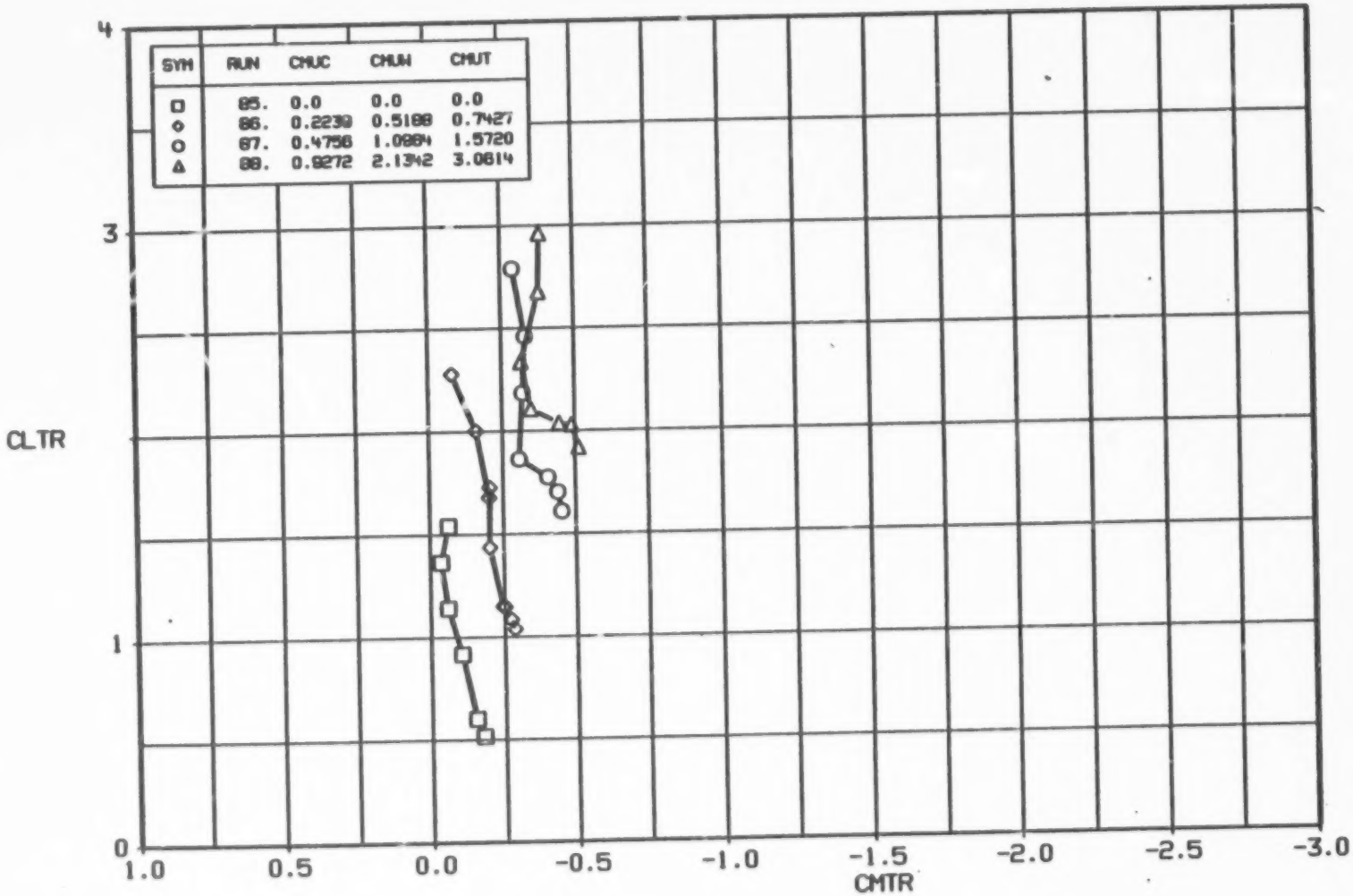

FIGURE A6f BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

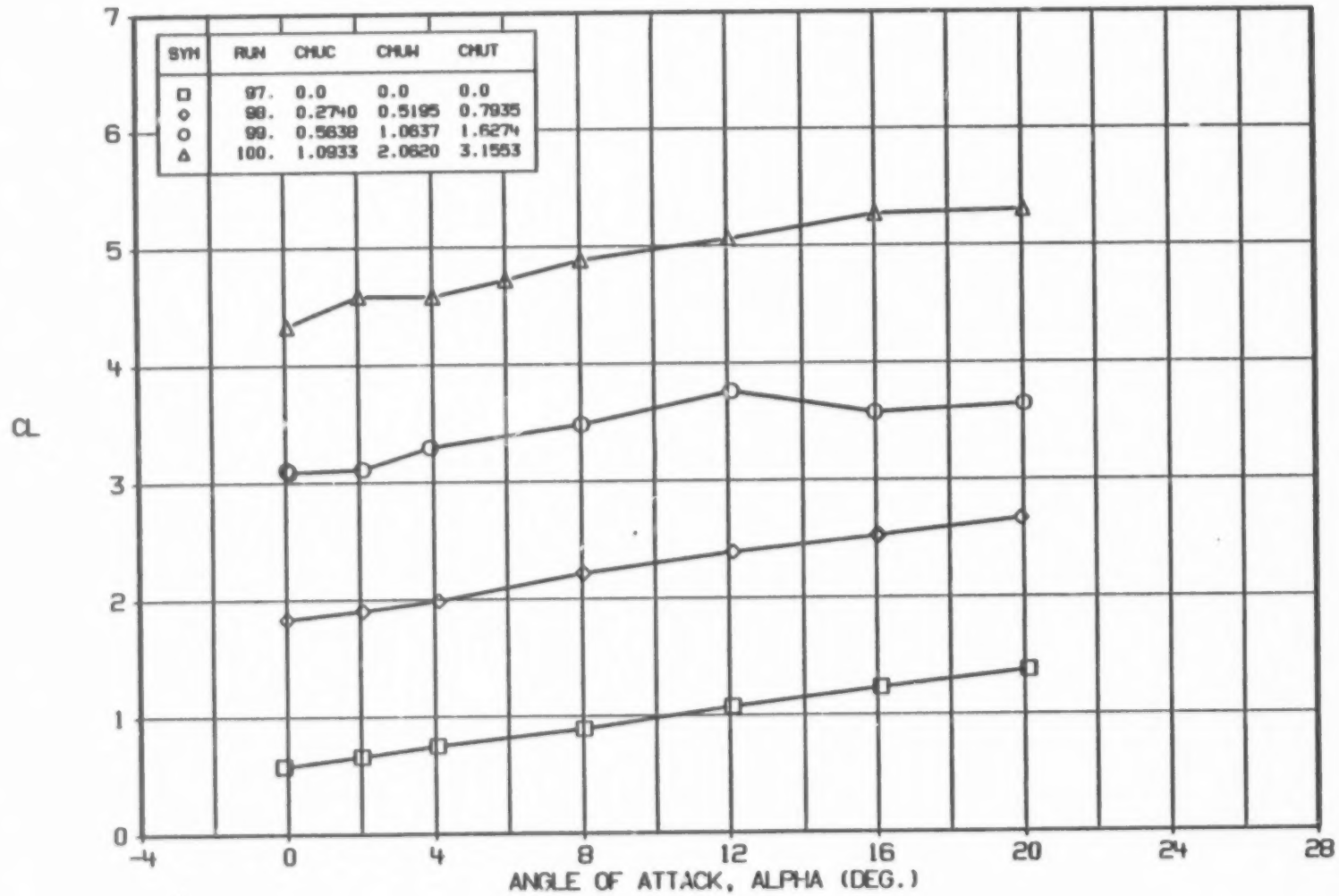

FIGURE A7a BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

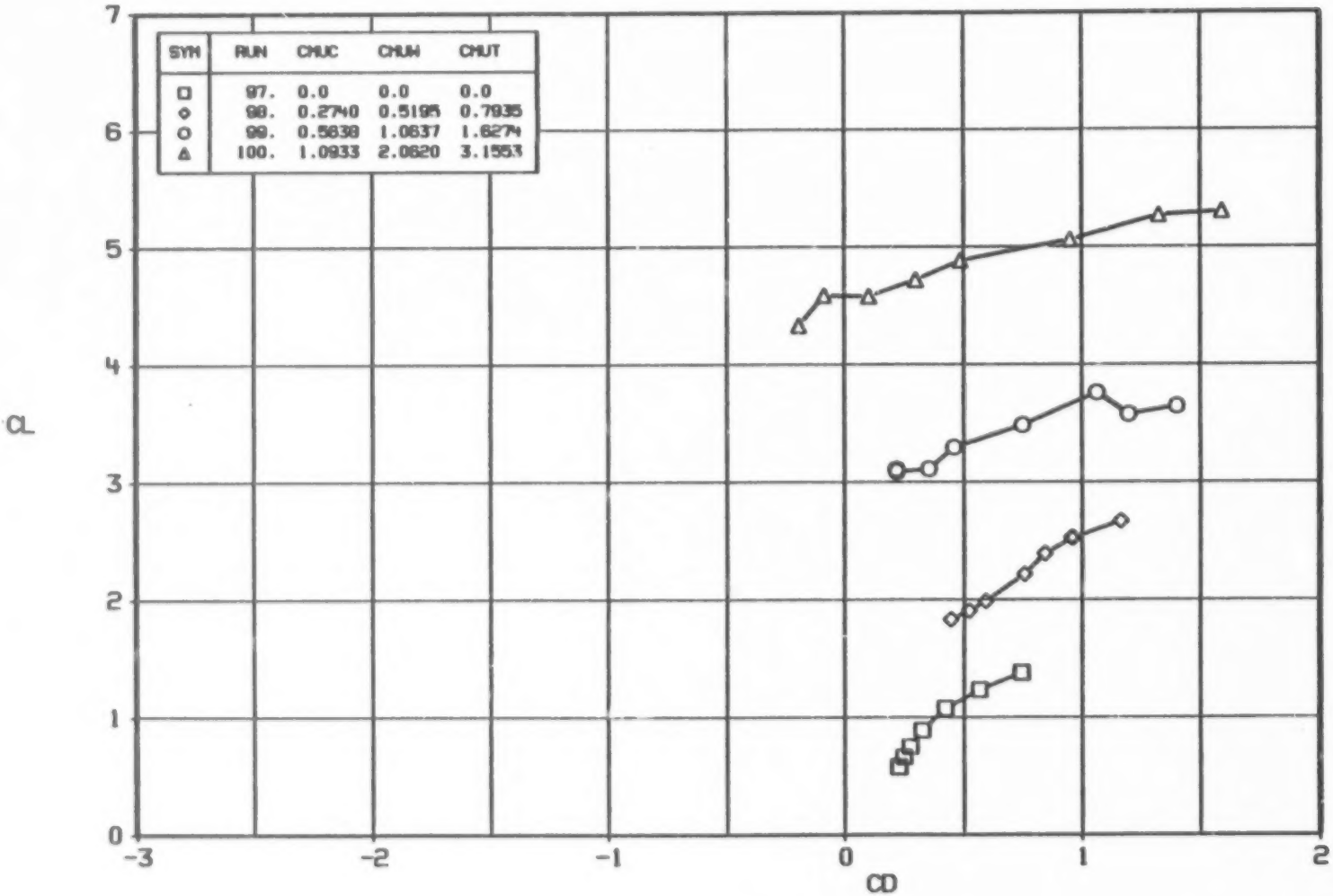

FIGURE A7b BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

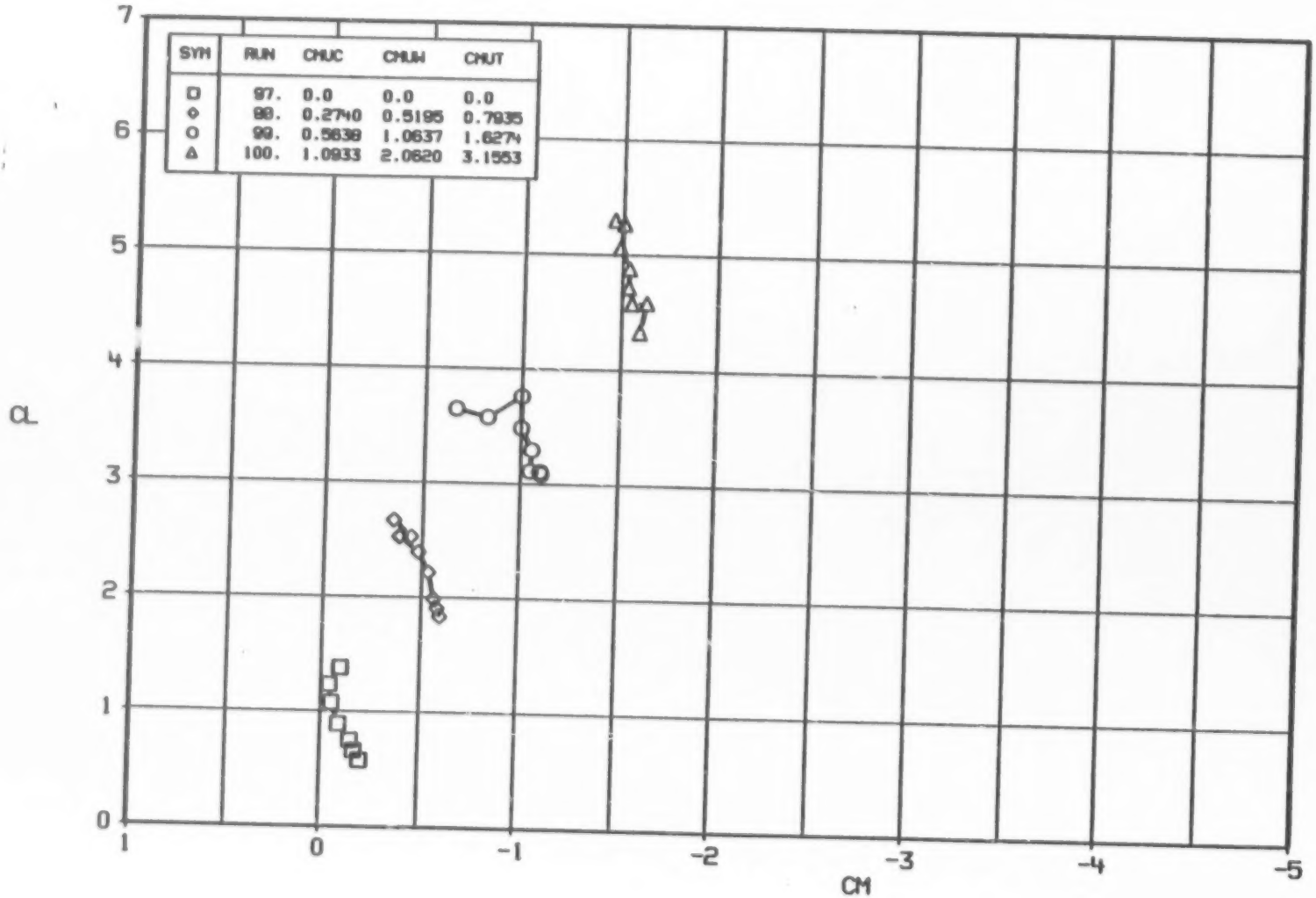

FIGURE A7c BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

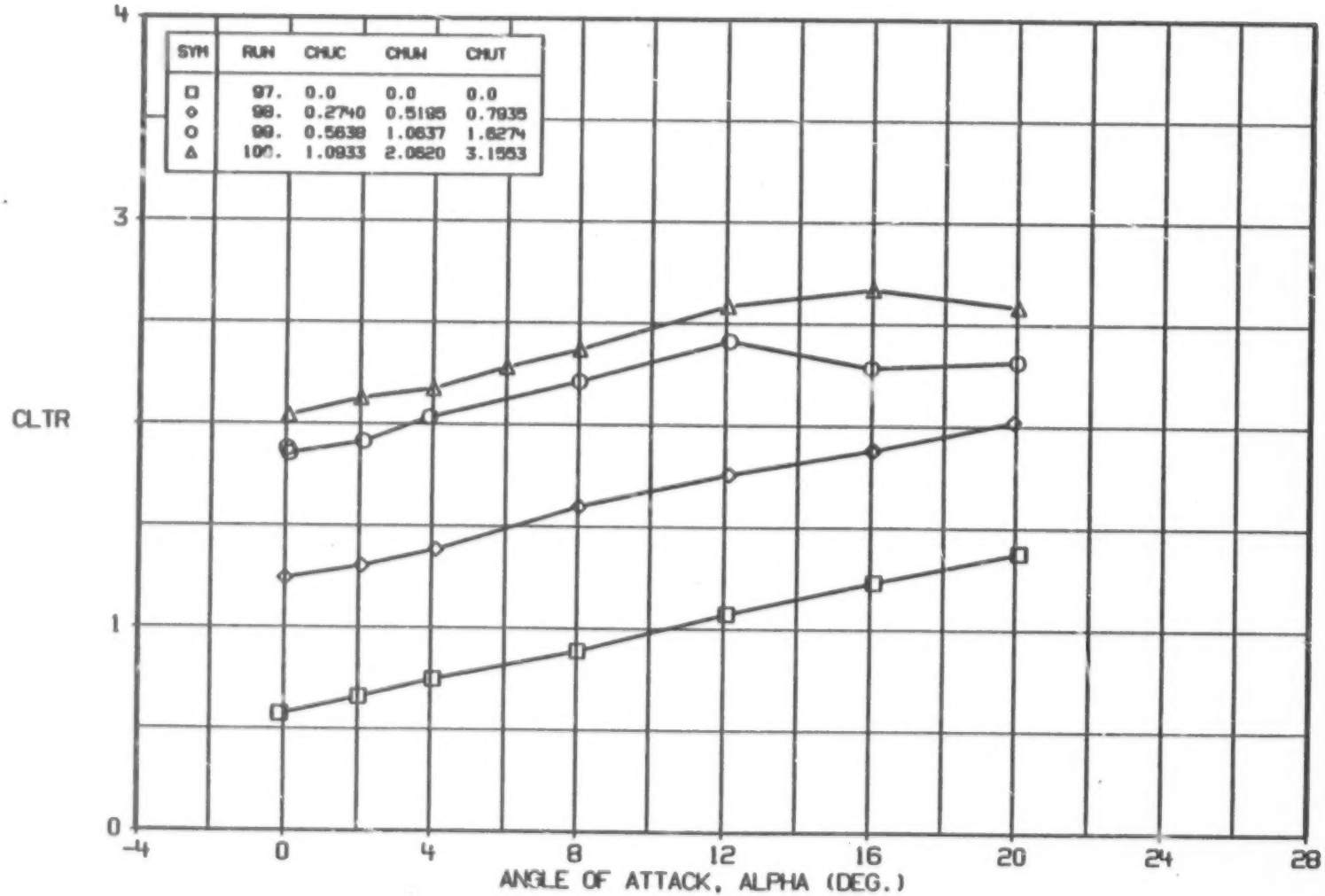

FIGURE A7d BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

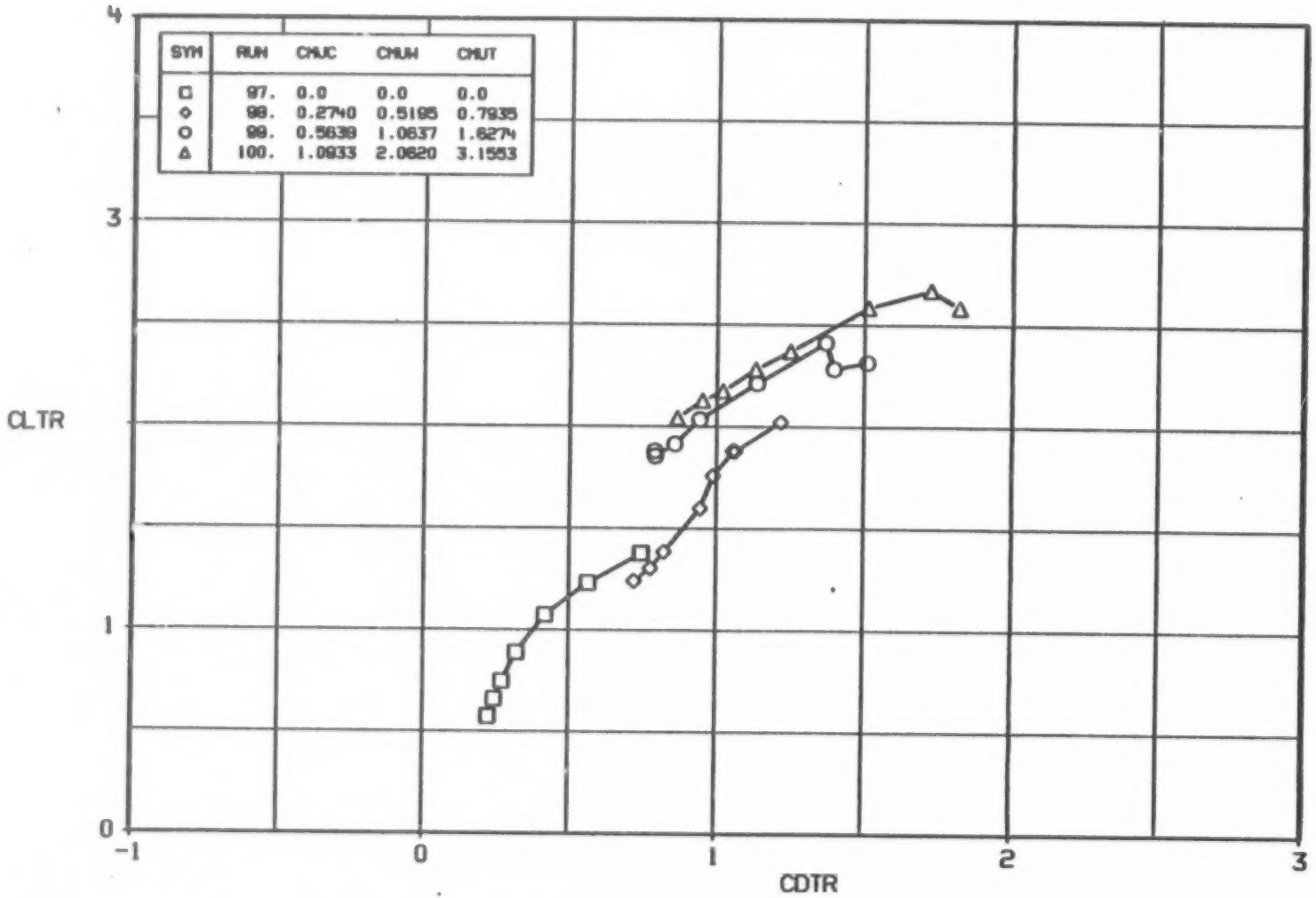

FIGURE A7e BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

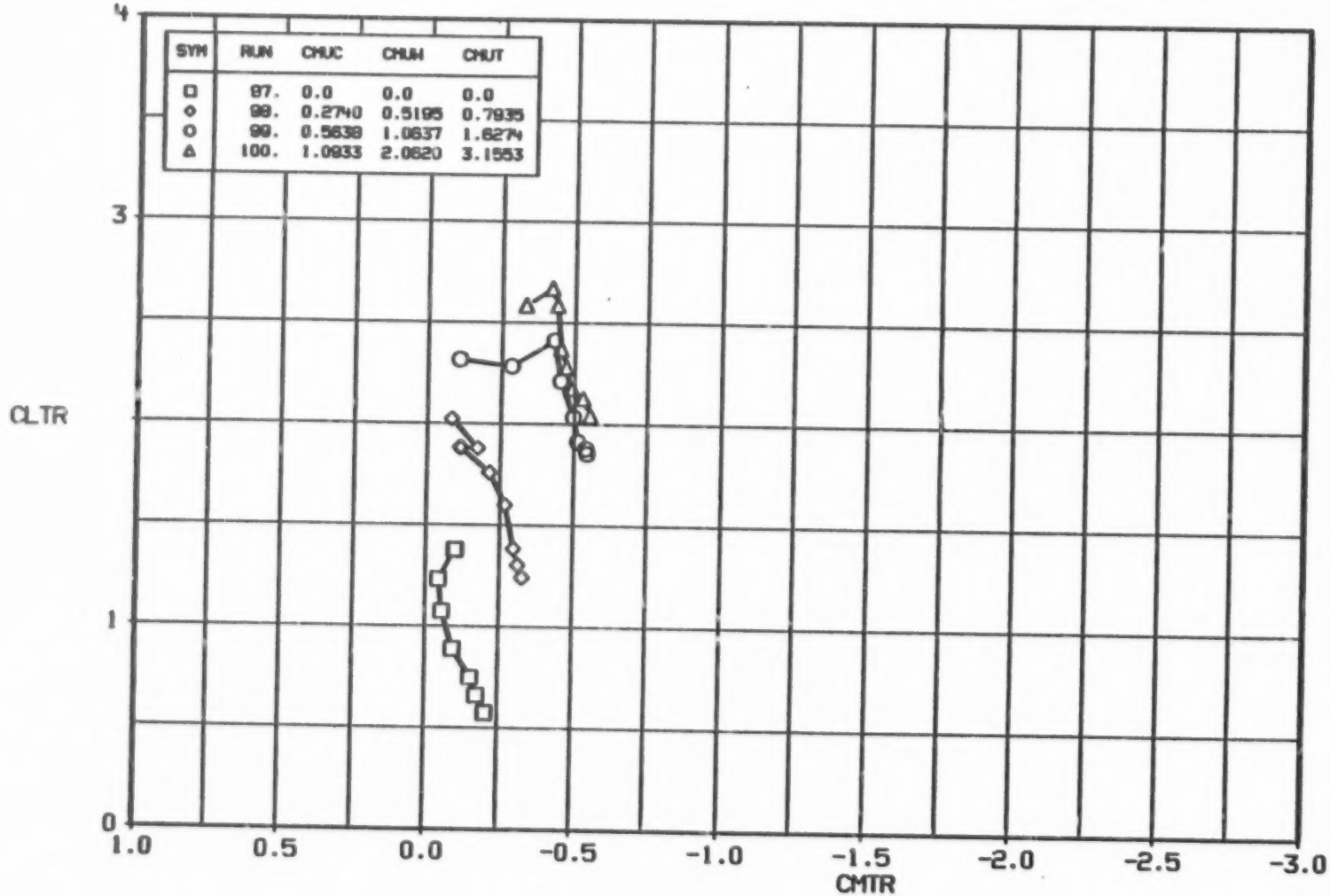

FIGURE A7f BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

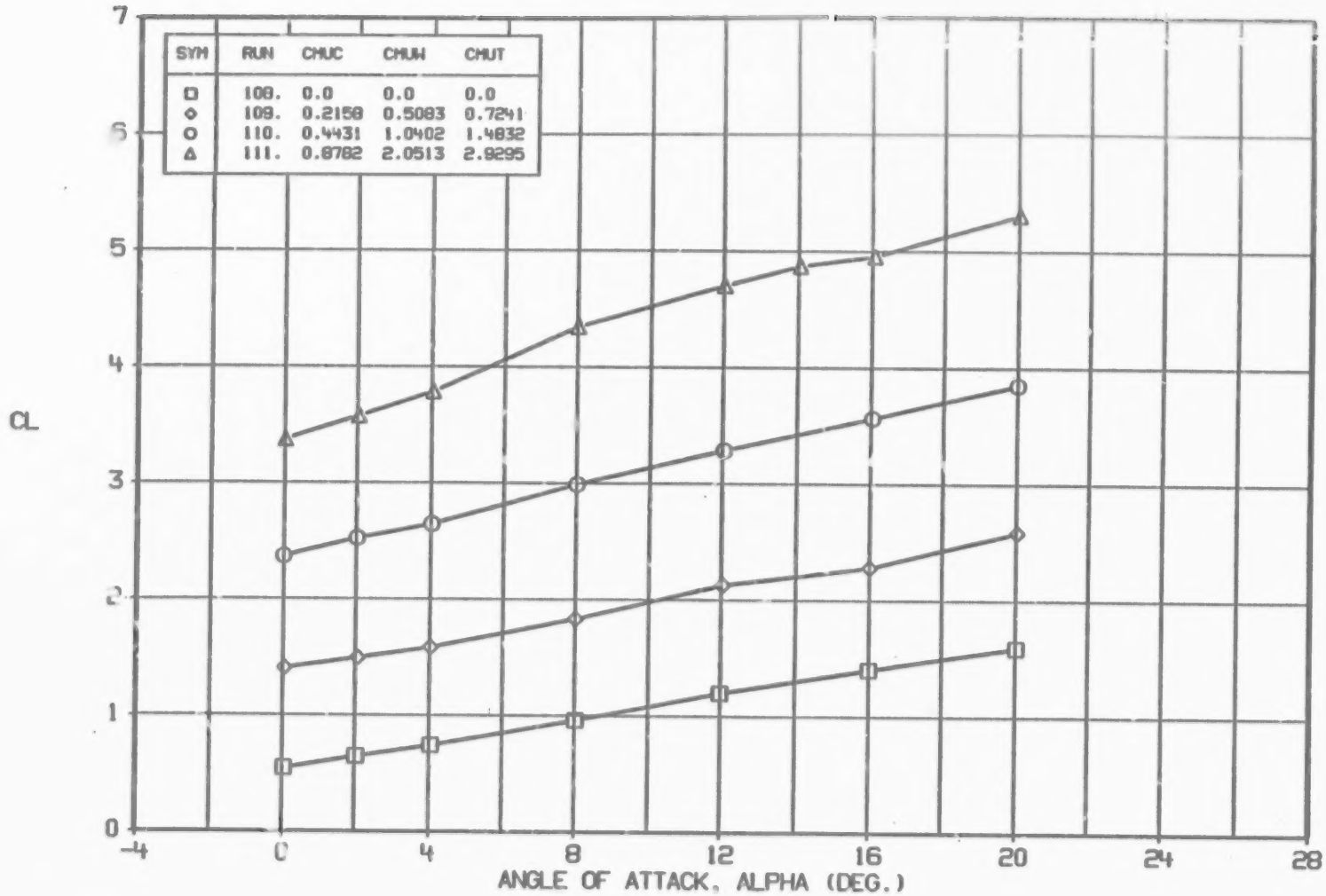

FIGURE A8a BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

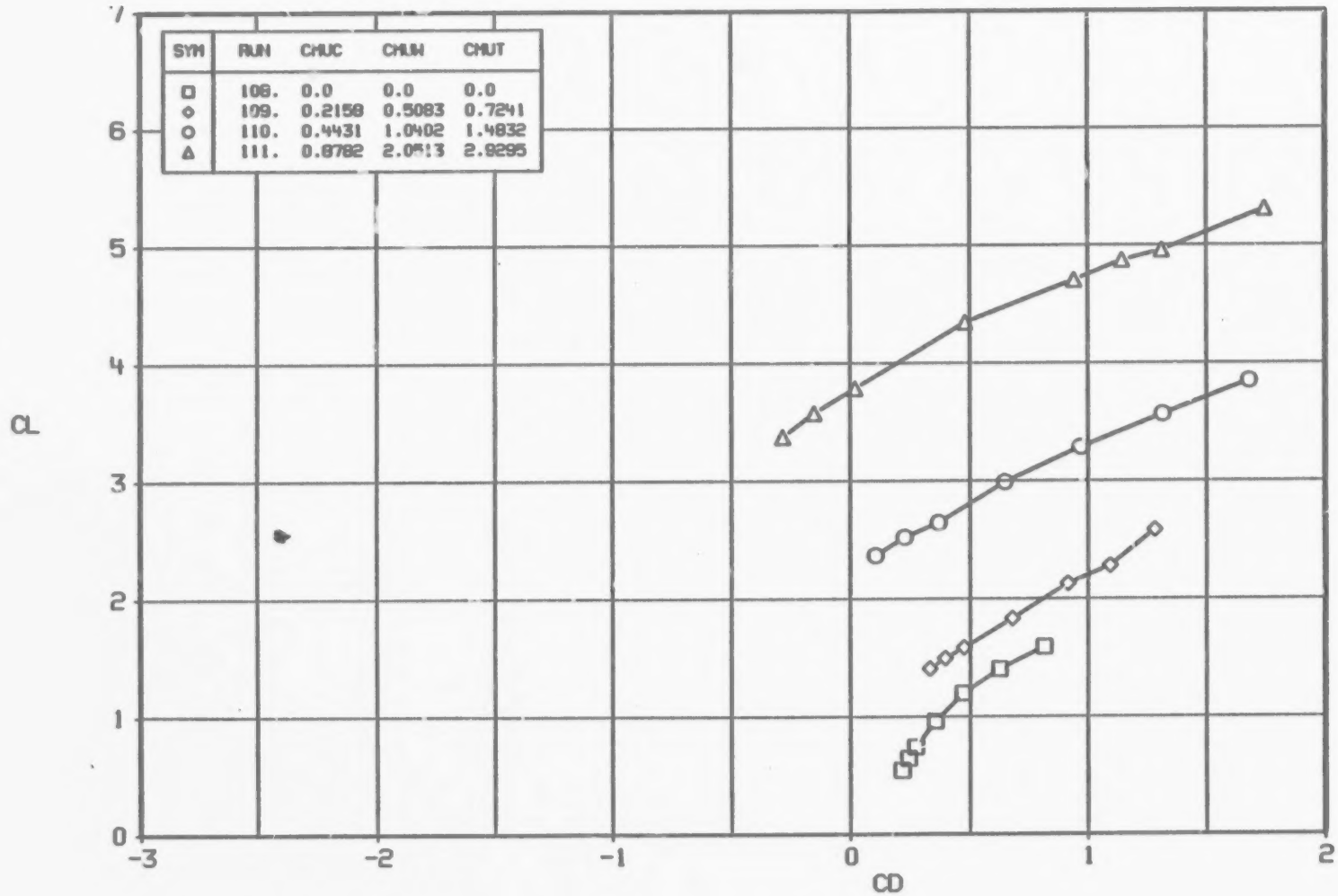

FIGURE A8b BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

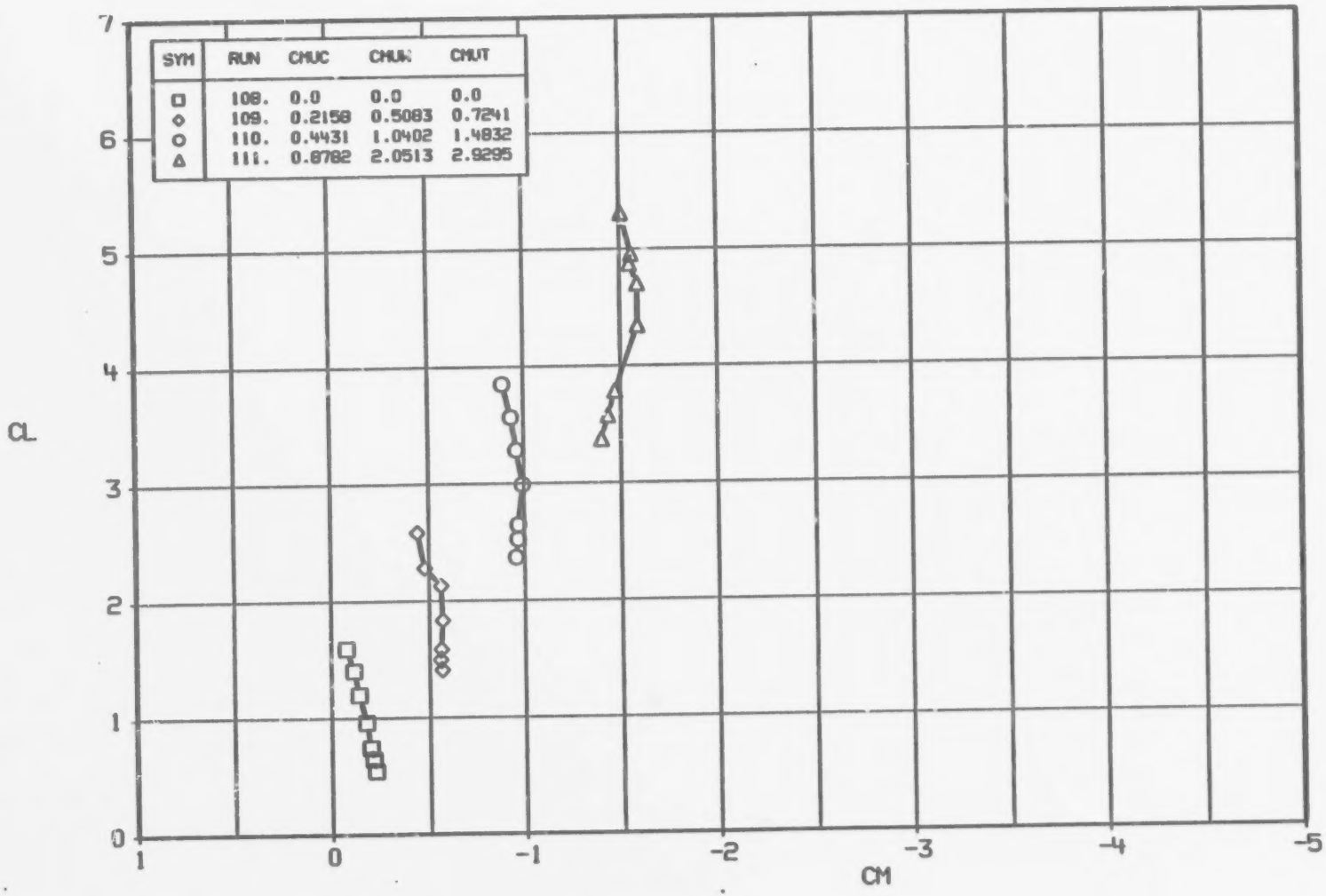

FIGURE A8c BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

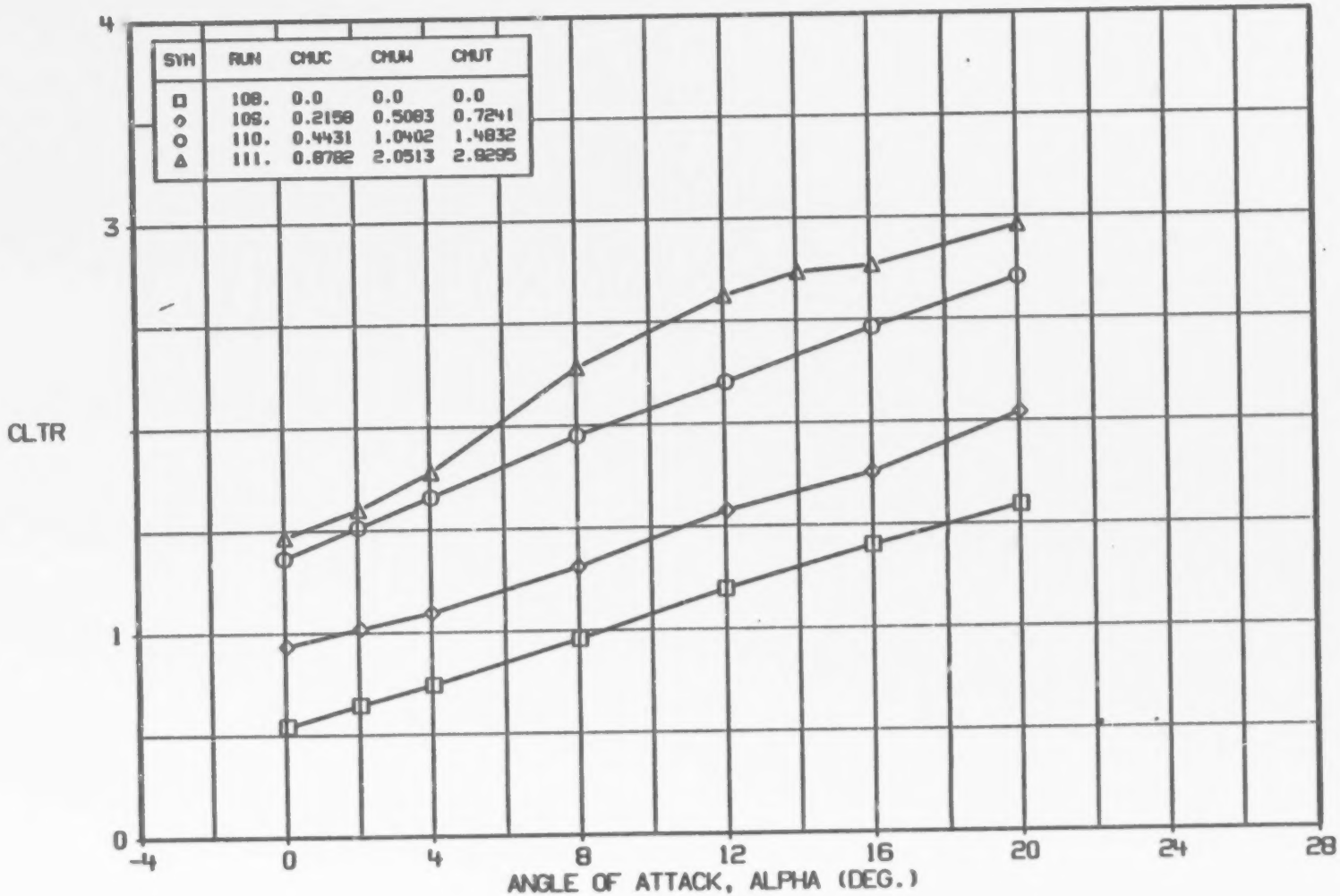

FIGURE A8d BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

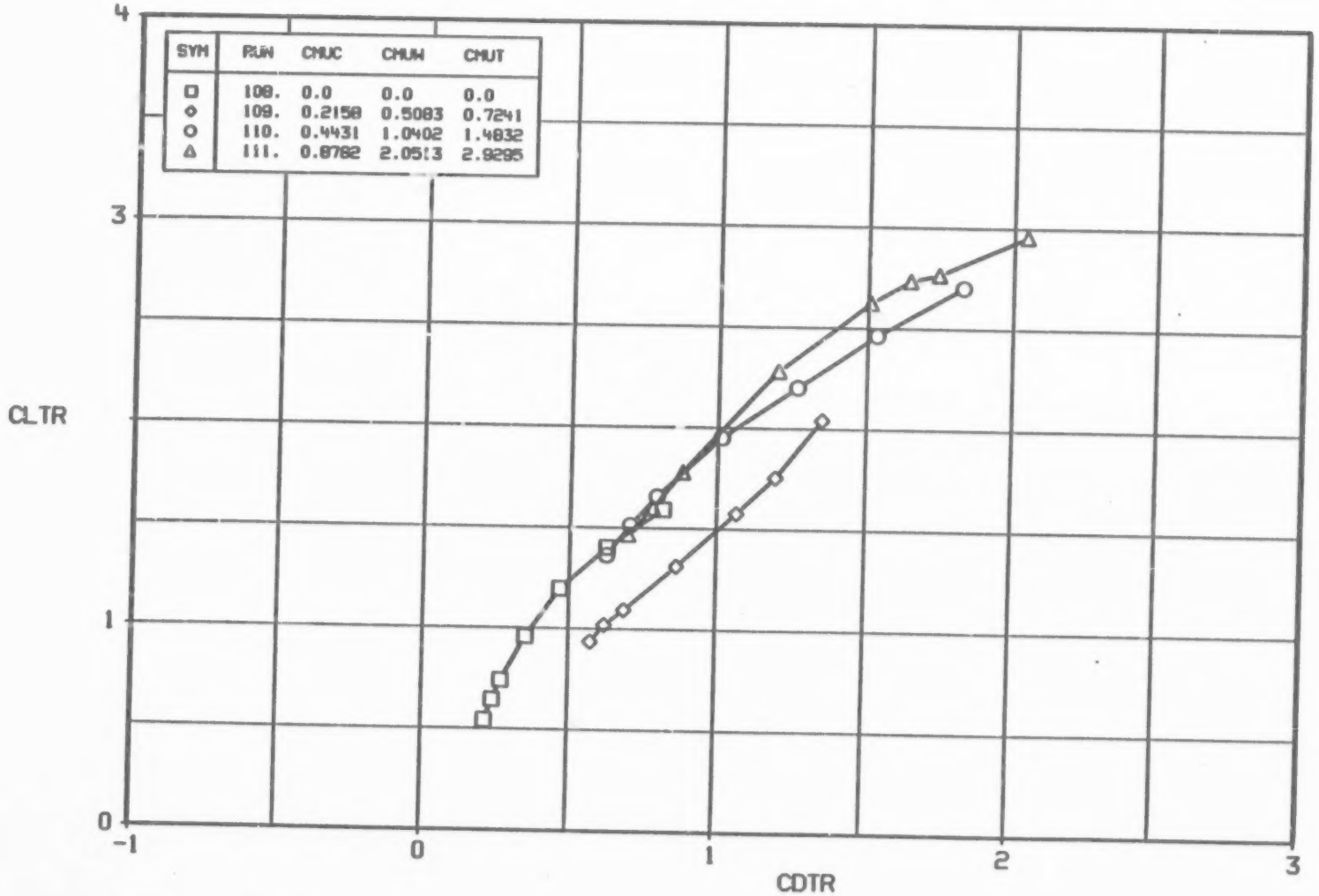

FIGURE A8e BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

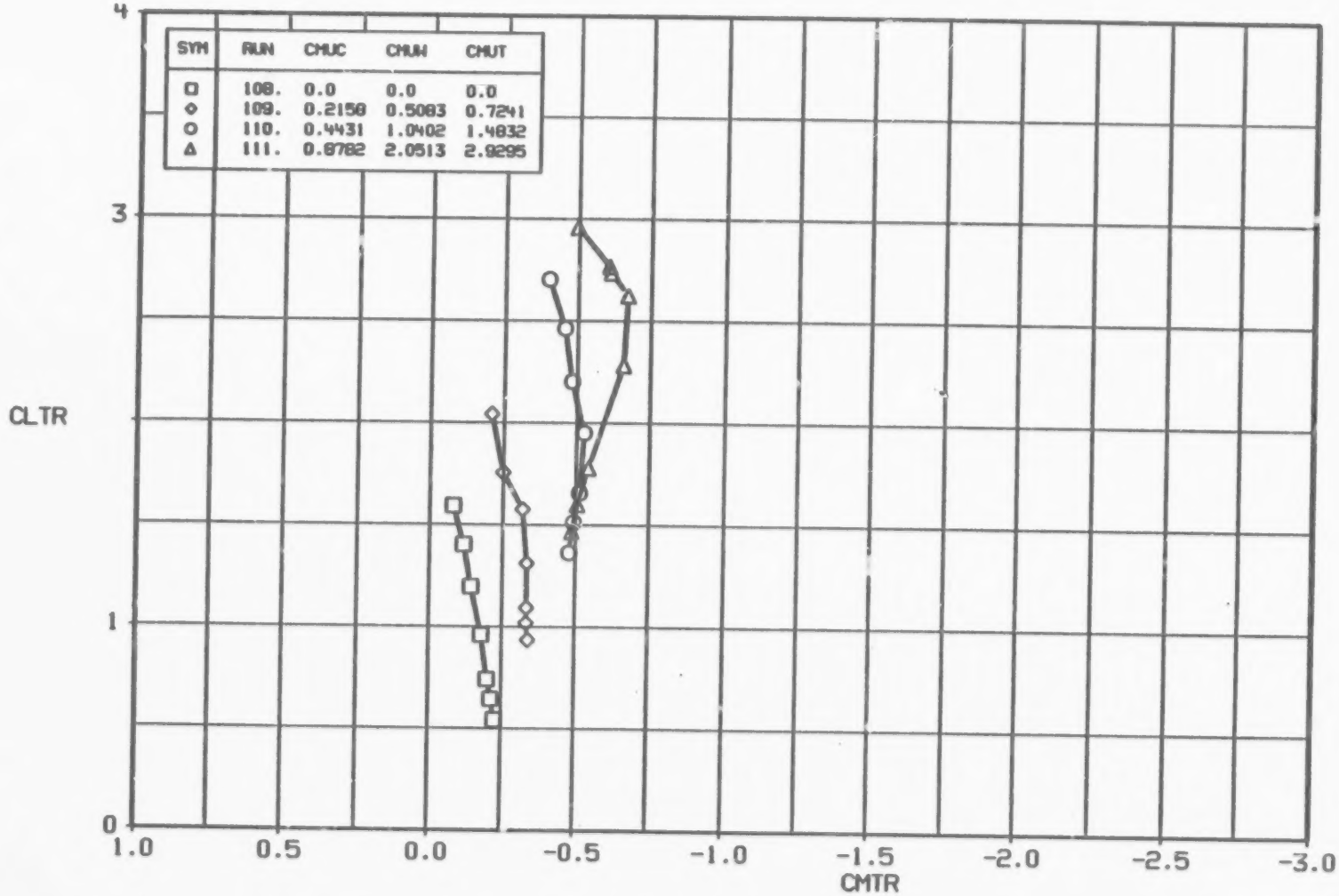

FIGURE A8f BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

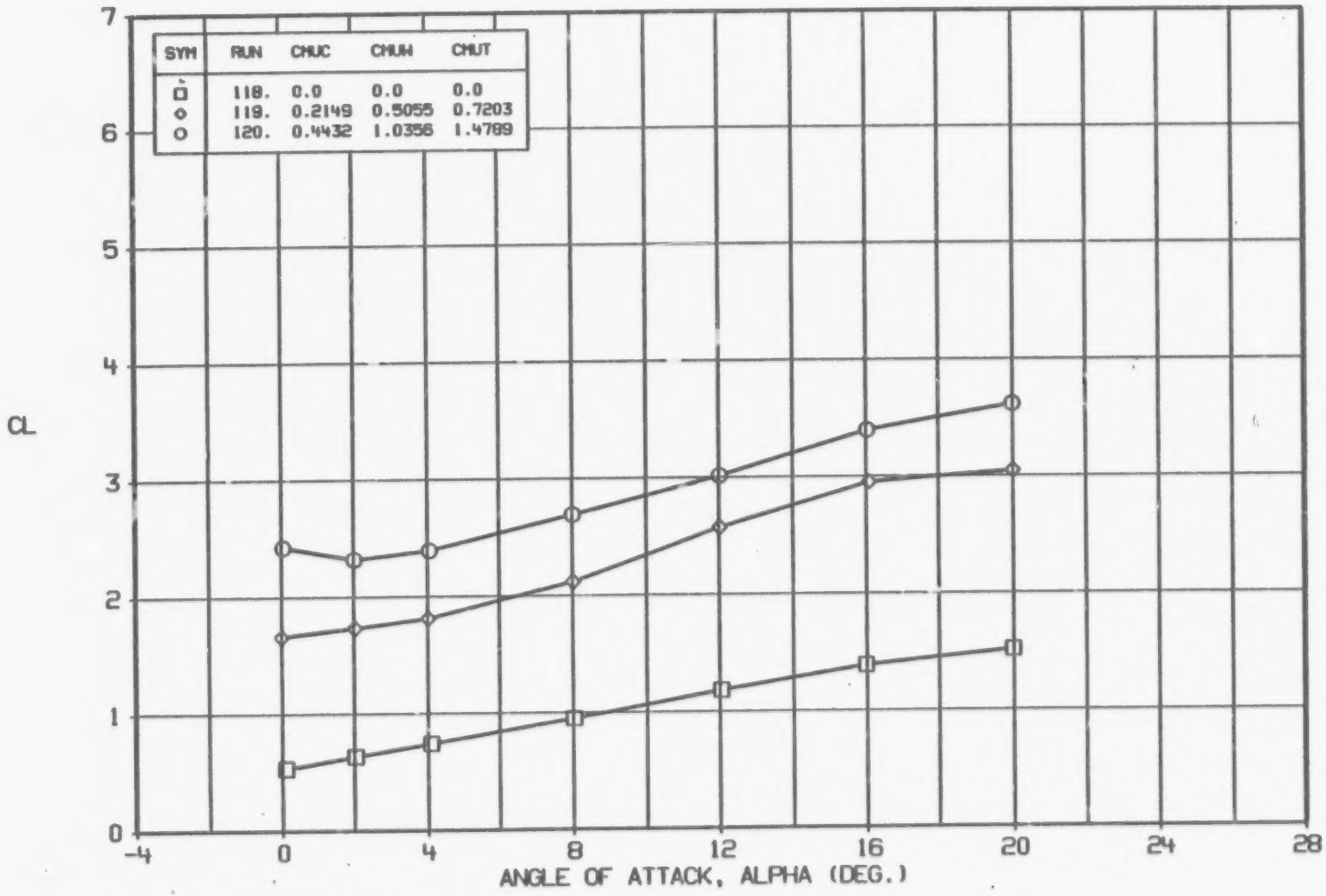

FIGURE A9a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

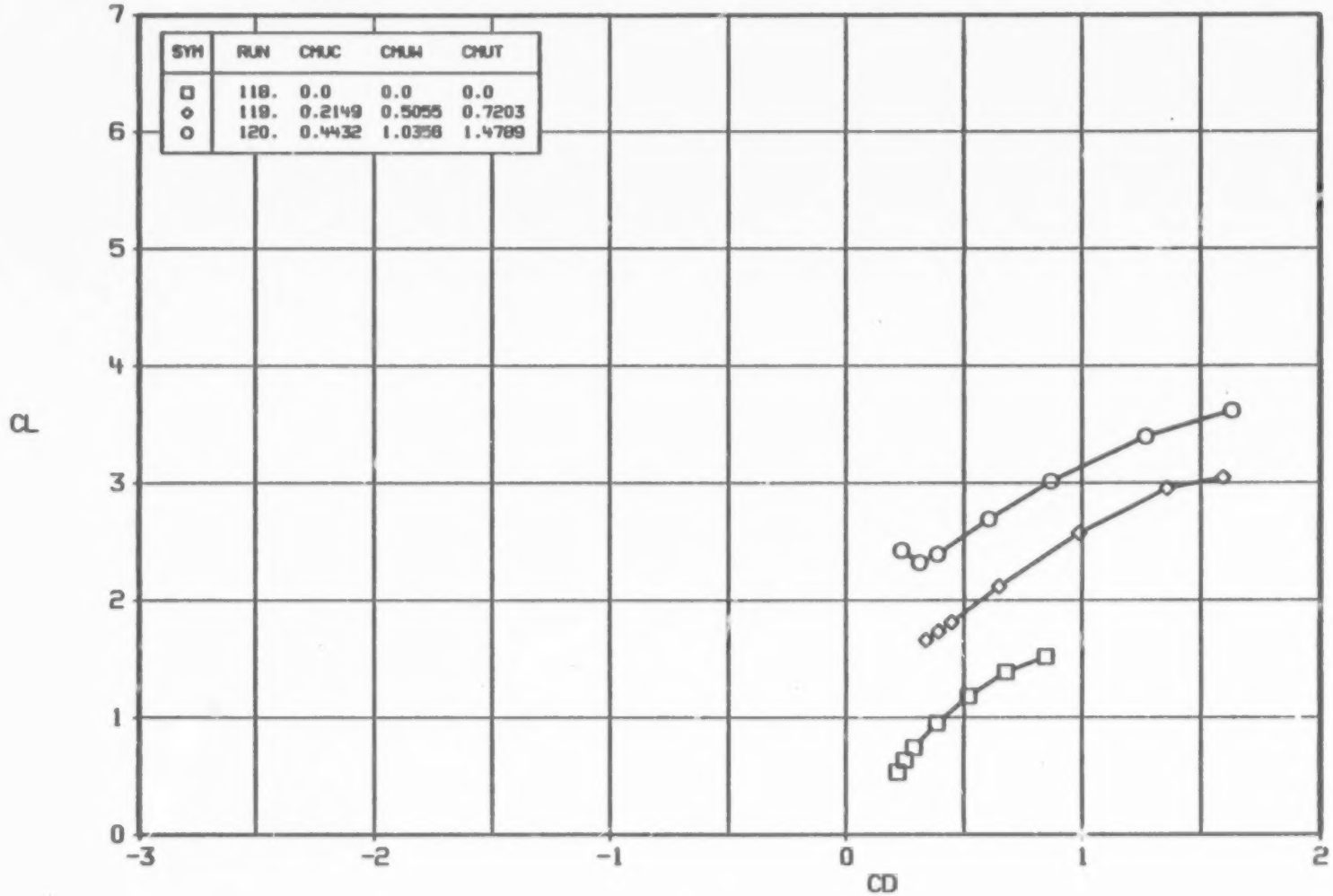

FIGURE A9b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

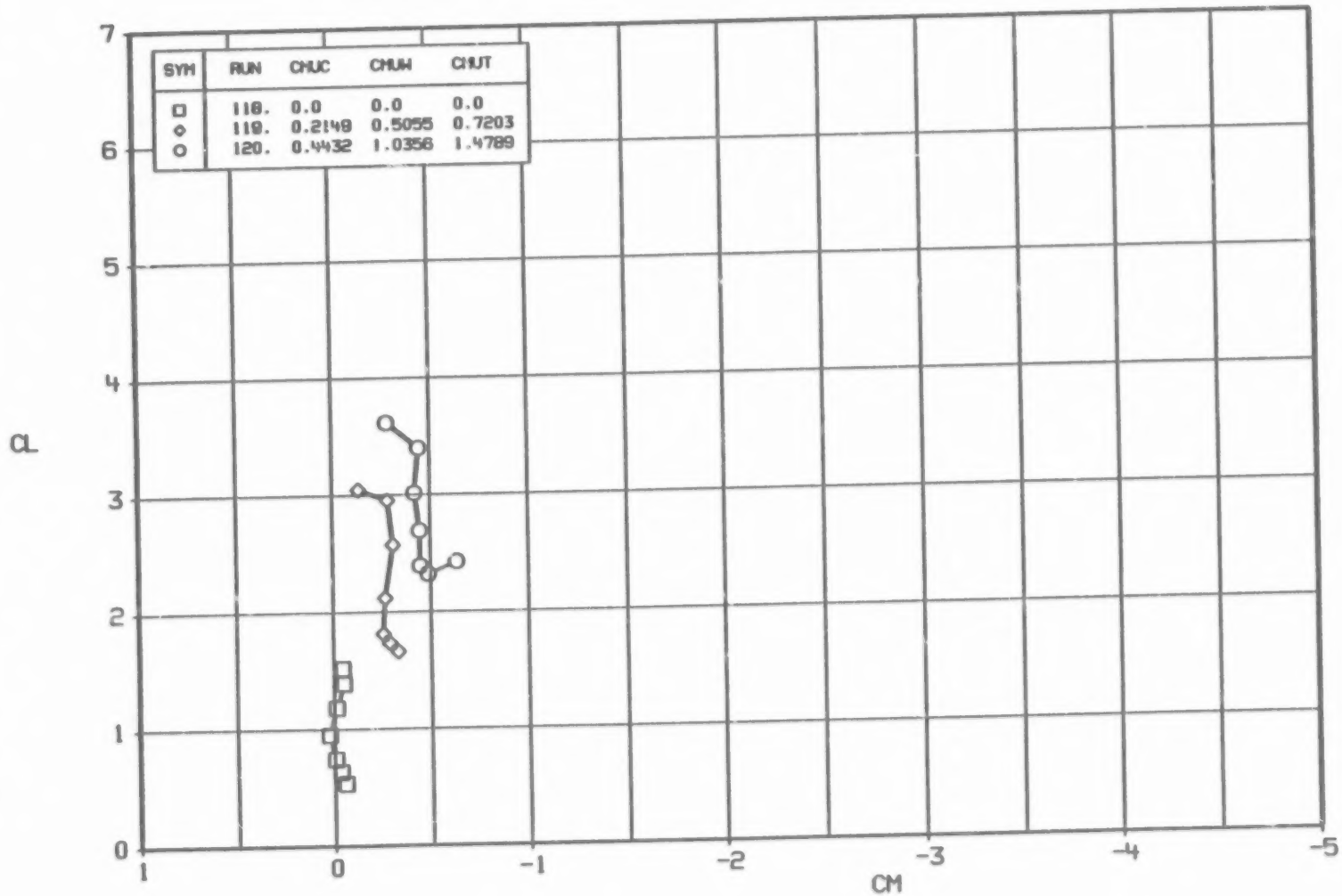

FIGURE A9c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

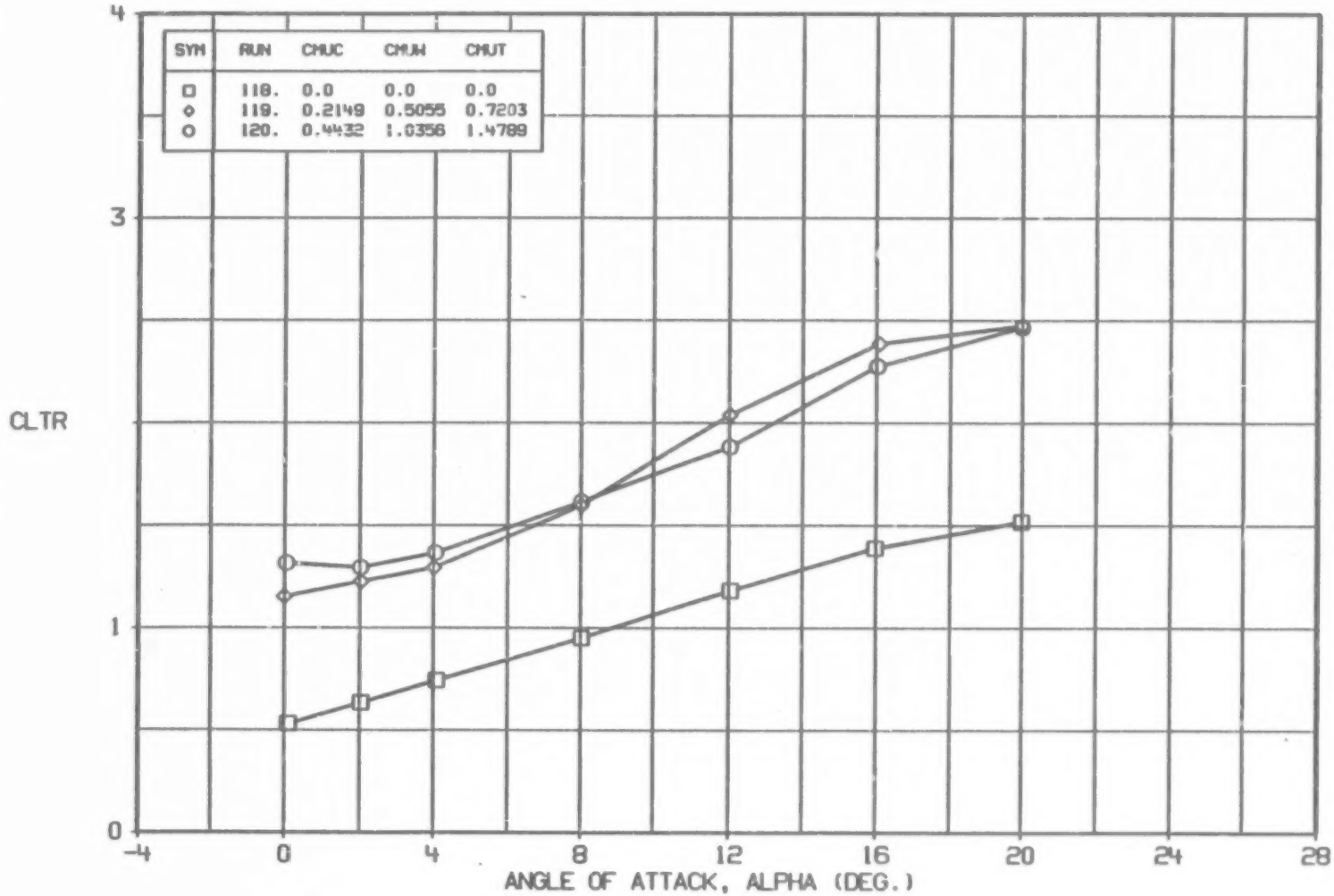

FIGURE A9d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1N6V, DELF=45, BN/B=1, DELC=10

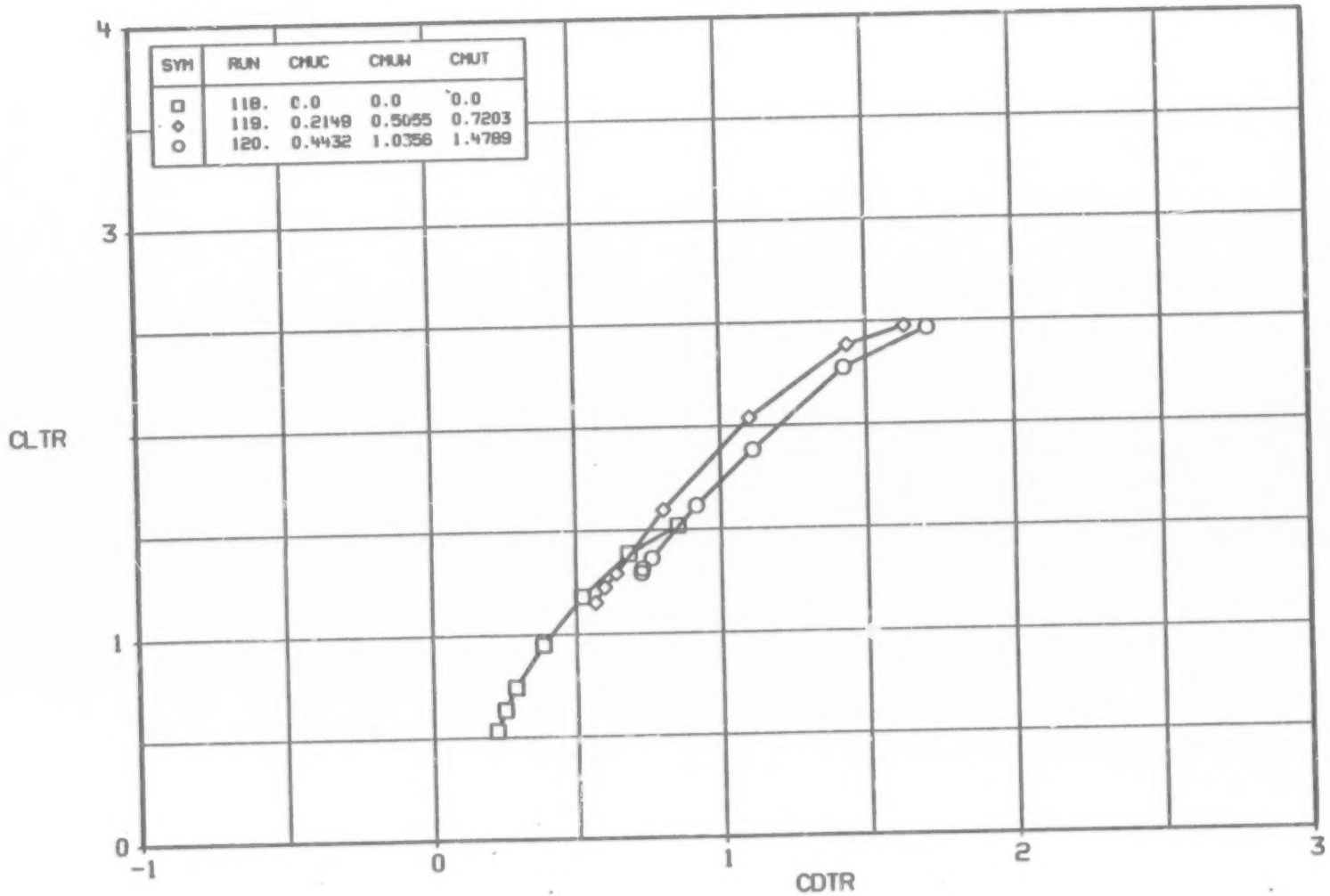

FIGURE A9e BASIC DATA EFFECT OF CMU
CONFIGURATION BC!W6V, DELF=45, BN/B=1, DELC=10

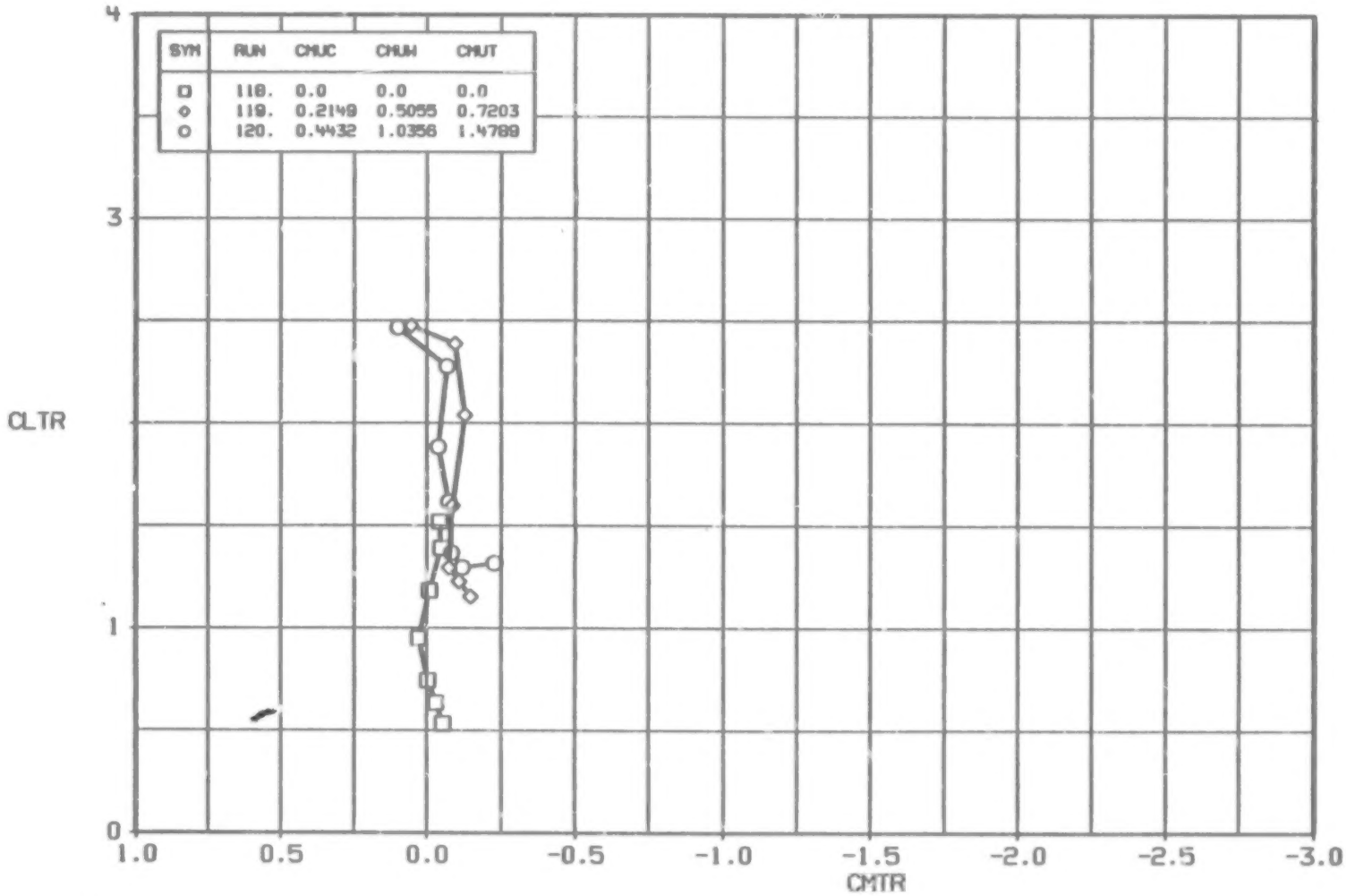

FIGURE A9f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10

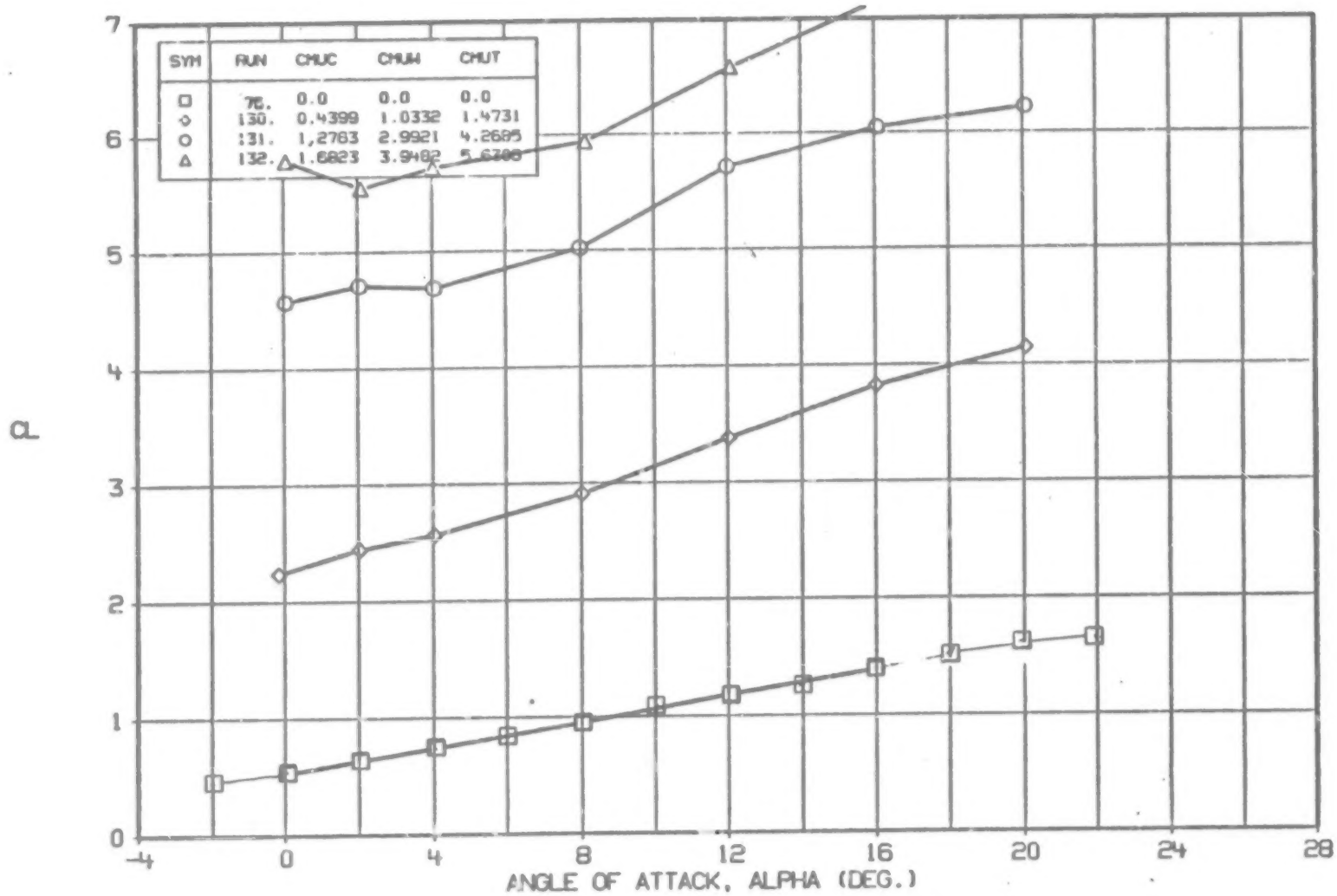

FIGURE A10a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, DELF=45, BN/B=1

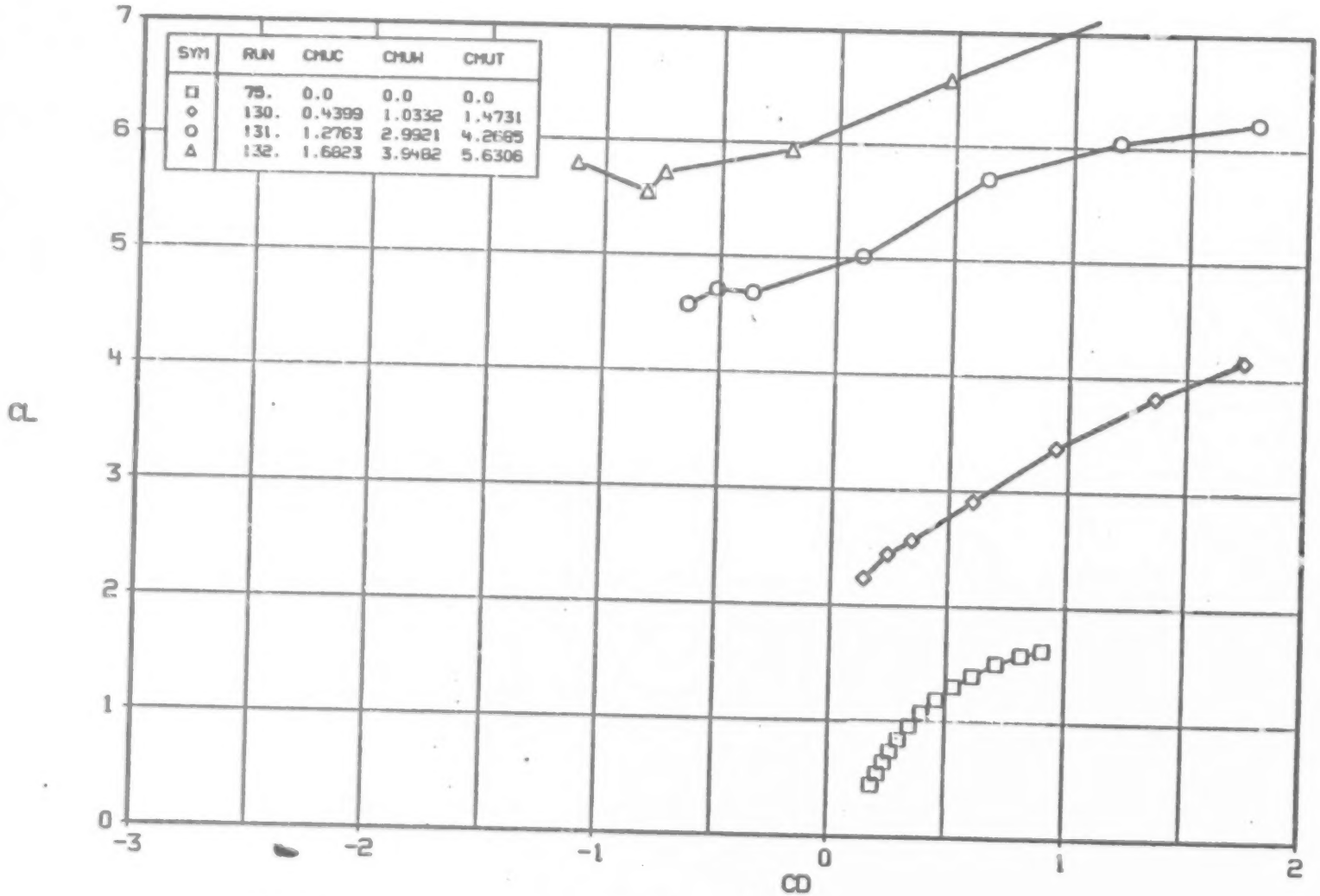

FIGURE A10b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, DELF=45, BN/B=1

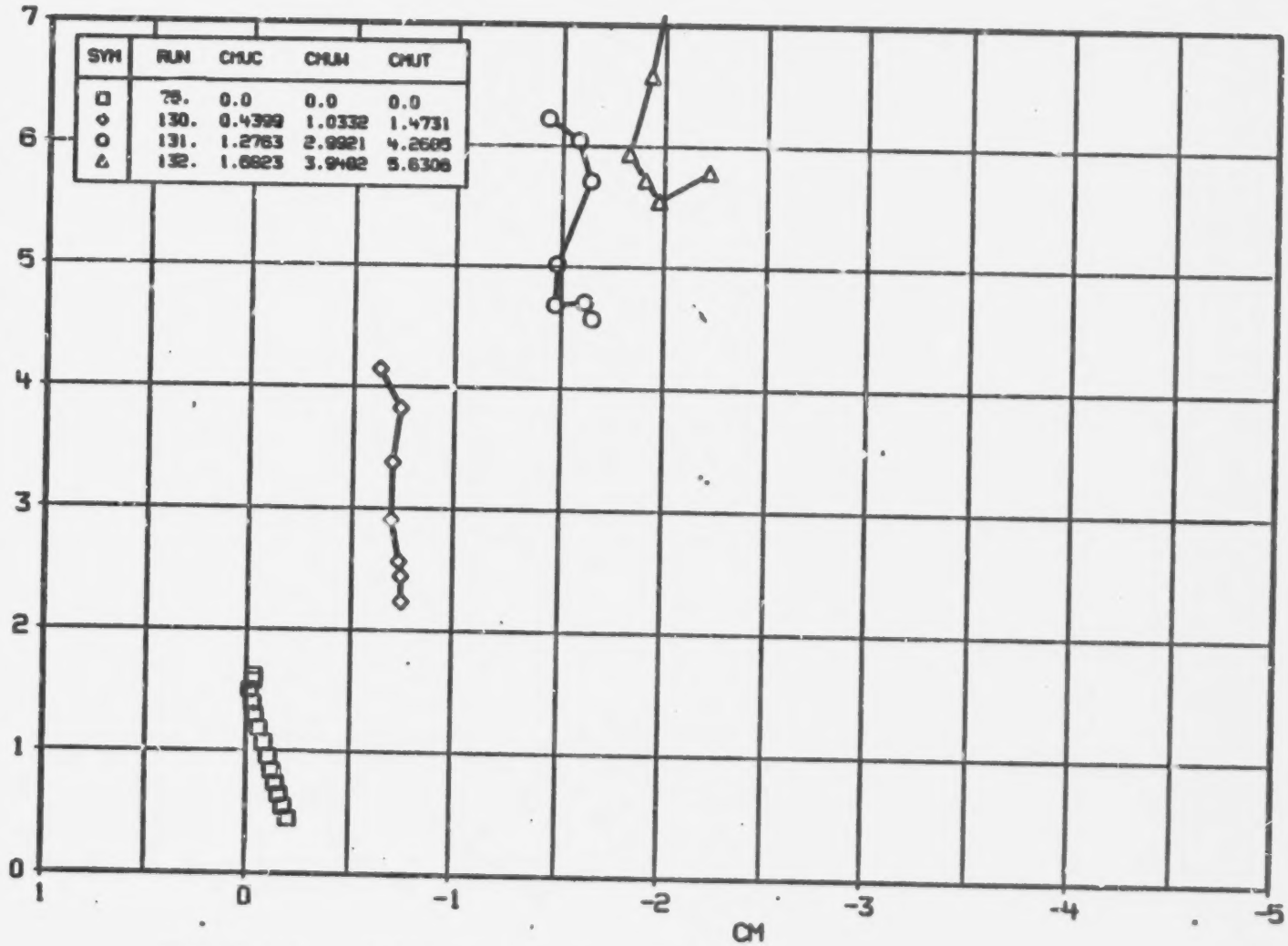

FIGURE A10c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

C-2

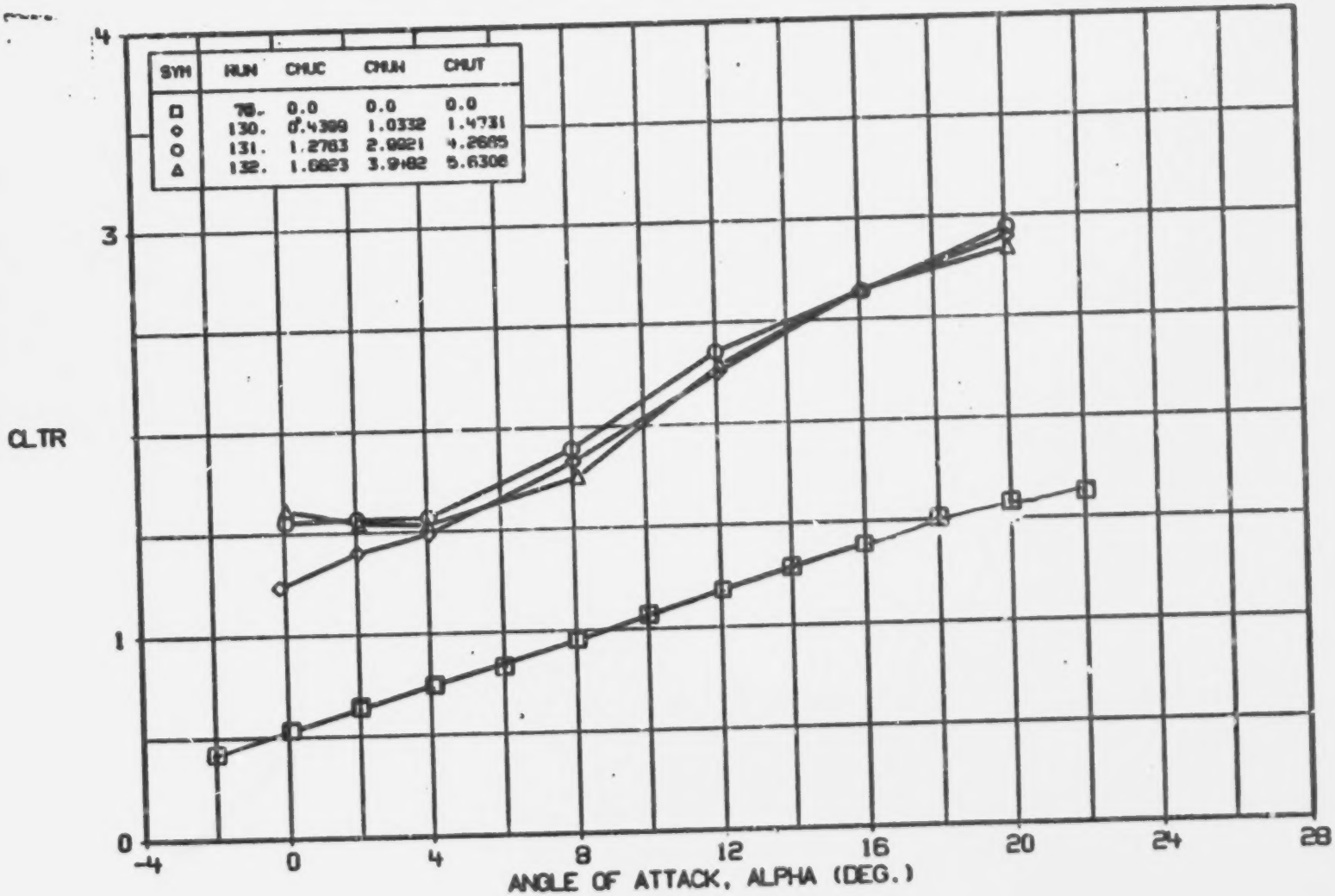

FIGURE A10d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

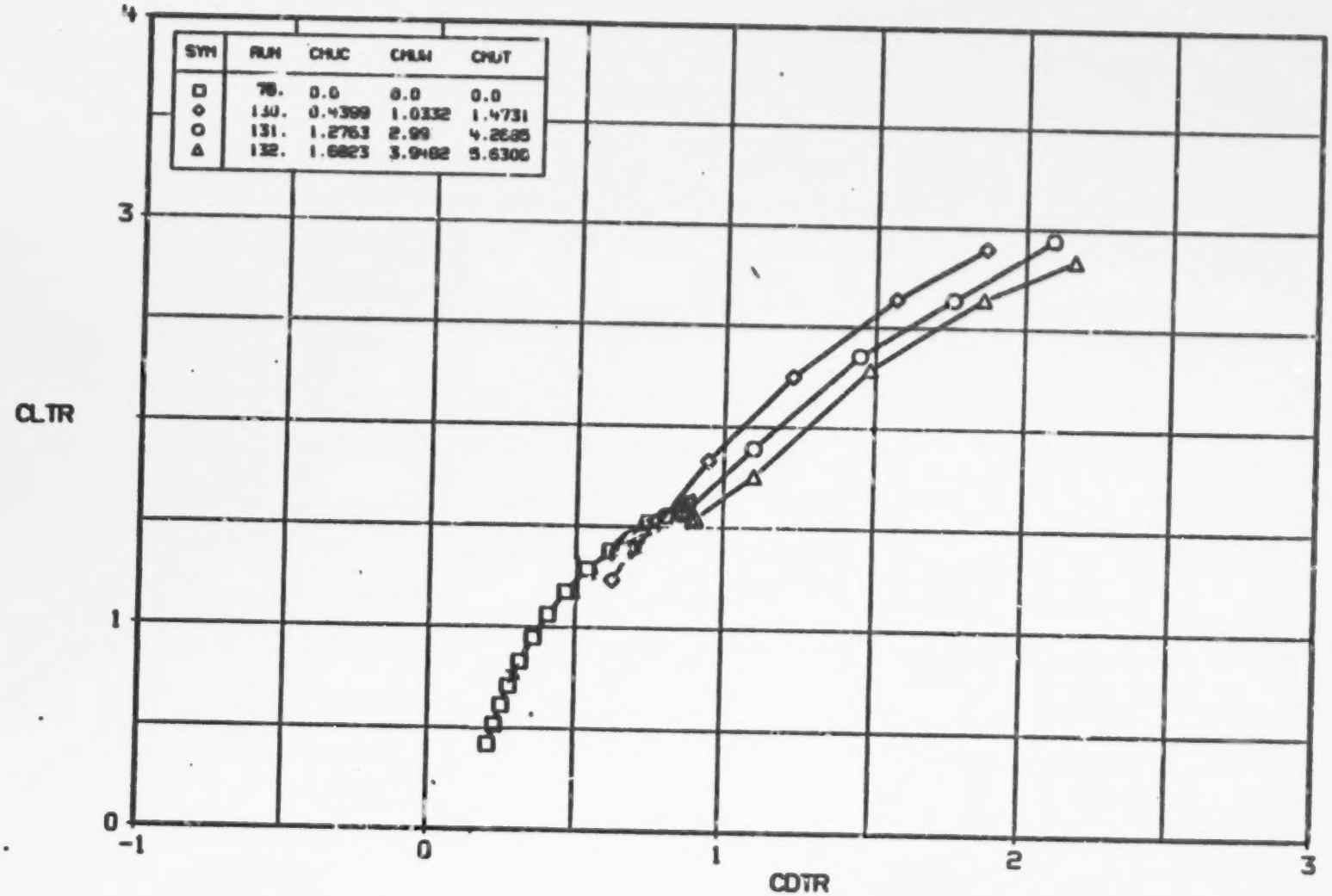

FIGURE A10e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1WSV, DELF=45, BN/B=1

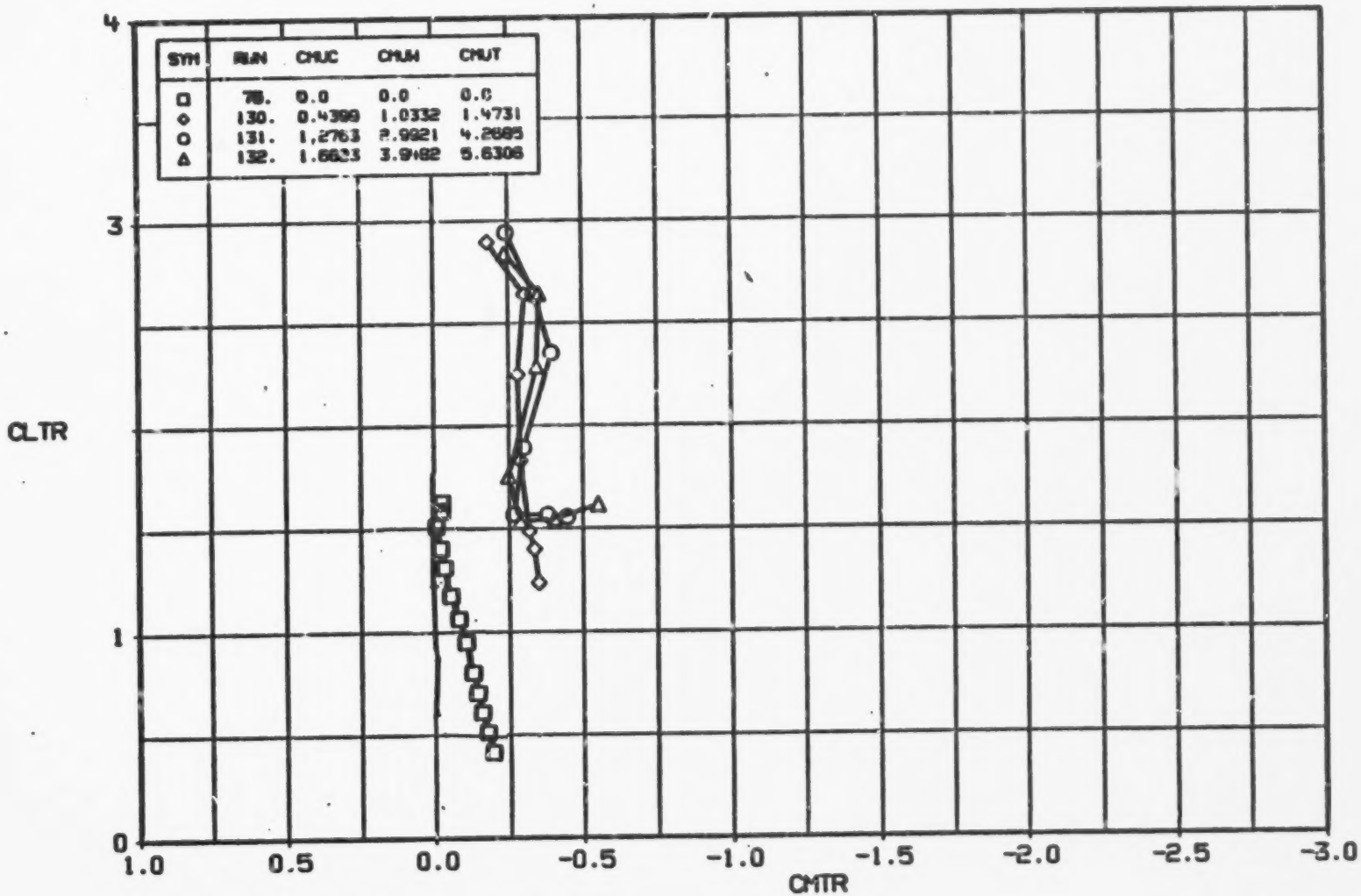

FIGURE A10f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

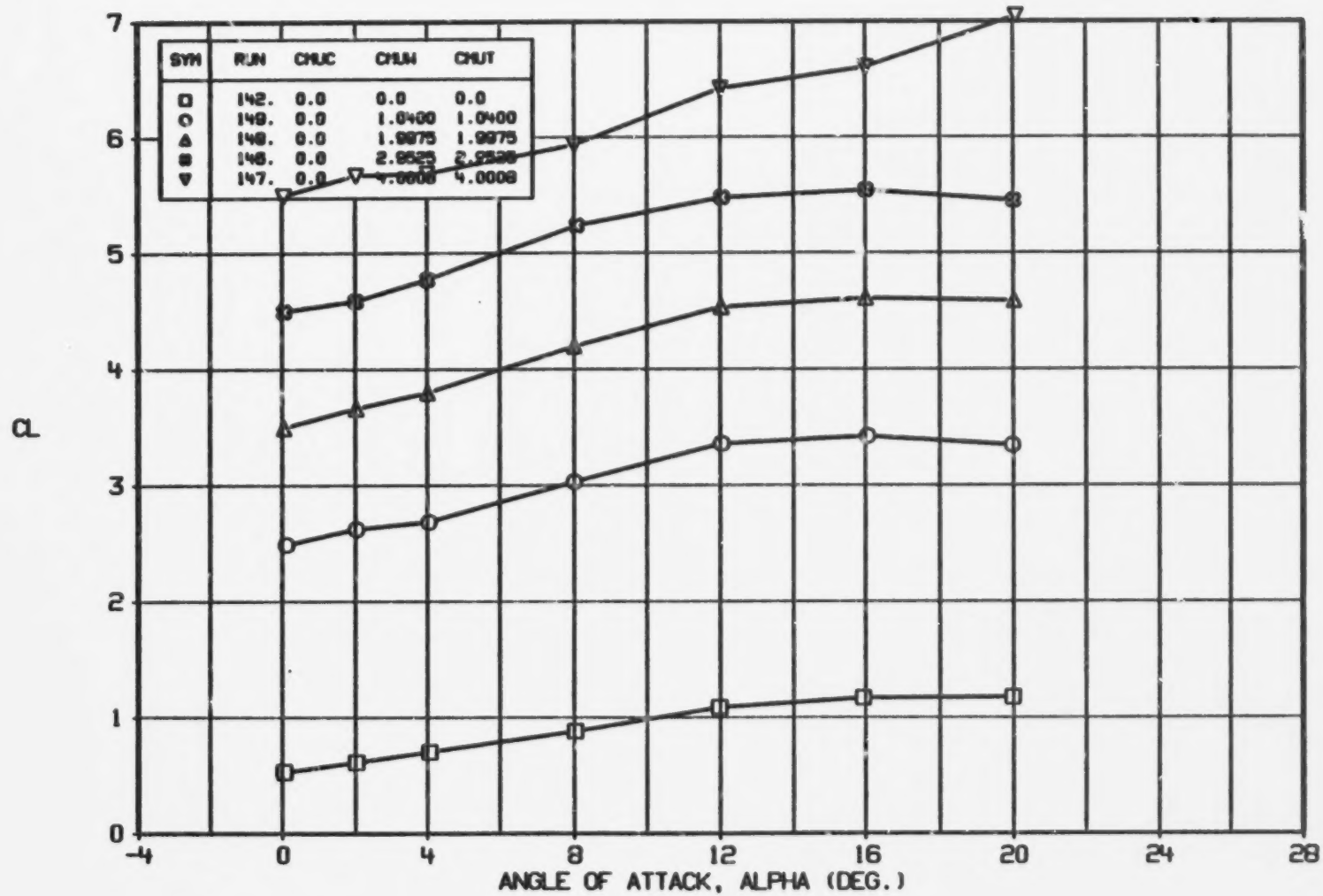

FIGURE A11a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

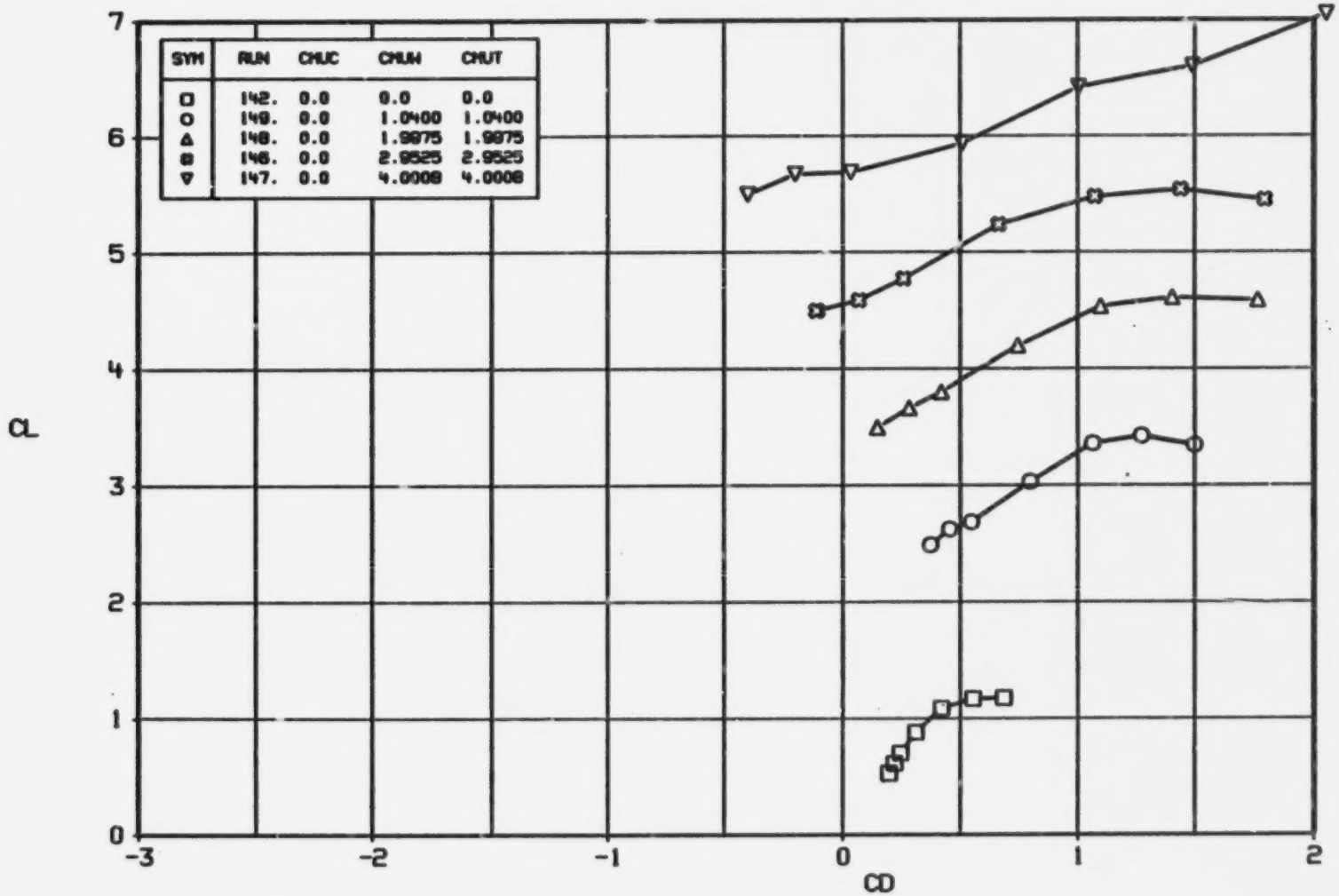

FIGURE A11b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

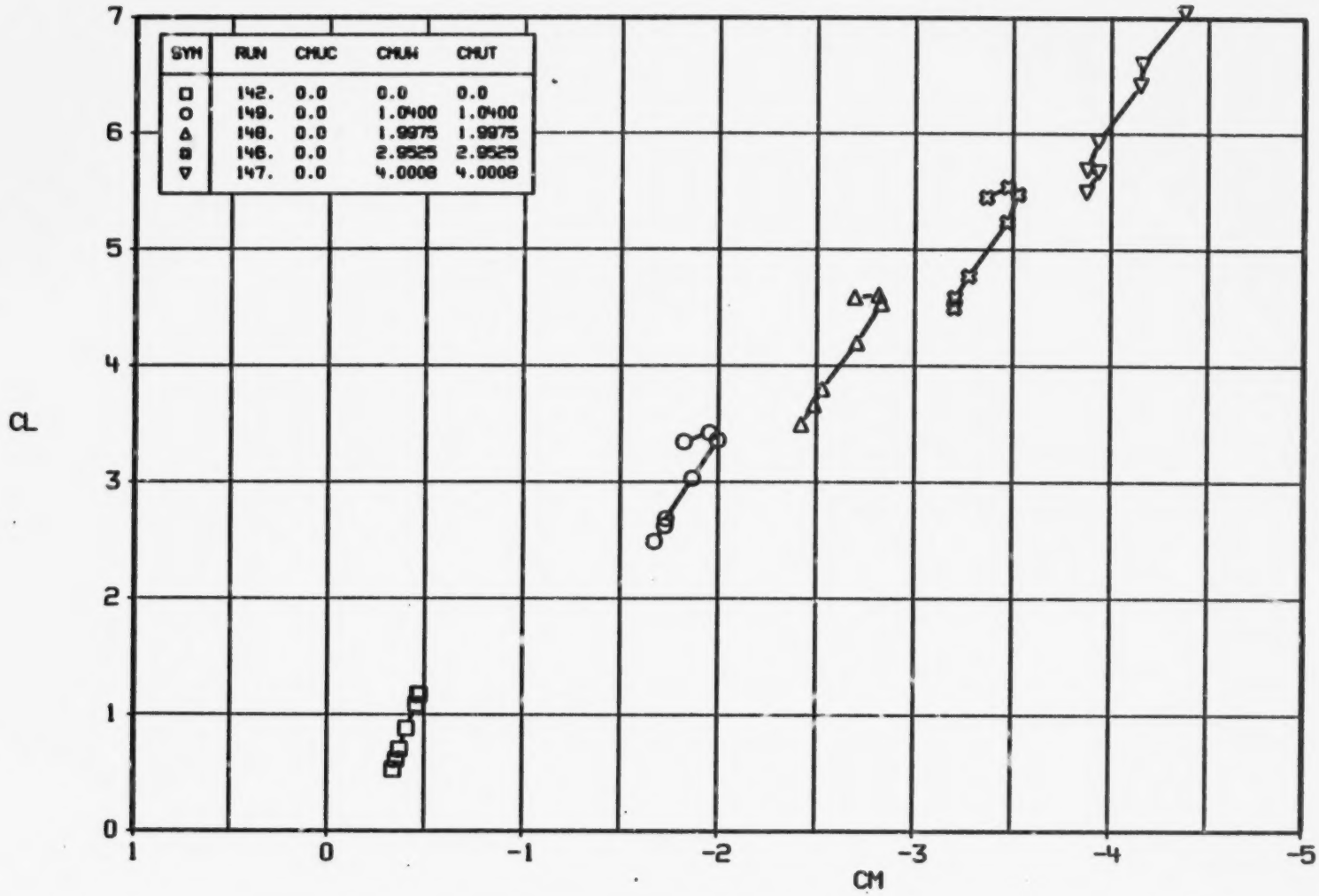

FIGURE A11c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

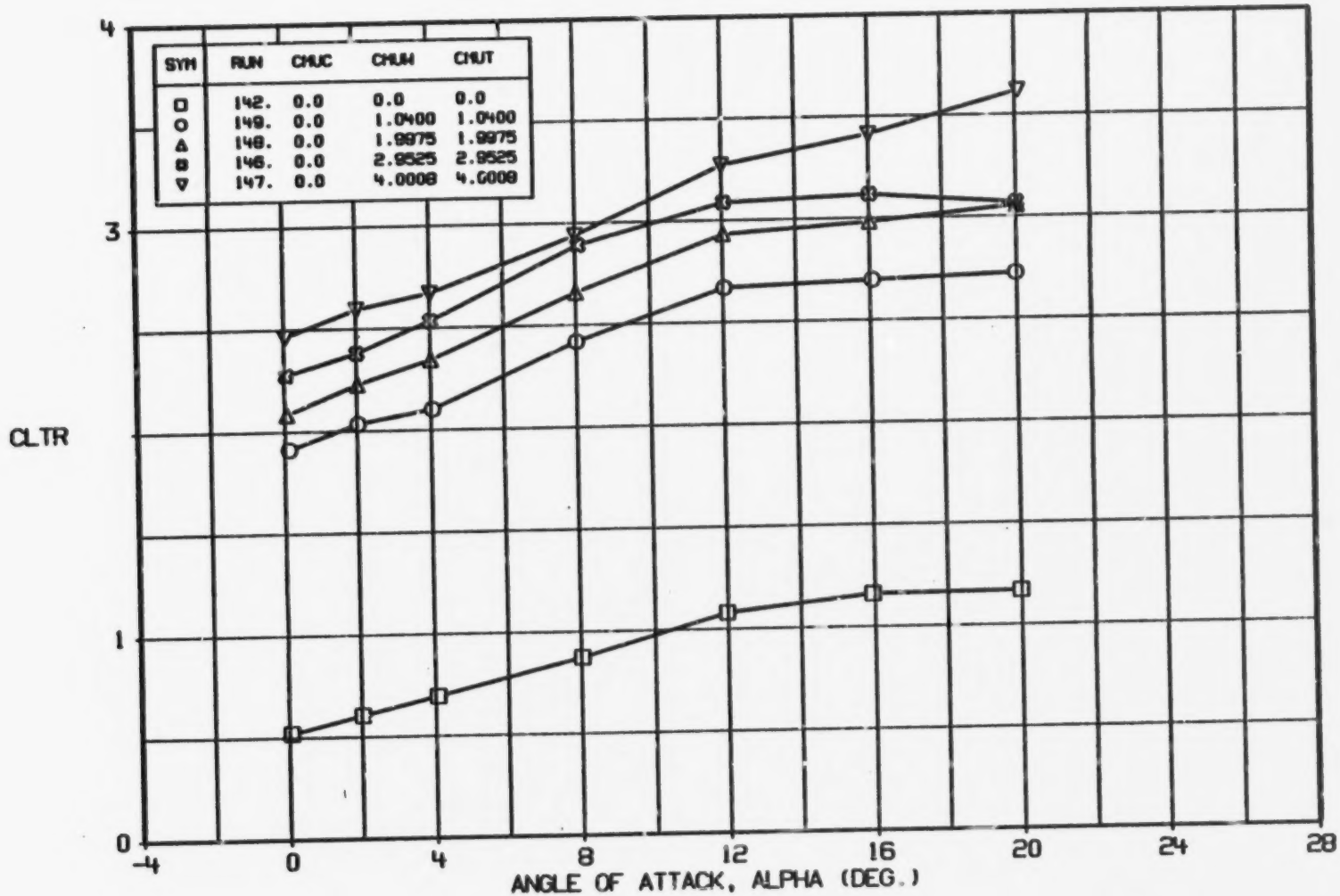

FIGURE A11d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

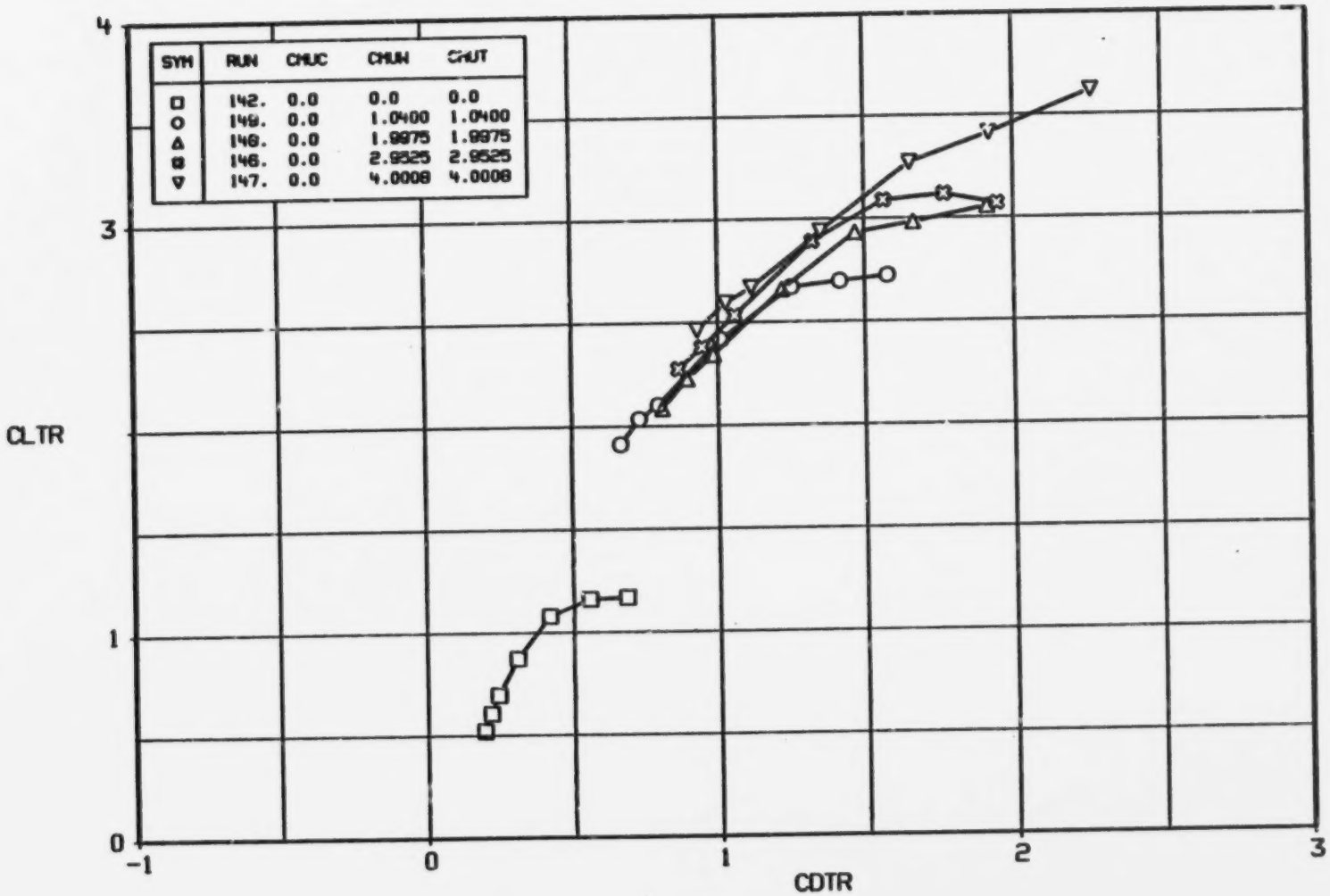

FIGURE A11e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

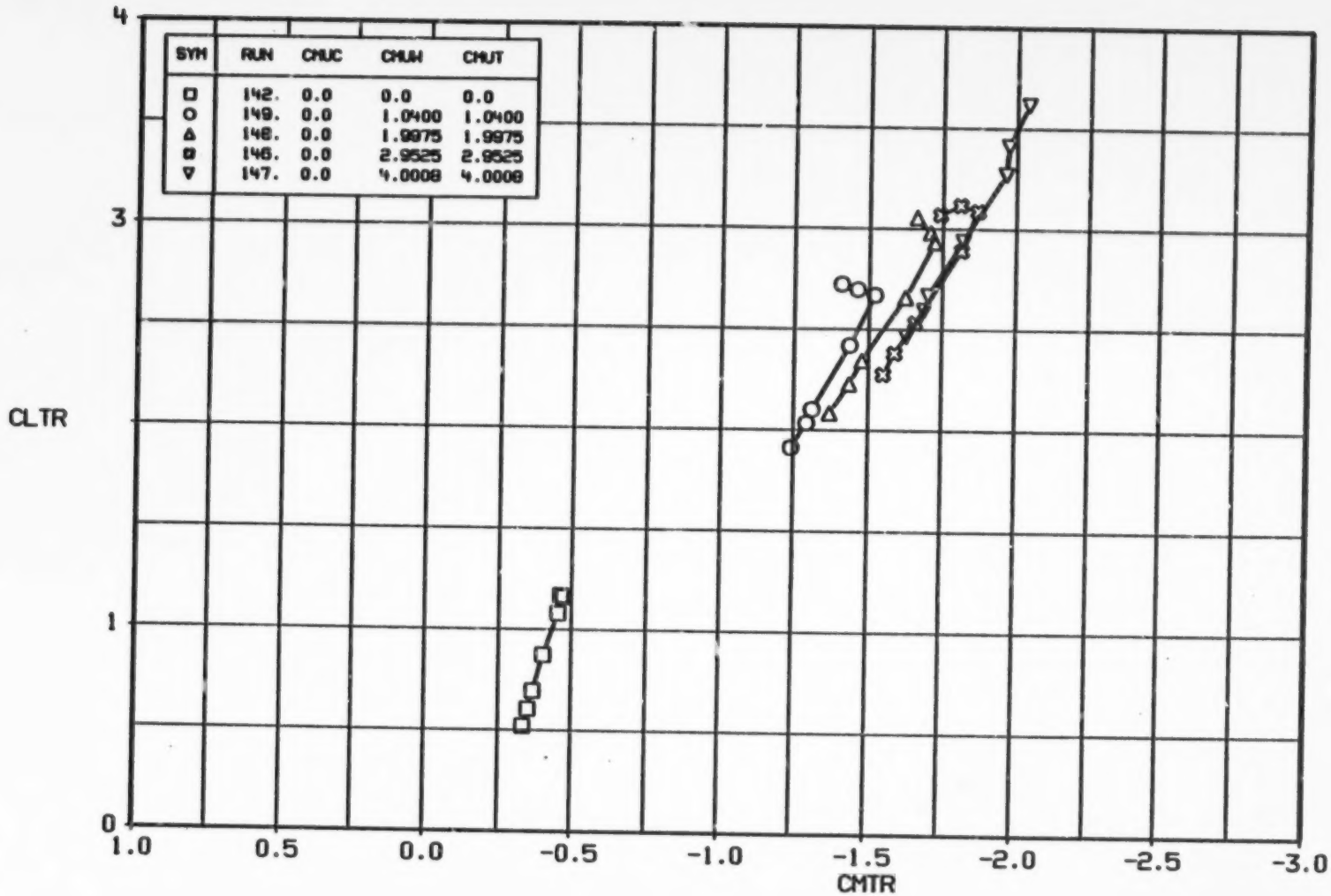

FIGURE A11f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

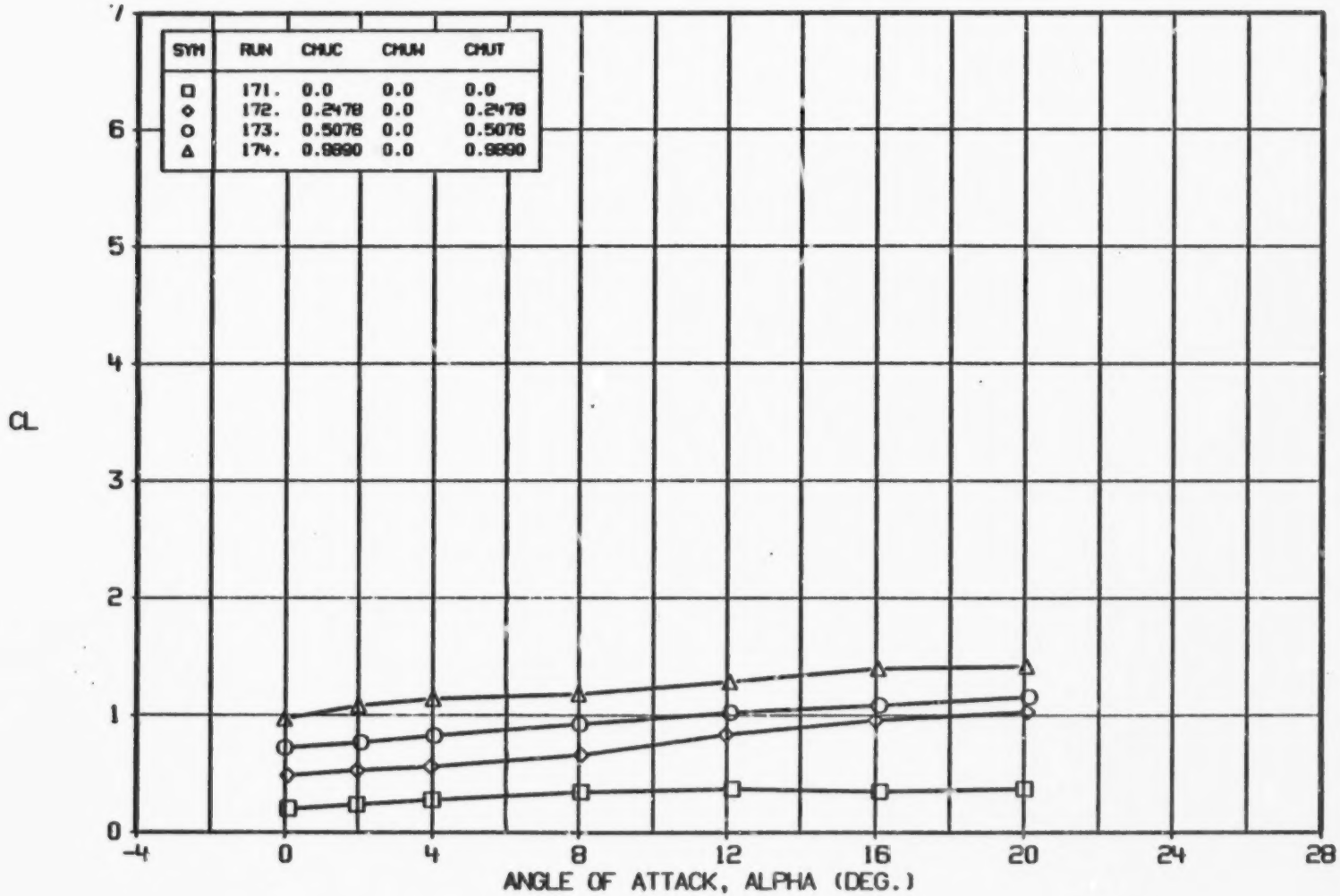

FIGURE A12a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1

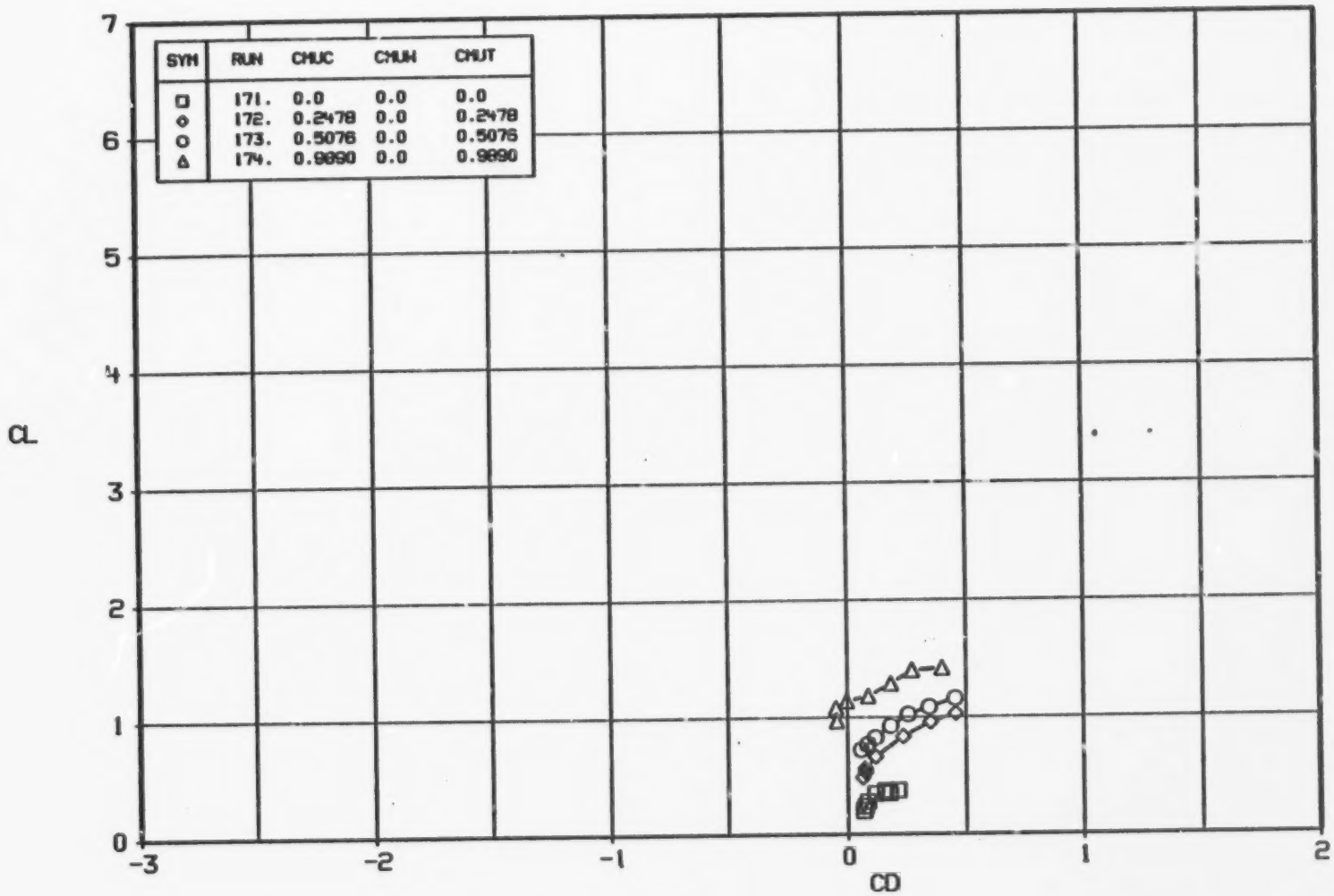

FIGURE A12b BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

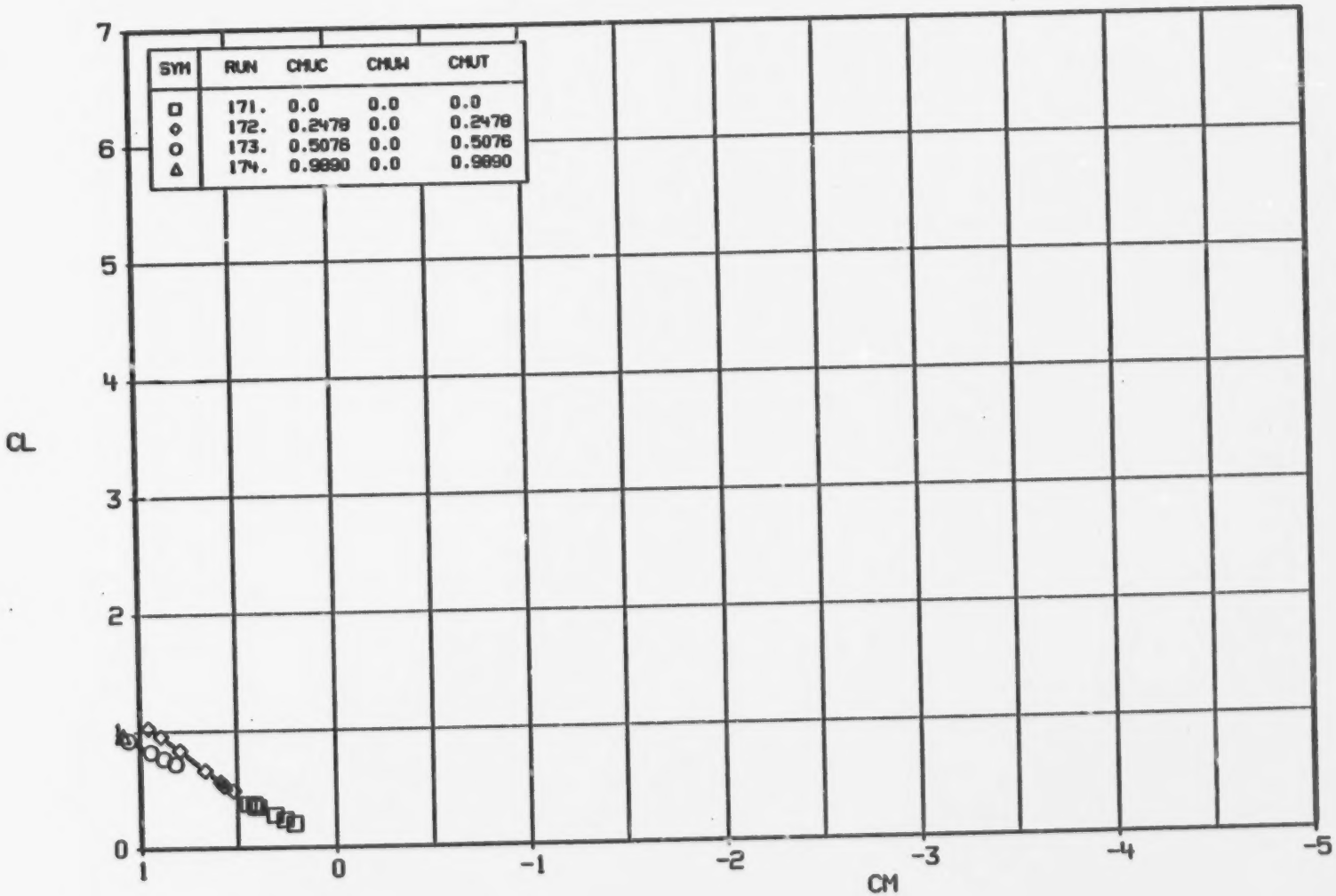

FIGURE A12c BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

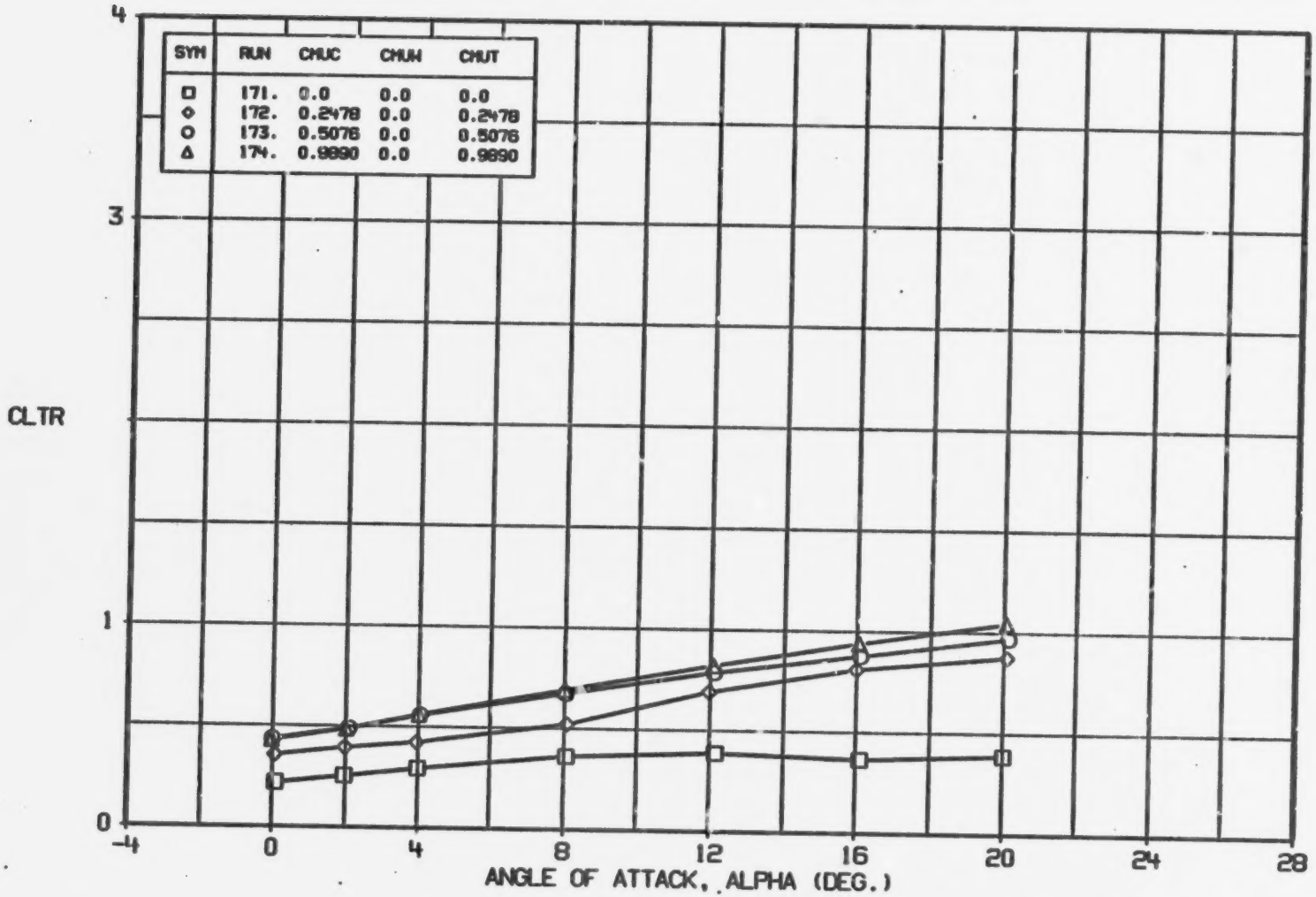

FIGURE A12d BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

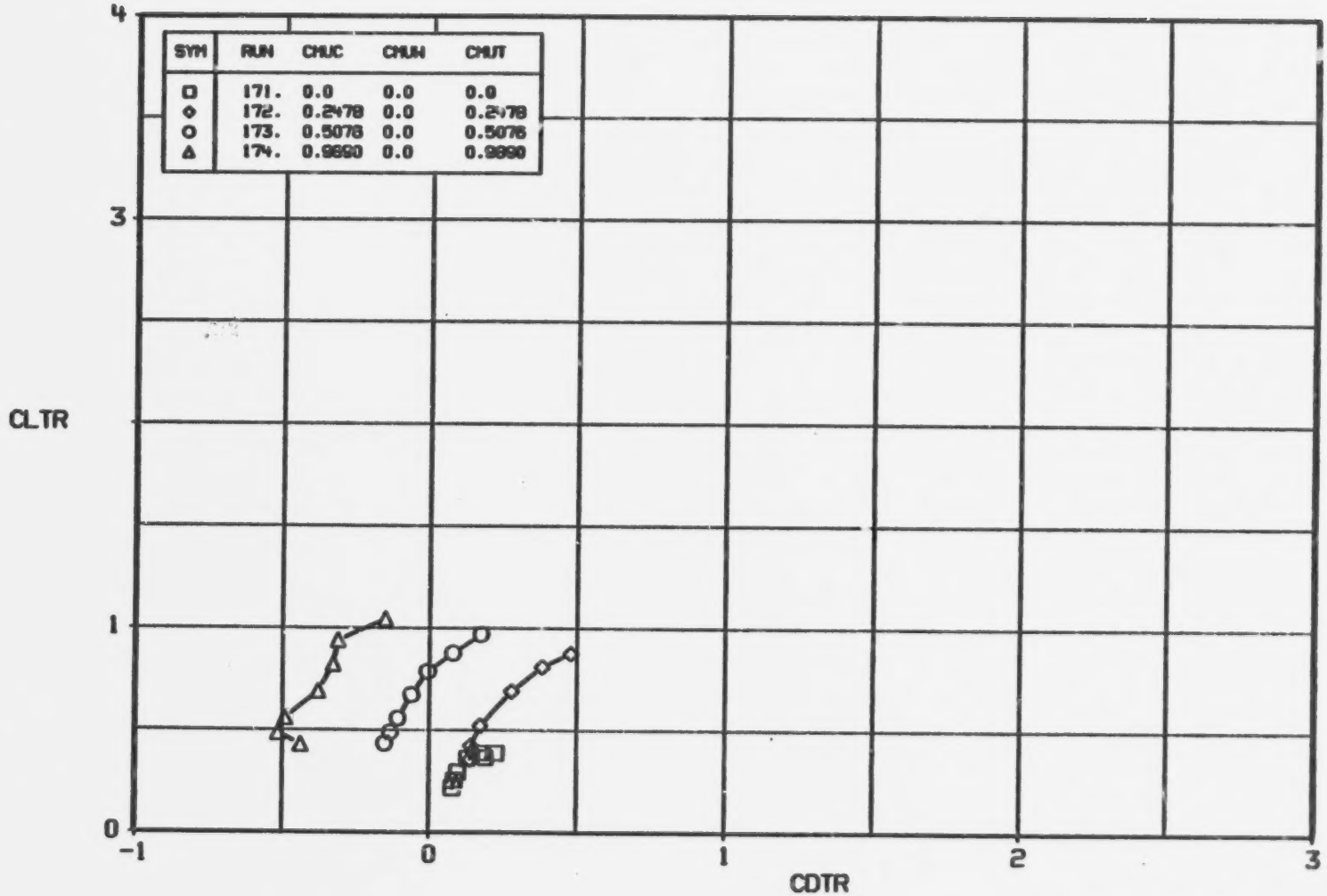

FIGURE A12e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1V, DELF=45, DELC=10, BN/B=1

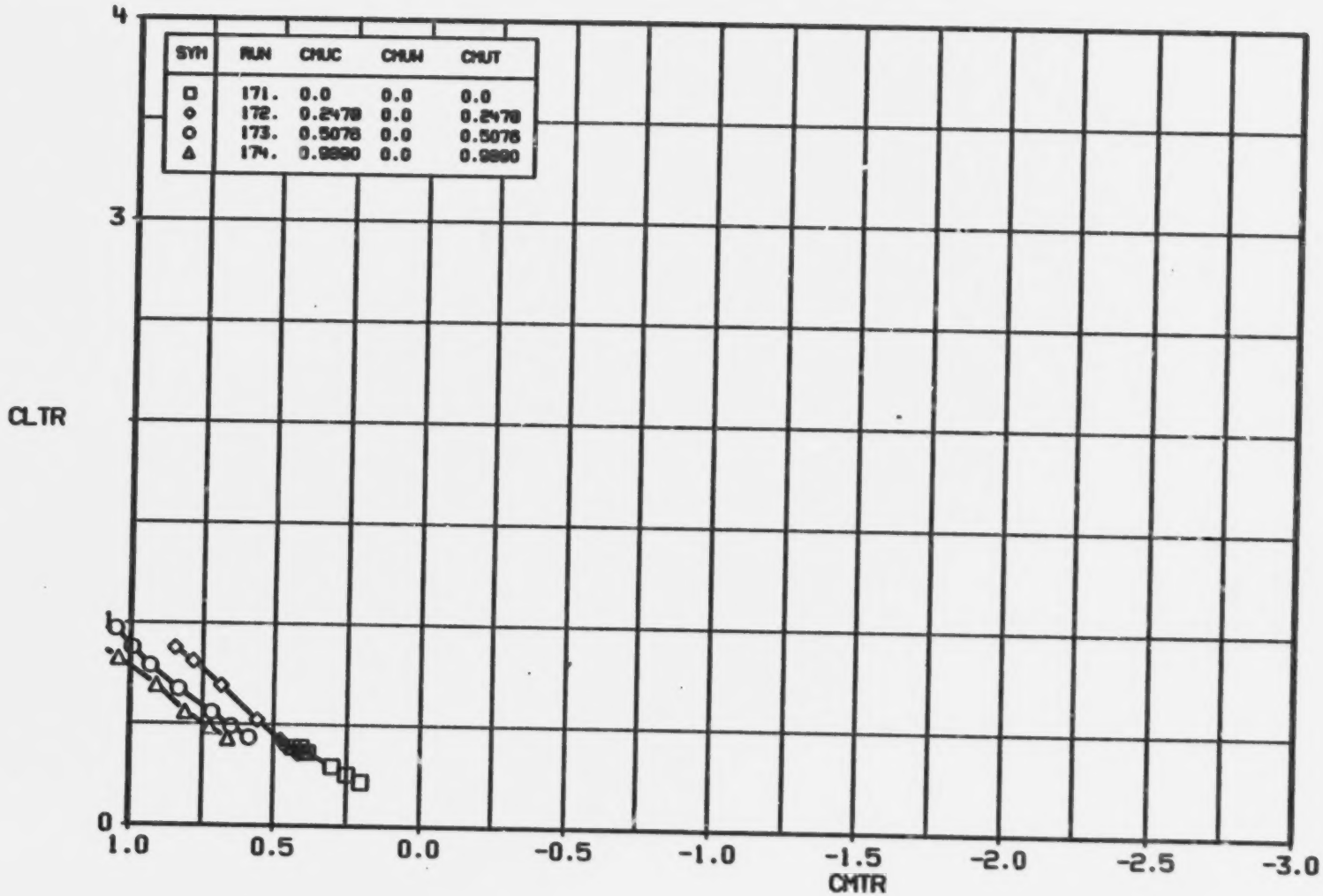

FIGURE A12f BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, DELC=10, BN/B=1

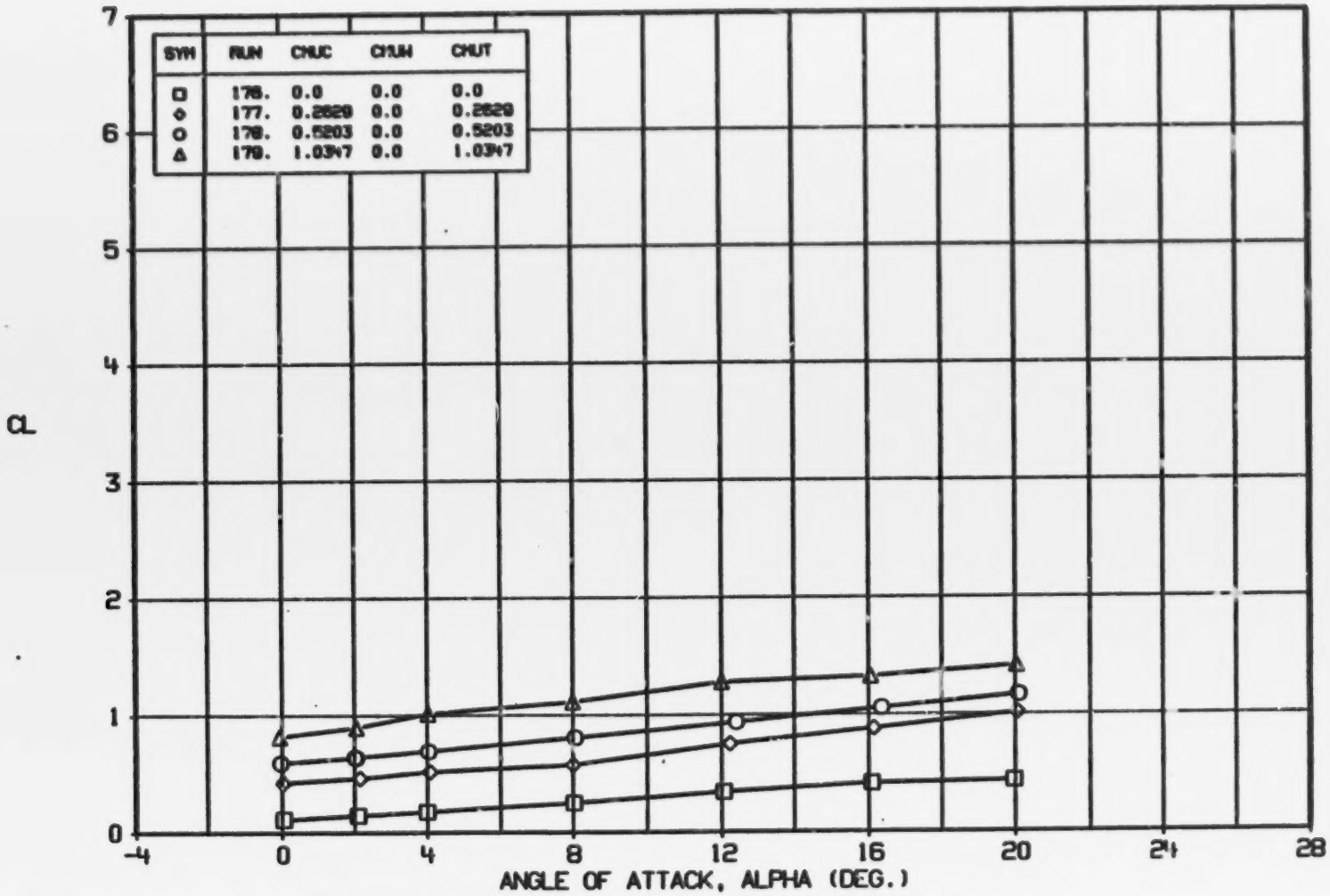

FIGURE A13a BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, BN/B=1

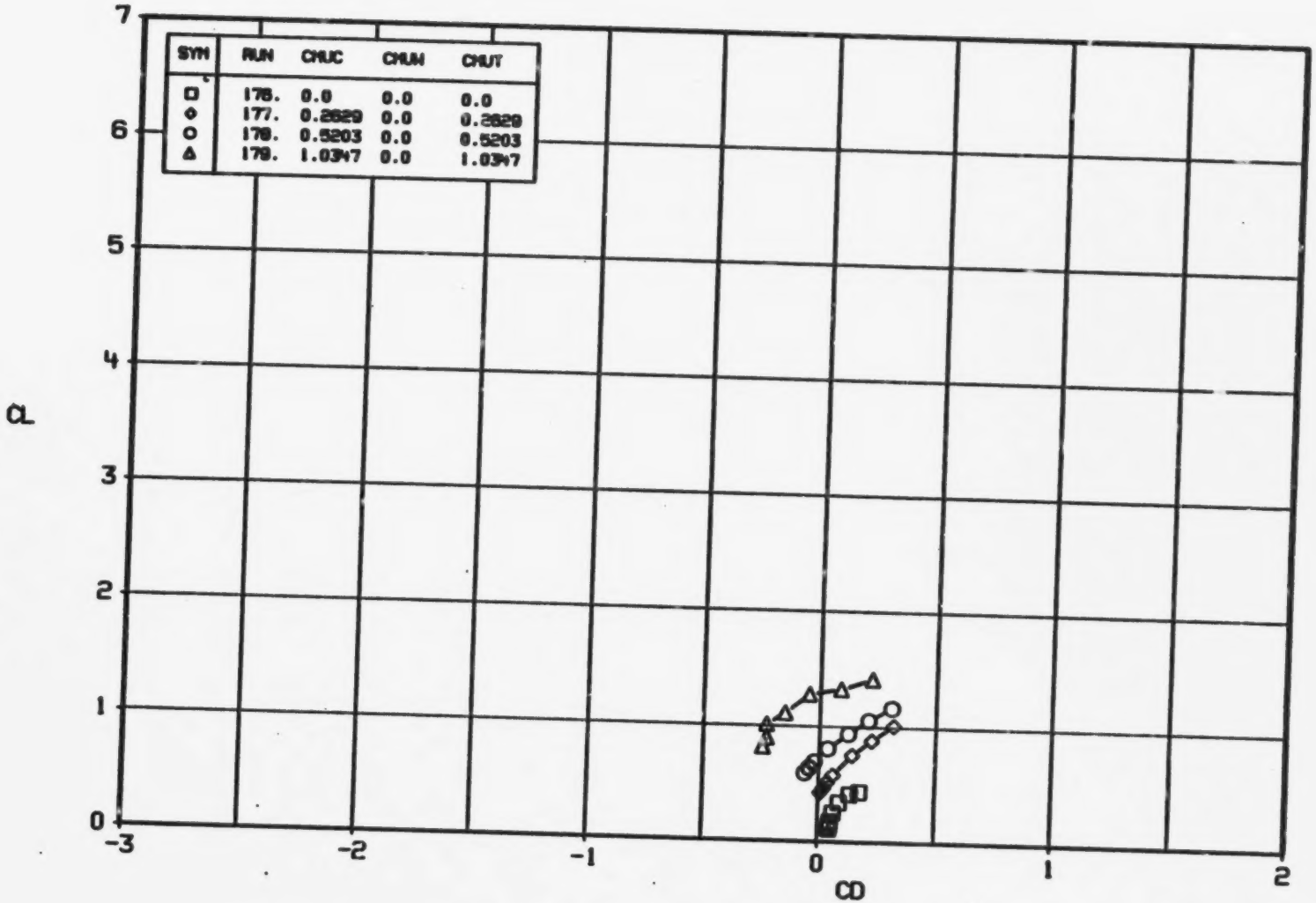

FIGURE A13b BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, BN/B=1

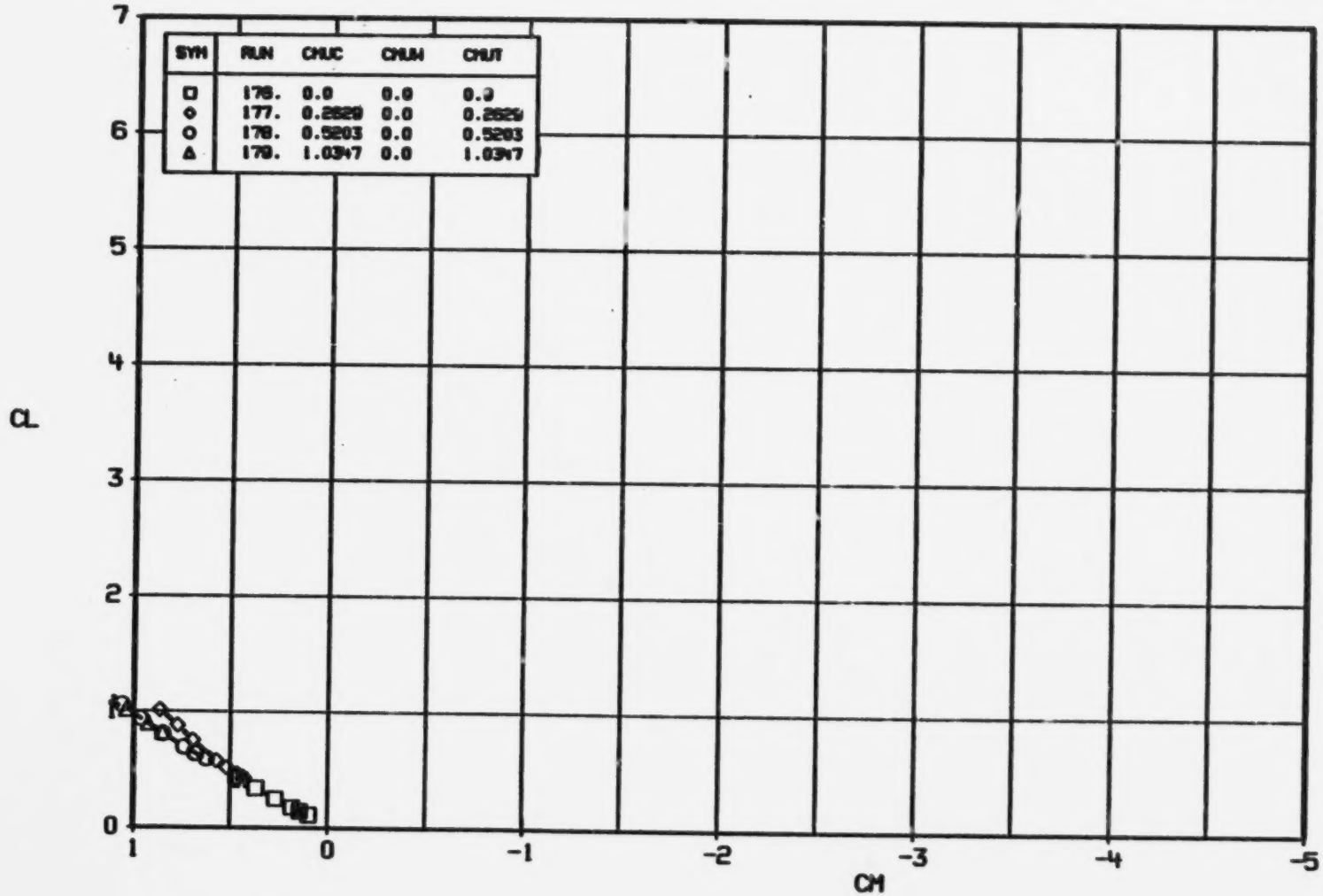

FIGURE A13c BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, BN/B=1

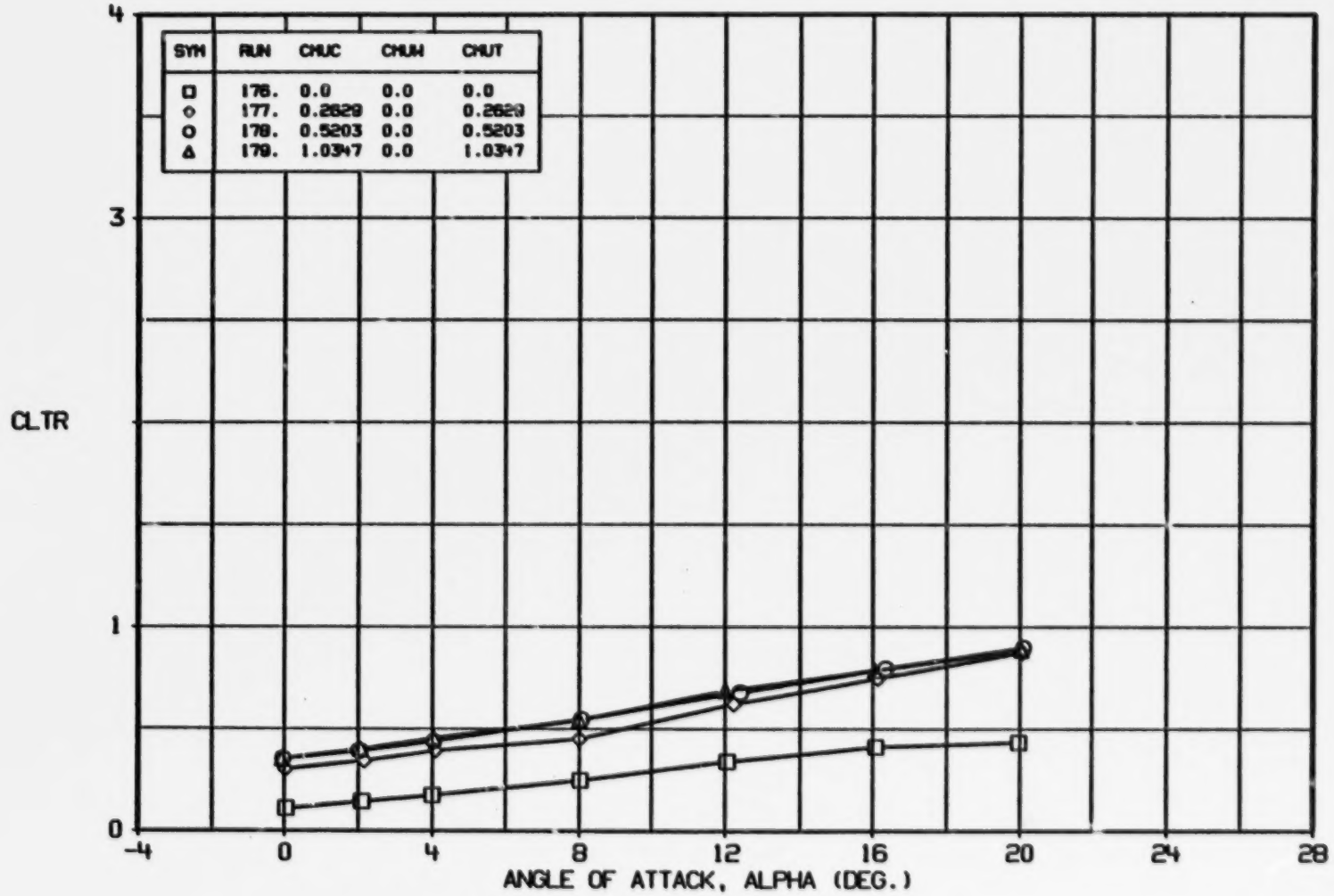

FIGURE A13d BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, BN/B=1

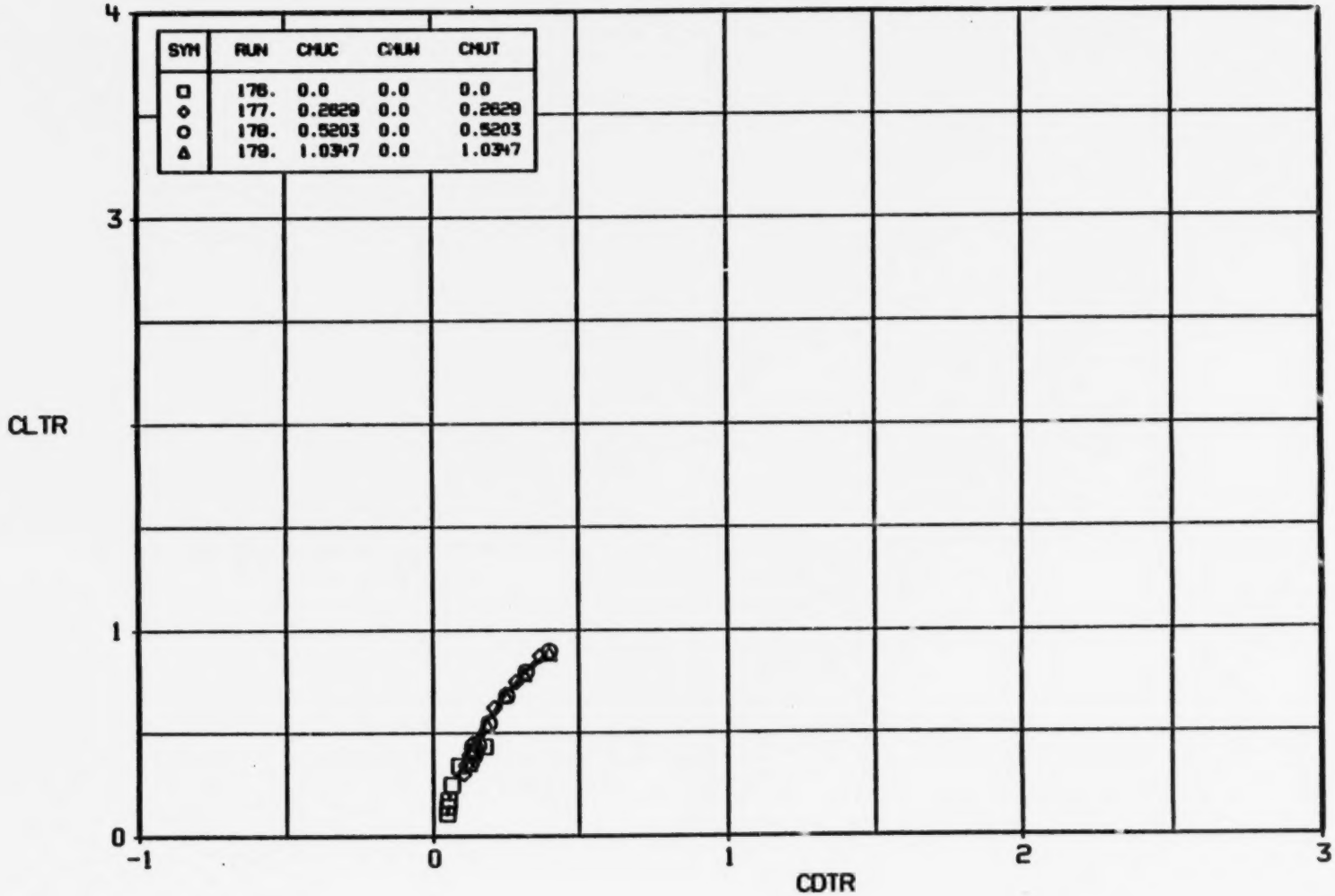

FIGURE A13e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1V, DELF=45, BN/B=1

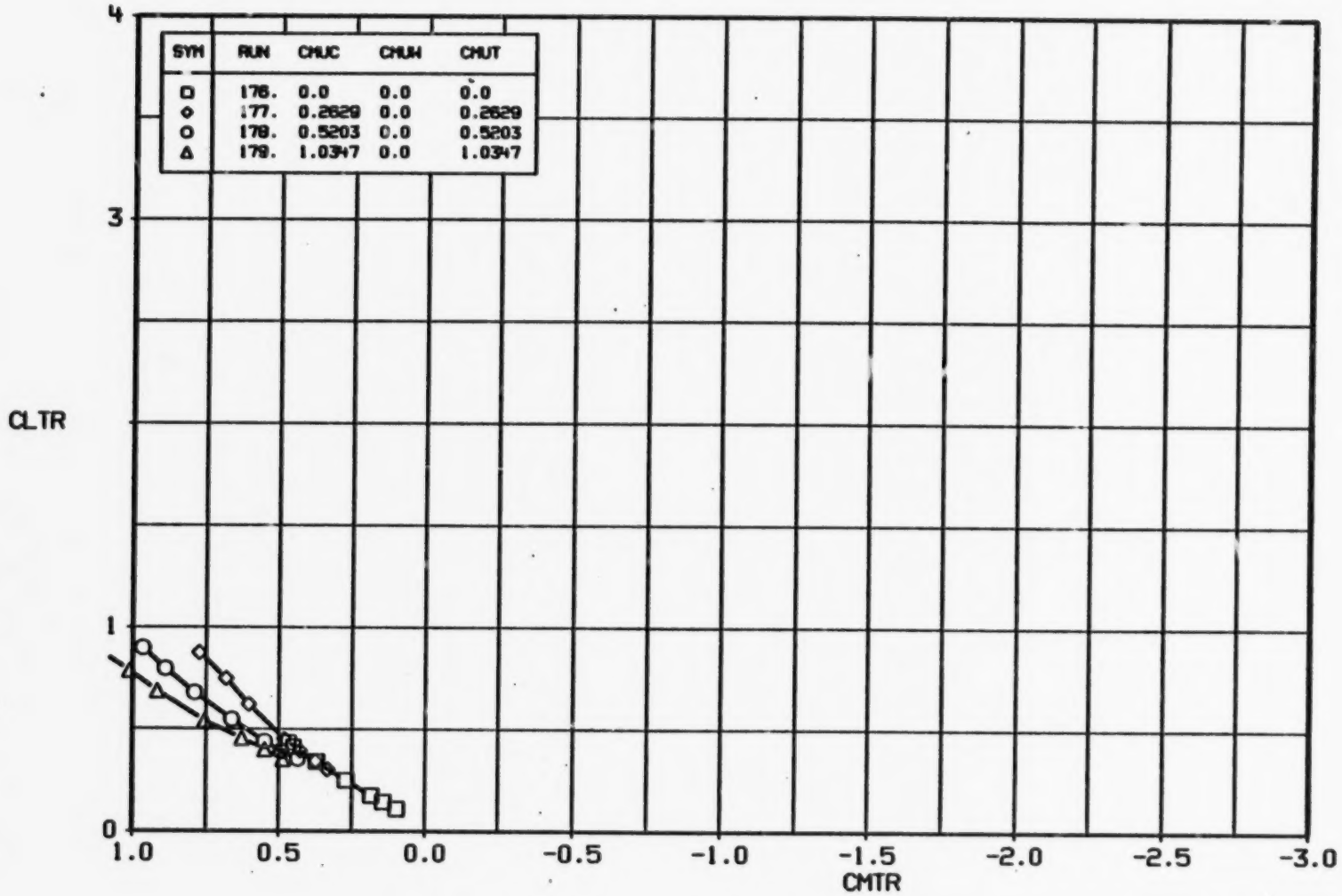

FIGURE A13f BASIC DATA EFFECT OF CMU
CONFIGURATION BCIV, DELF=45, BN/B=1

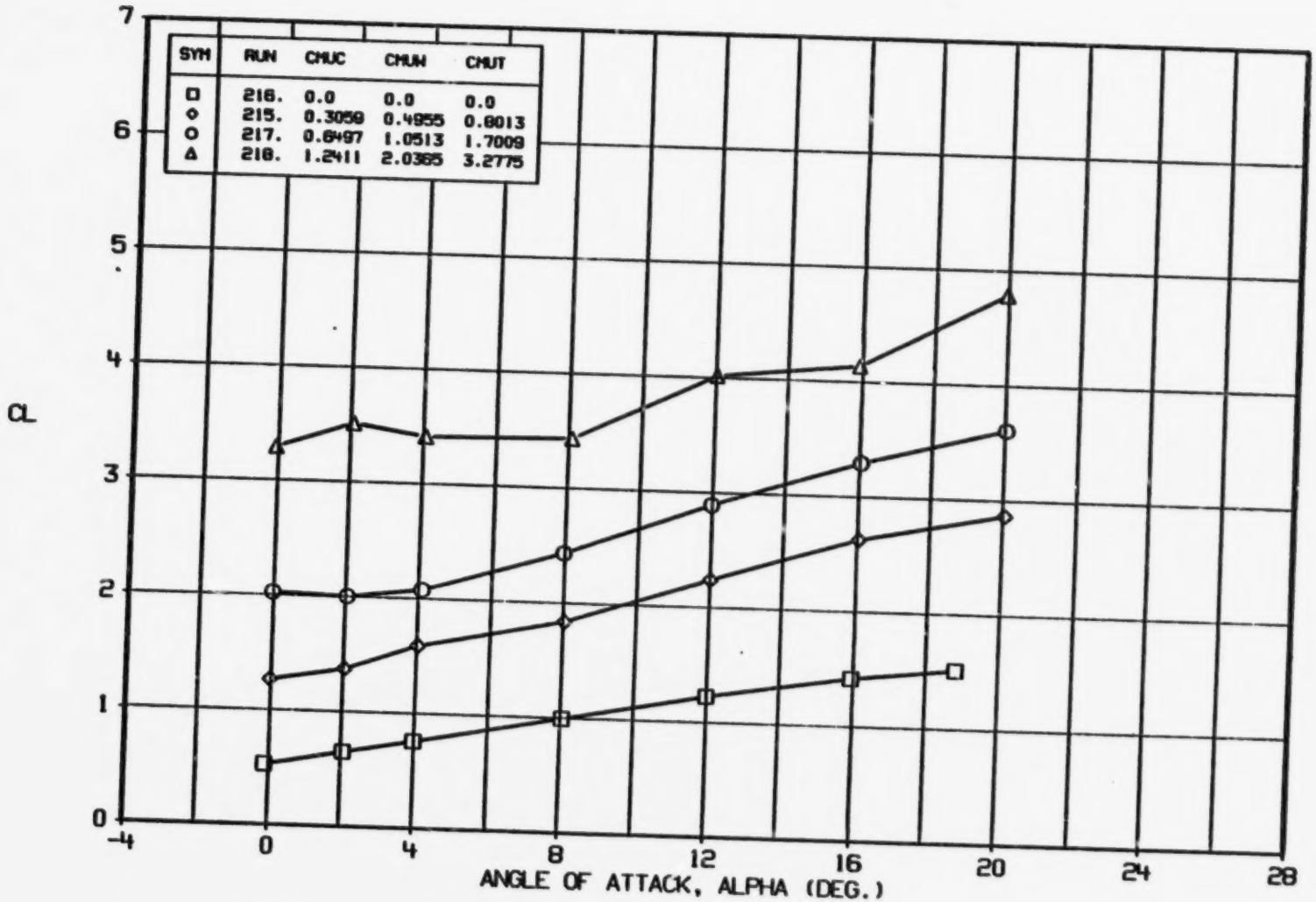

FIGURE A14a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V,(BN/B)C=1,(BN/B)W=0.5, DELF=45

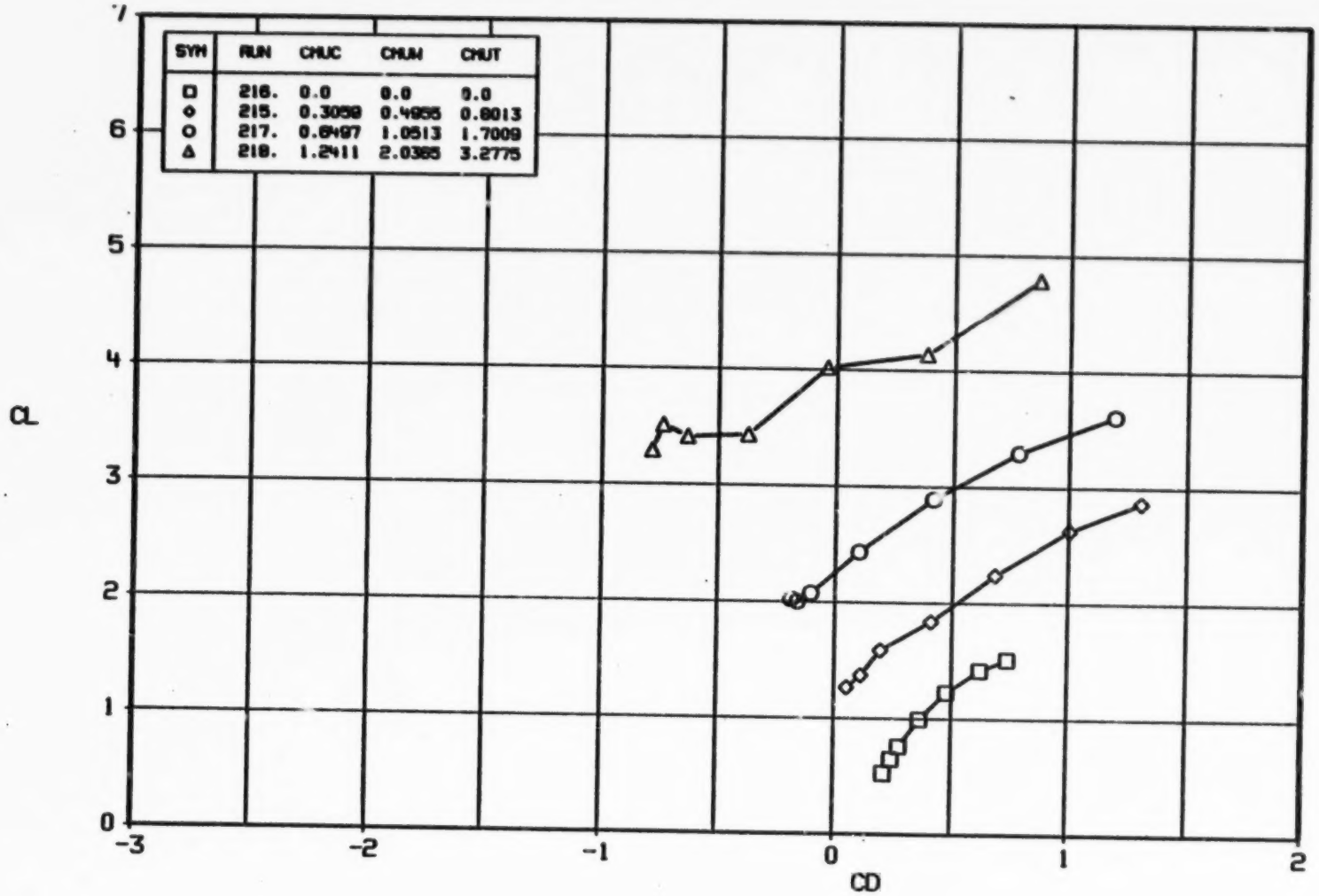

FIGURE A14b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

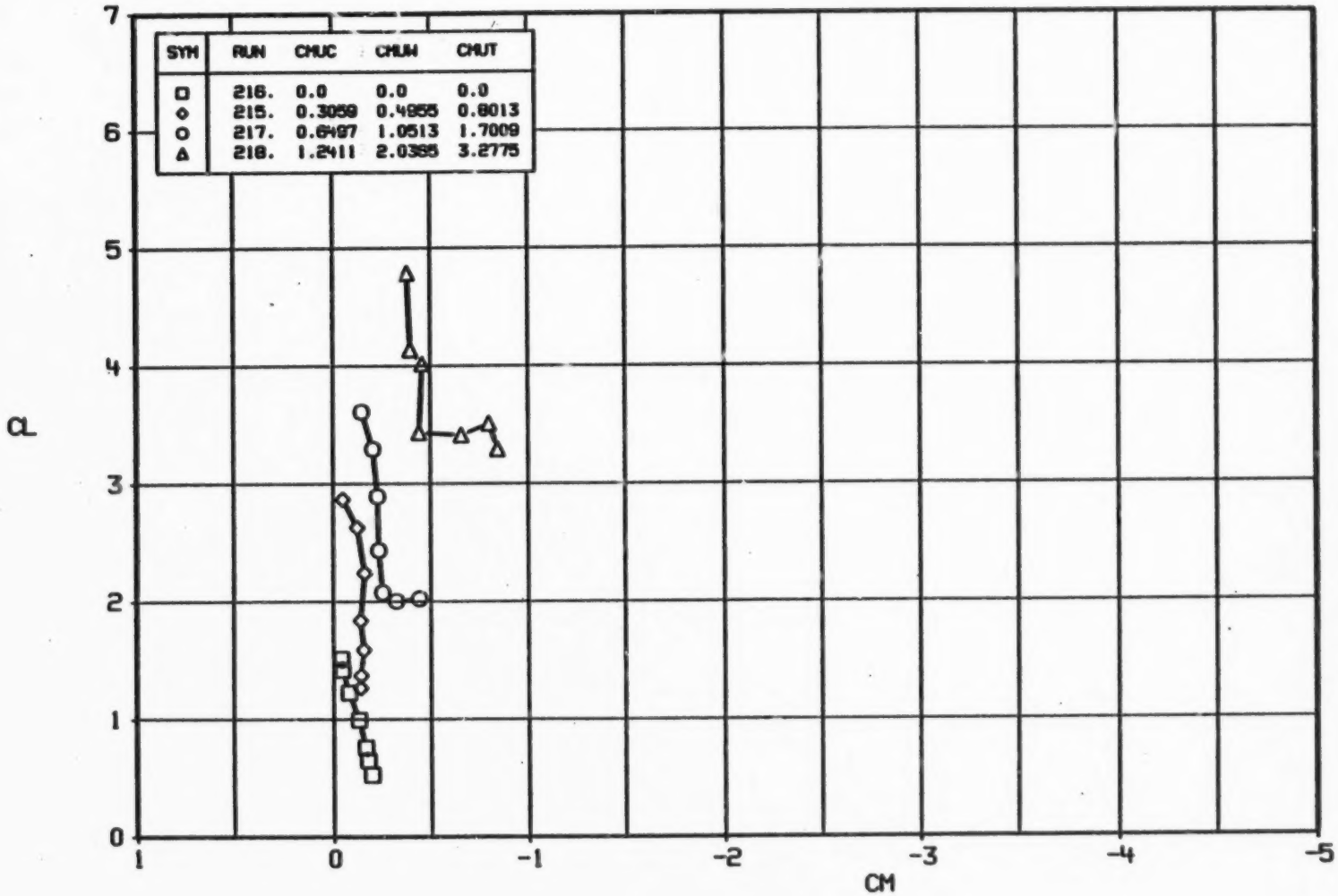

FIGURE A14c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V,(BN/B)C=1,(BN/B)W=0.5 , DELF=45

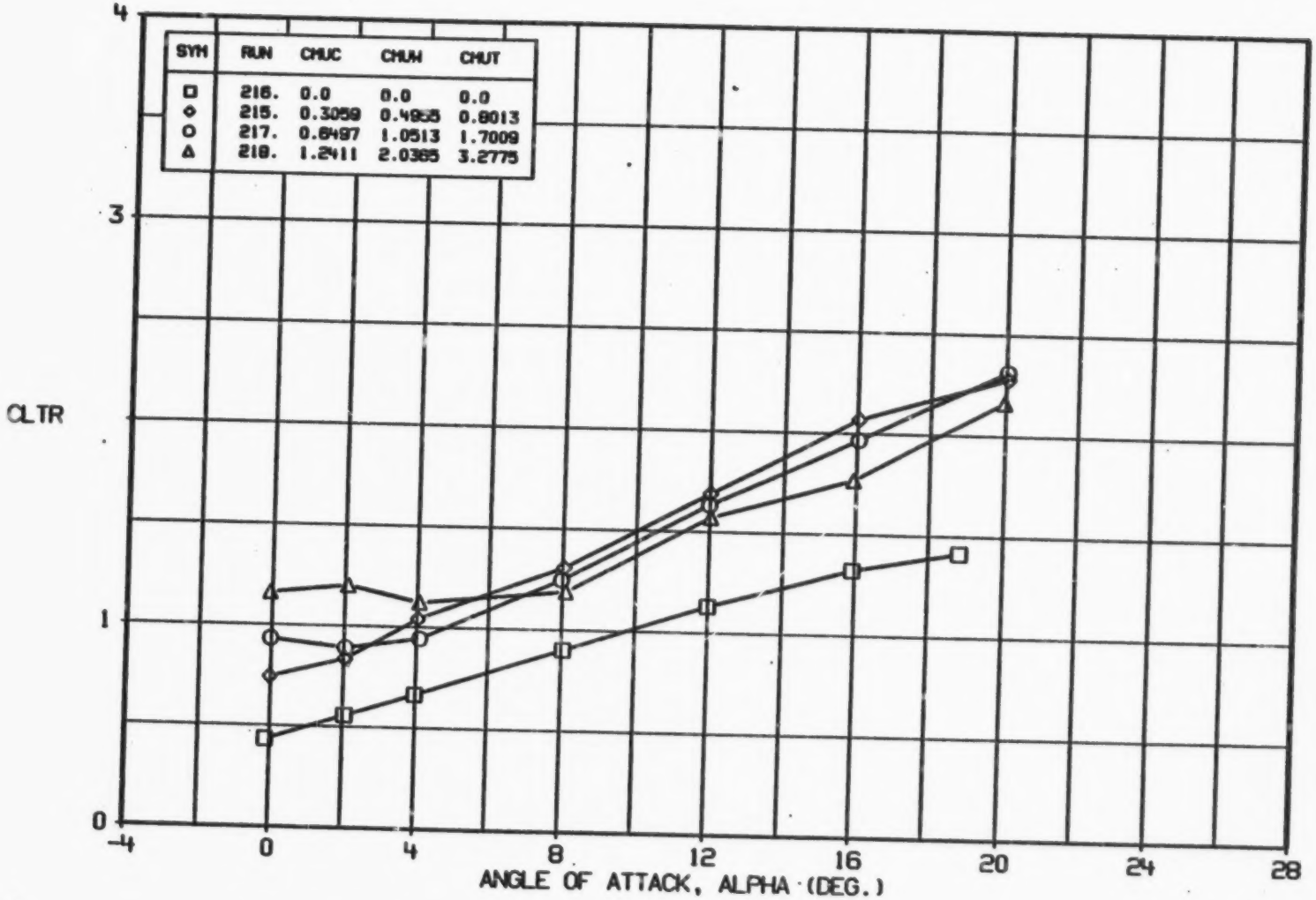

FIGURE A14d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

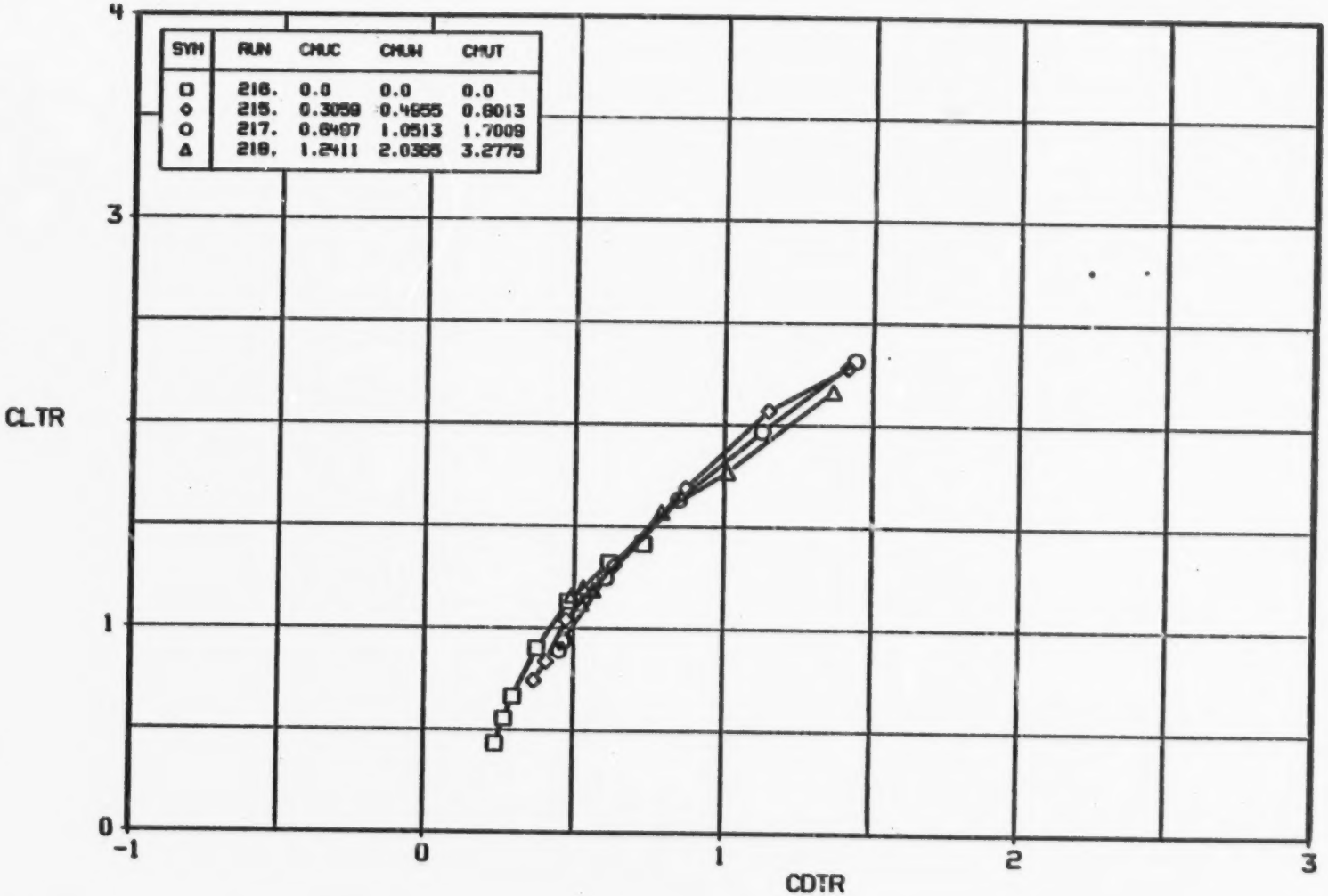

FIGURE A14e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V,(BN/B)C=1,(BN/B)W=0.5, DELF=45

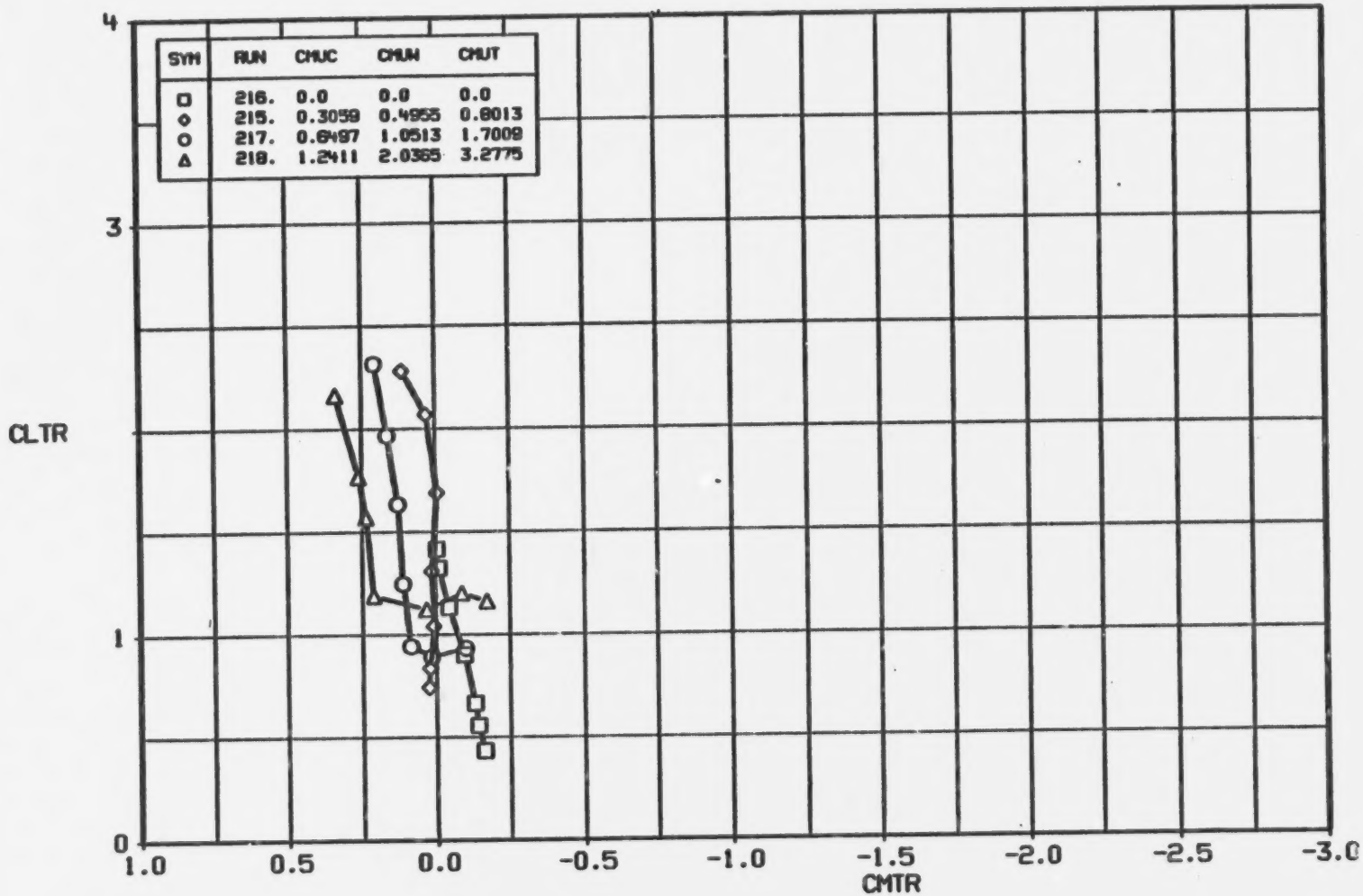

FIGURE A14f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=45

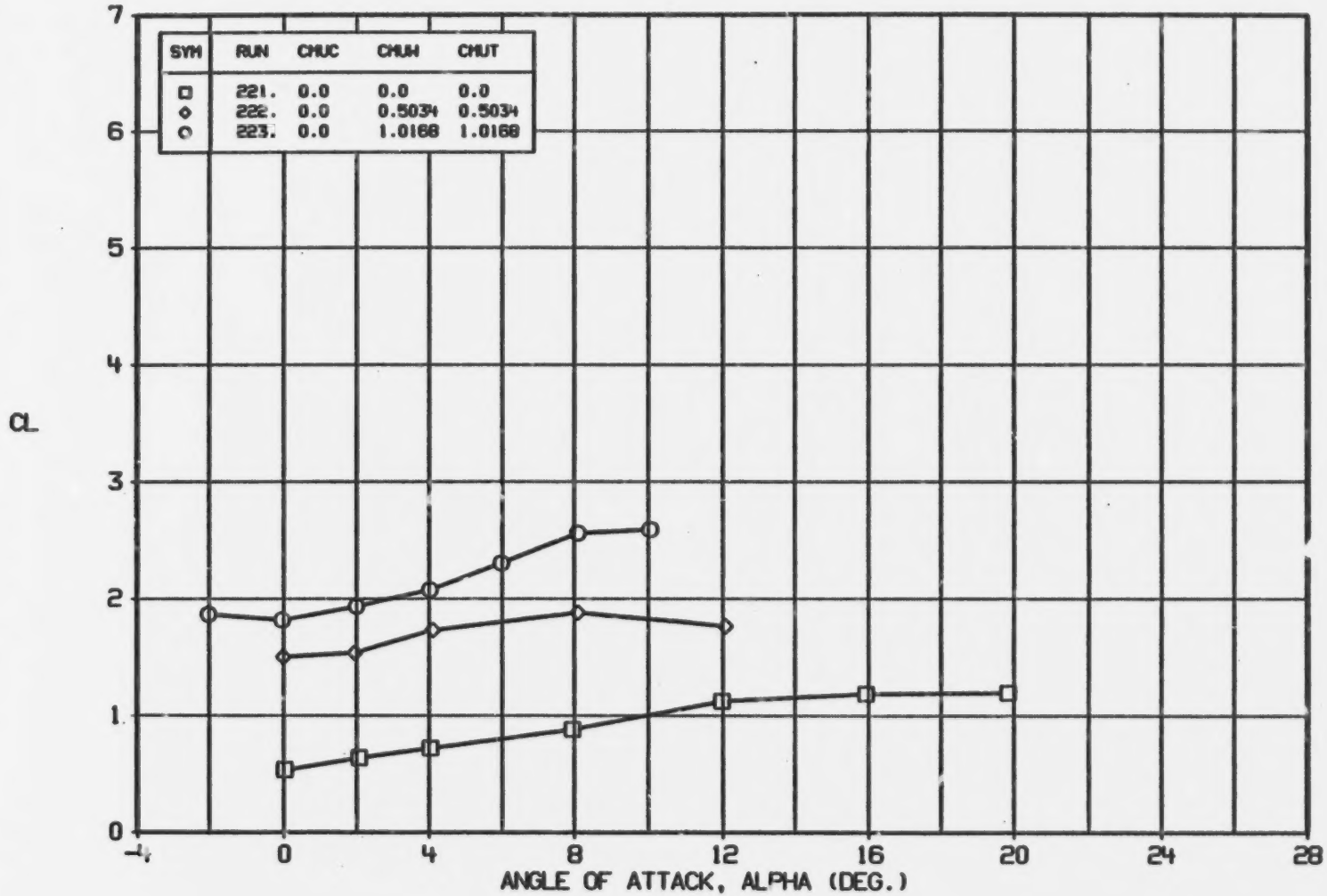

FIGURE A15a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

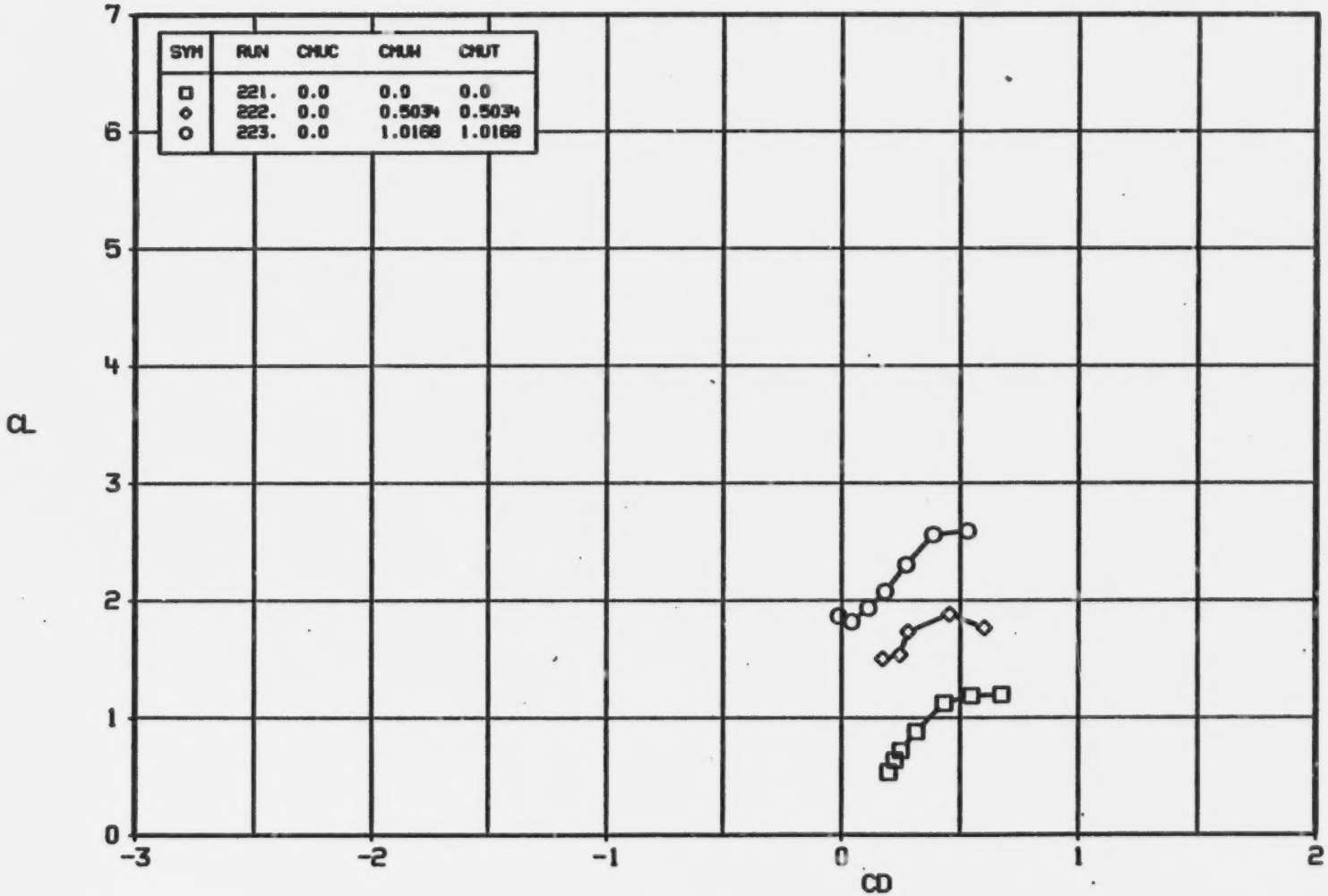

FIGURE A15b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

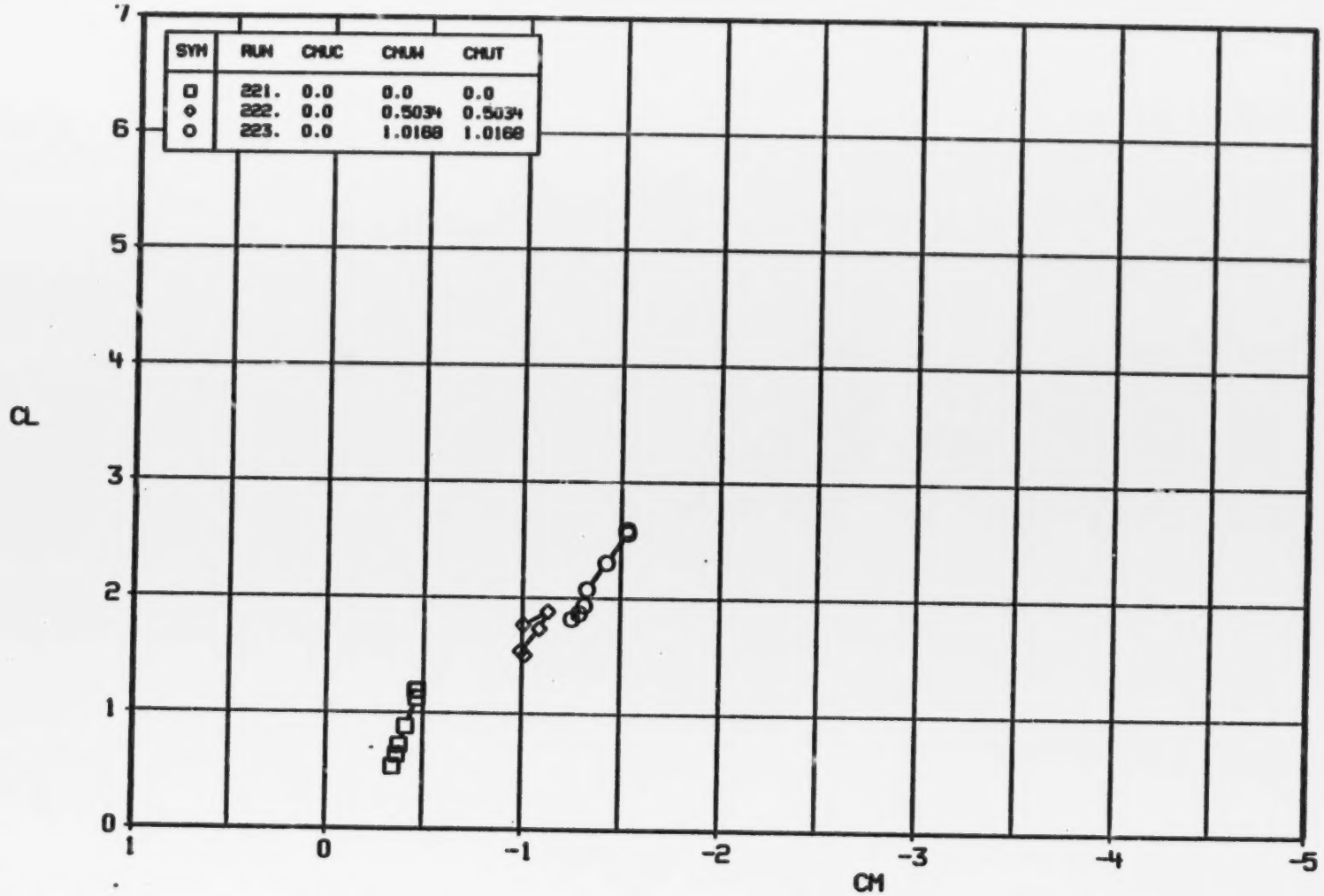

FIGURE A15c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

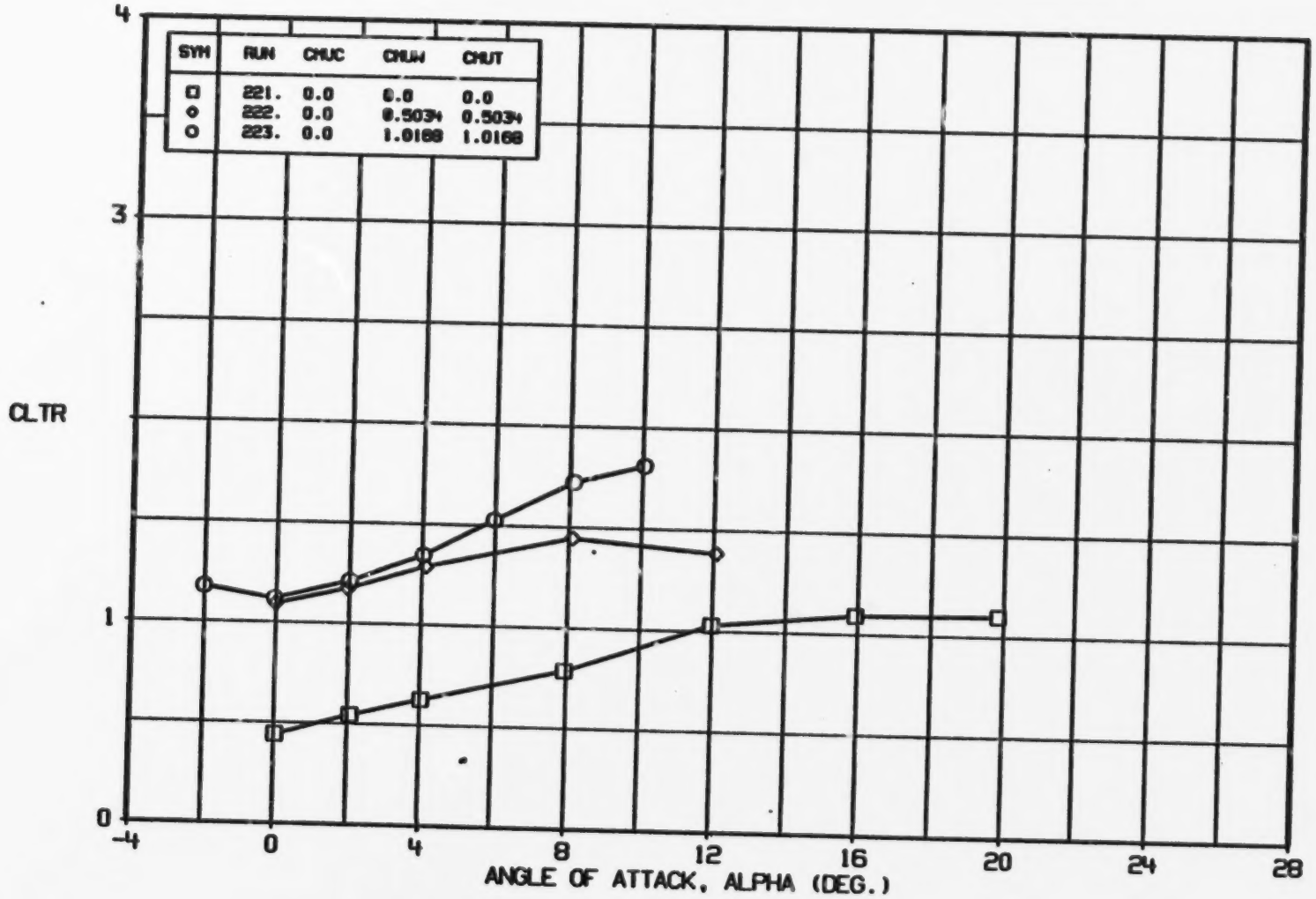

FIGURE A15d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

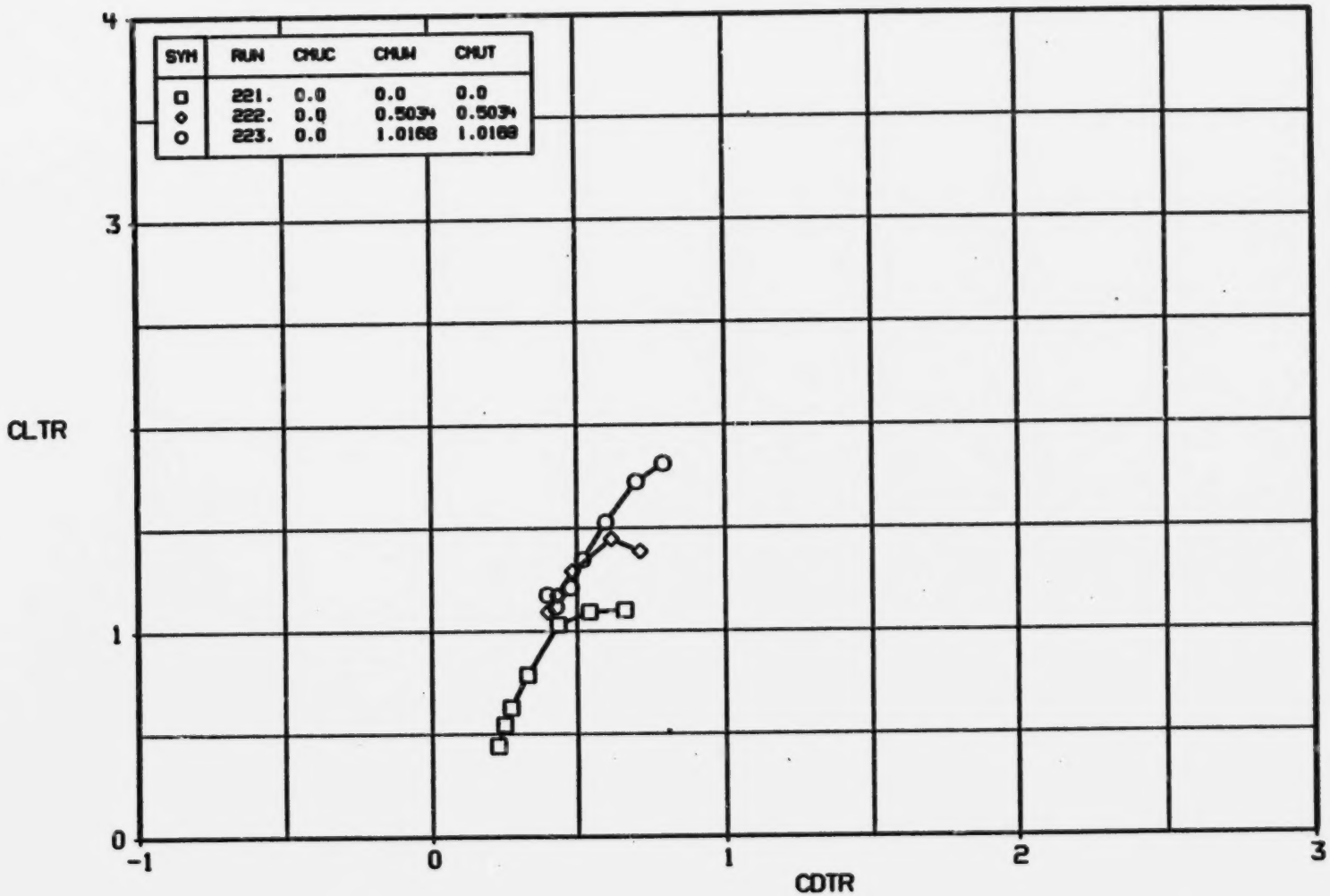

FIGURE A15e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

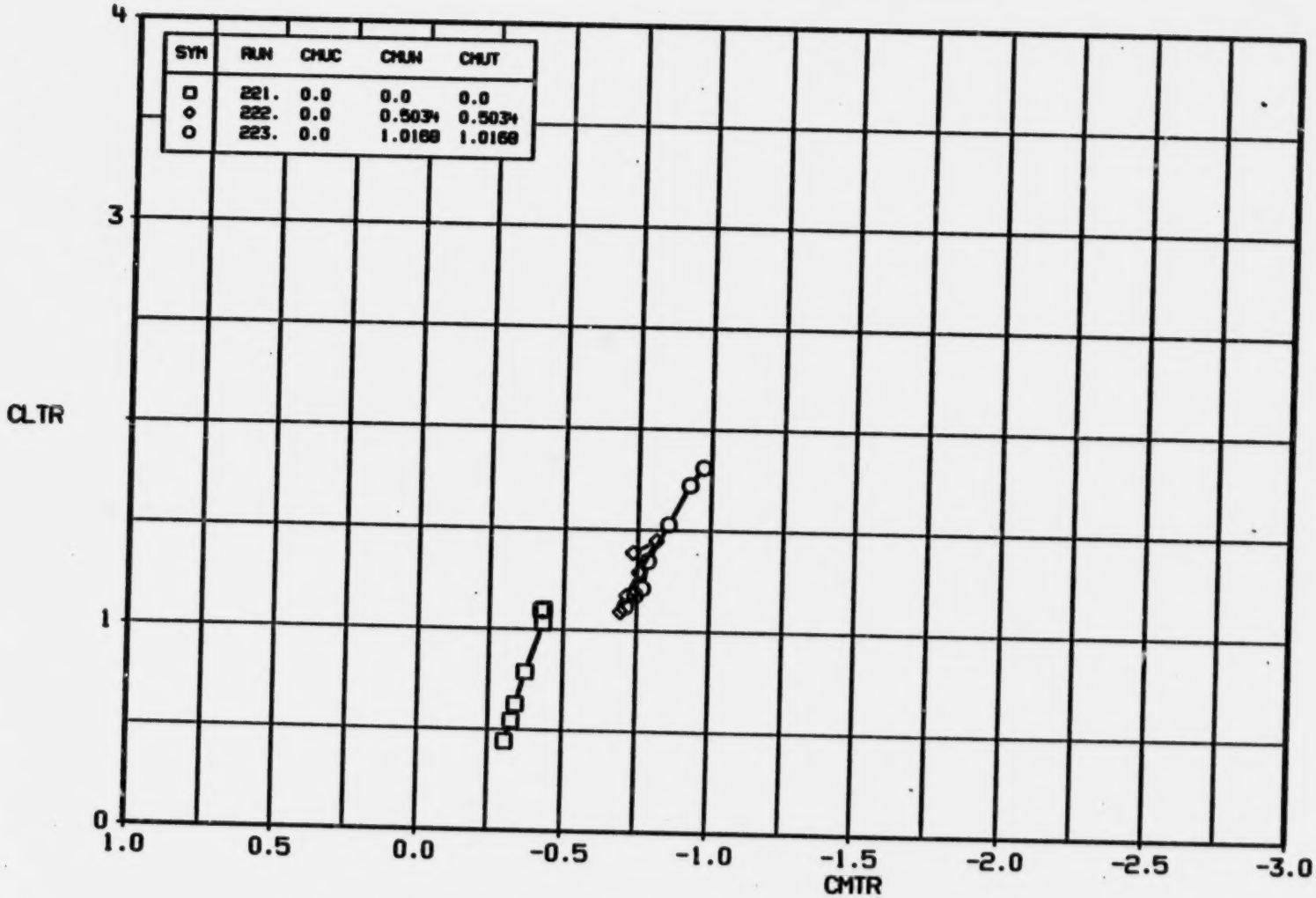

FIGURE A15f BASIC DATA EFFECT OF CMU
CONFIGURATION BN6V, BN/B=0.5 , DELF=45

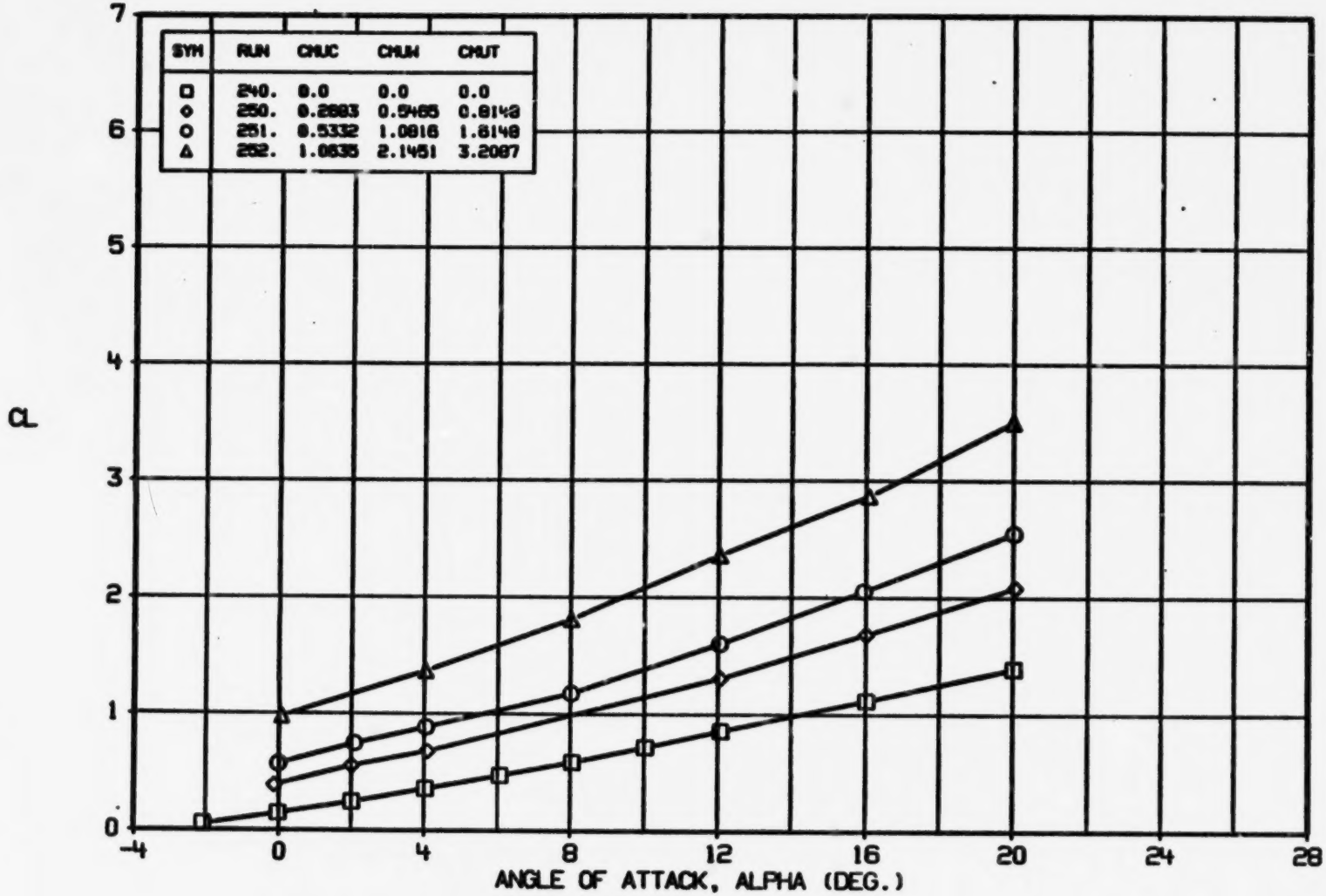

FIGURE A16a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V,(BN/B)C=1,(BN/B)W=0.5, DELF=0

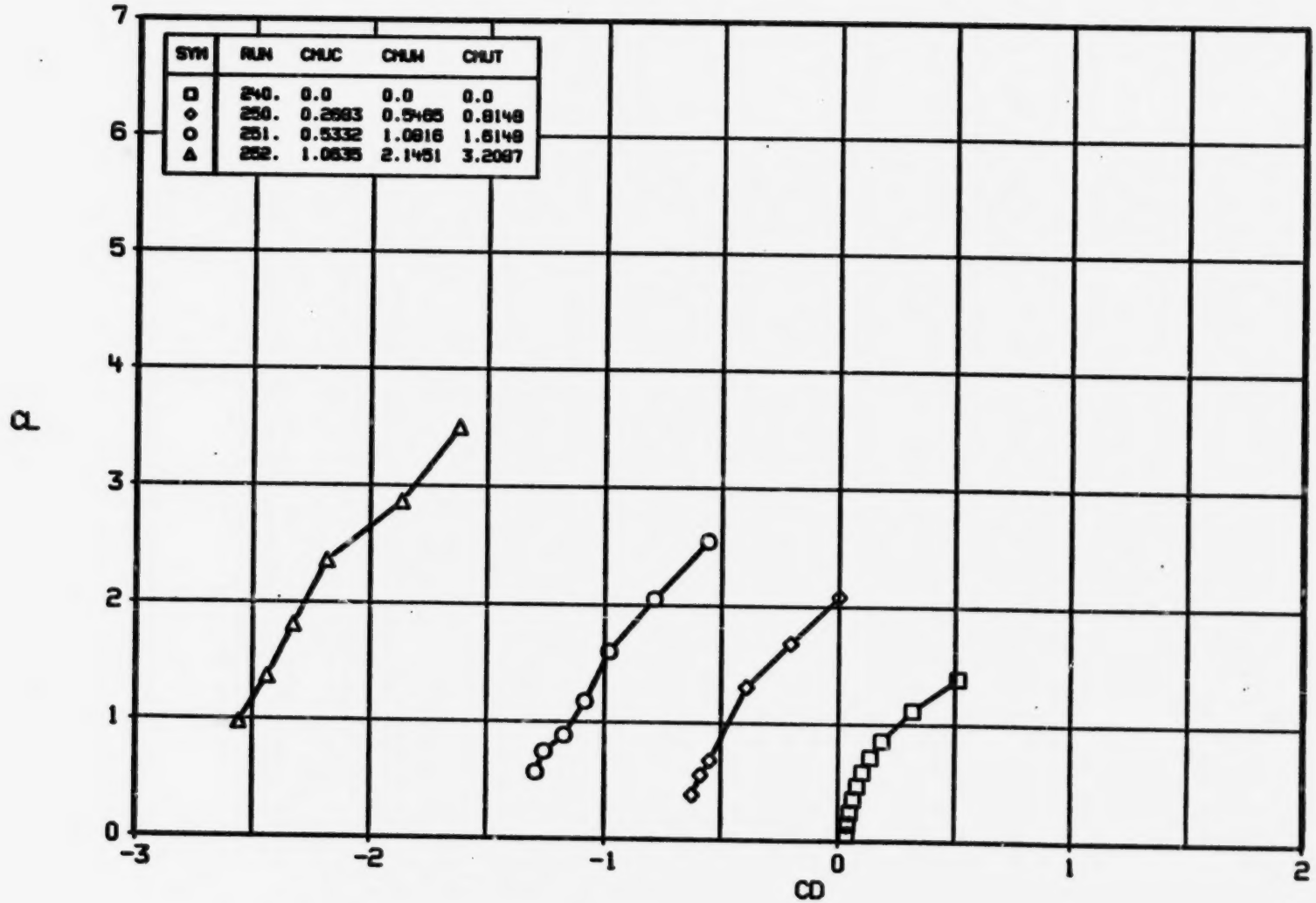

FIGURE A16b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, (BN/B)C=1, (BN/B)W=0.5 , DELF=0

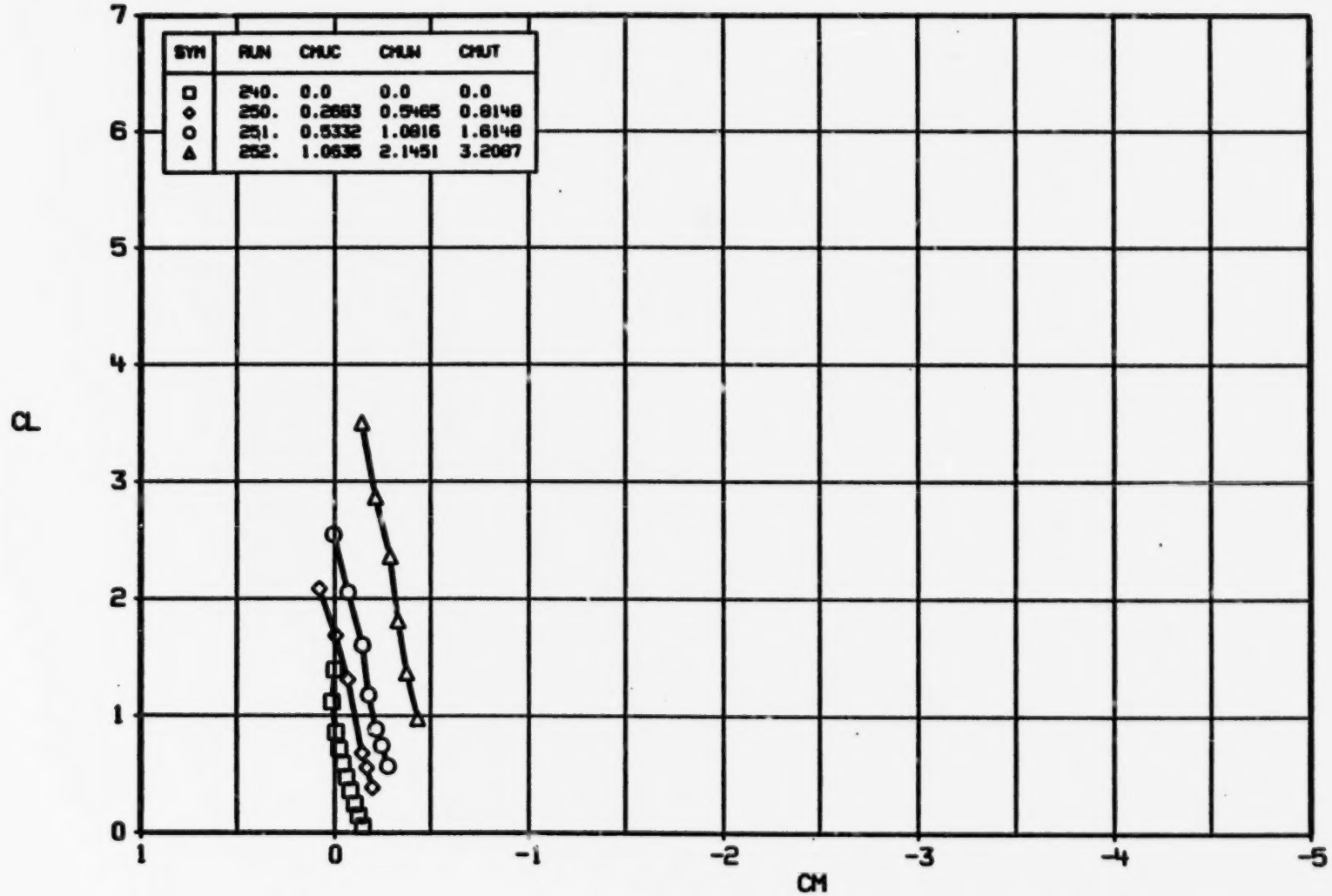

FIGURE A16c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.5, DELF=0

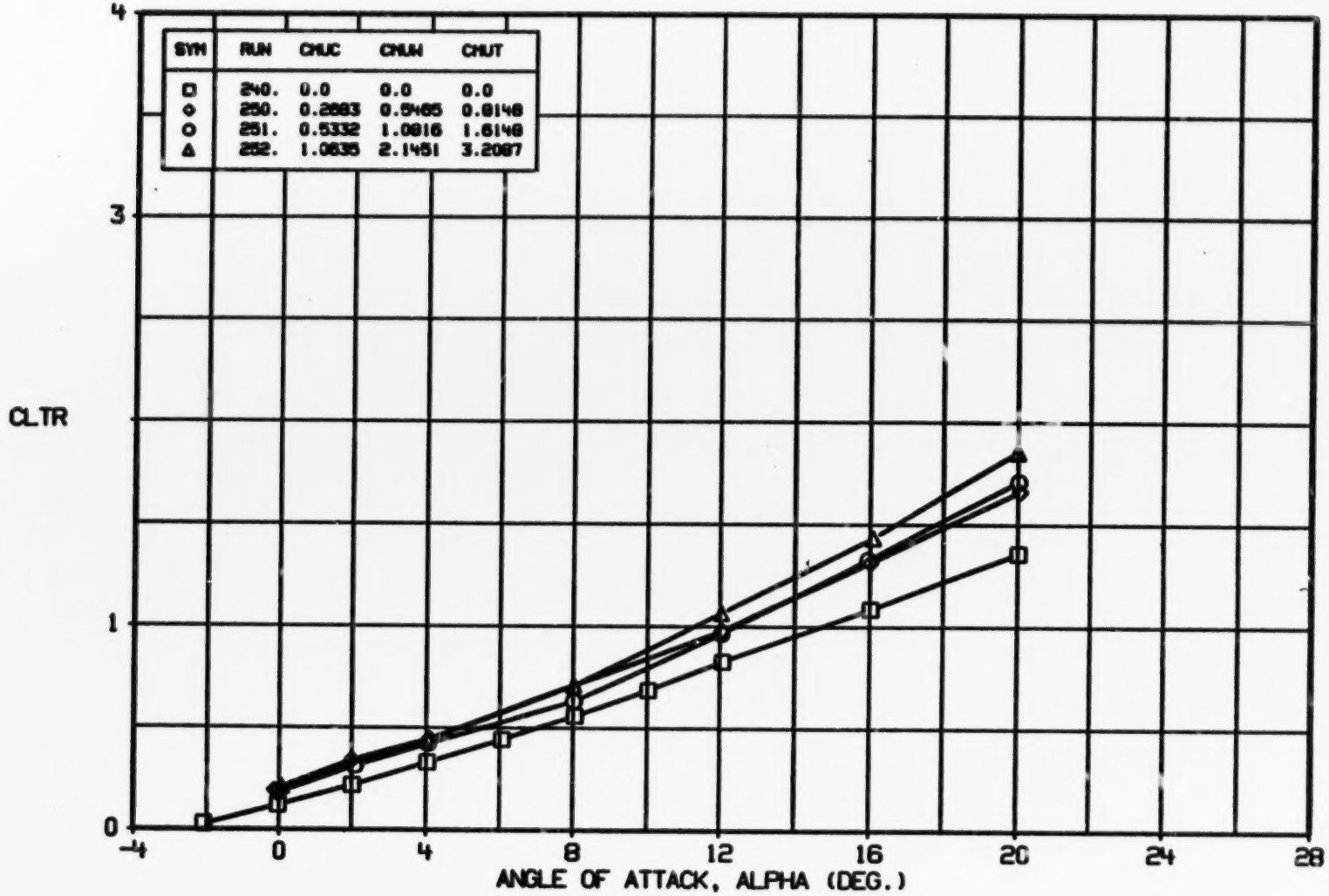

FIGURE A16d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V,(IBW/B)C=1,(BW/B)W=0.5, DELF=0

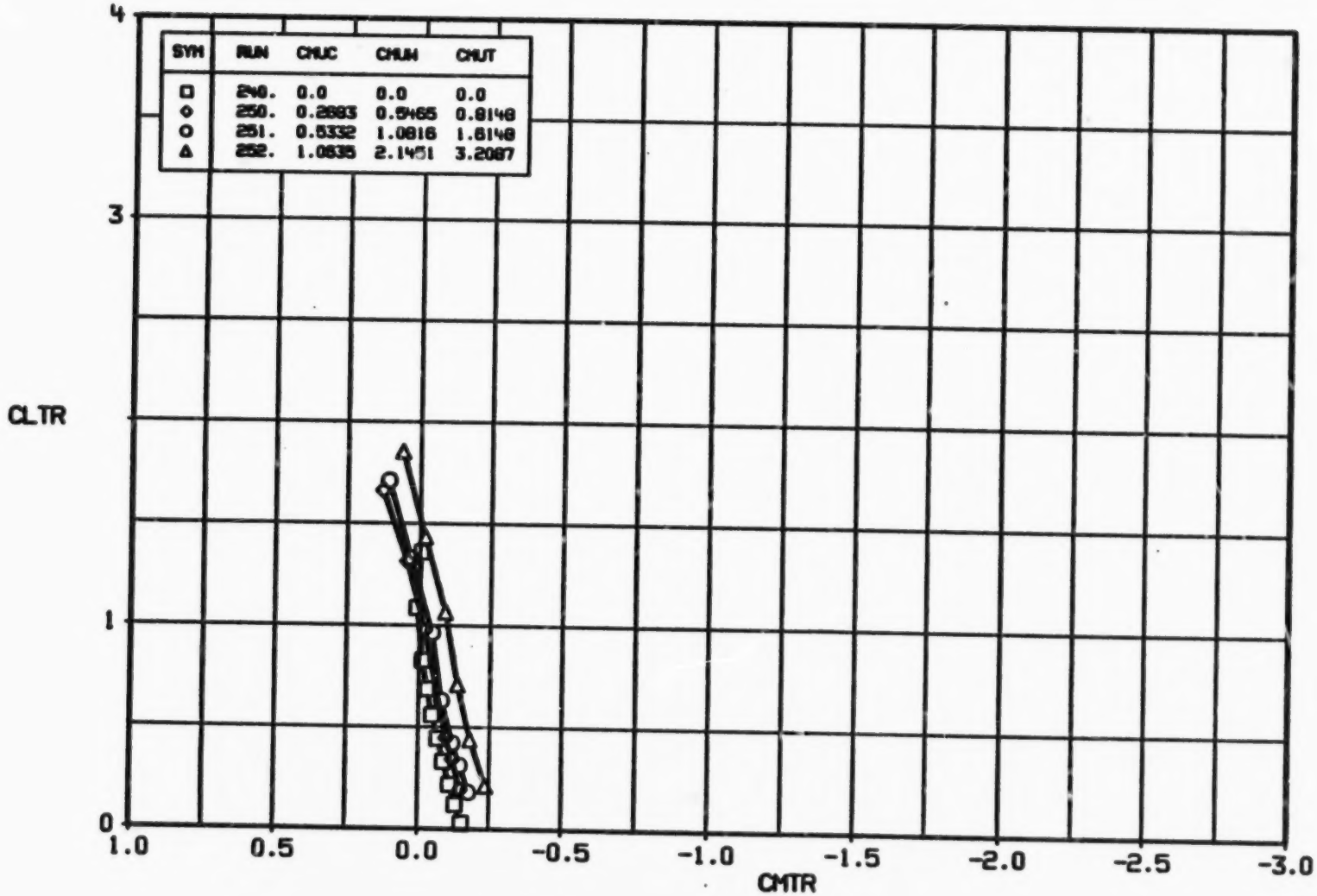

FIGURE A16e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V,(BN/B)C=1,(BN/B)W=0.5, DELF=0

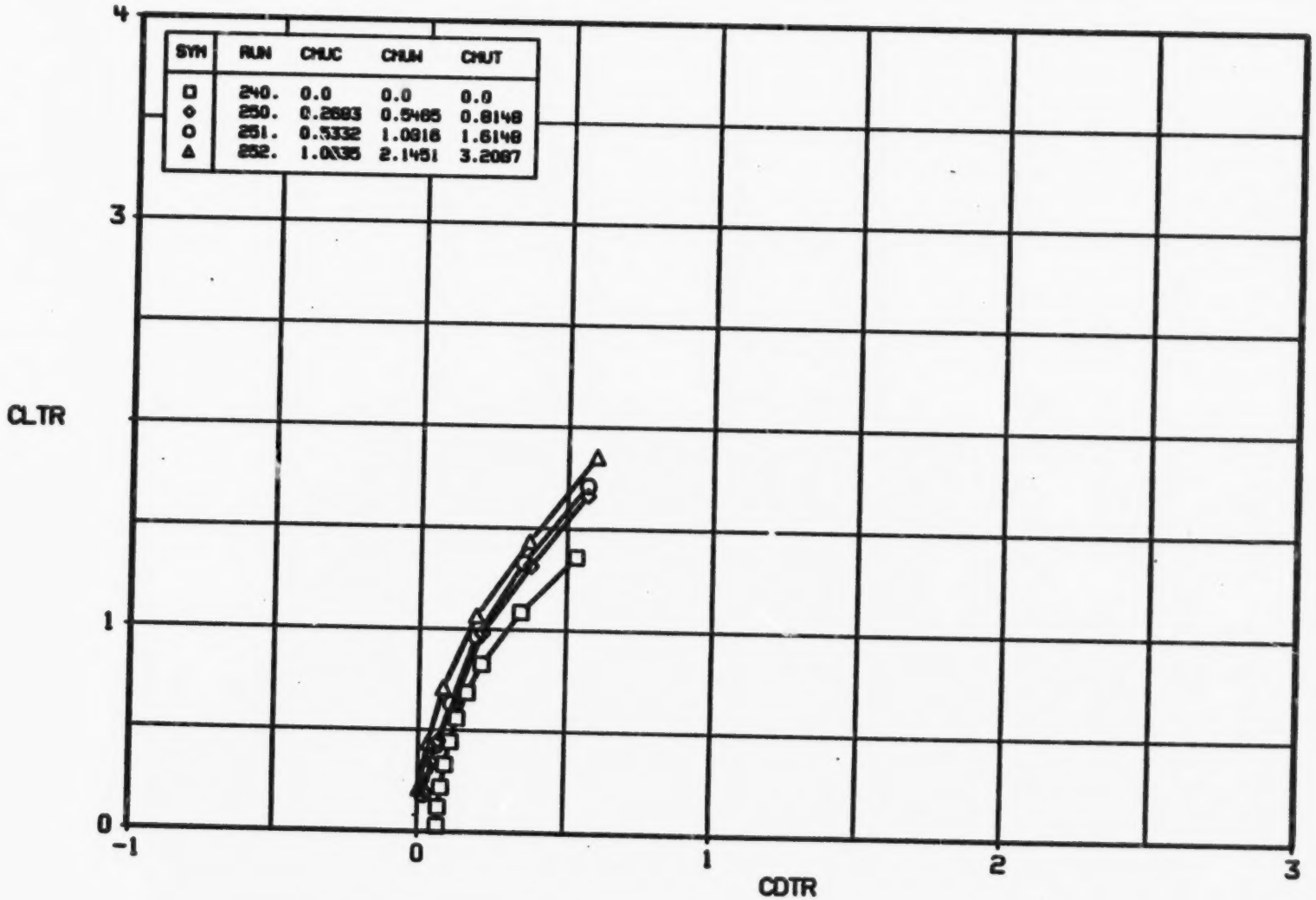

FIGURE A16f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W5V, (BN/B)C=1, (BN/B)W=0.5, DELF=0

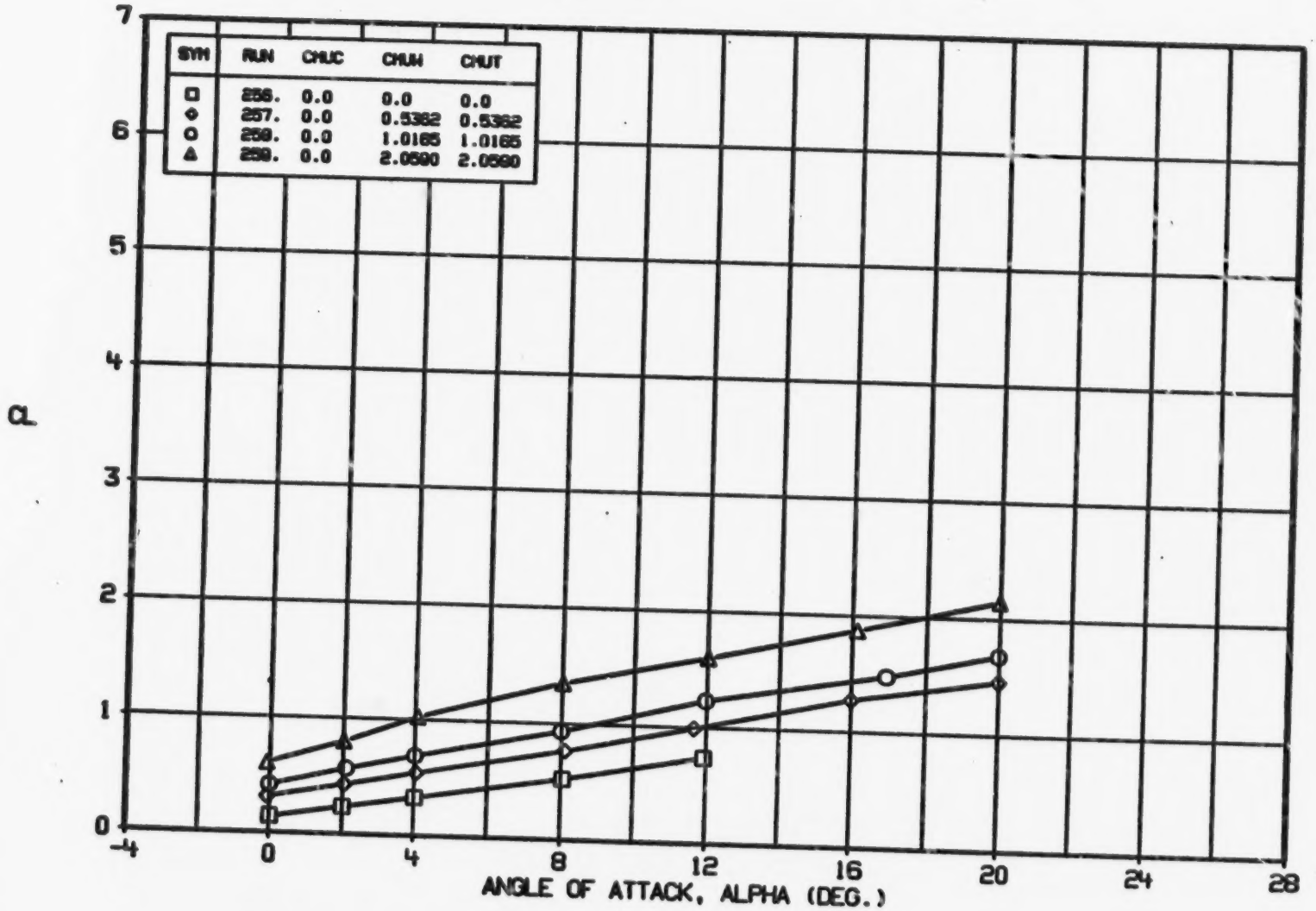

FIGURE A17a BASIC DATA EFFECT OF CMU
CONFIGURATION BW5V, BW/B=0.5, DELF=0

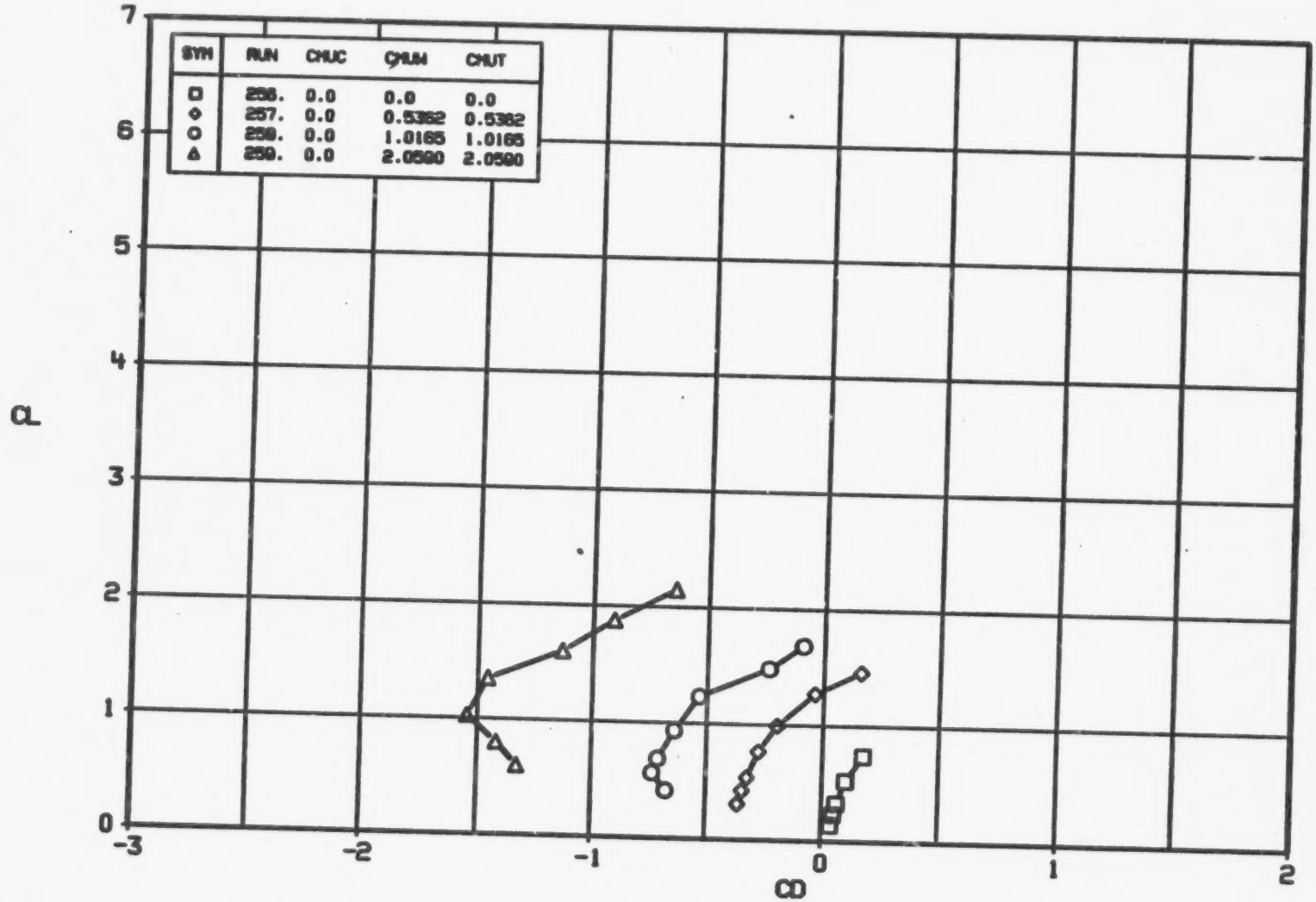

FIGURE A17b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=0

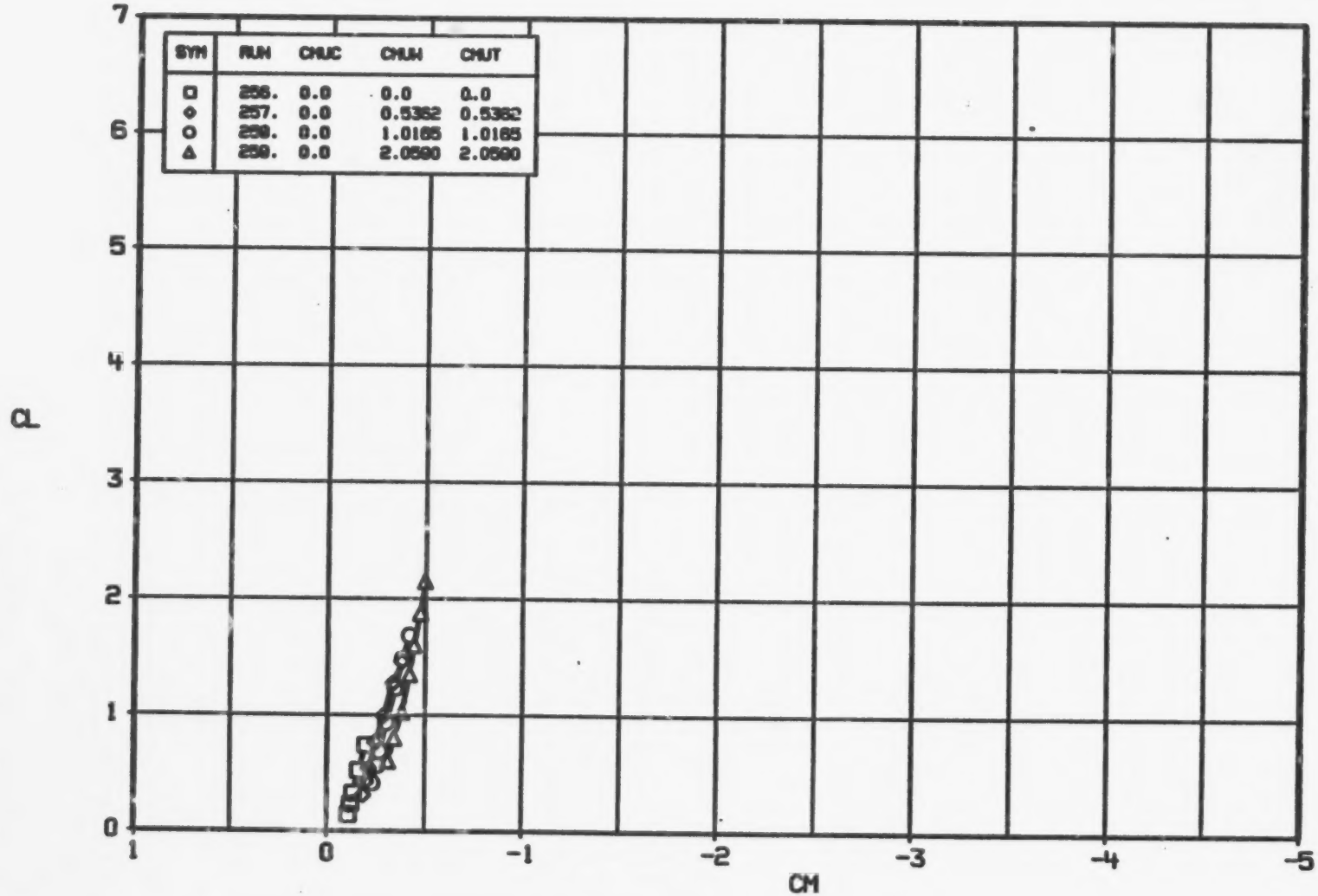

FIGURE A17c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 , DELF=0

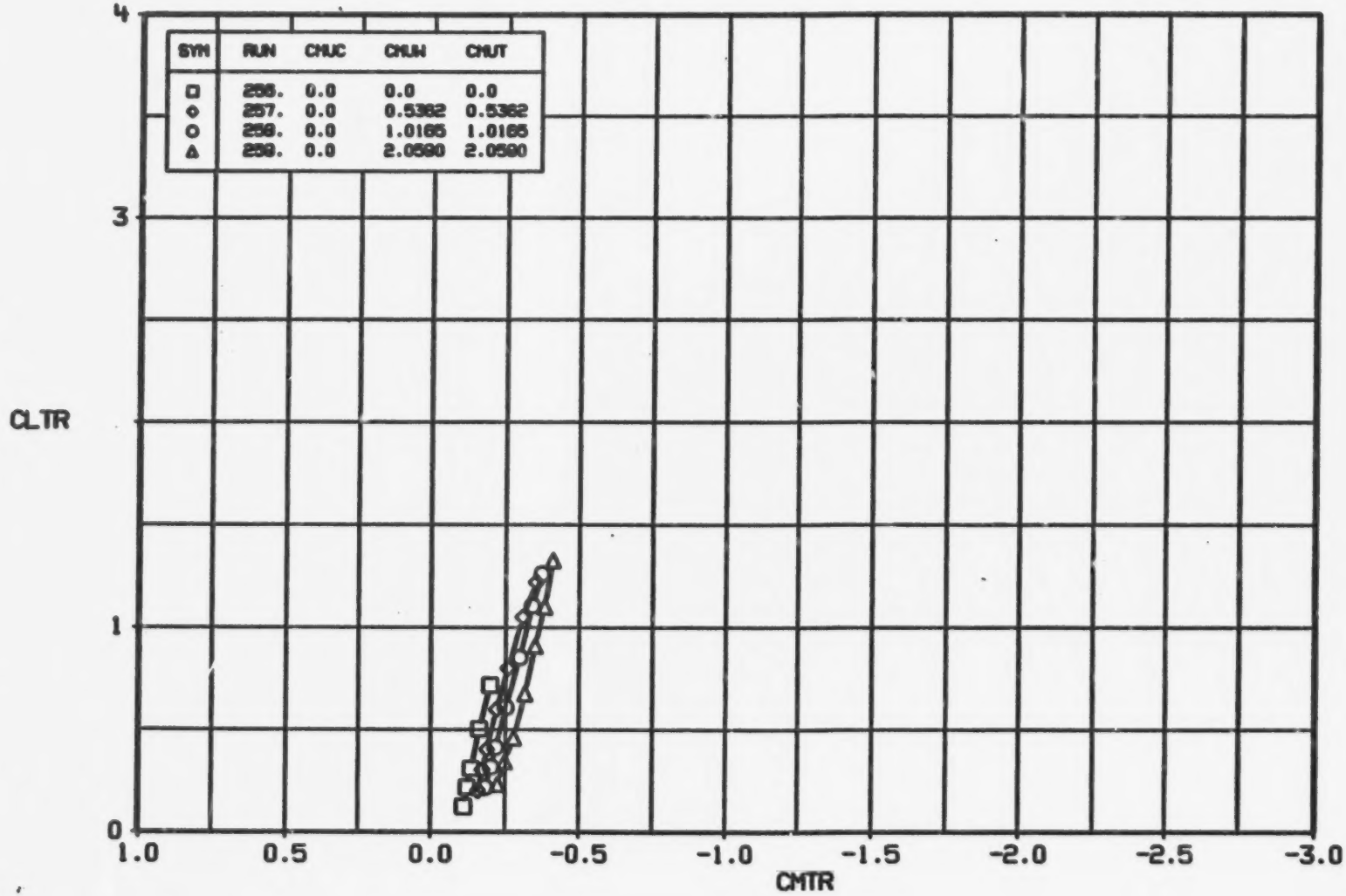

FIGURE A17d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=0

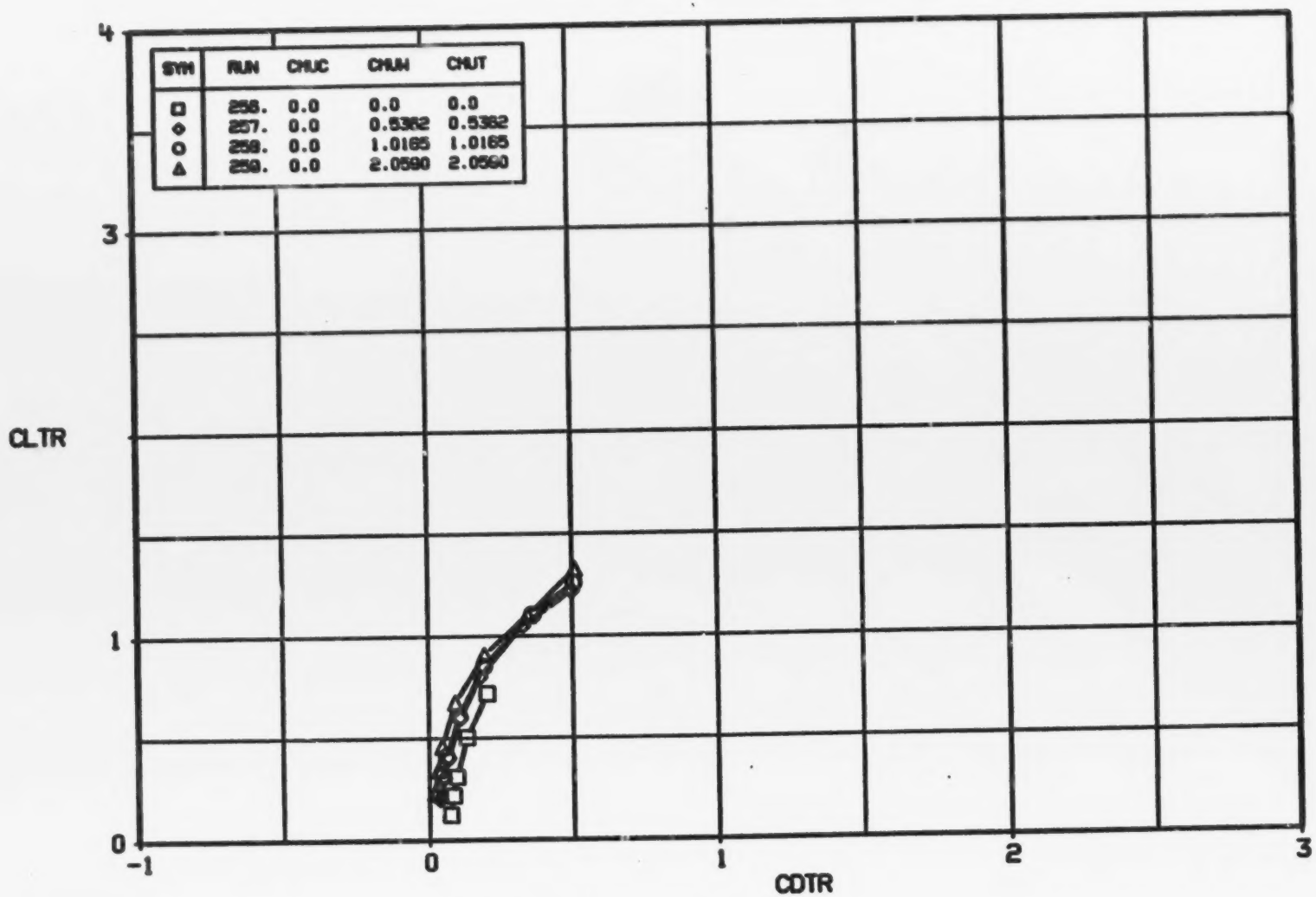

FIGURE A17e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 , DELF=0

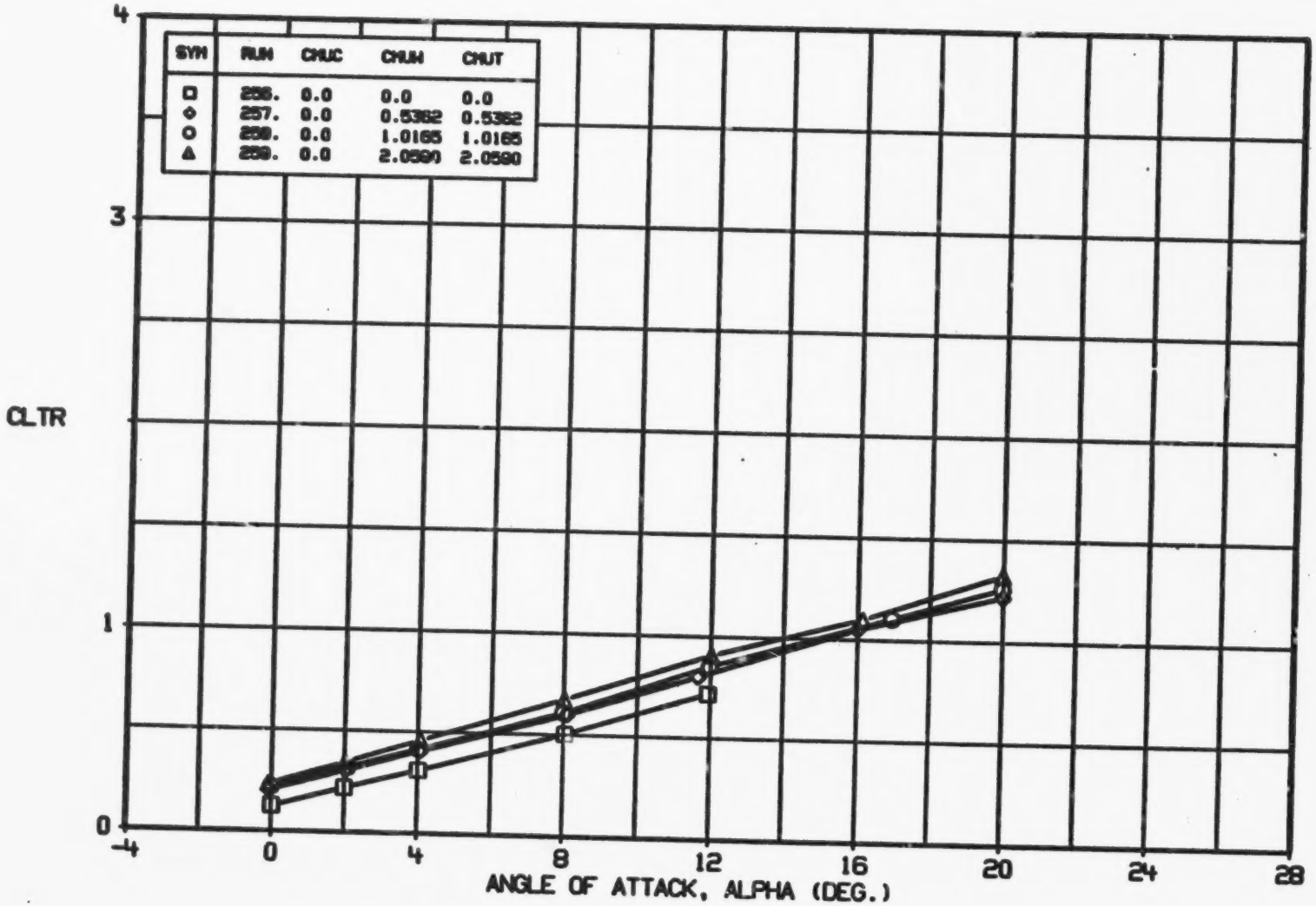

FIGURE A17f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=0

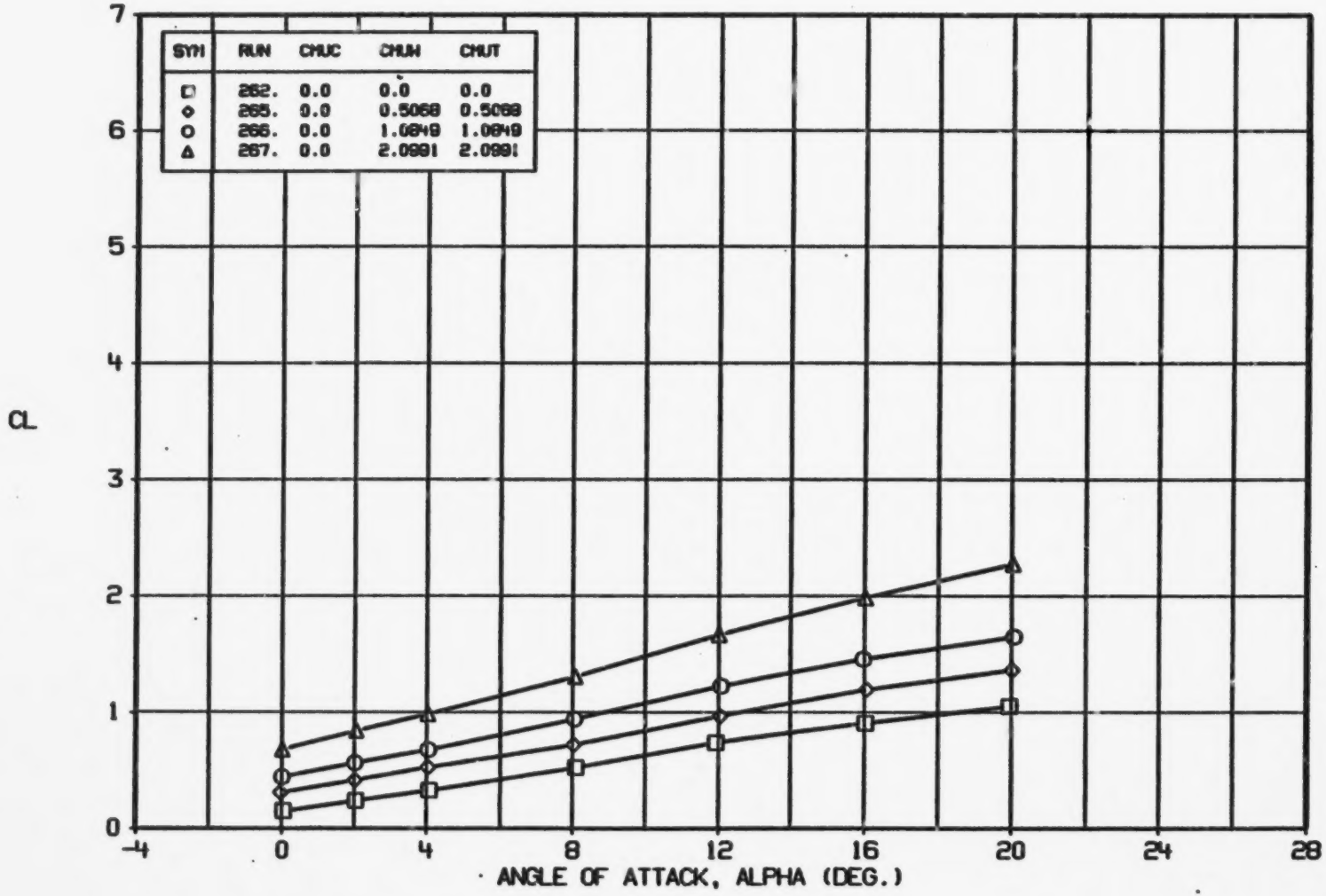

FIGURE A18a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V; BN/B=0.25 , DELF=0

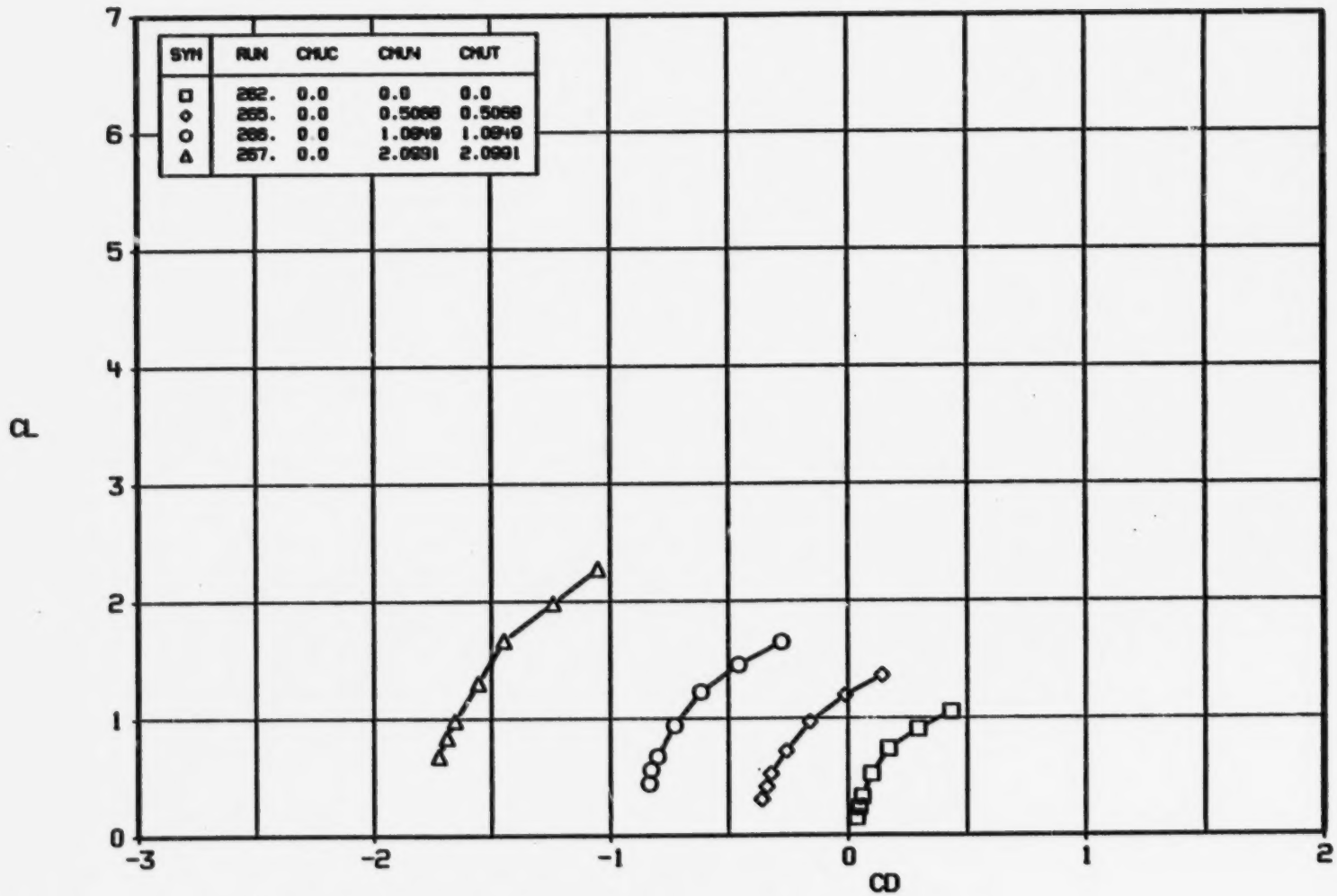

FIGURE A18b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.25, DELF=0

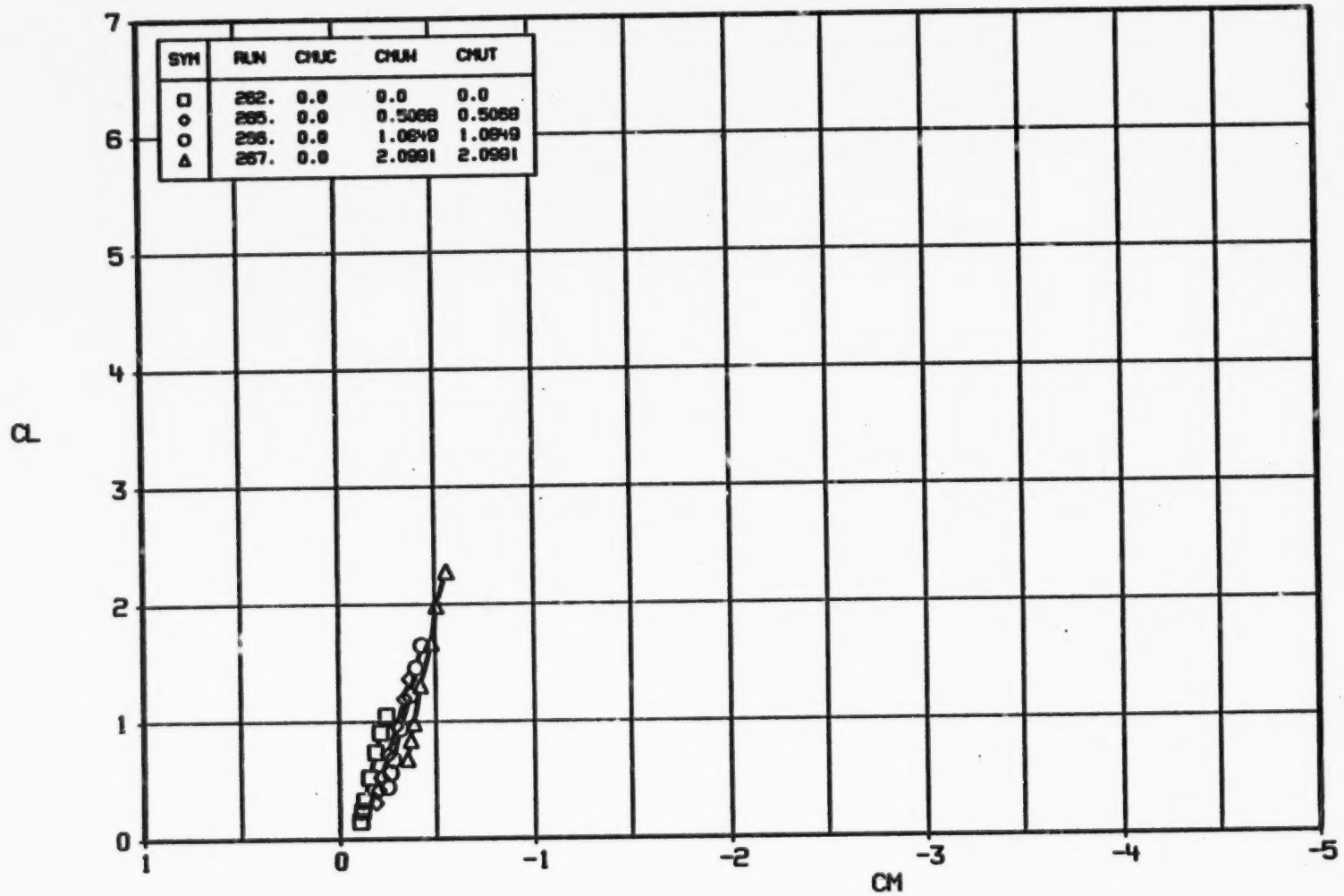

FIGURE A18c BASIC DATA EFFECT OF CMU
CONFIGURATION BW5V, BN/B=0.25, DELF=0

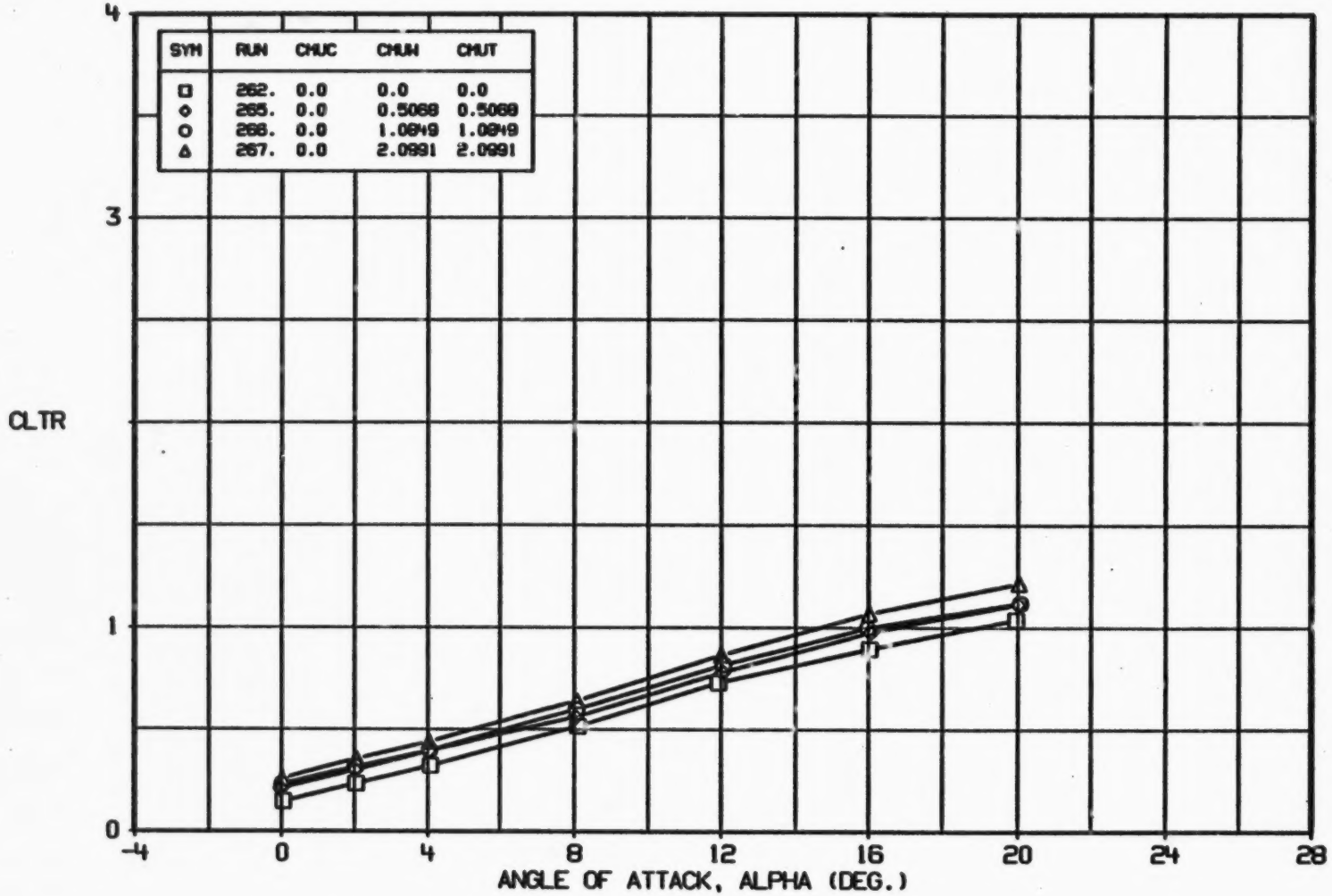

FIGURE A18d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.25, DELF=0

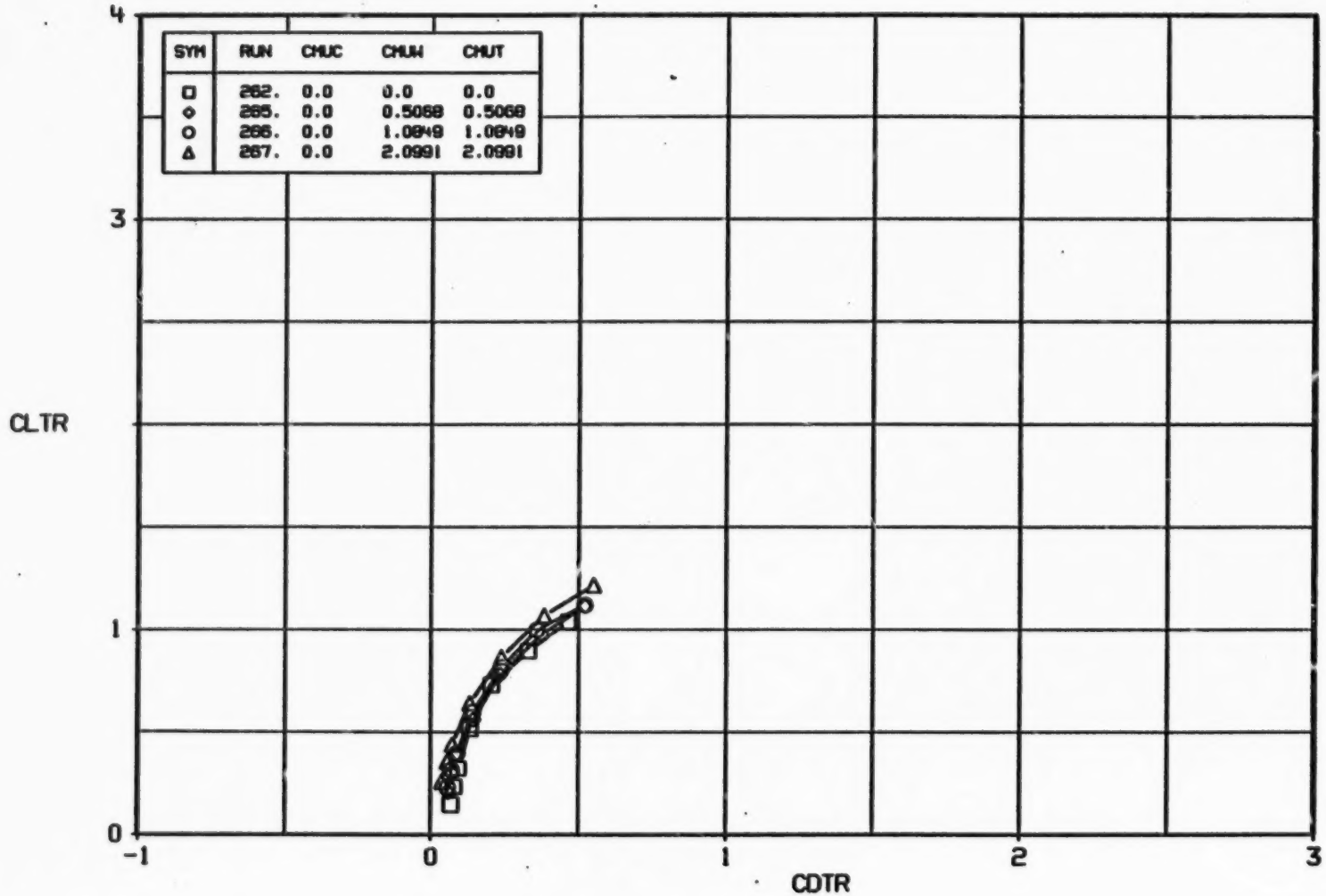

FIGURE A18e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.25, DELF=0

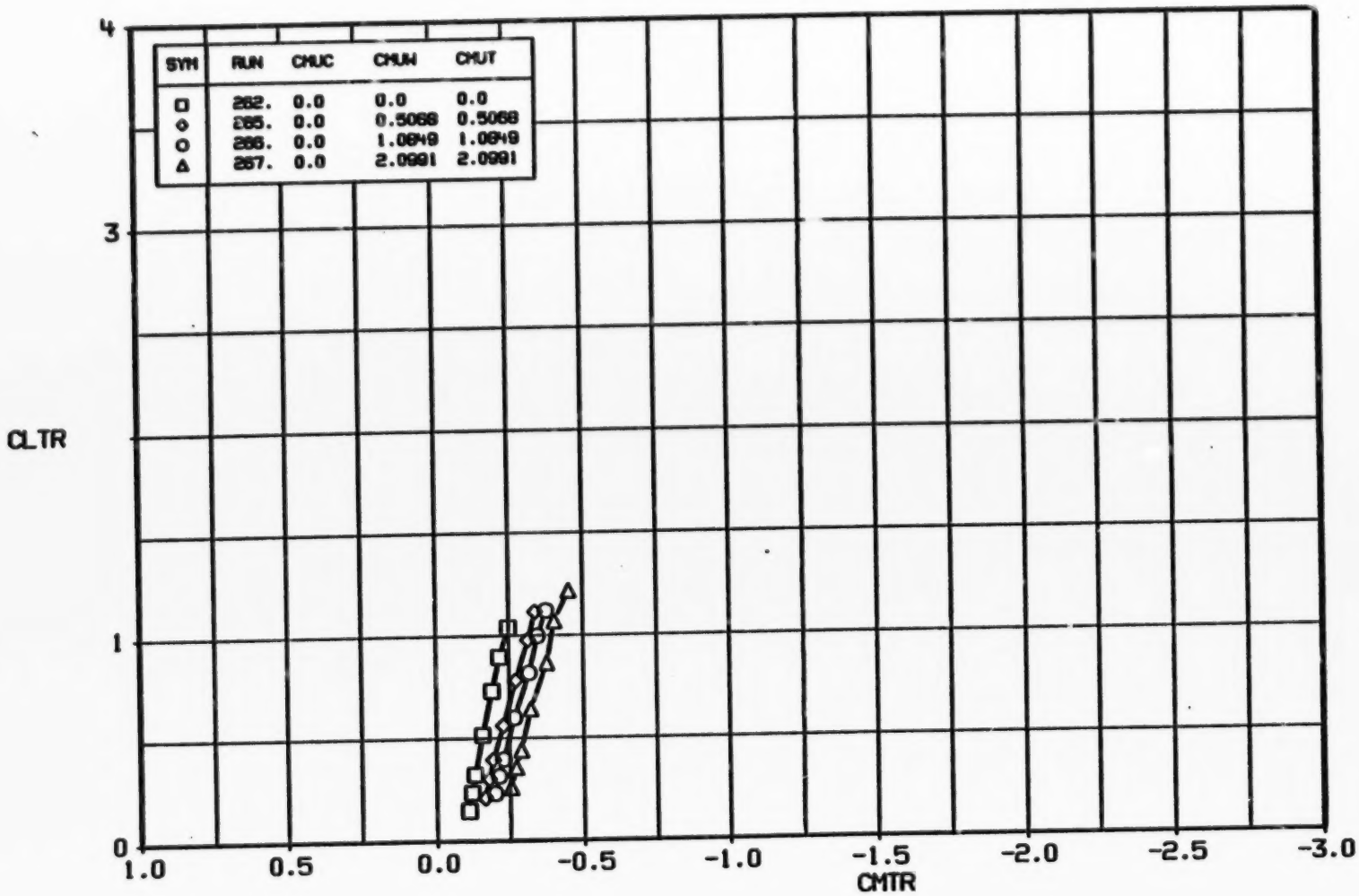

FIGURE A18f BASIC DATA EFFECT OF CMU
CONFIGURATION BN6V, BN/B=0.25, DELF=0

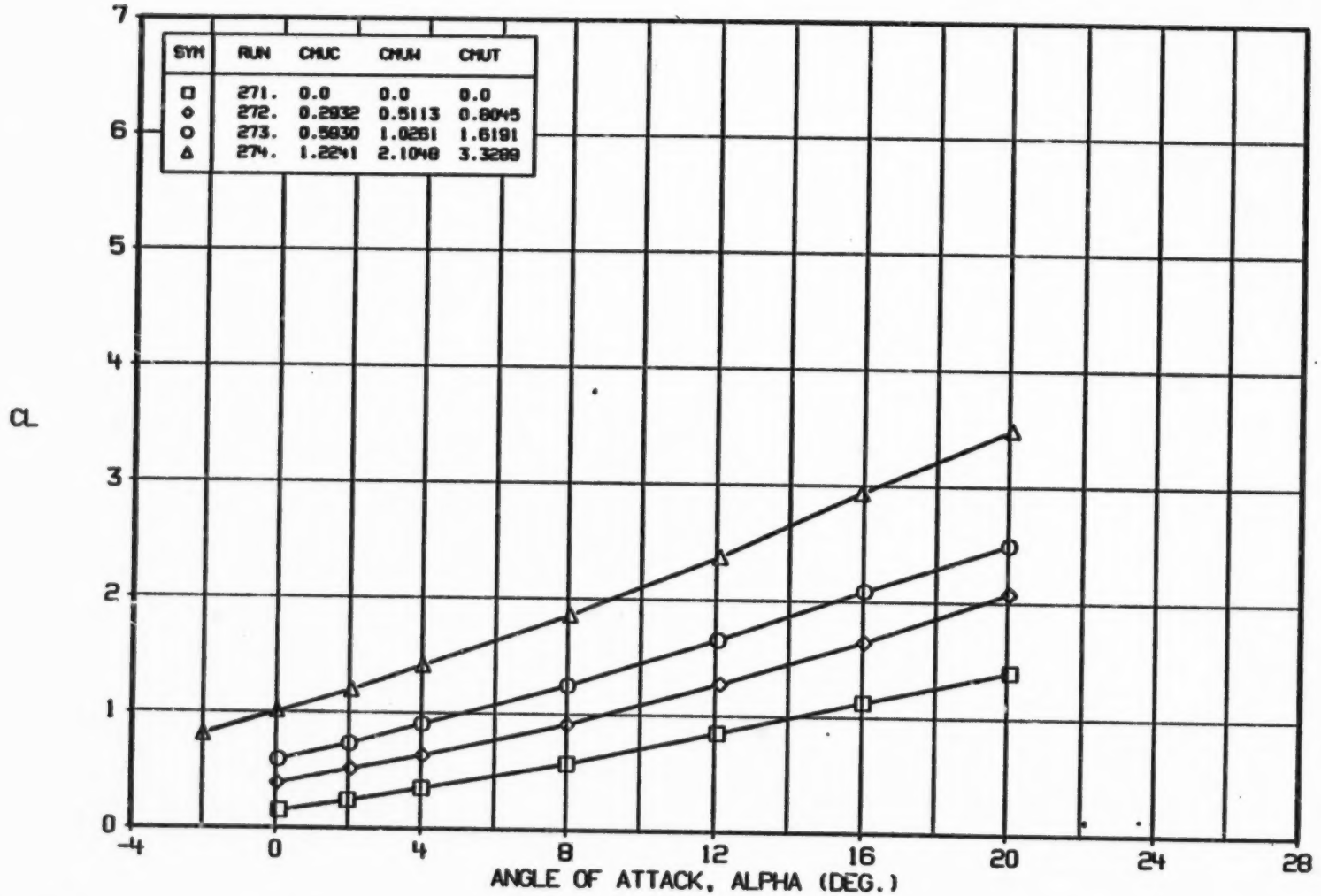

FIGURE A19a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V,(BN/B)C=1,(BN/B)W=0.25. DELF=0

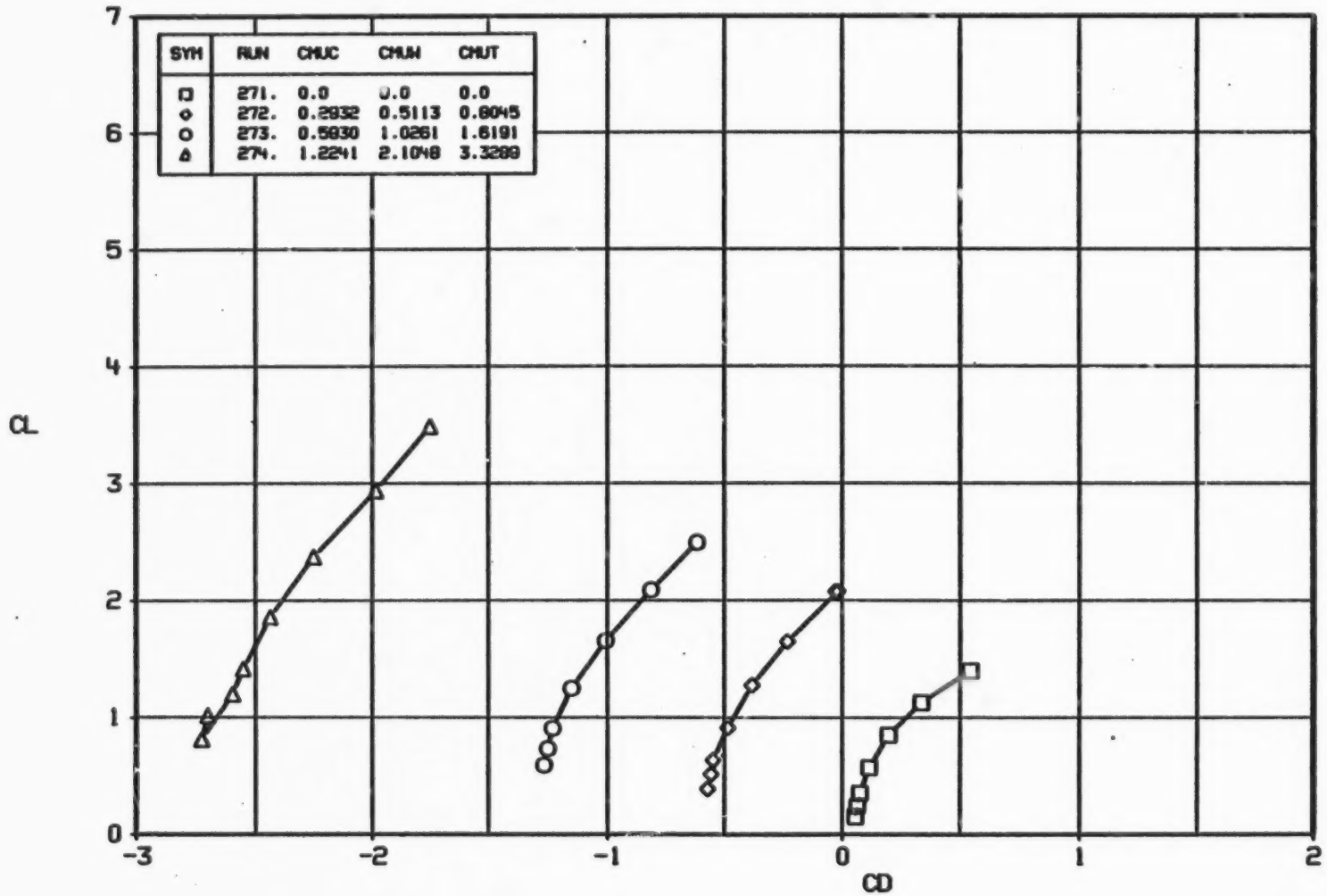

FIGURE A19b BASIC DATA EFFECT OF CMU
CONFIGURATION BCIWSV,(BN/B)C=1,(BN/B)W=0.25, DELF=0

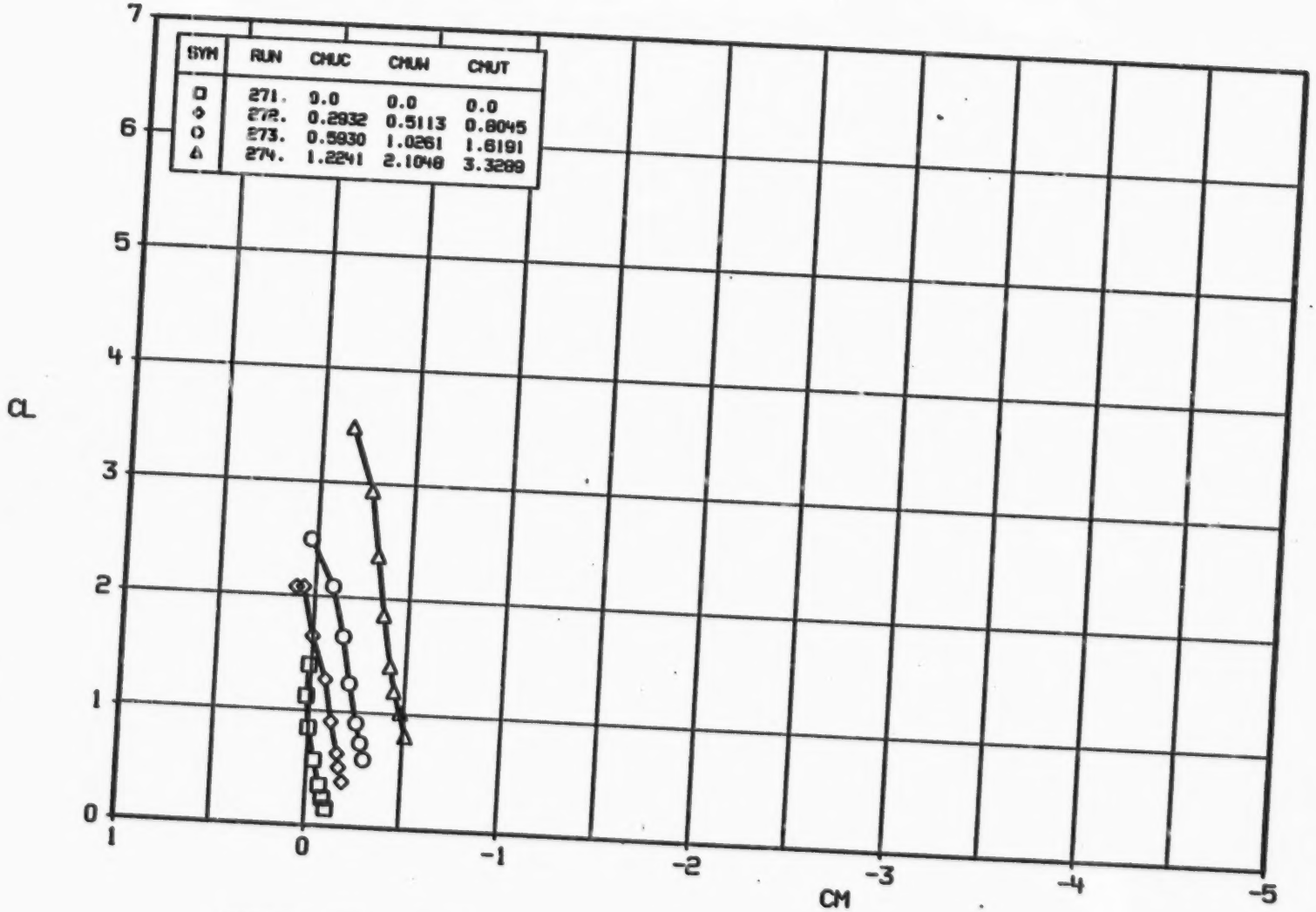

FIGURE A19c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

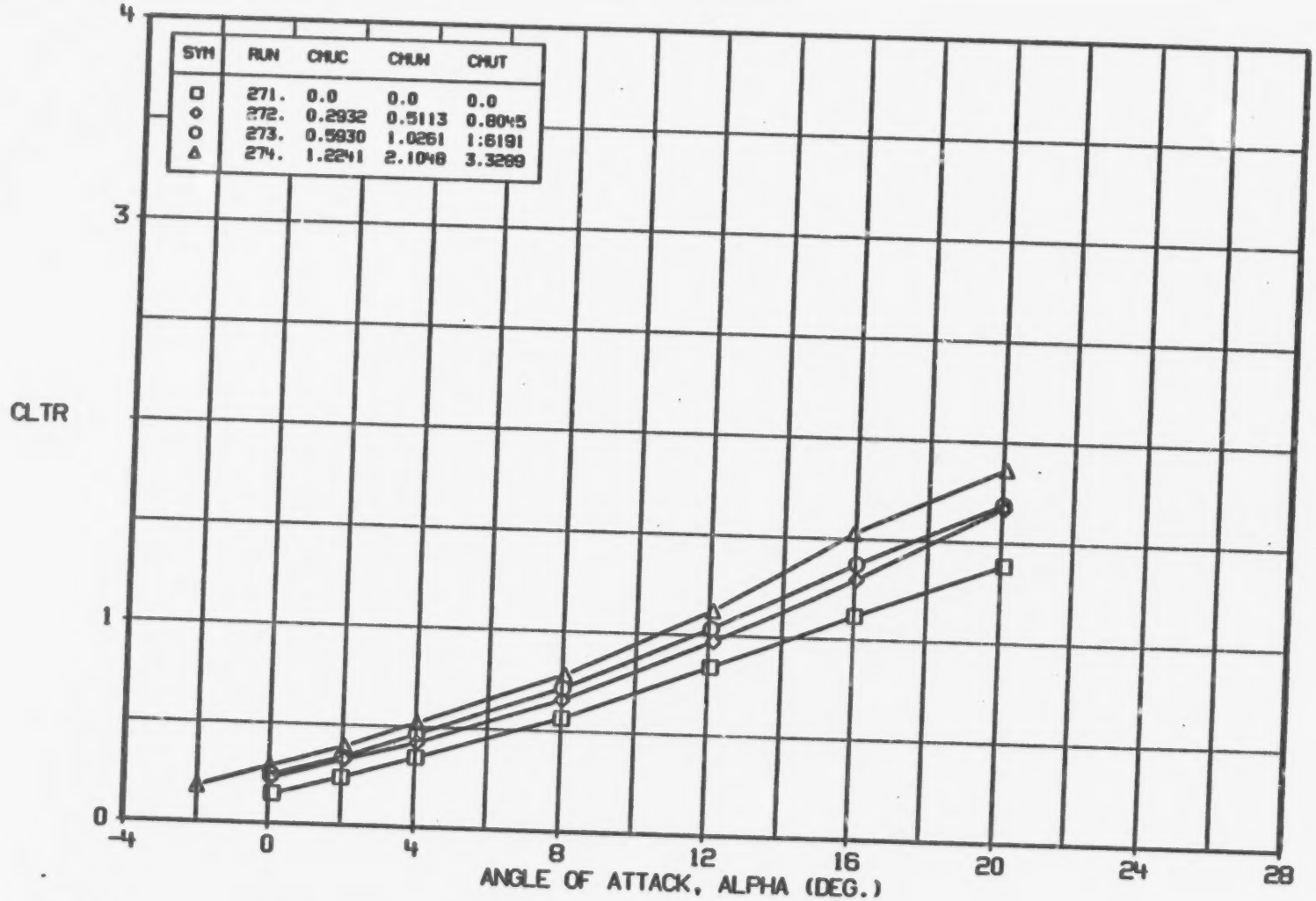

FIGURE A19d BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

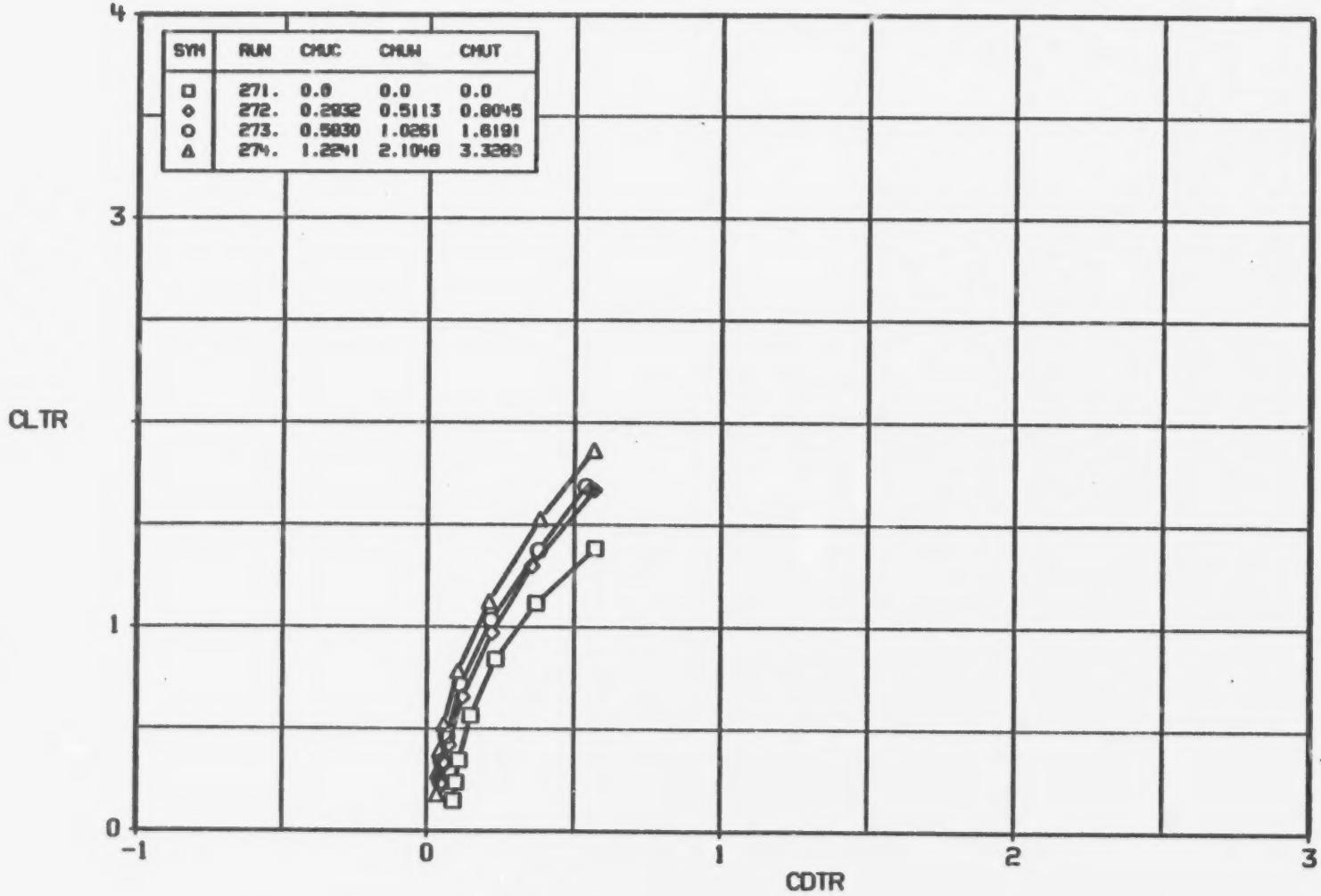

FIGURE A19e BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V,(BN/B)C=1,(BN/B)W=0.25, DELF=0

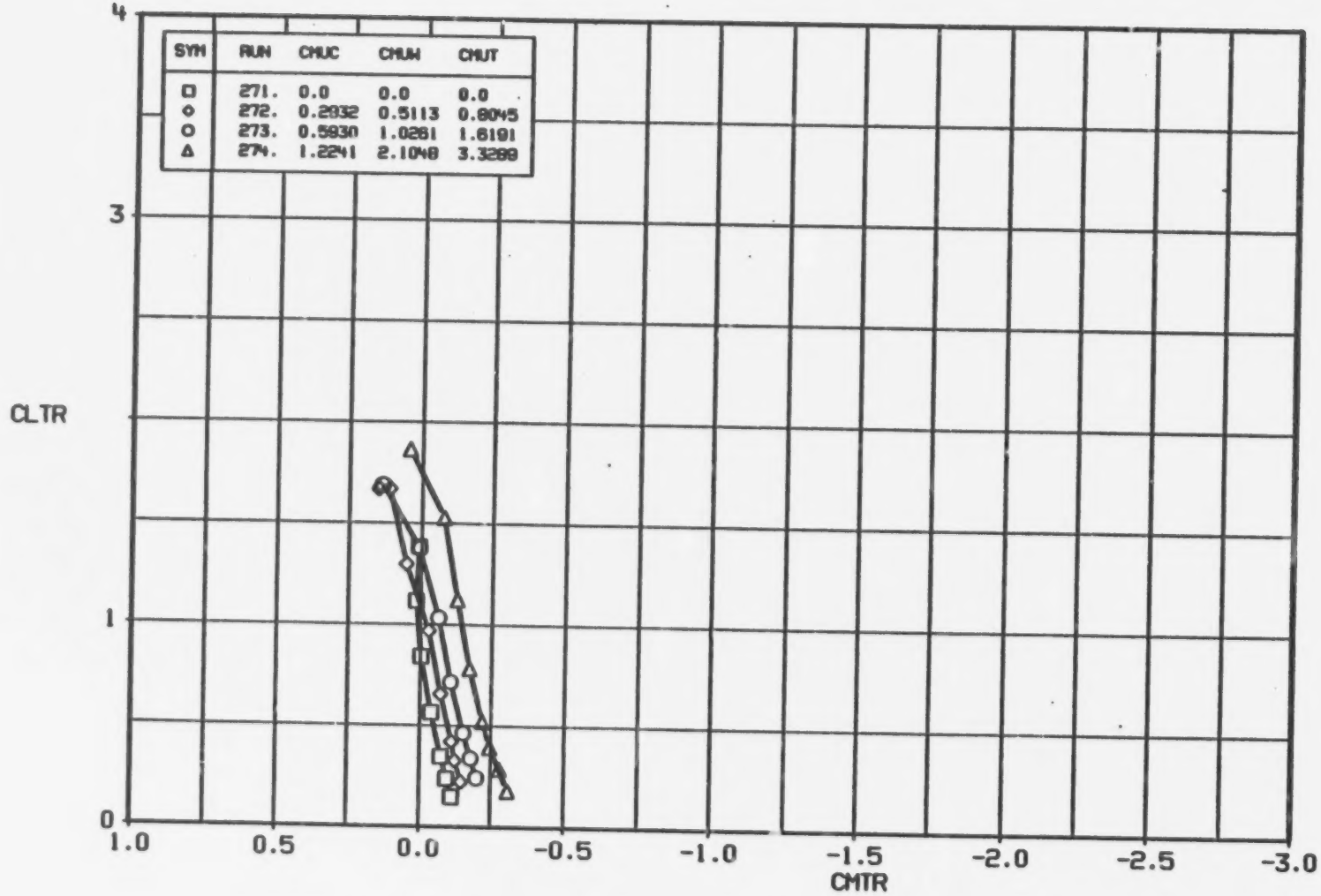

FIGURE A19f BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, (BN/B)C=1, (BN/B)W=0.25, DELF=0

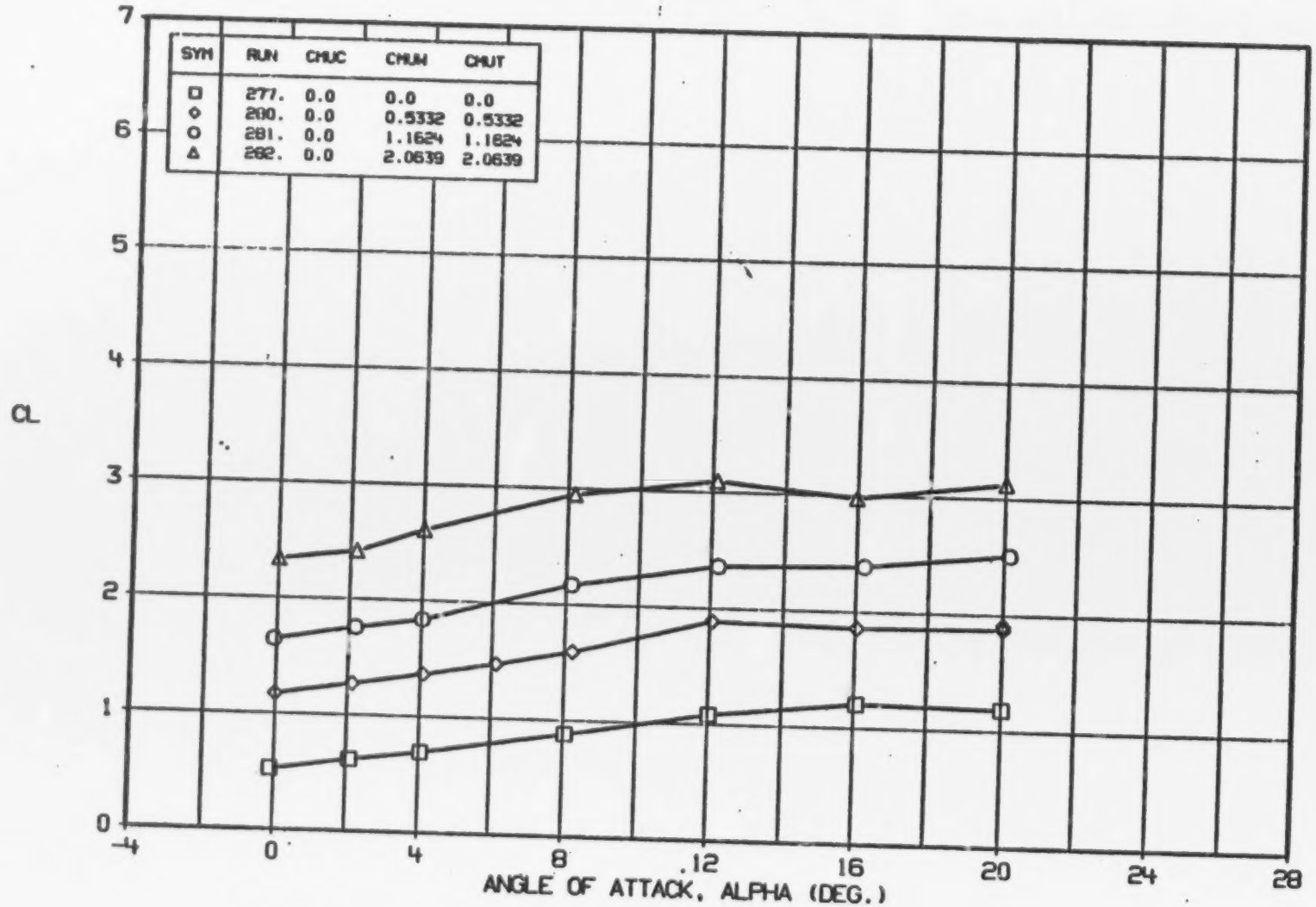

FIGURE A20a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

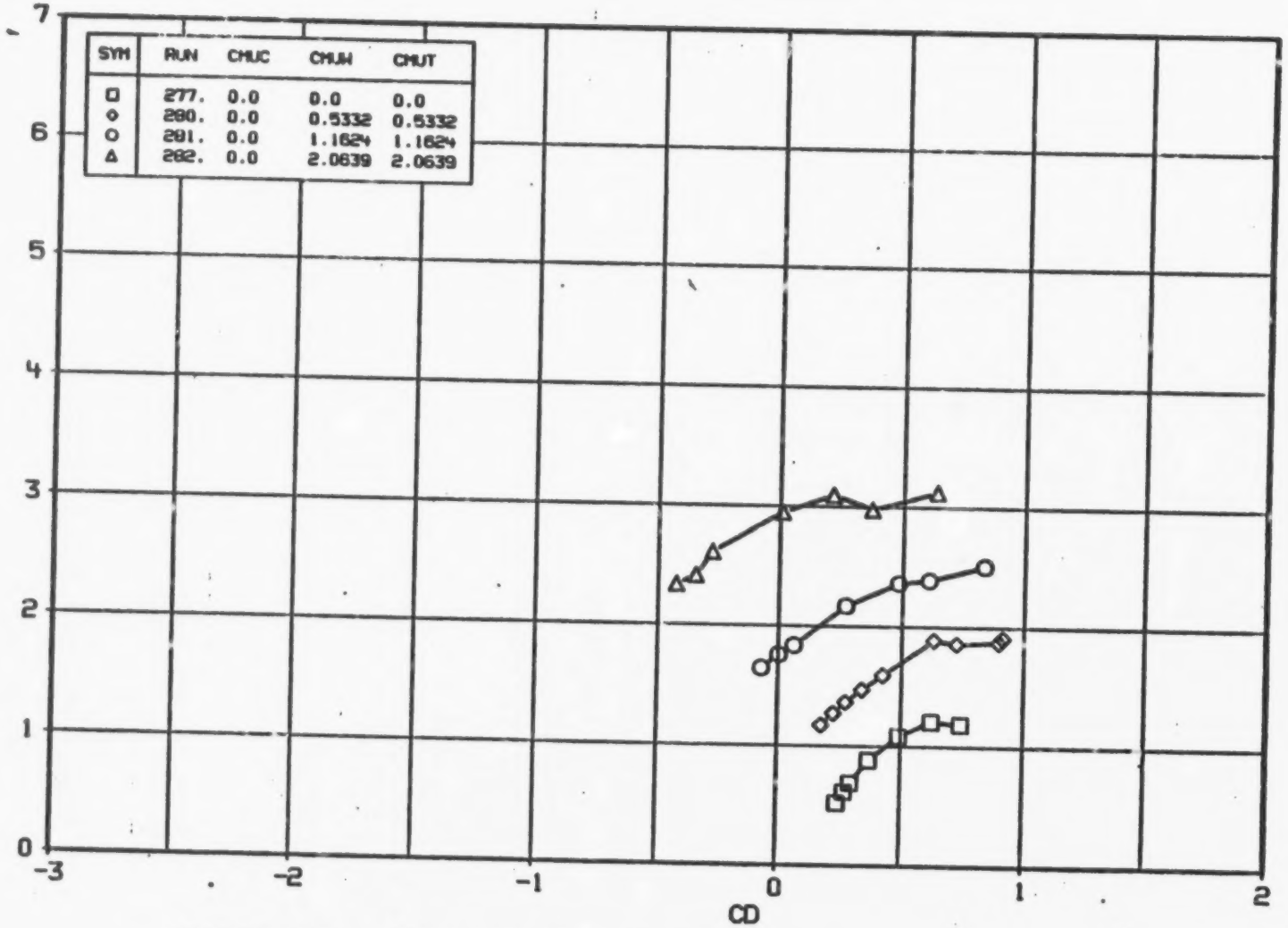

FIGURE A20b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

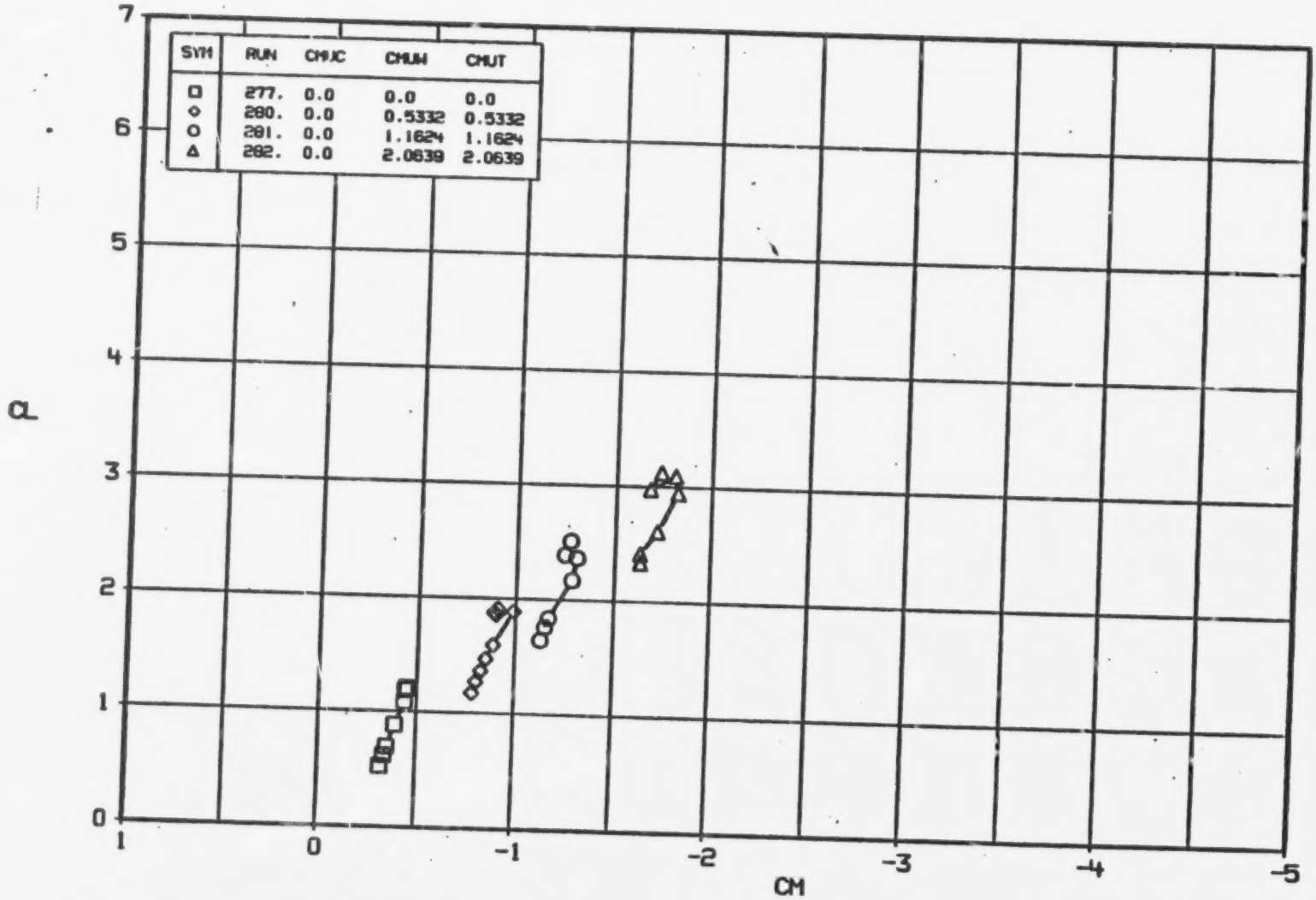

FIGURE A20c BASIC DATA EFFECT OF CMU
CONFIGURATION BN6V, DELF=45,BN/B=.25

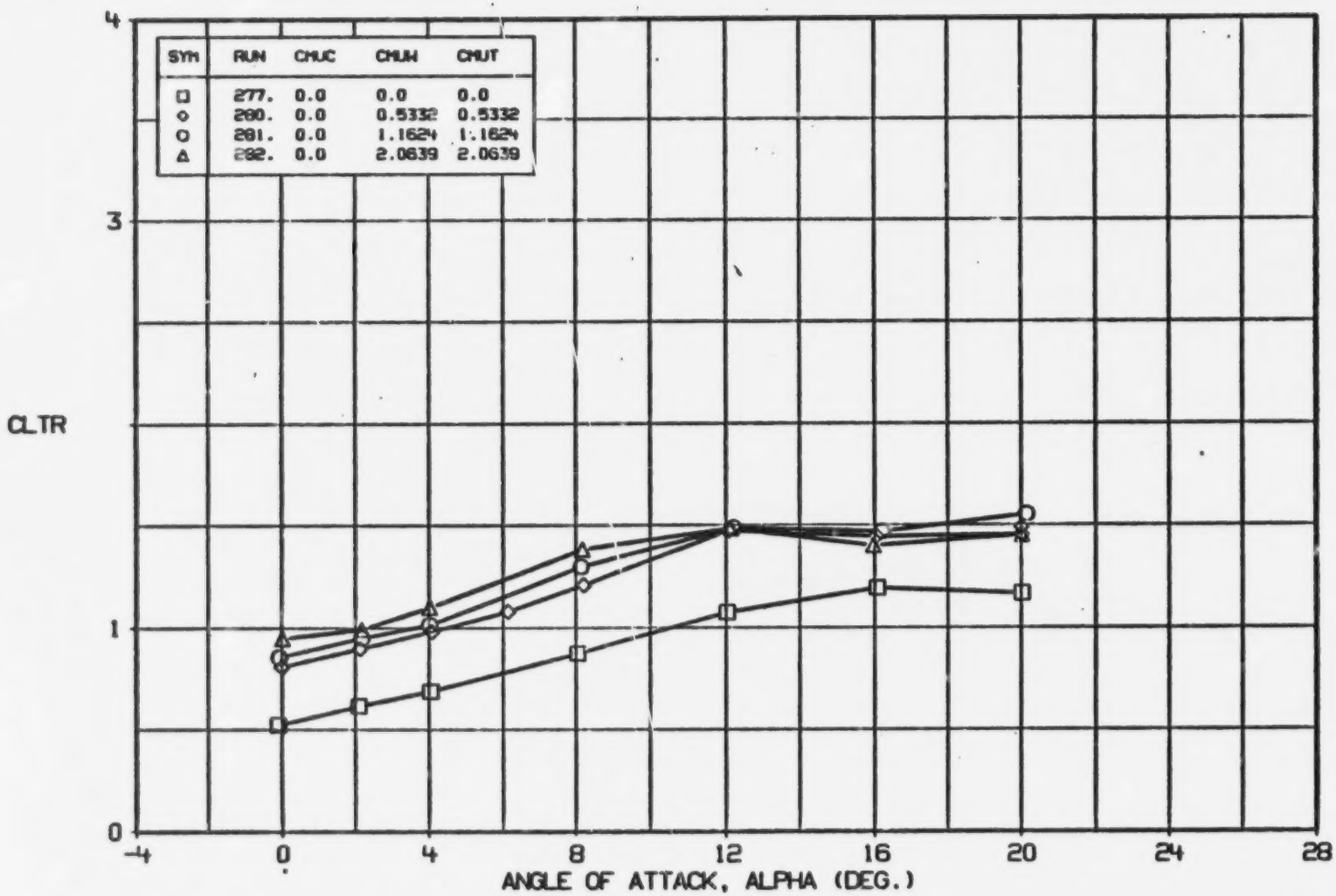

FIGURE A20d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

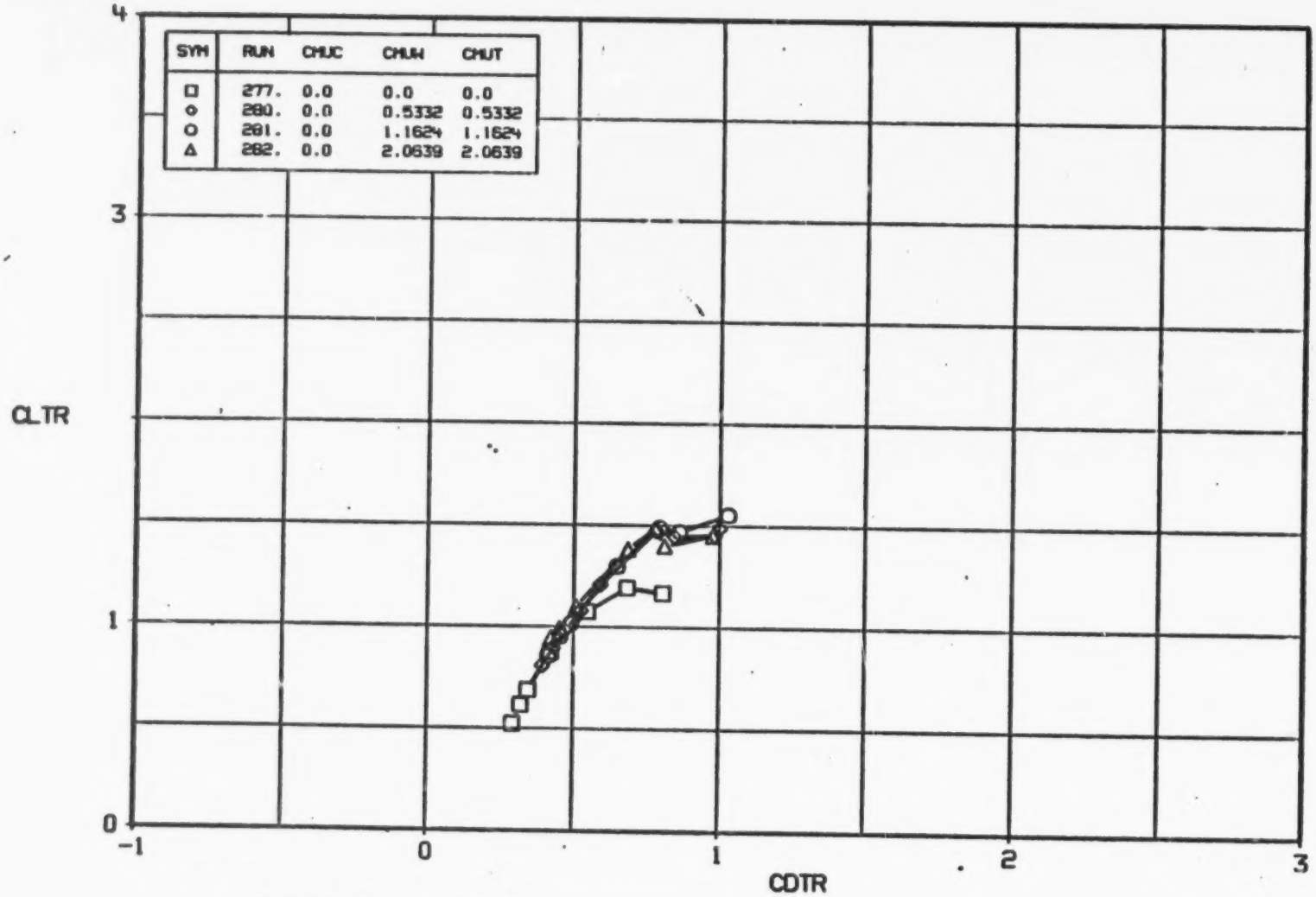

FIGURE A20e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

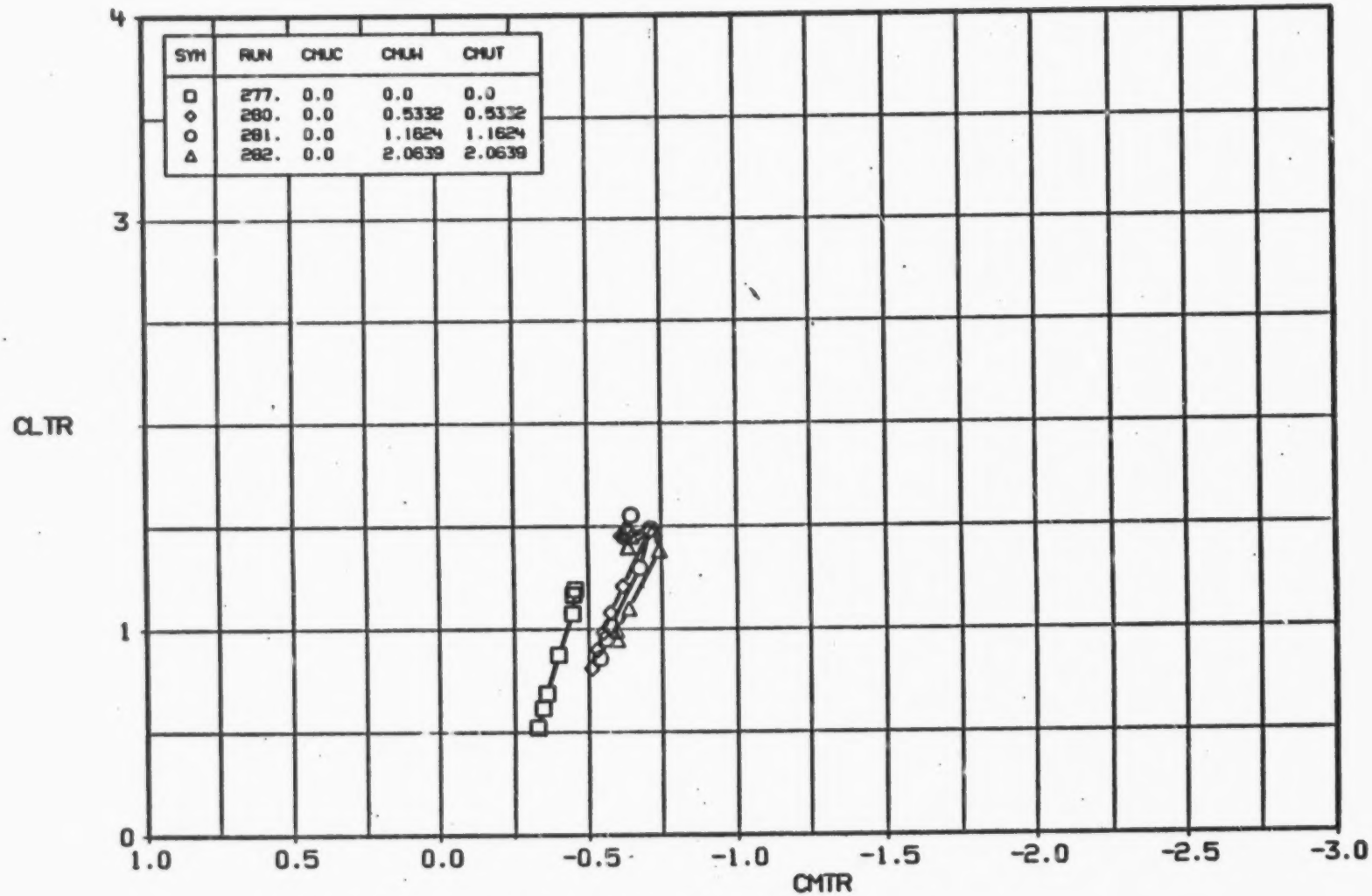

FIGURE A20f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

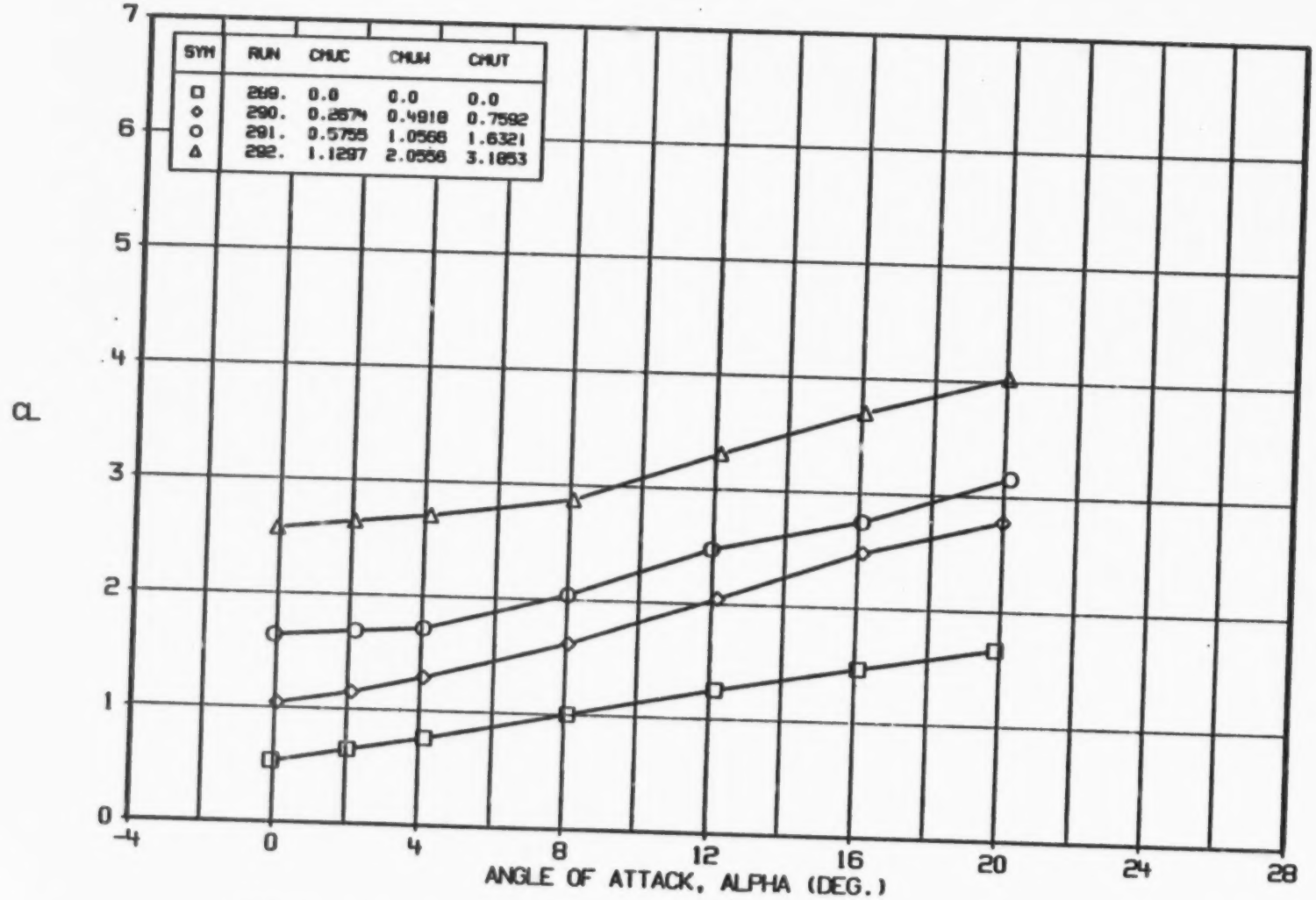

FIGURE A21a BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

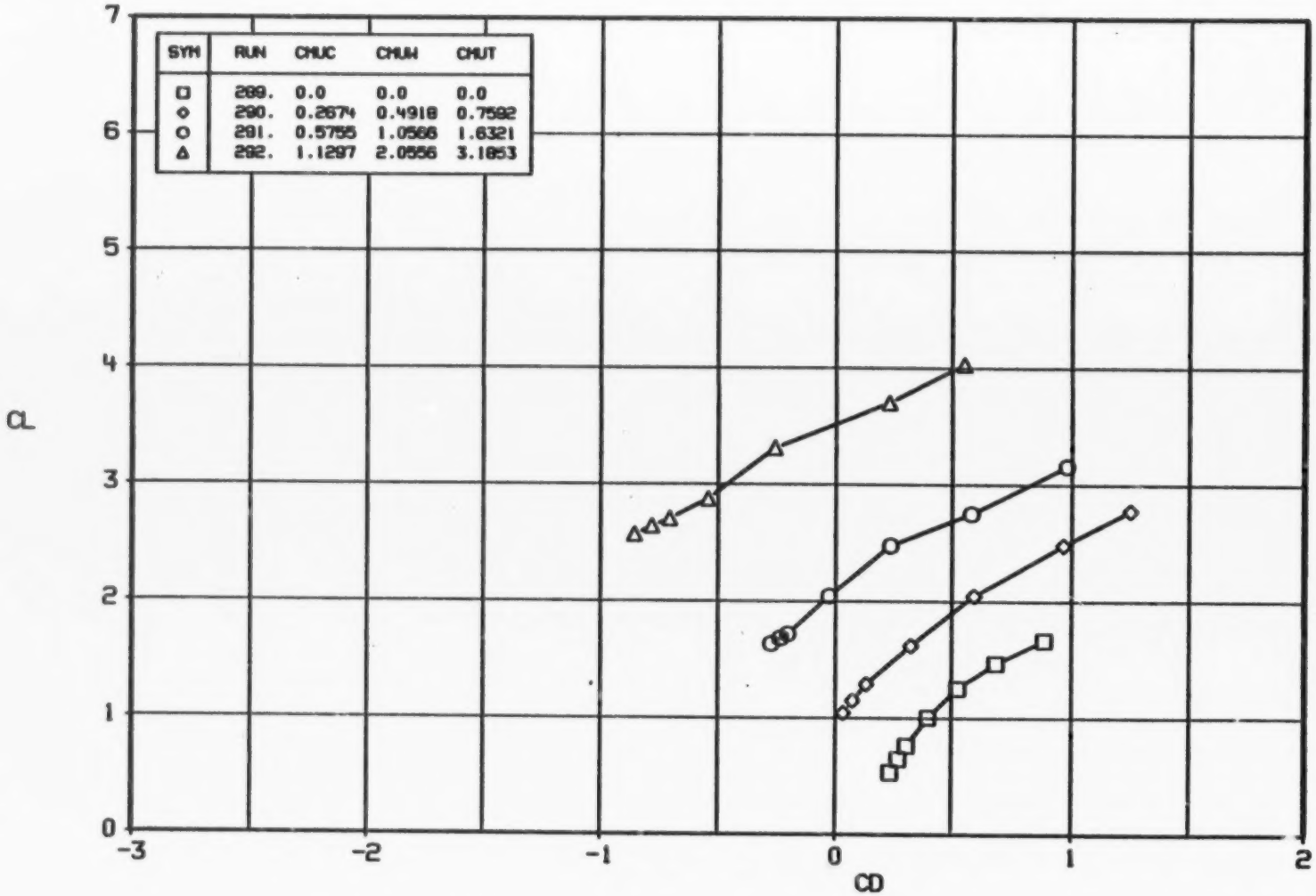

FIGURE A21b BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

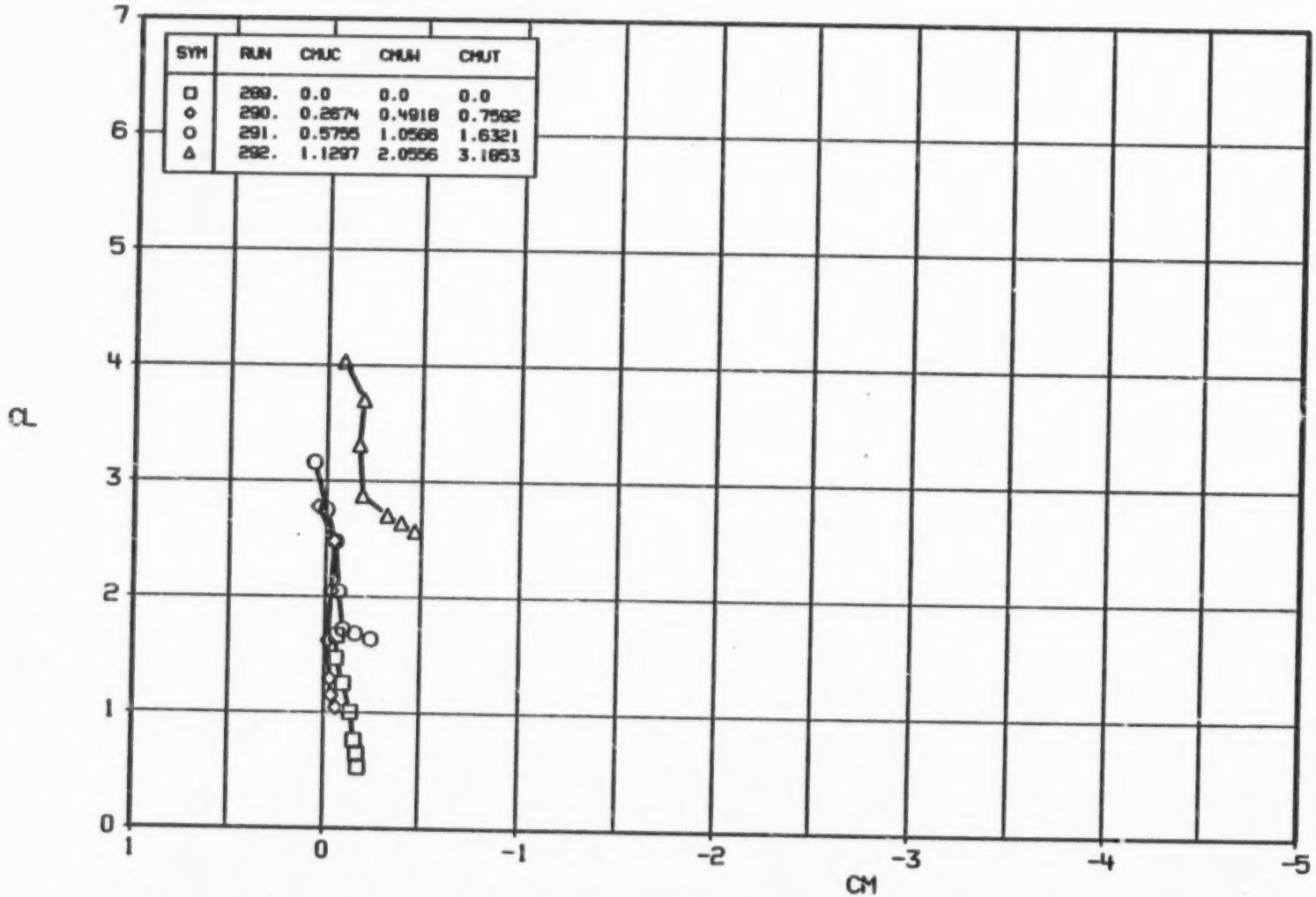

FIGURE A21c BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

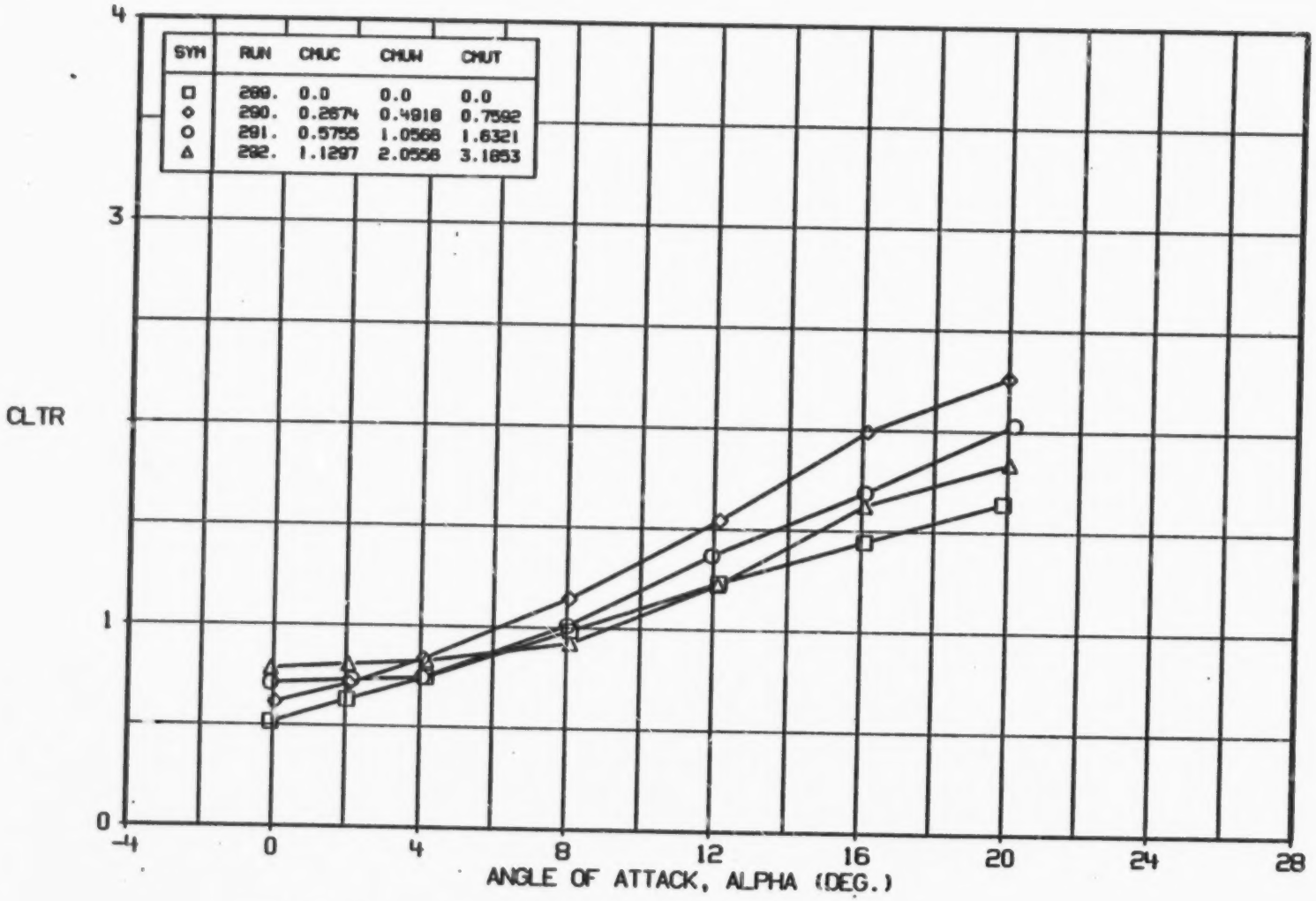

FIGURE A21d BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

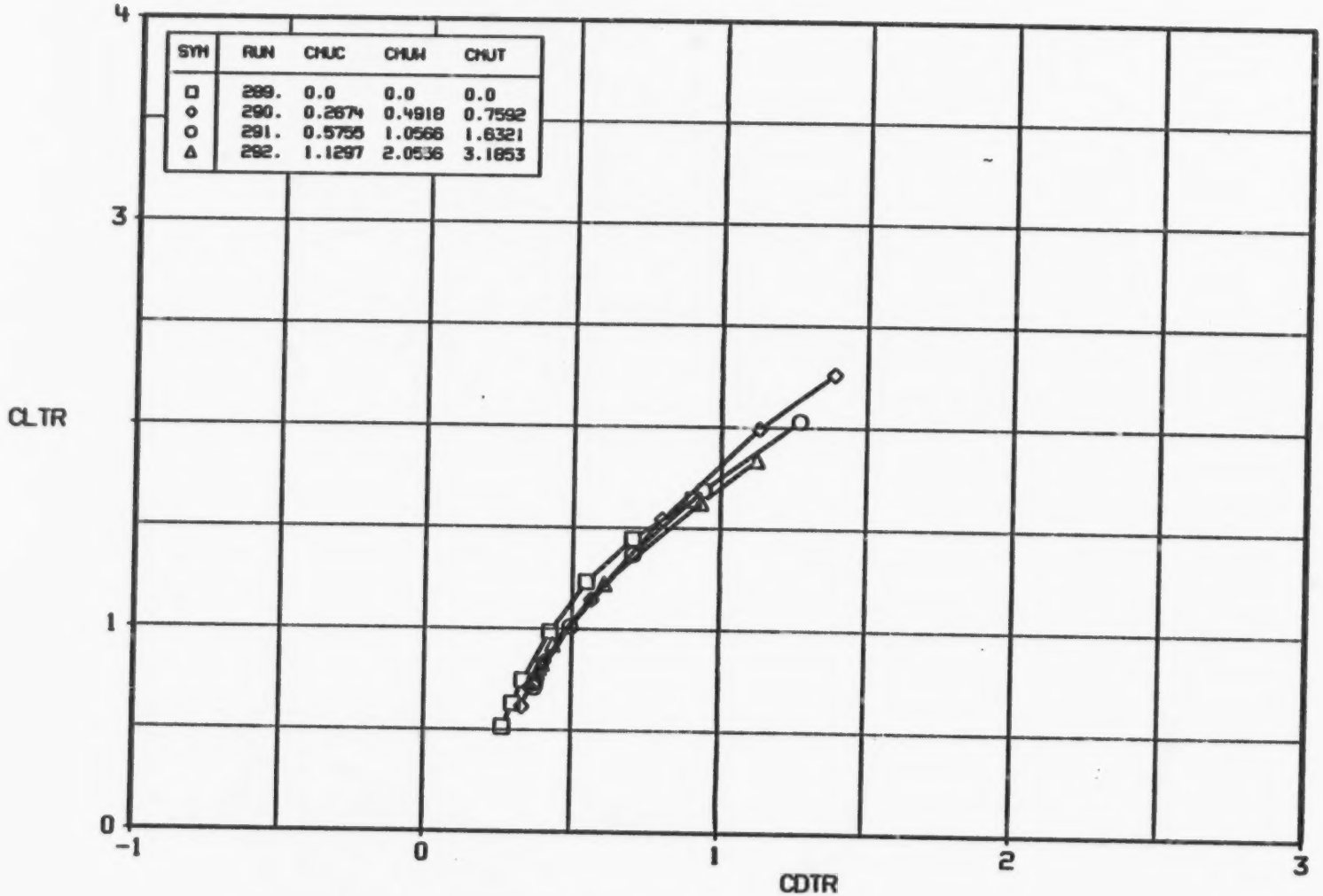

FIGURE A21e BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

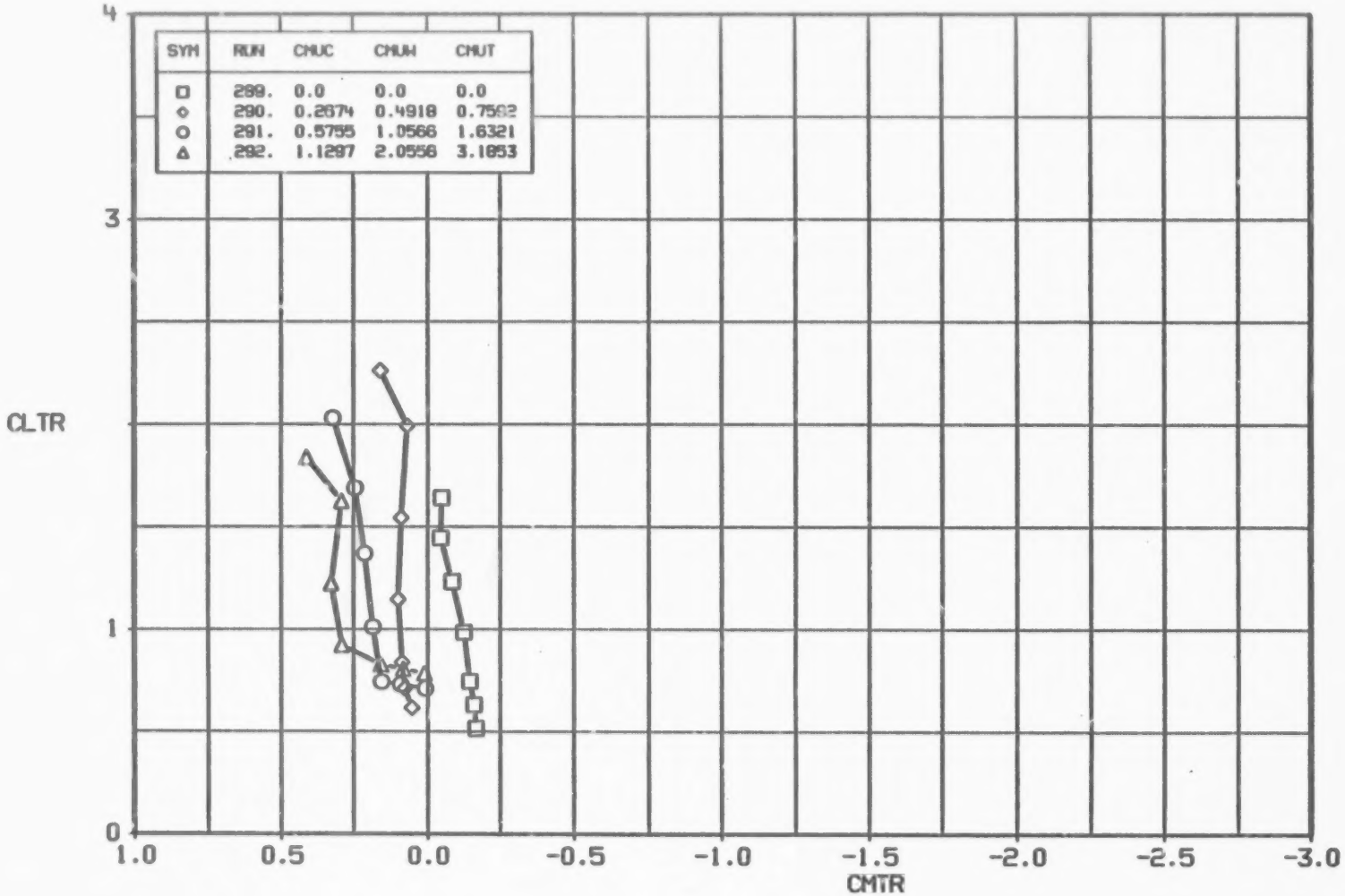

FIGURE A21f BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V,DELF=45,(BN/B)C=1,(BN/B)W=0.25

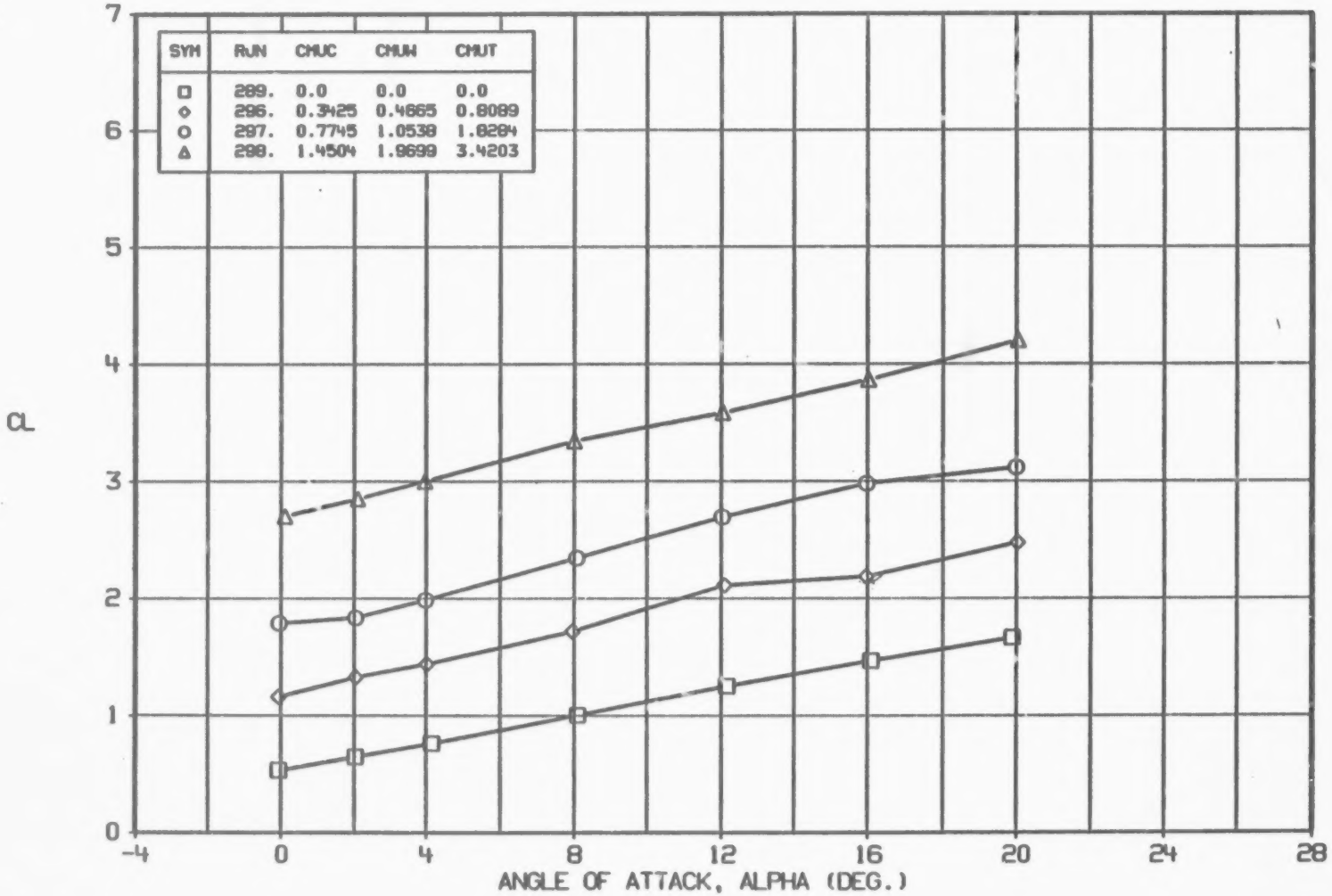

FIGURE A22a BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

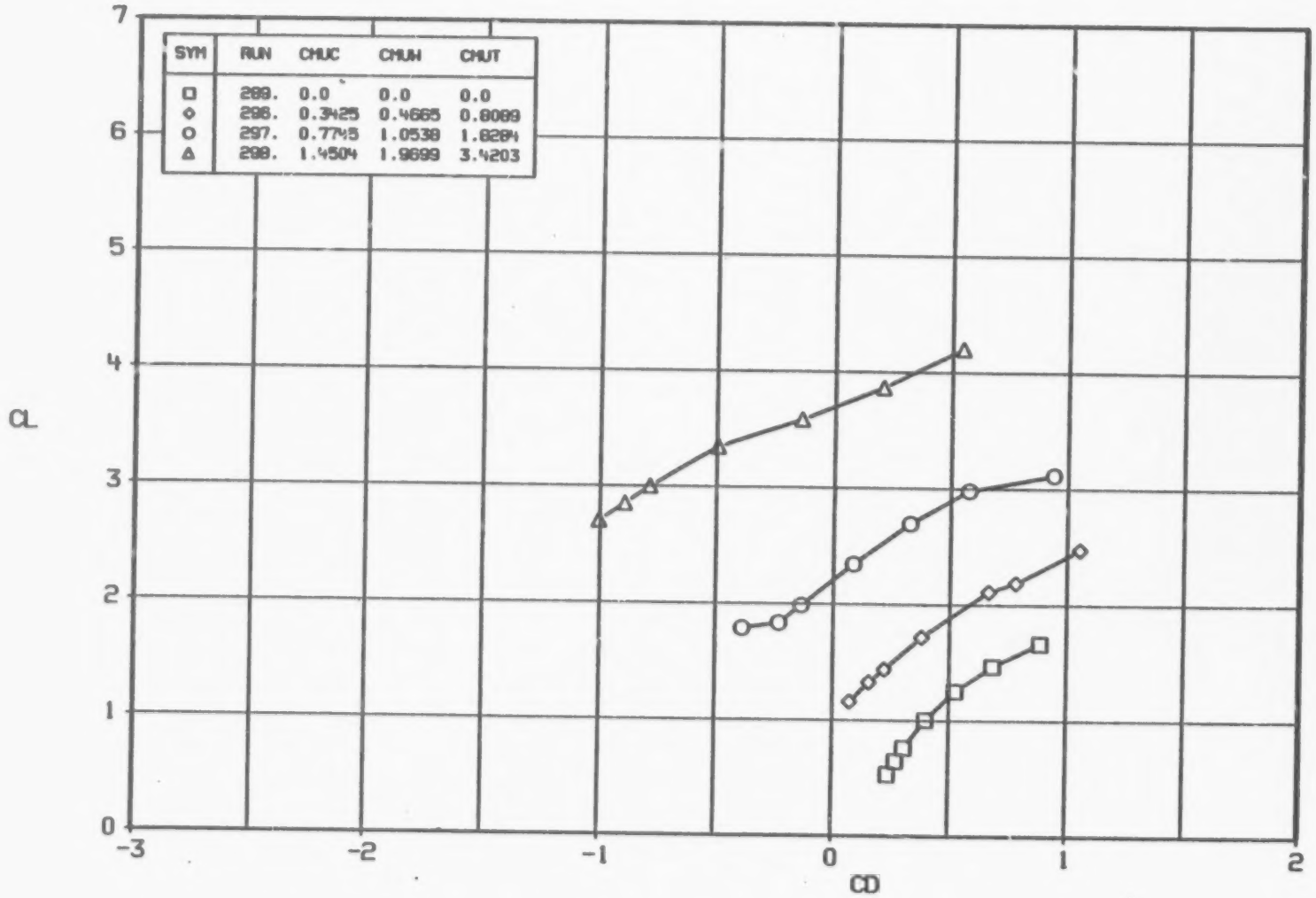

FIGURE A22b BASIC DATA EFFECT OF CMU
CONFIG. BCW6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

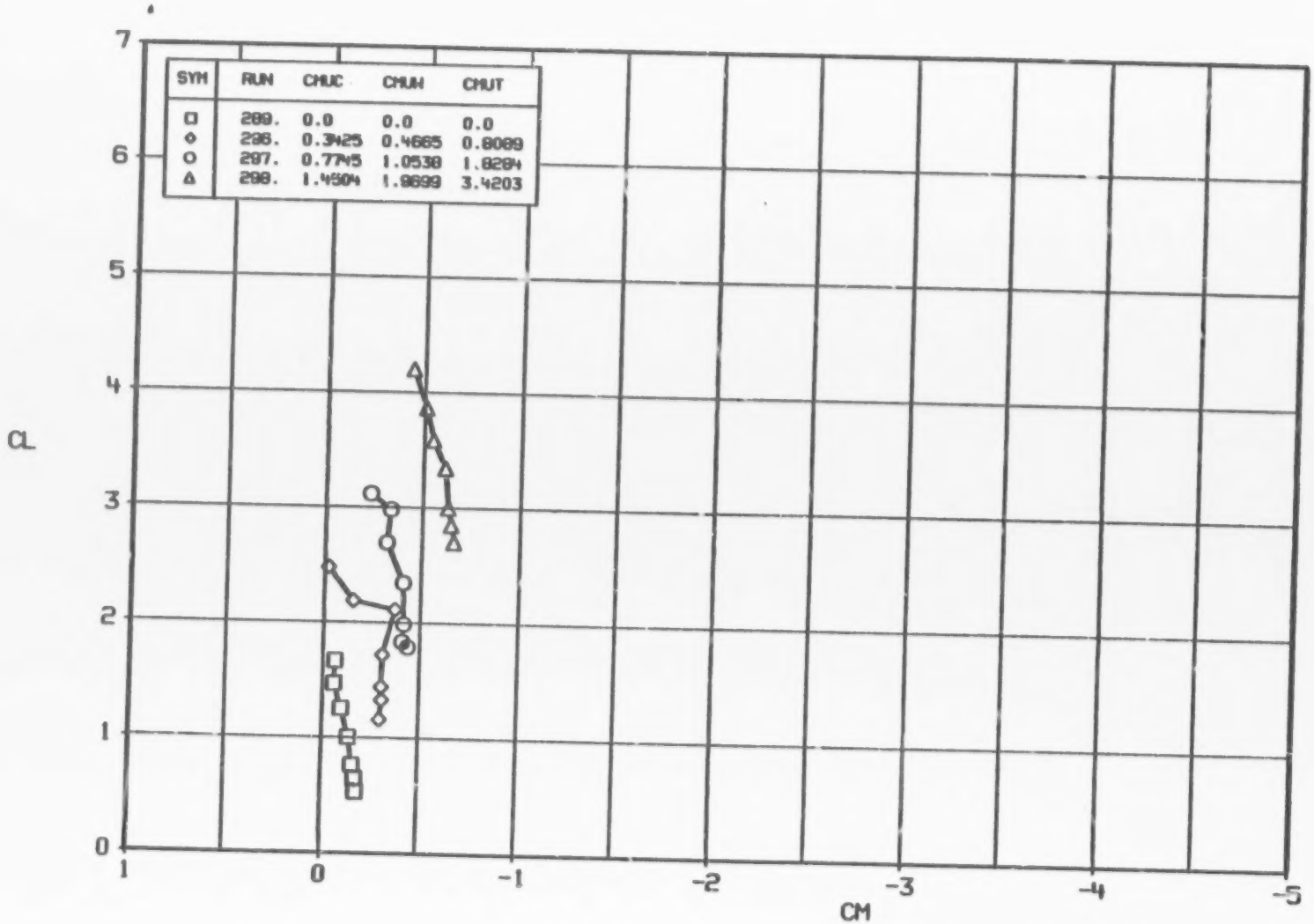

FIGURE A22c BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

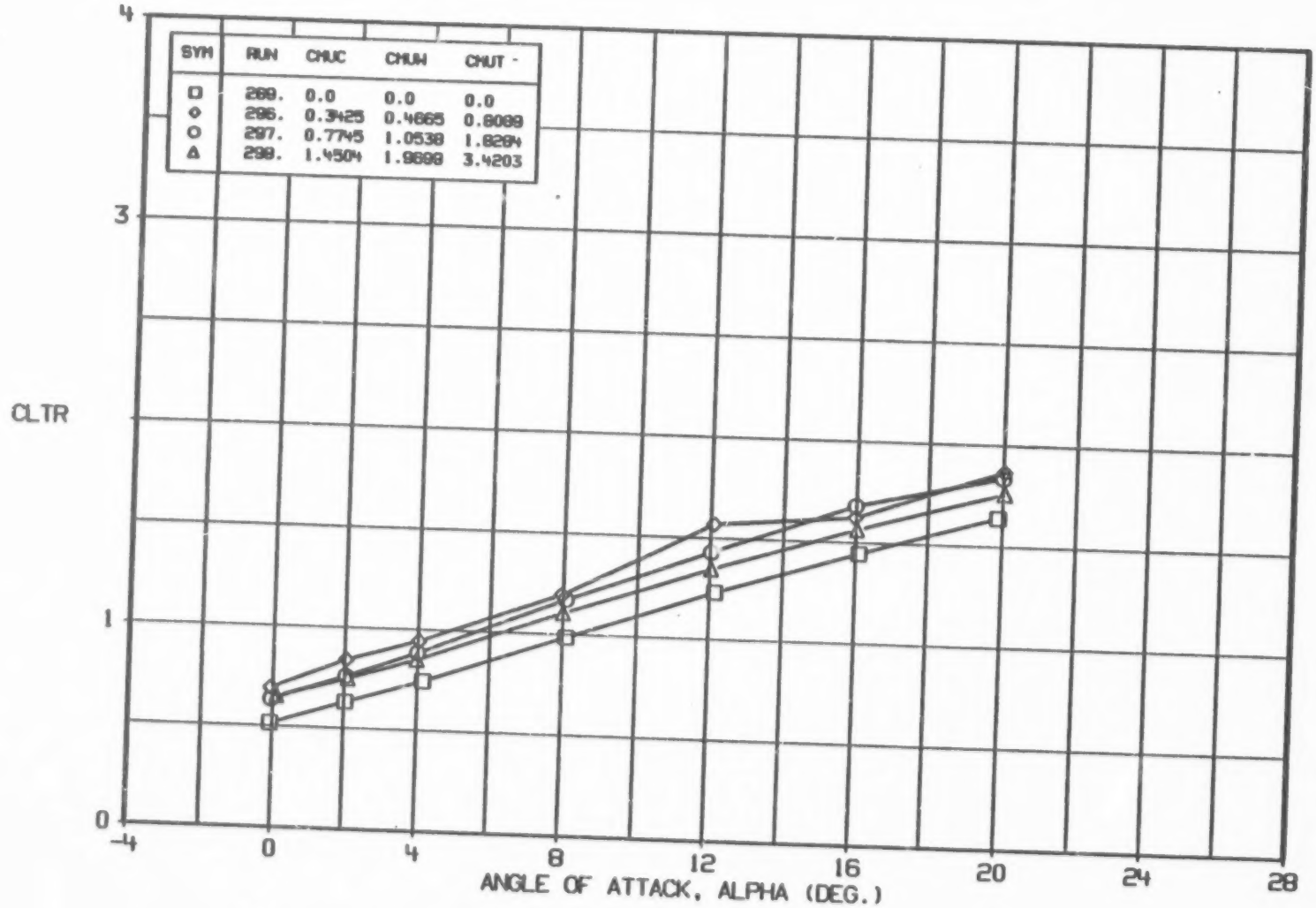

FIGURE A22d BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

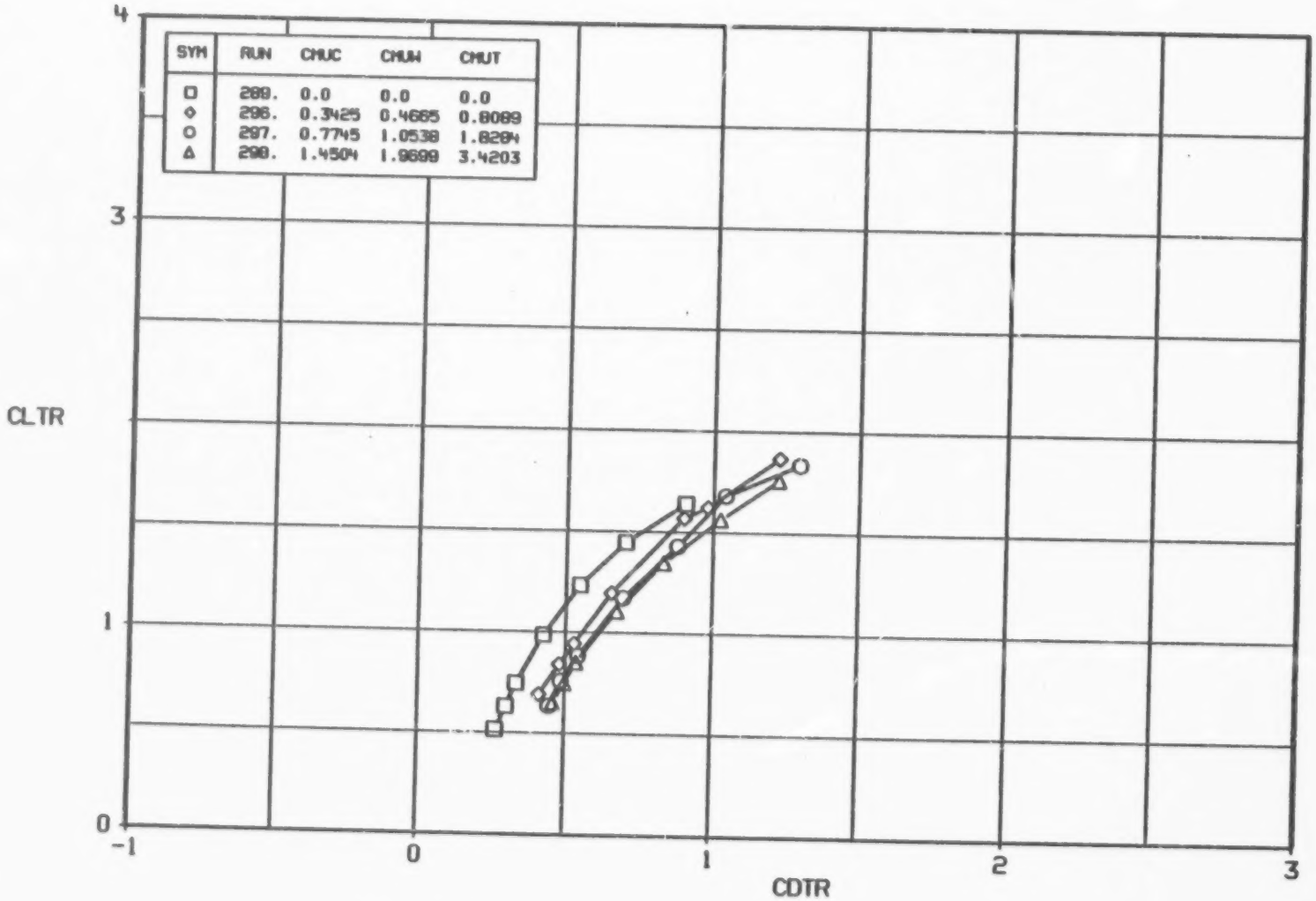

FIGURE A22e BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

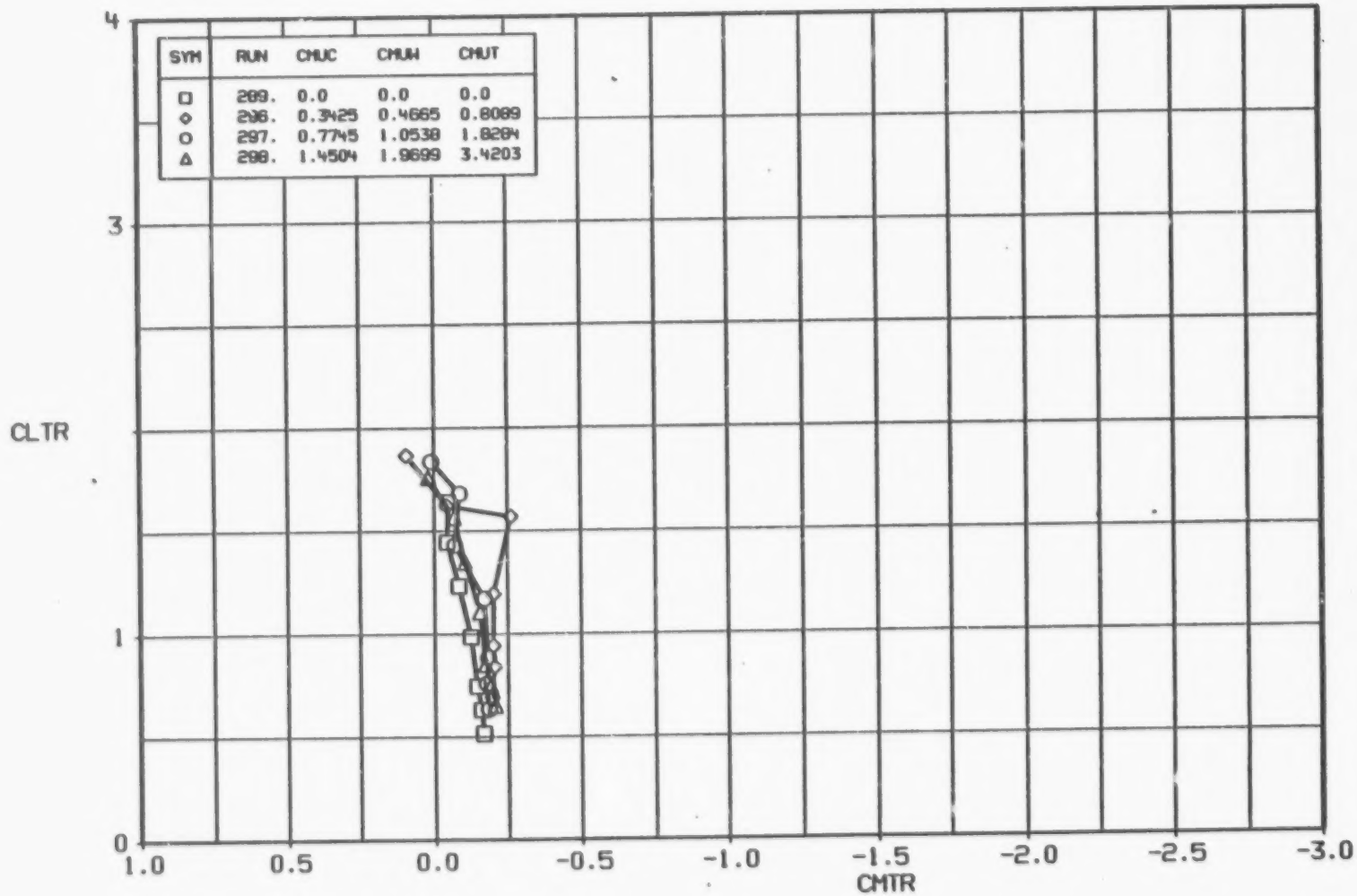

FIGURE A22f BASIC DATA EFFECT OF CMU
CONFIG. BC1W6V DELF=45 $(BN/B)_C$=0.5 $(BN/B)_W$=0.25

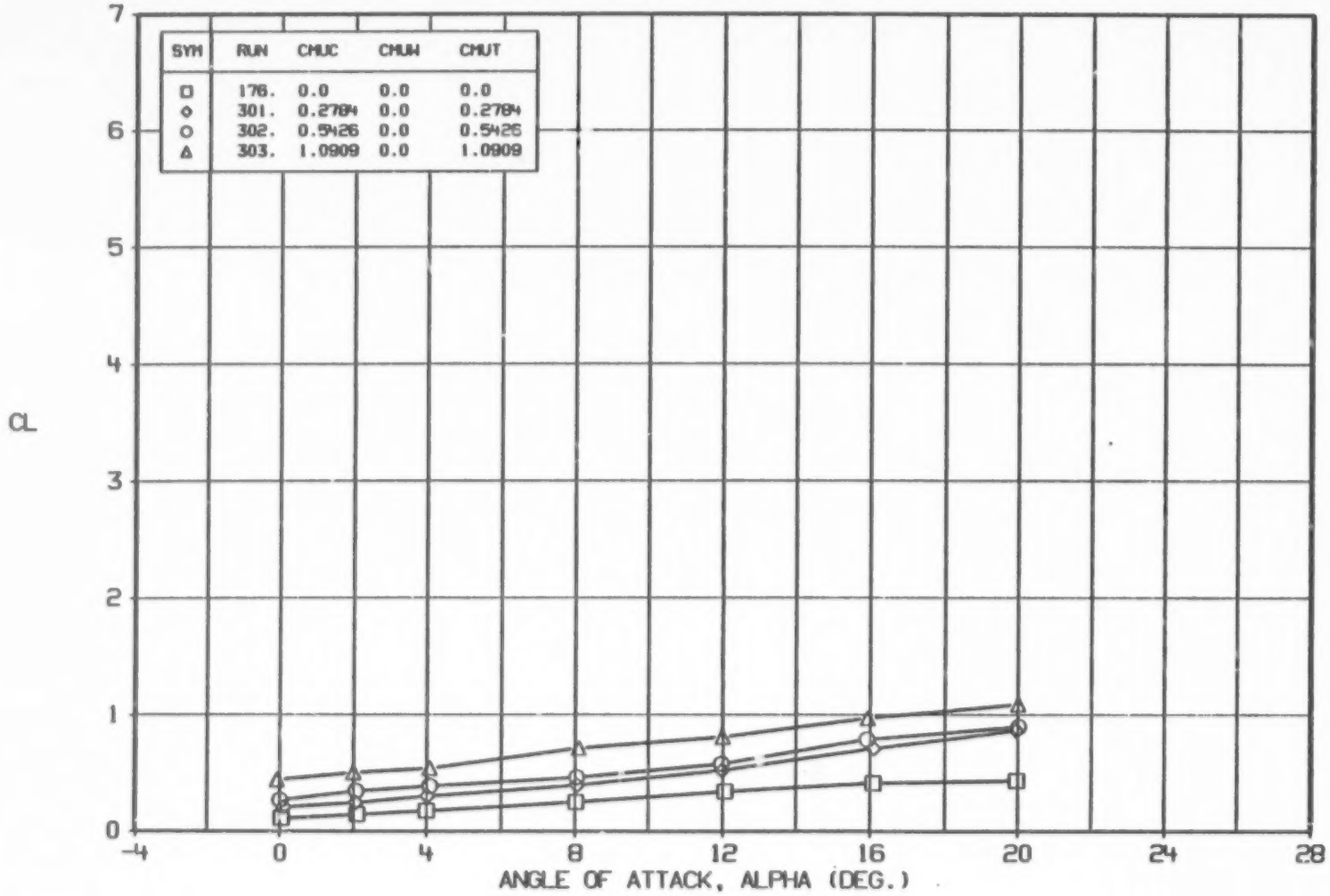

FIGURE A23a BASIC DATA EFFECT OF CMU
CONFIG. BCIV, DELF=45,(BN/B)C=0.5

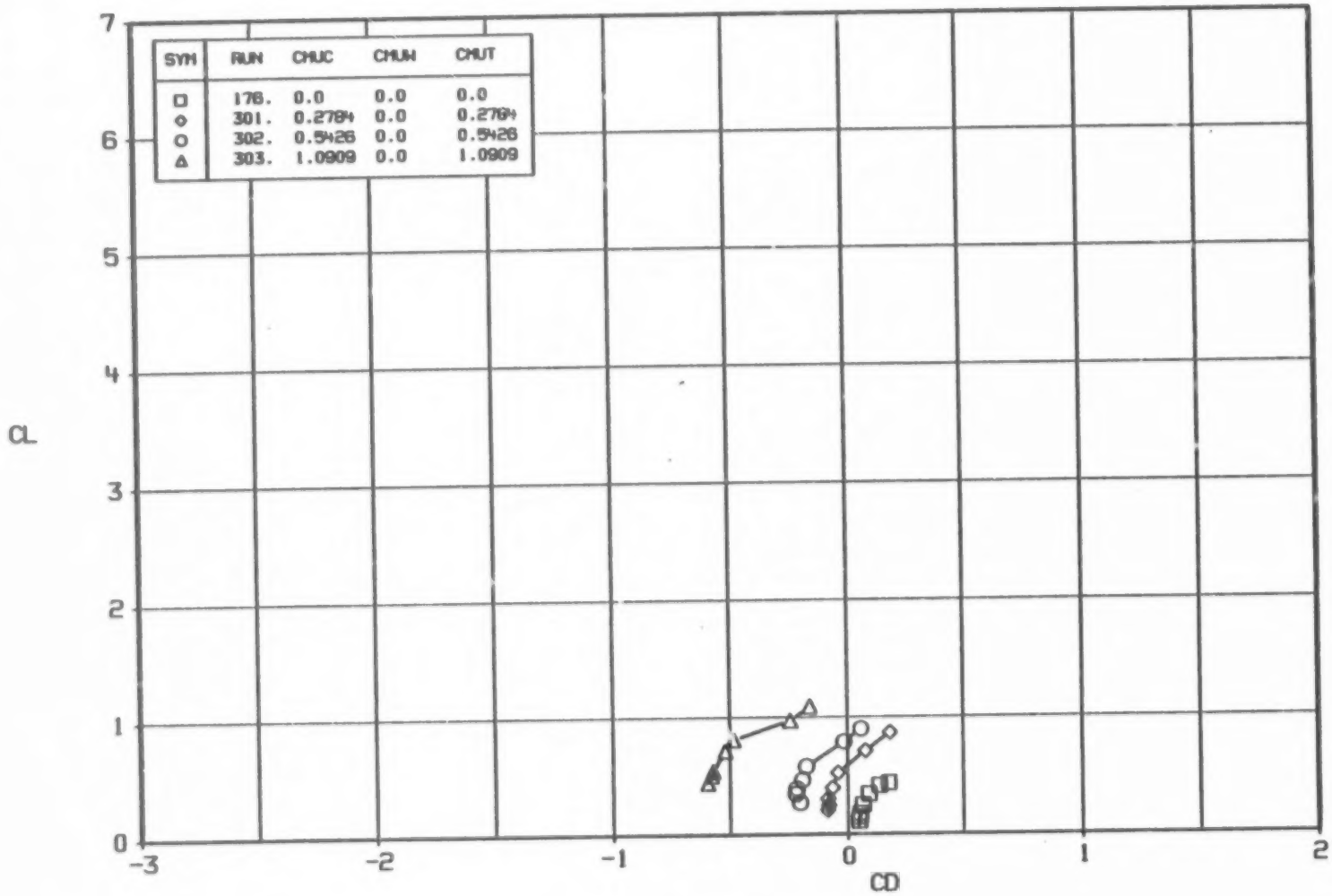

FIGURE A23b BASIC DATA EFFECT OF CMU
CONFIG. BC1V, DELF=45,(BN/B)C=0.5.

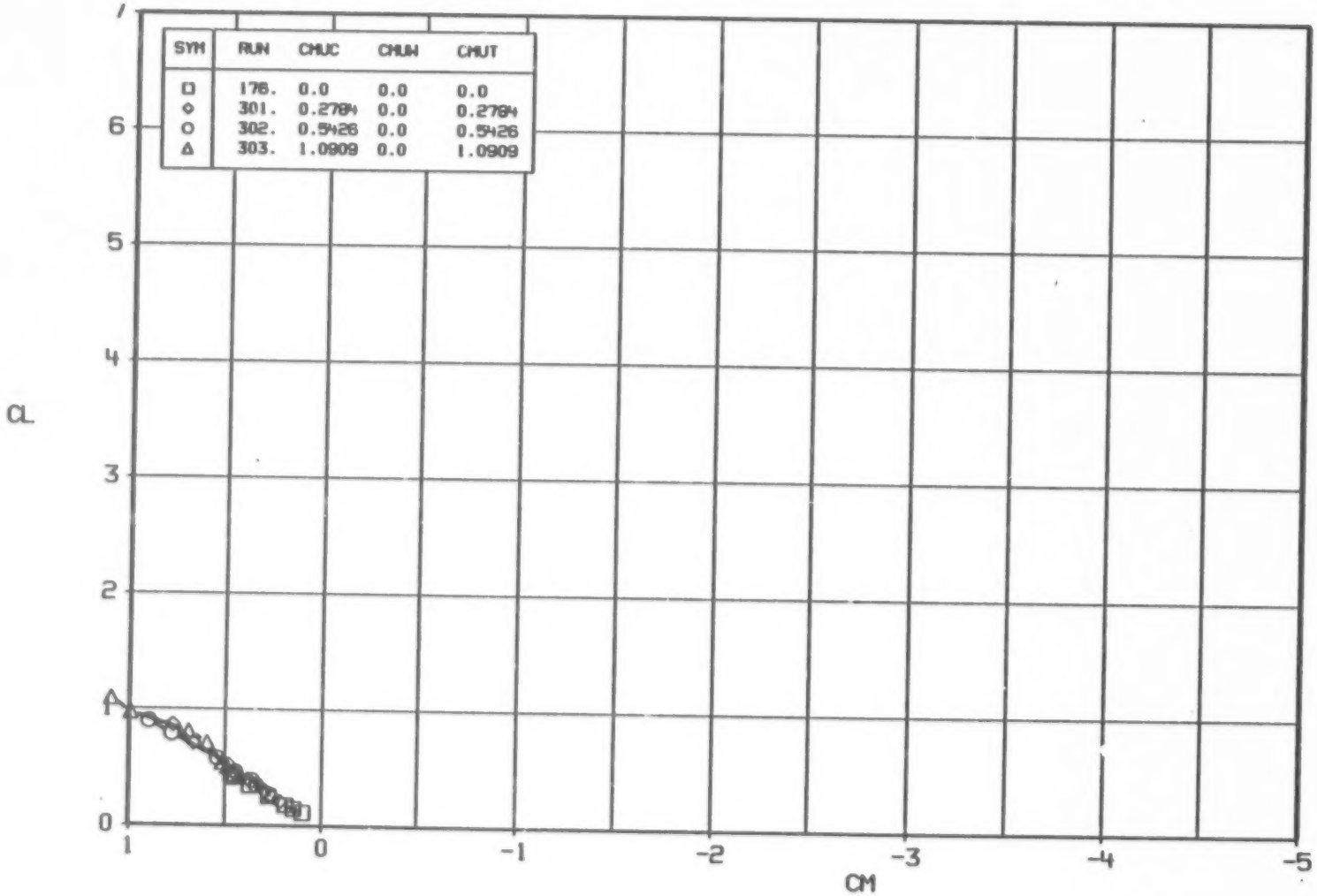

FIGURE A23c BASIC DATA EFFECT OF CMU
CONFIG. BCIV, DELF=45, (BN/B)C=0.5

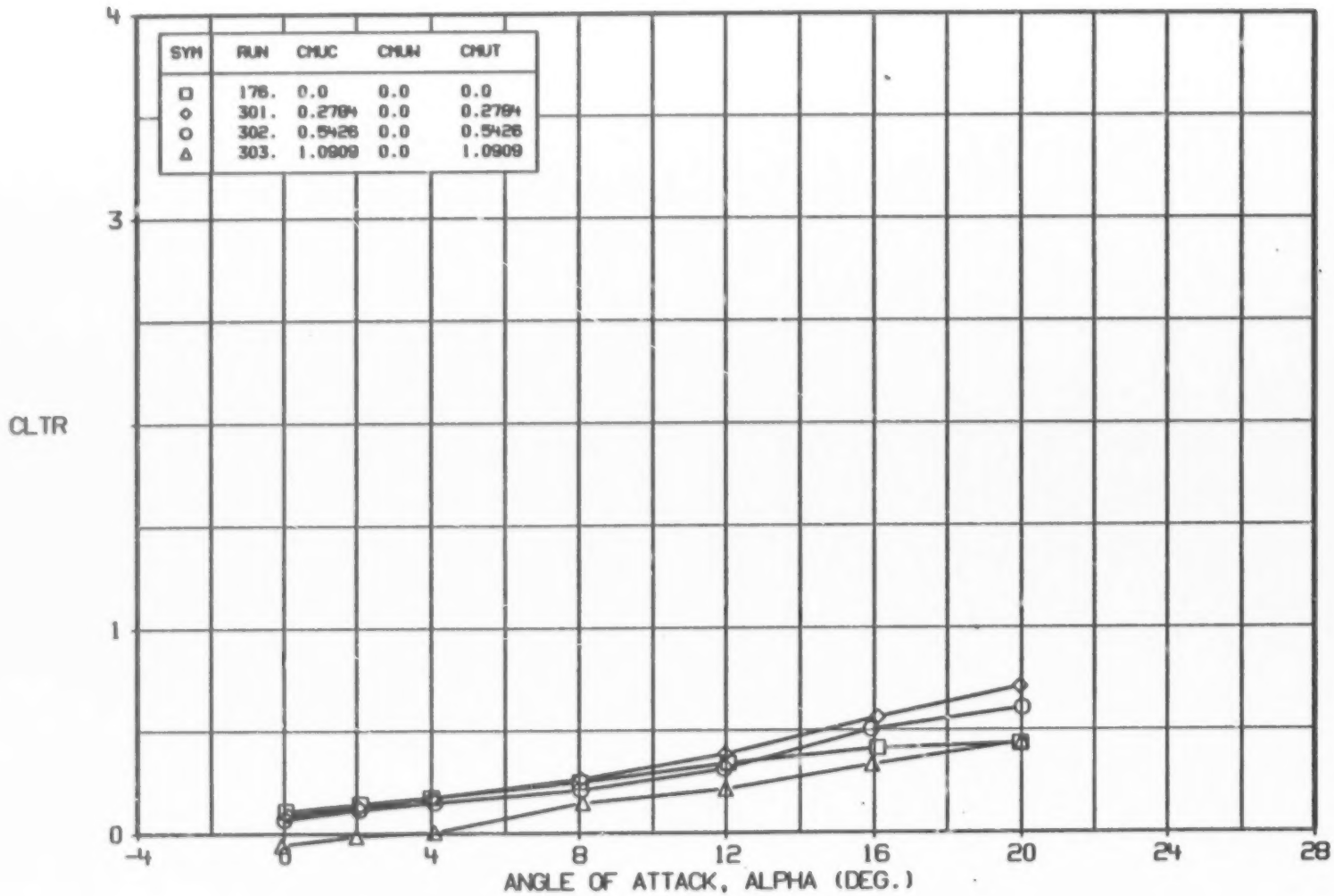

FIGURE A23d BASIC DATA EFFECT OF CMU
CONFIG. BC1V, DELF=45,(BN/B)C=0.5,(BN/B)W=0

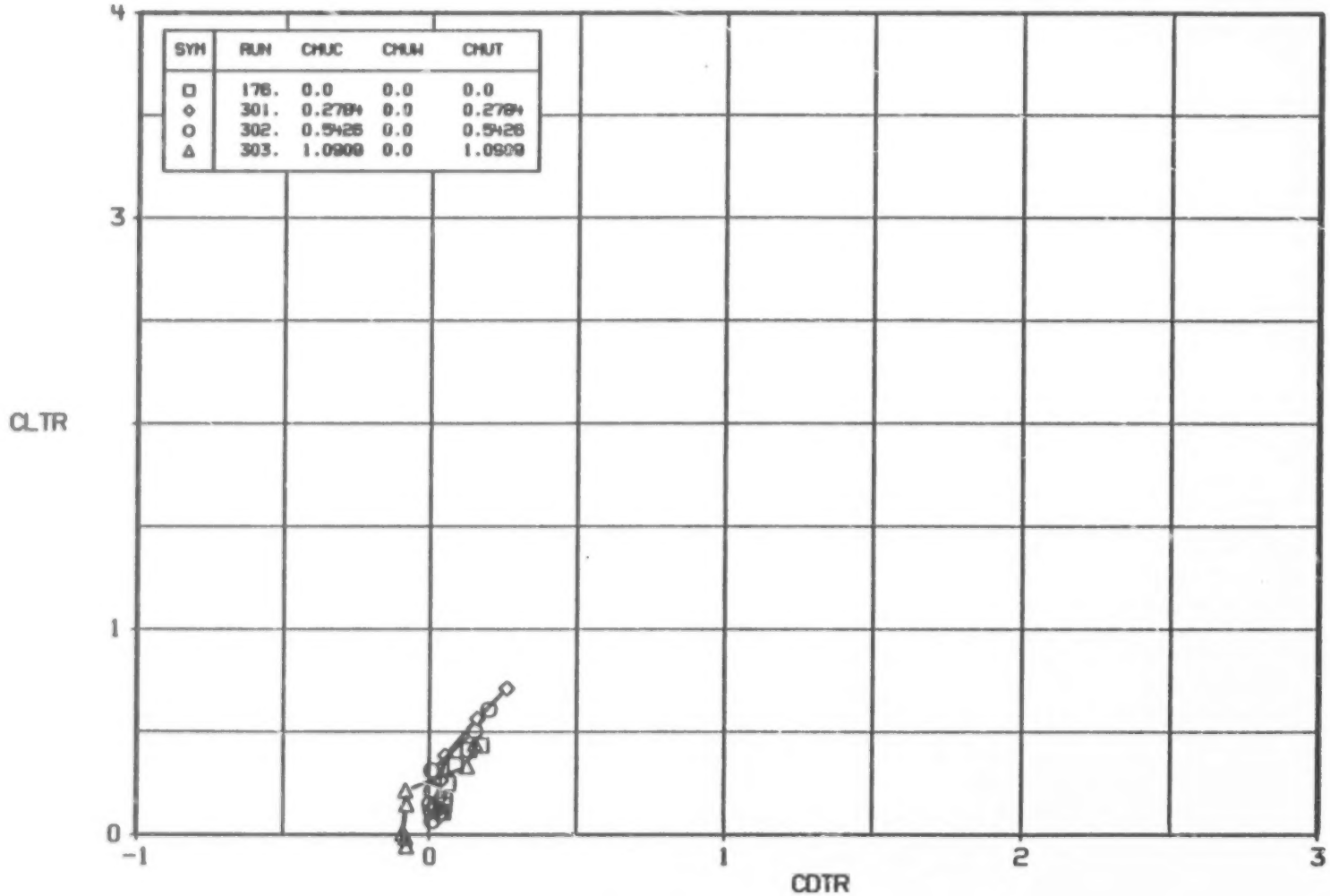

FIGURE A23e BASIC DATA EFFECT OF CMU
CONFIG. BCIV, DELF=45,(BN/B)C=0.5

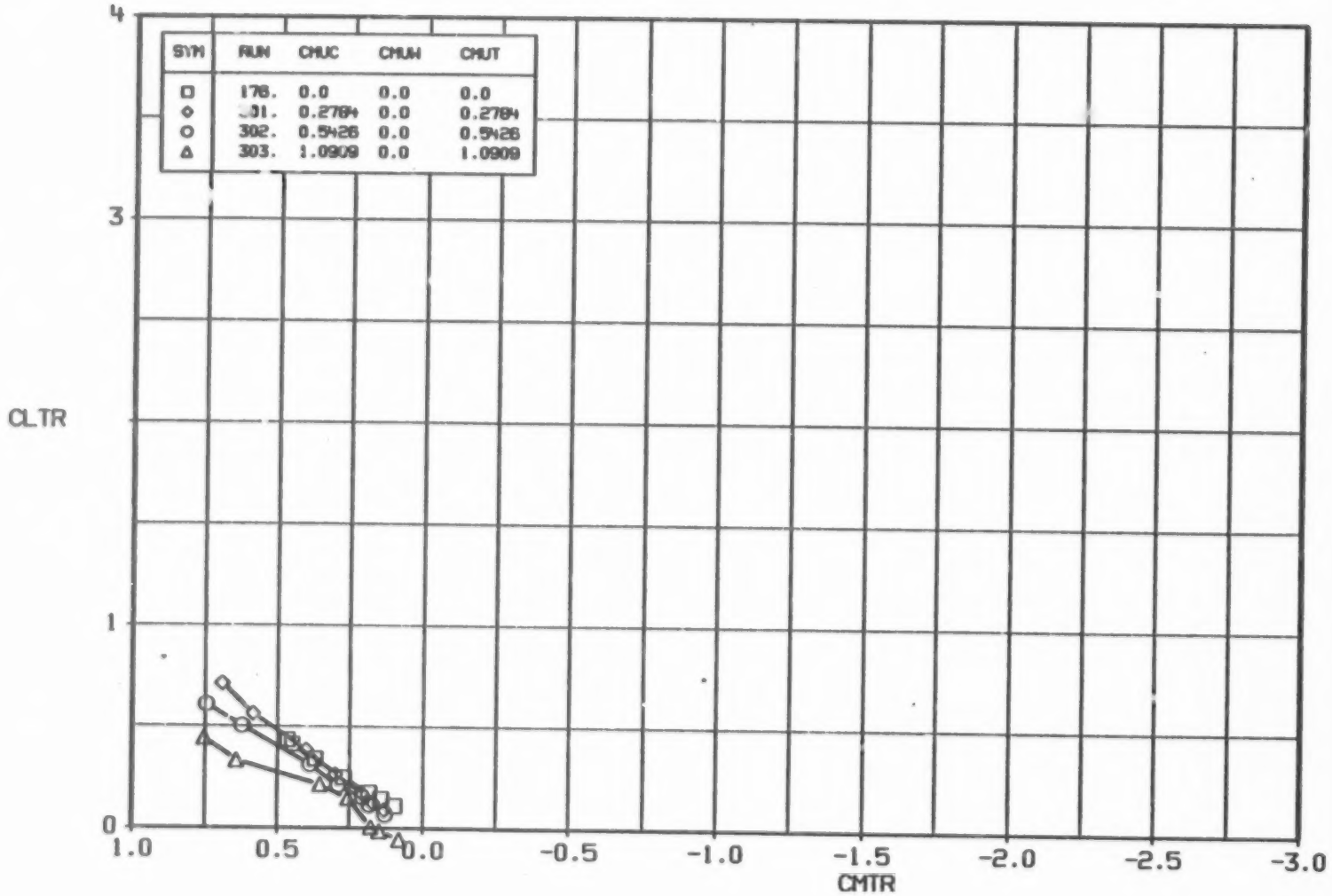

FIGURE A23f BASIC DATA EFFECT OF CMU
CONFIG. BCIV, DELF=45,(BN/B)C=0.5,

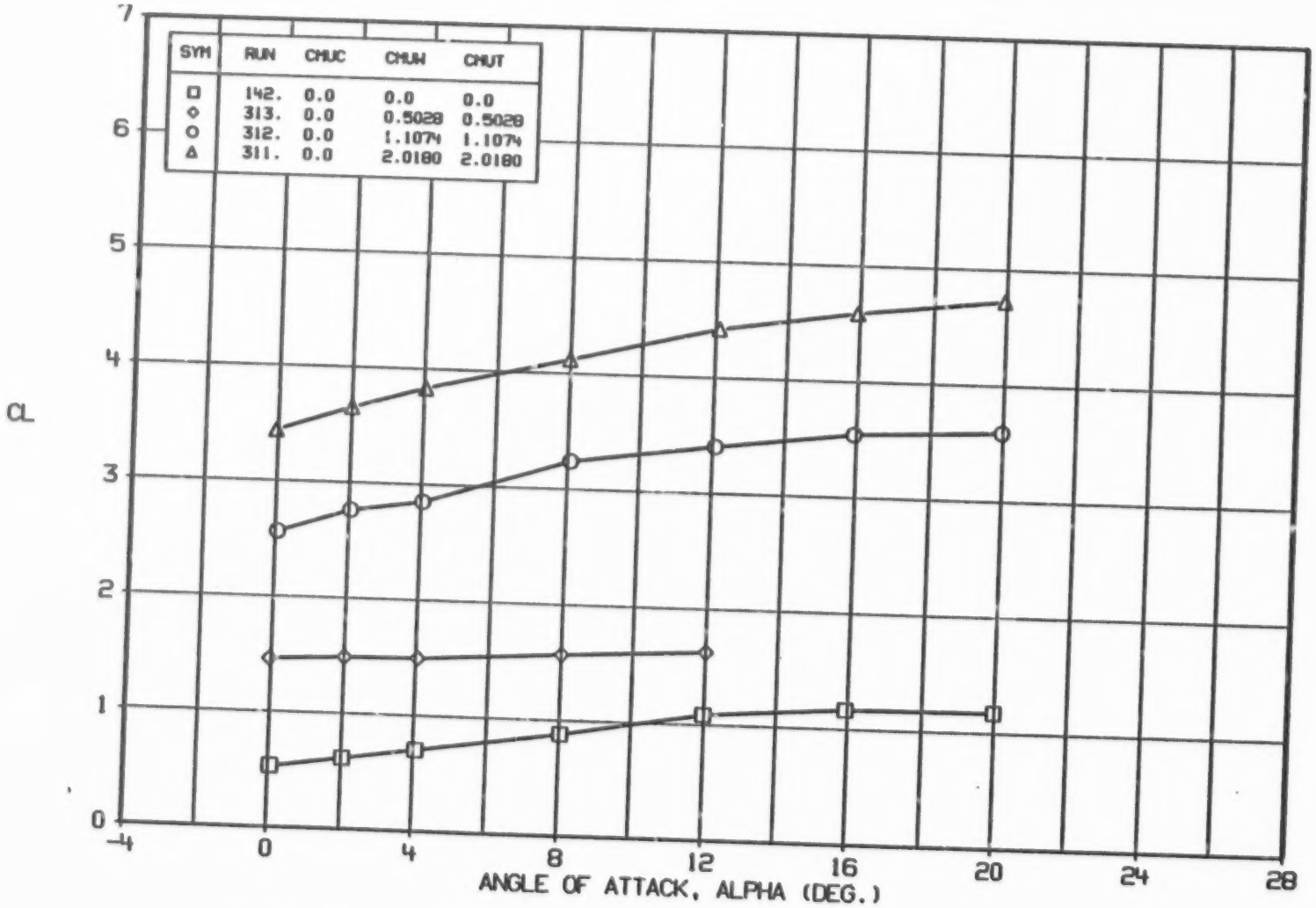

FIGURE A24a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

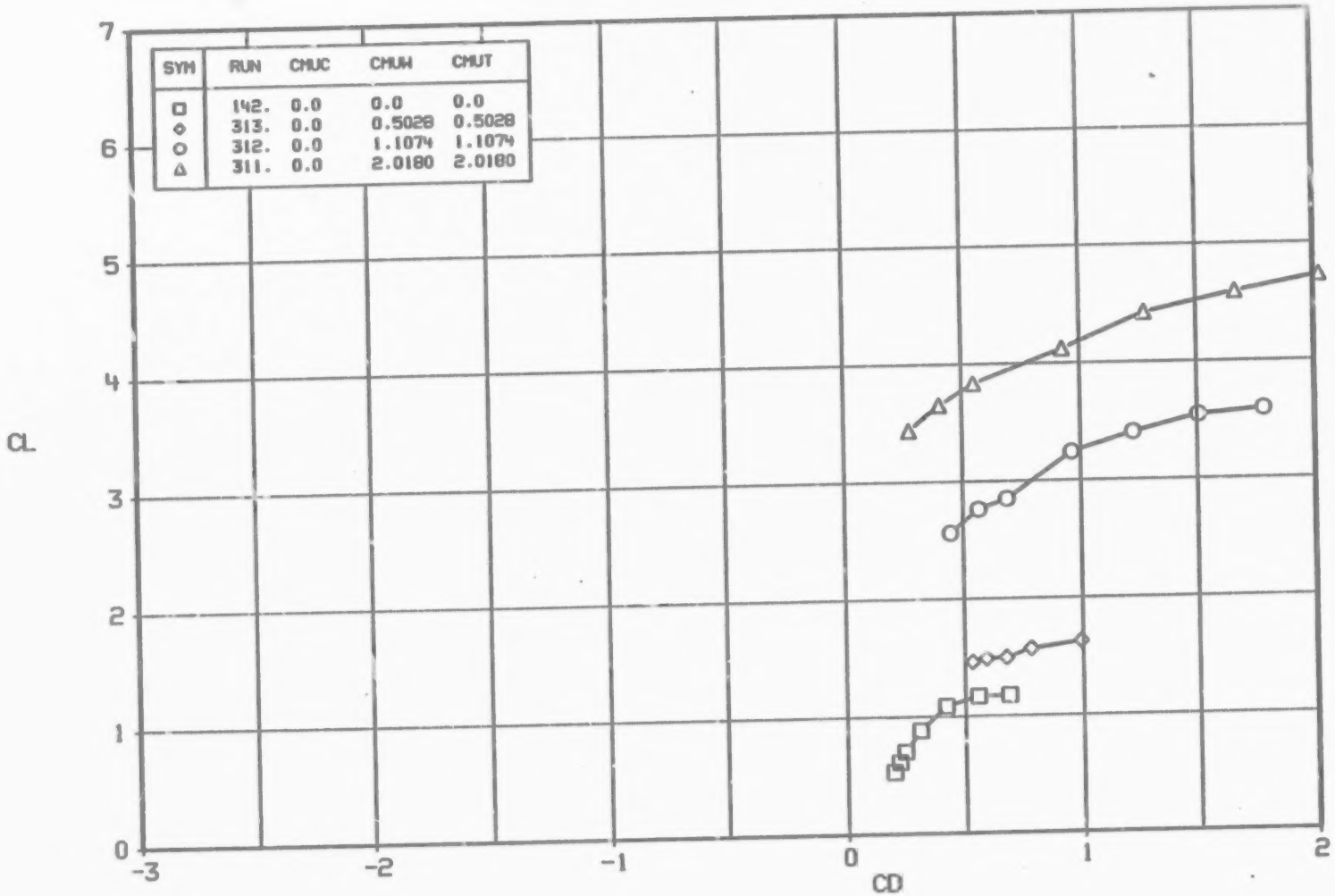

FIGURE A24b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

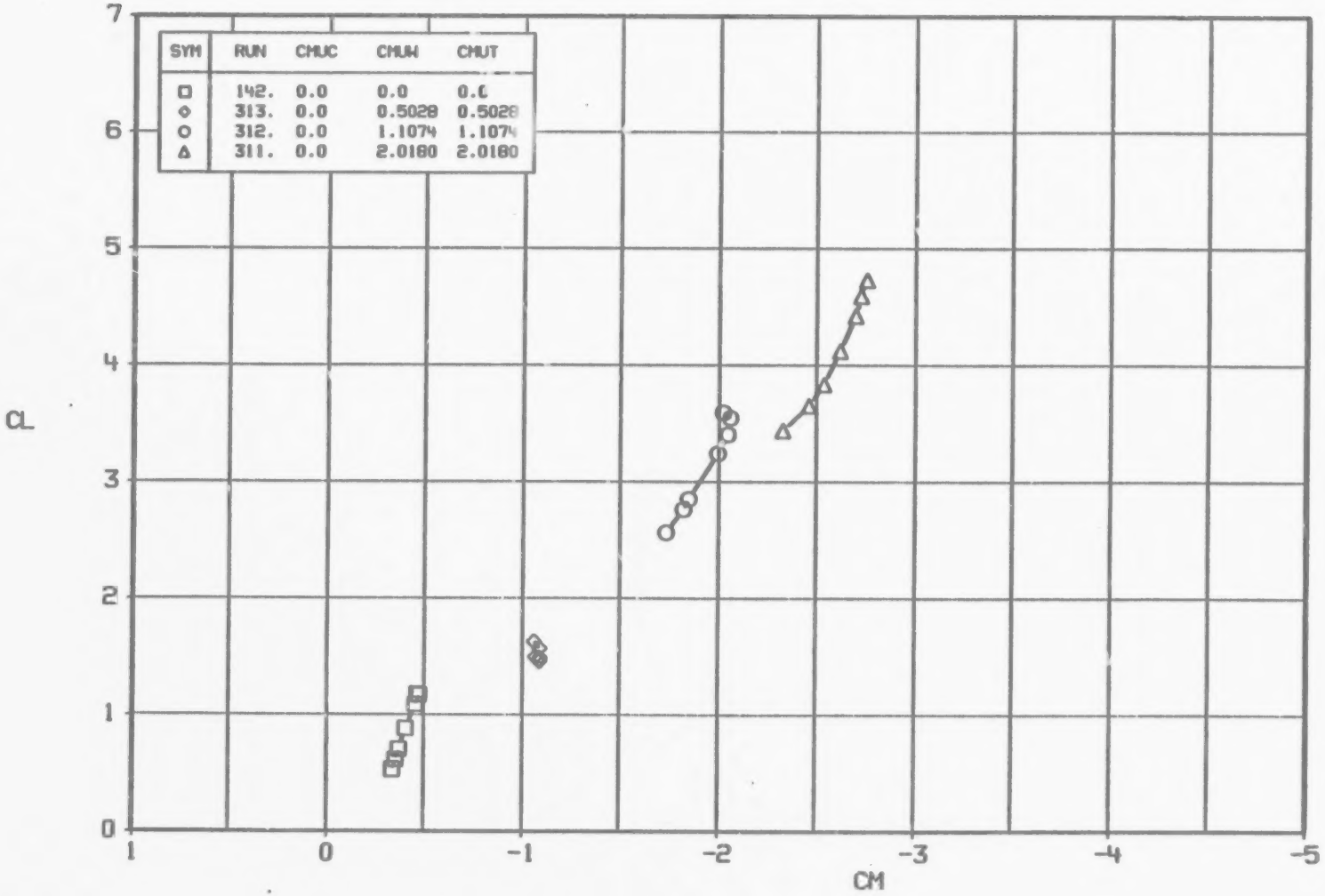

FIGURE A24c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

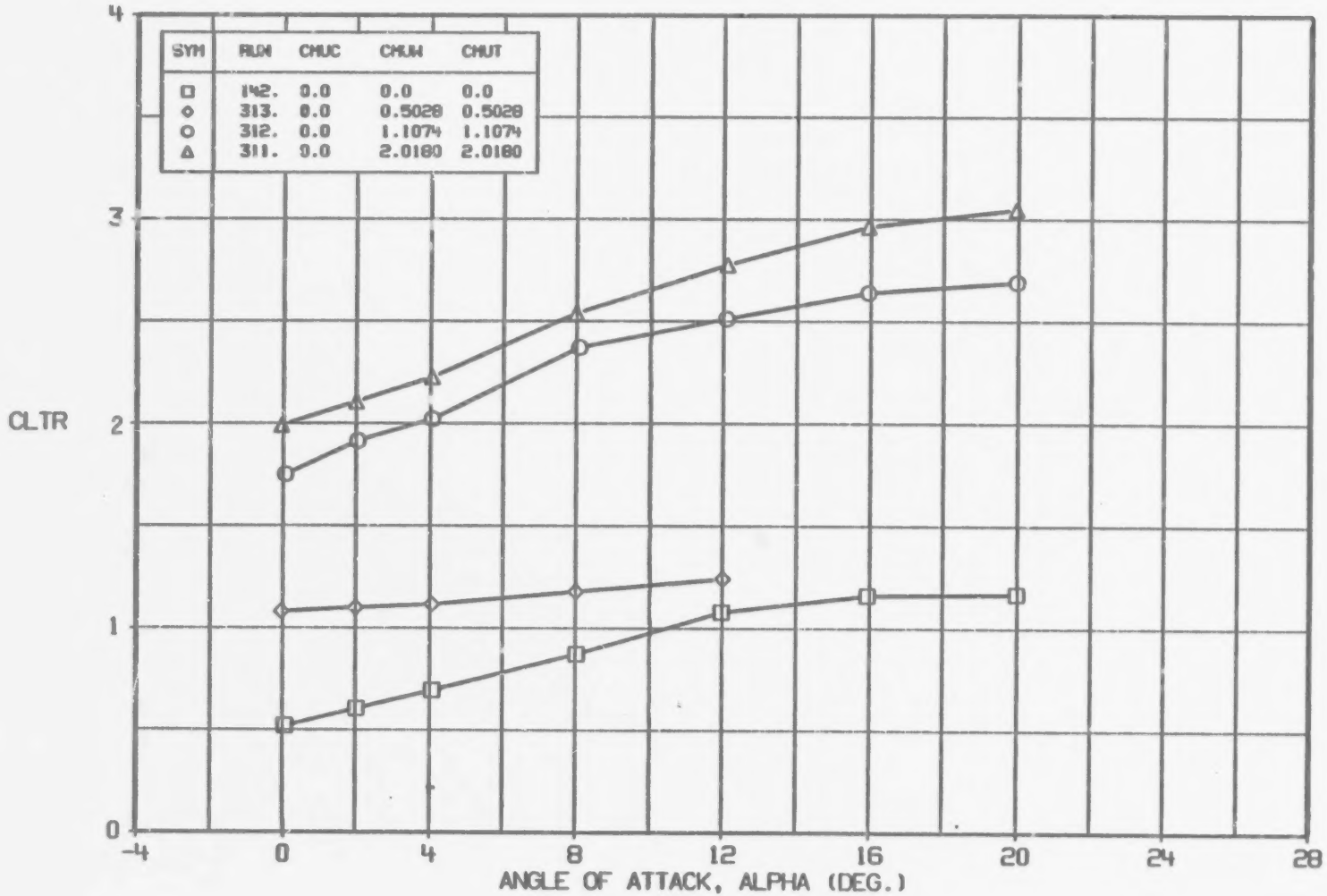

FIGURE A24d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

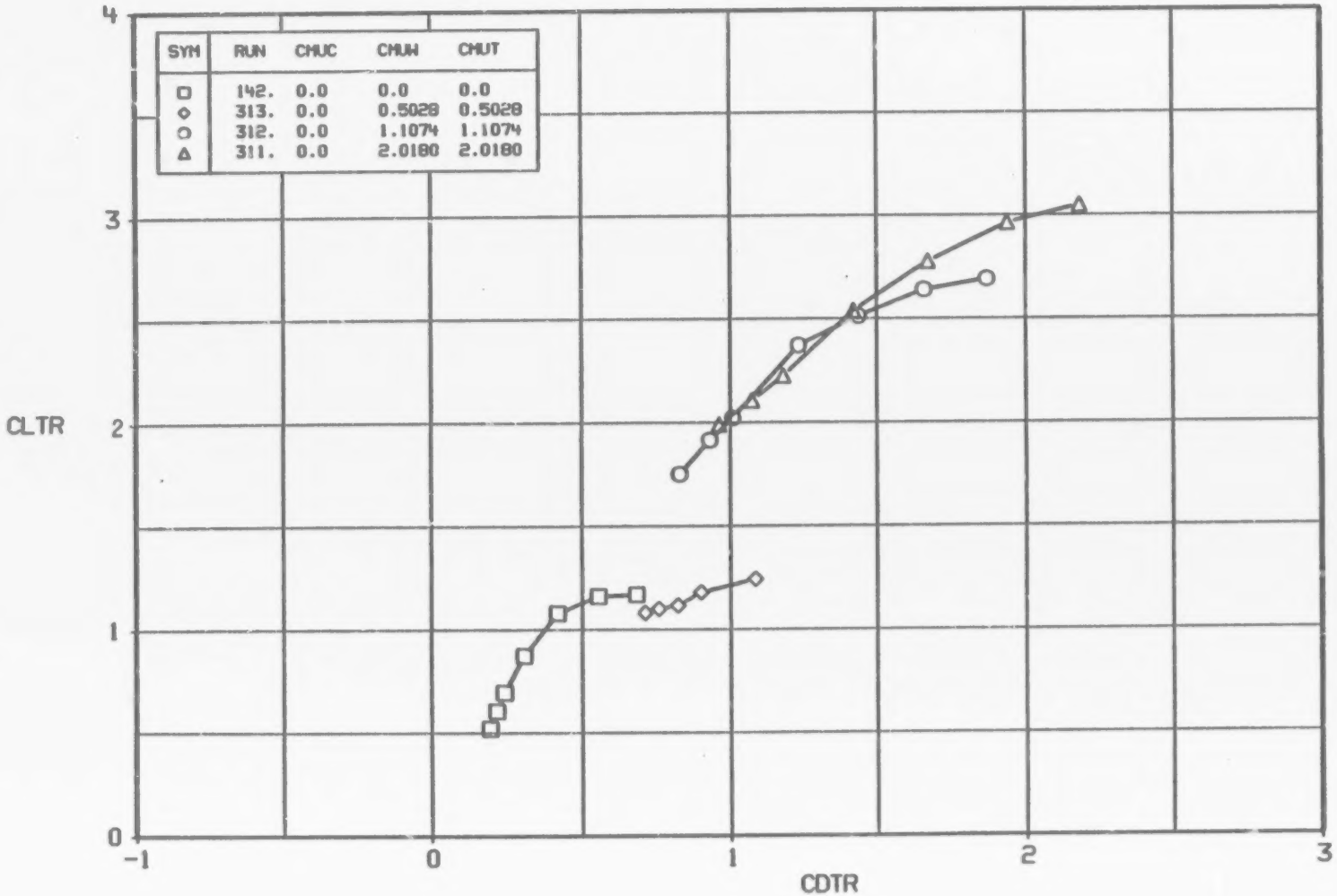

FIGURE A24e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

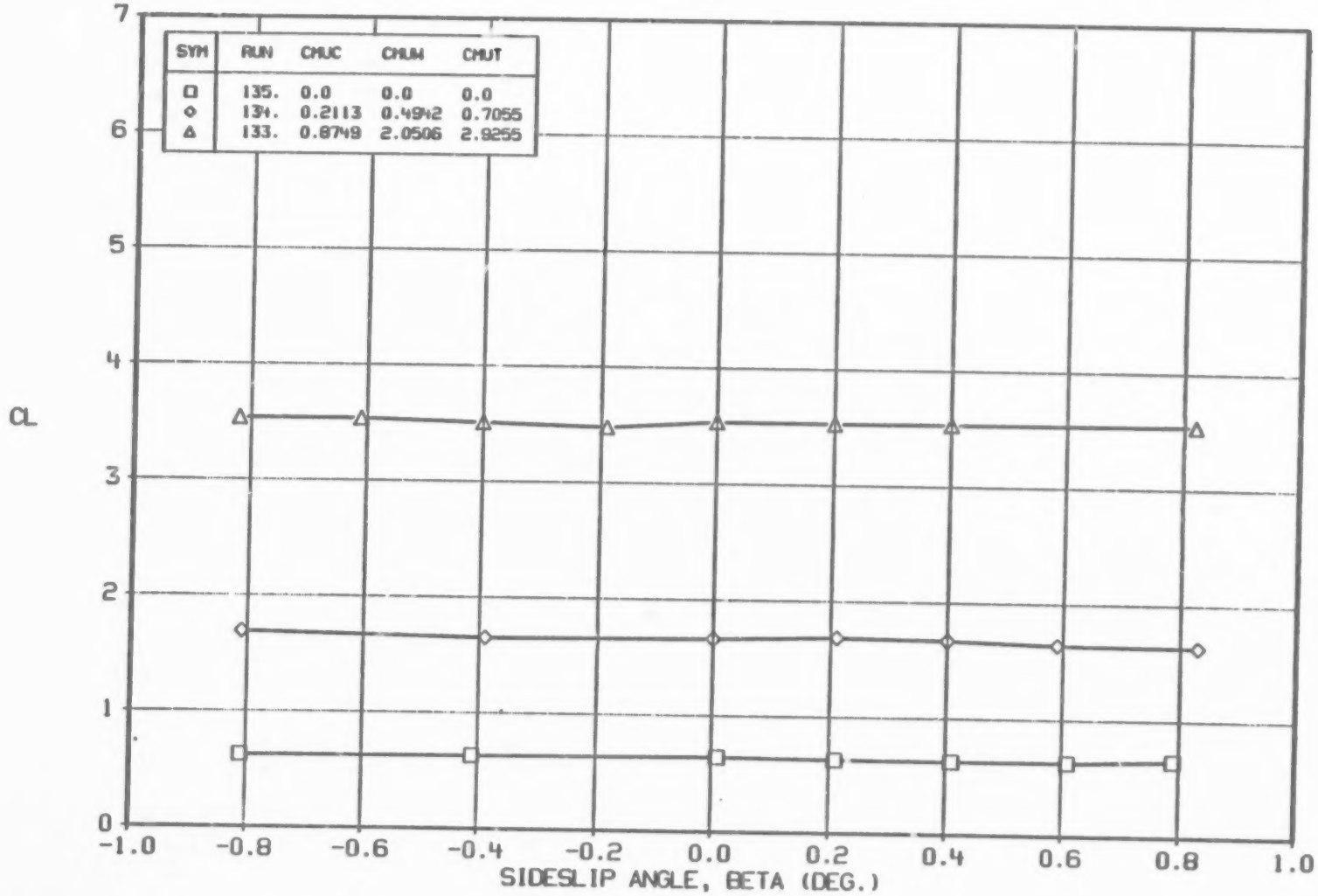

FIGURE A25a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=1.0 , DELF=45, α = 2 .

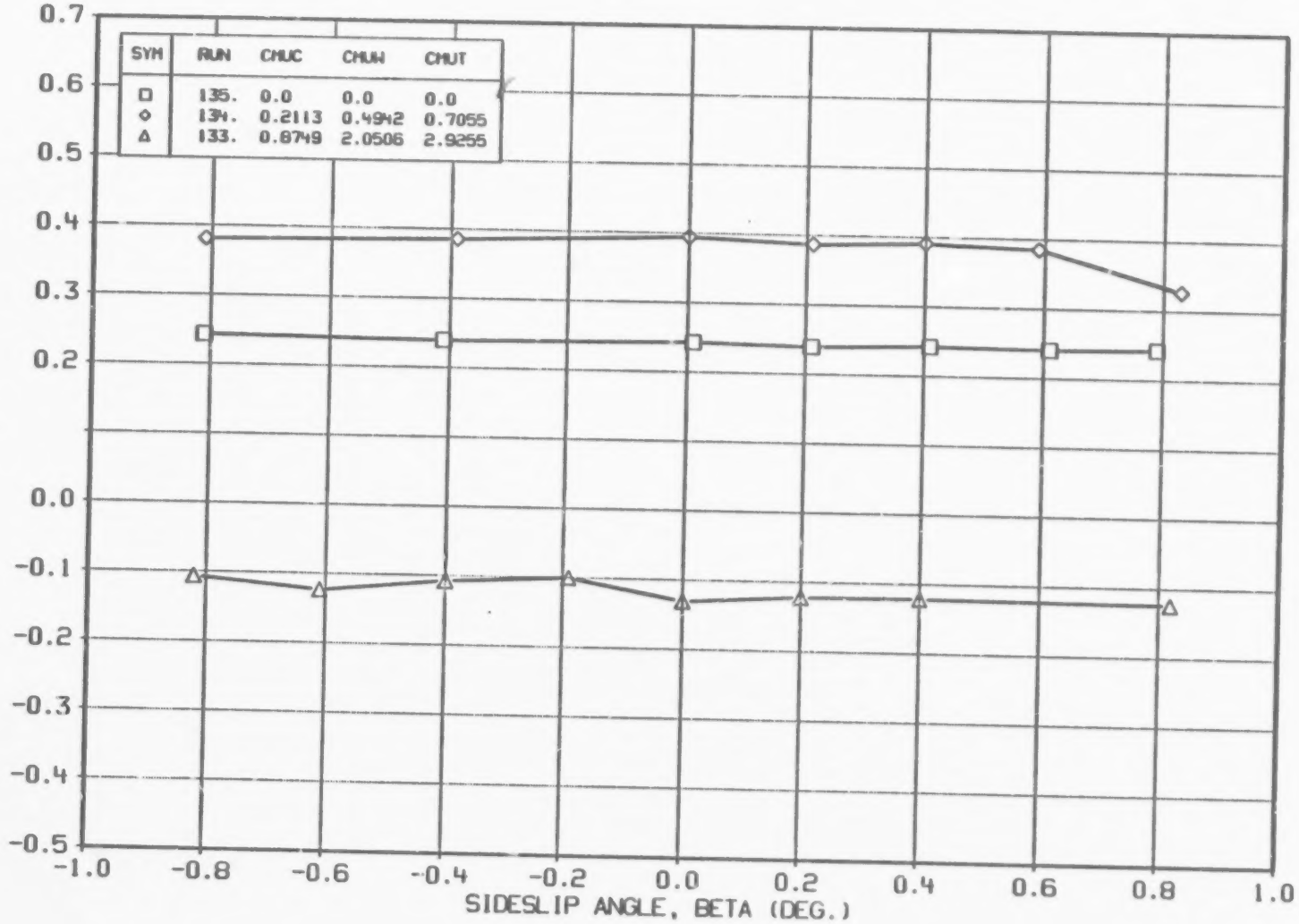

FIGURE A25b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=1.0, DELF=45, α = 2 .

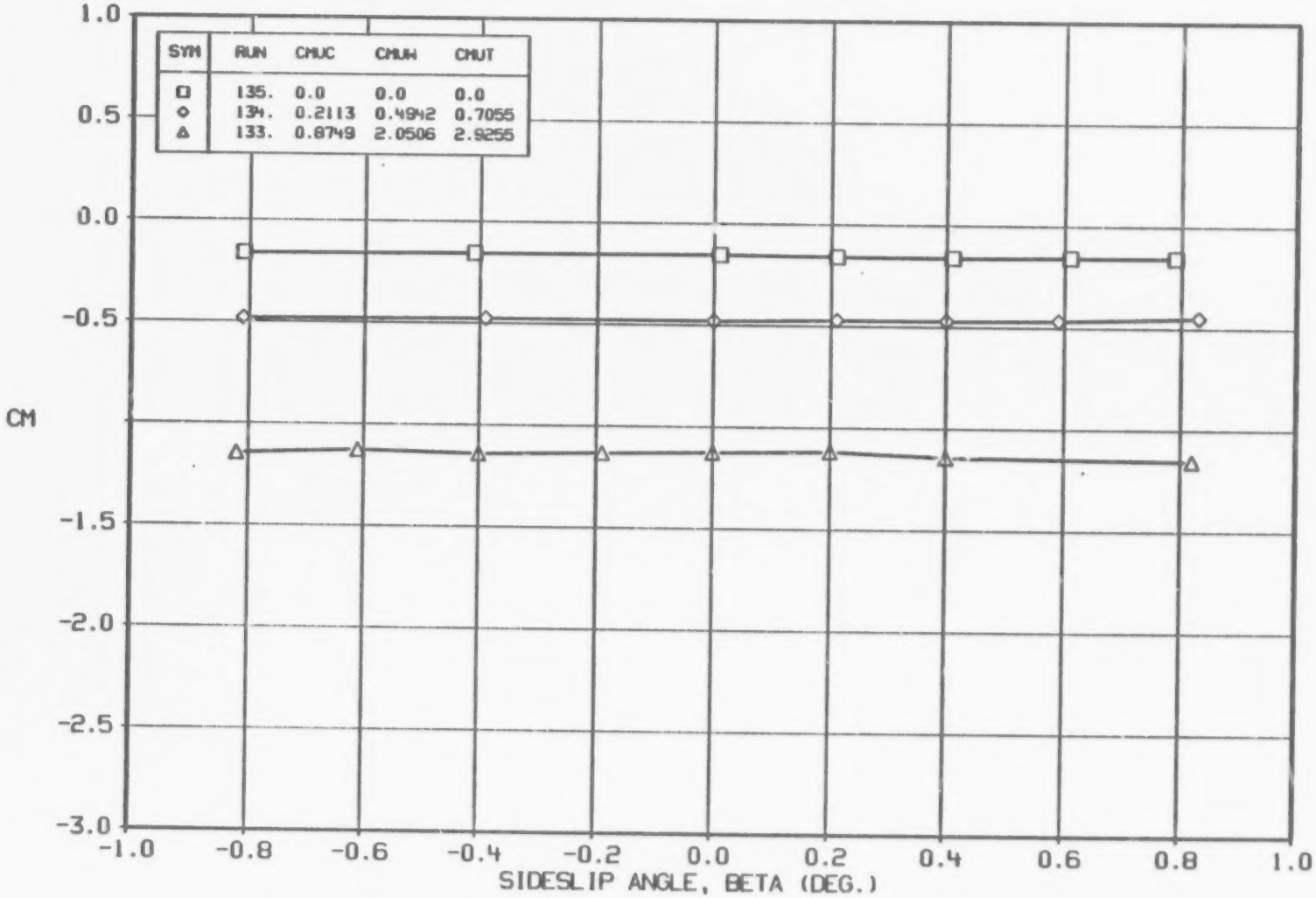

FIGURE A25c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=1.0, DELF=45, α = 2 .

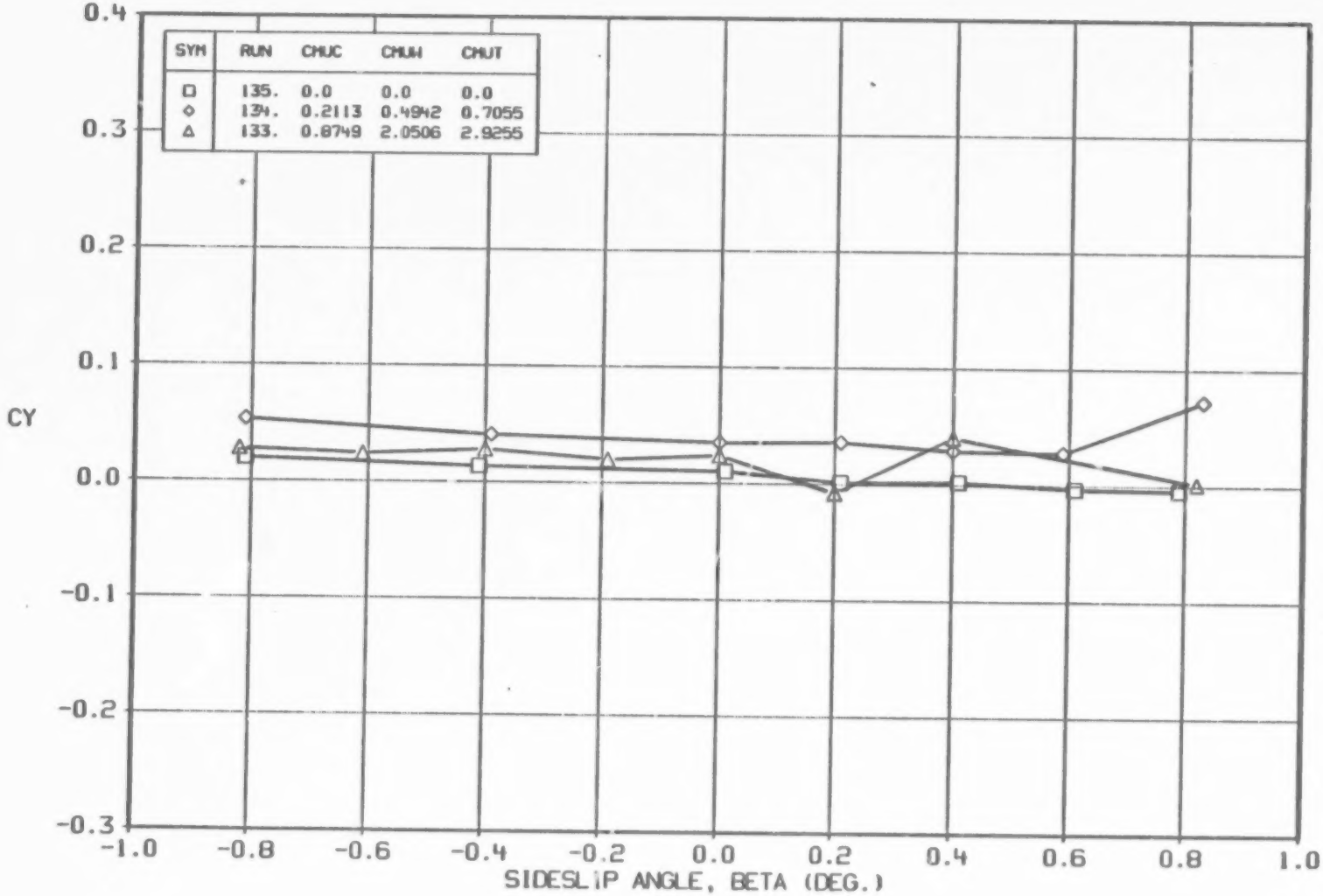

FIGURE A25d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=1.0, DELF=45, α = 2

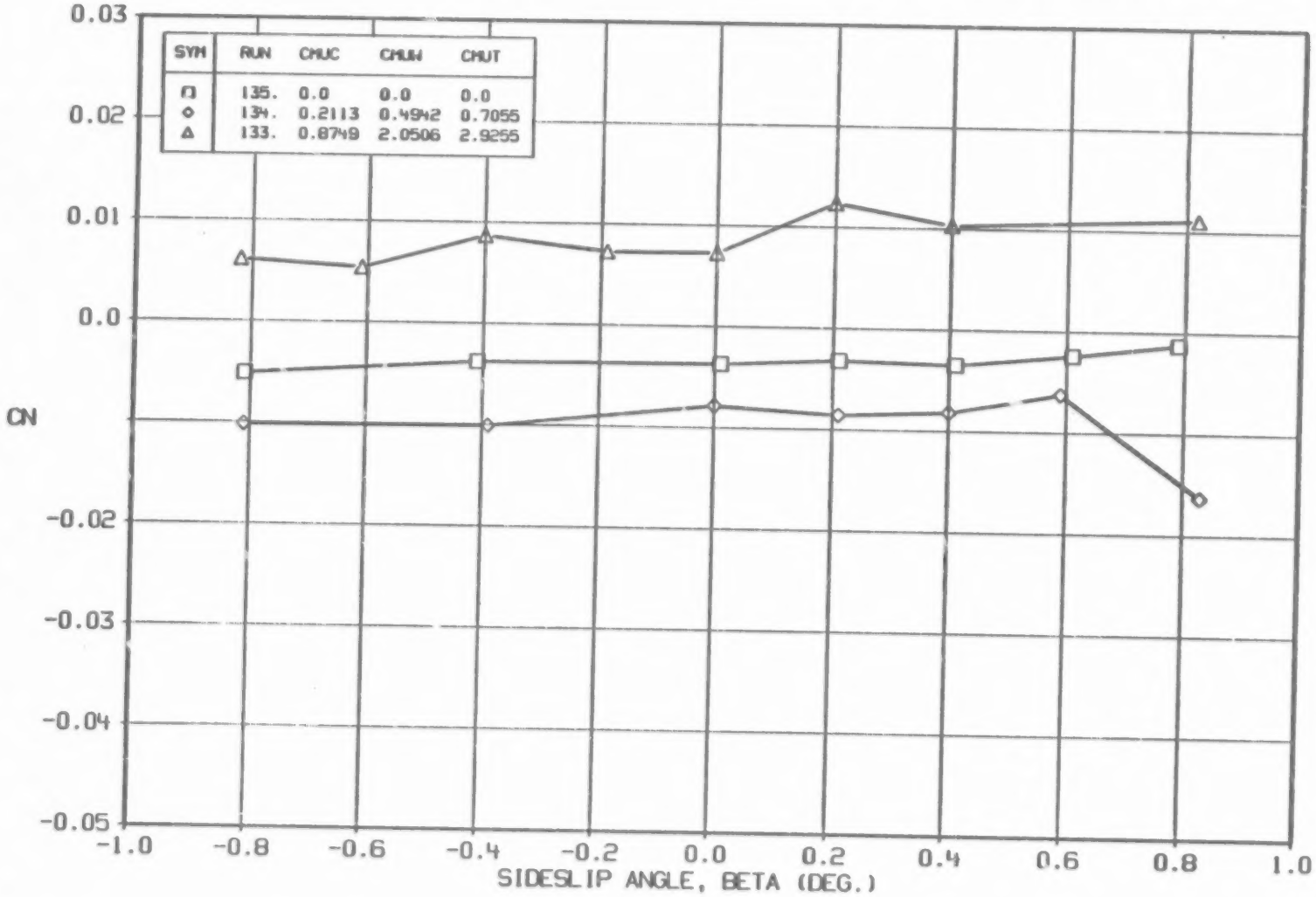

FIGURE A25c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=1.0, DELF=45, α = 2

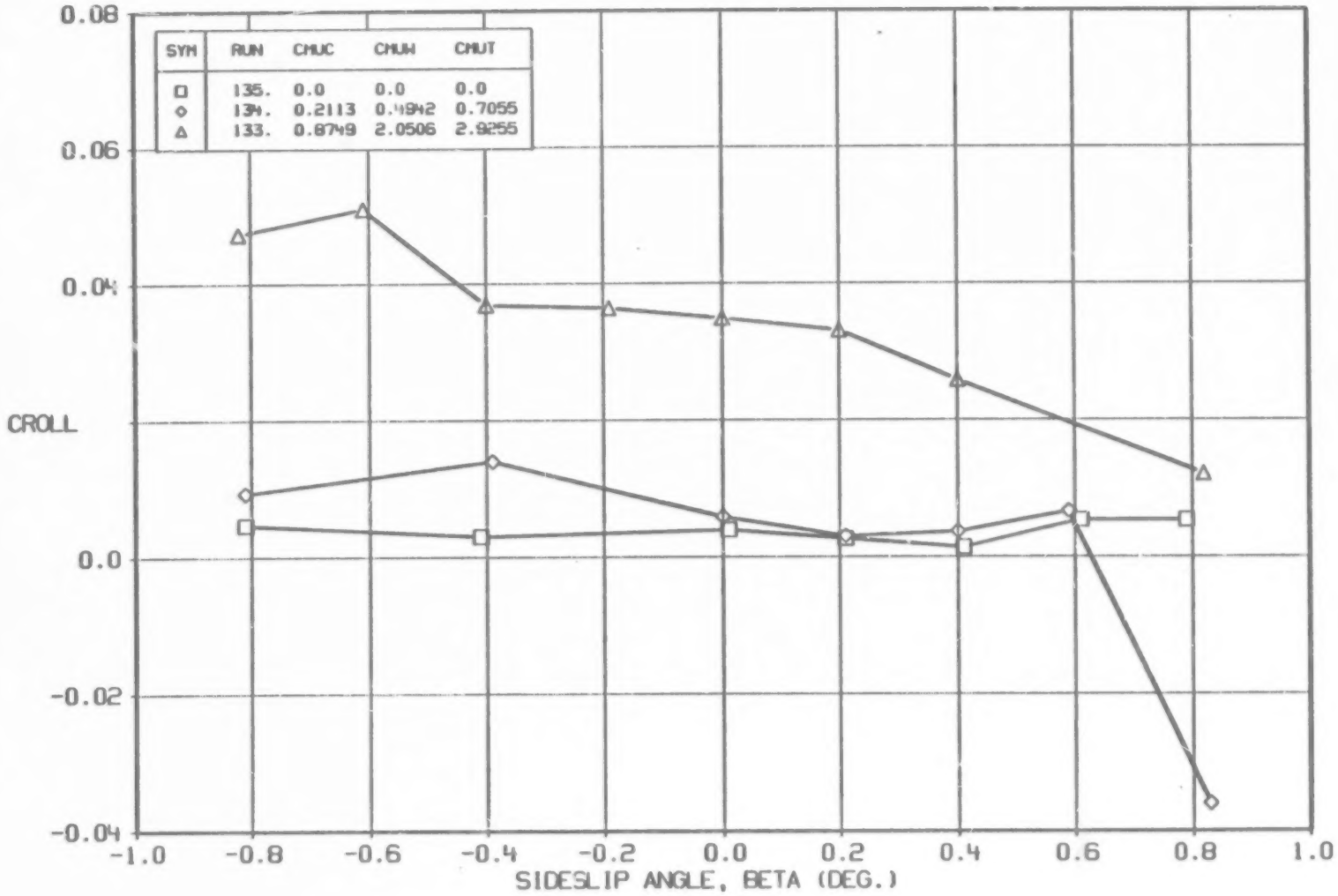

FIGURE A25f BASIC DATA EFFECT OF CMU
CONFIGURATION BWGV, BN/B=1.0, DELF=45, α = 2

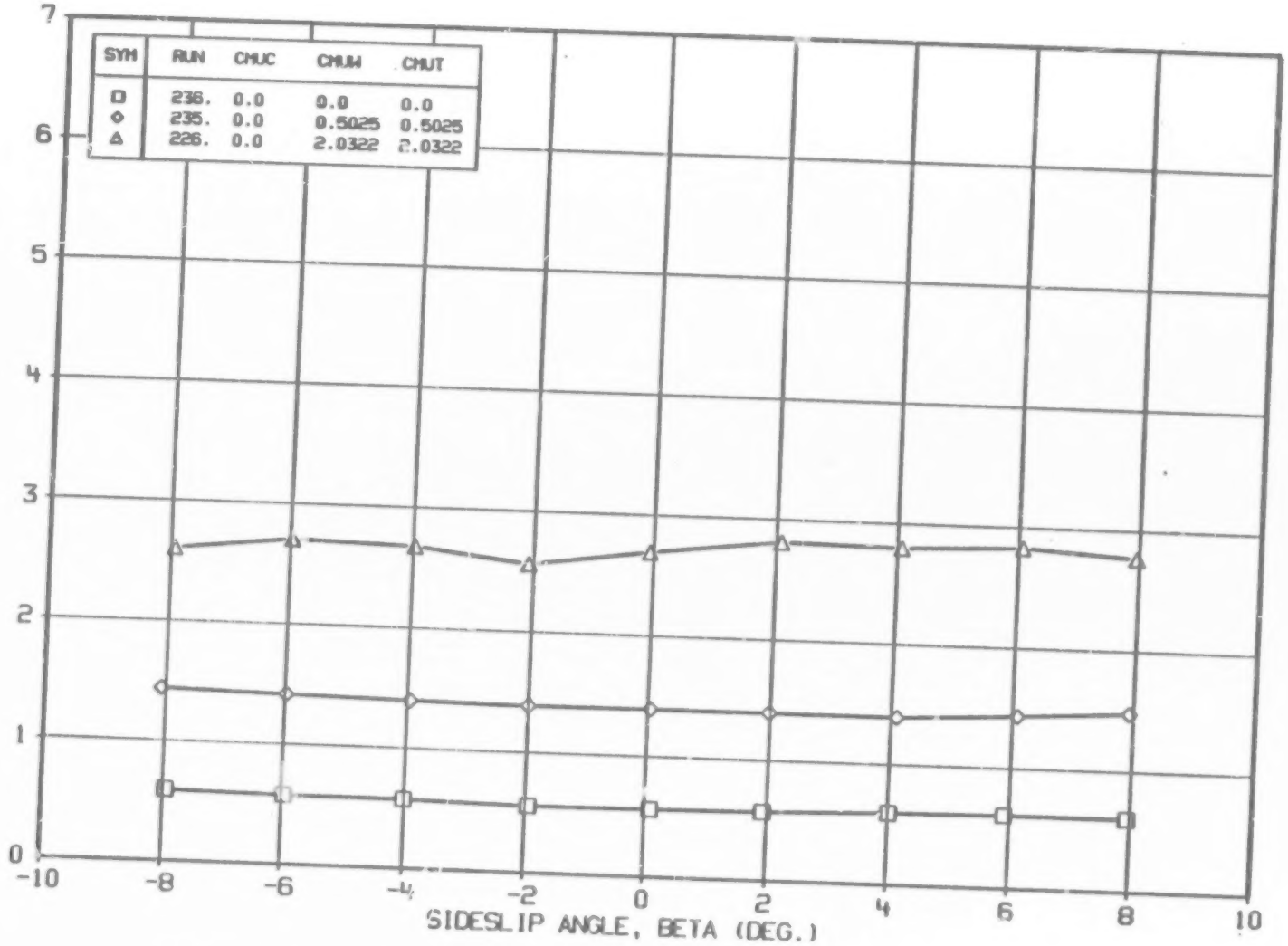

FIGURE A26a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

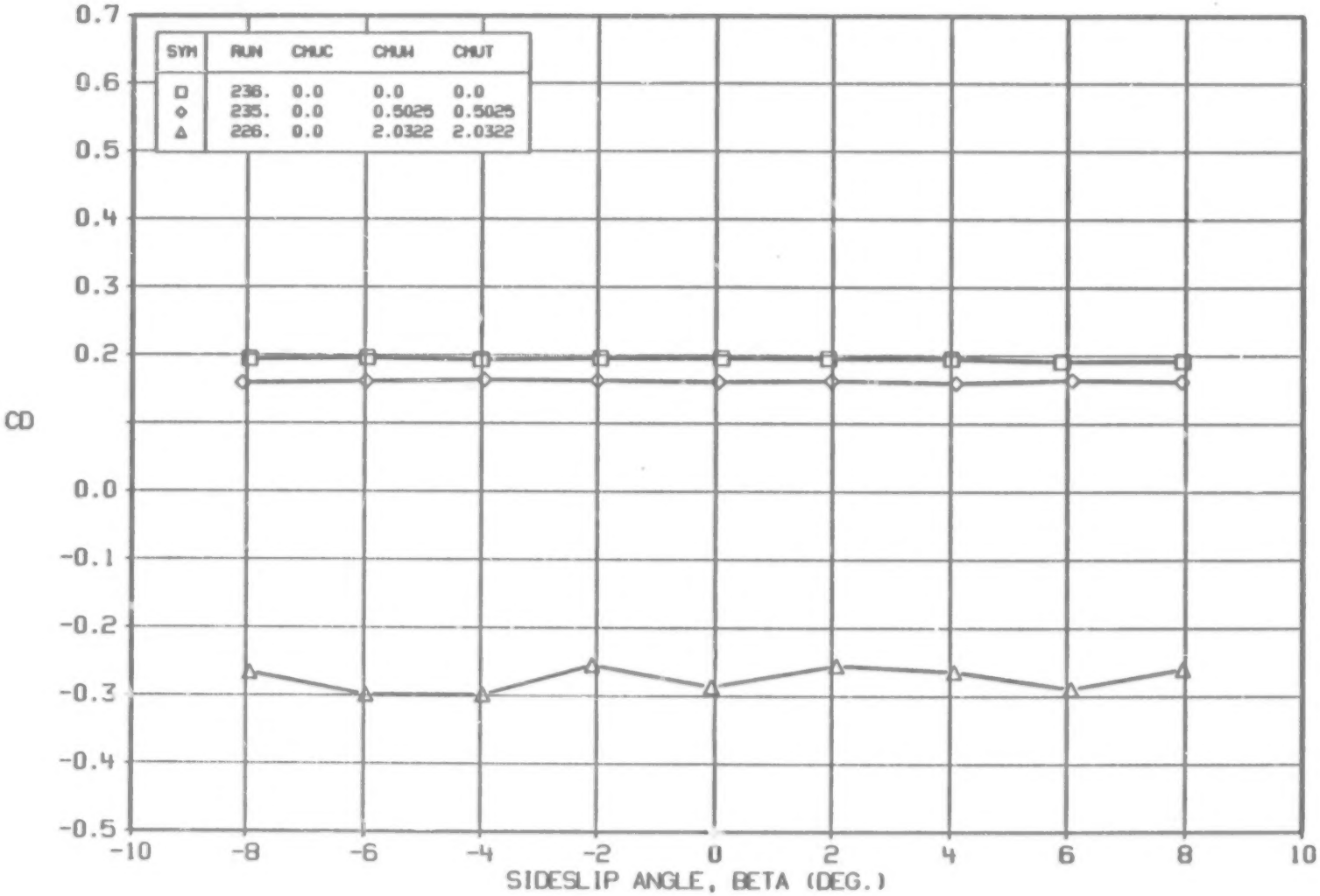

FIGURE A26b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

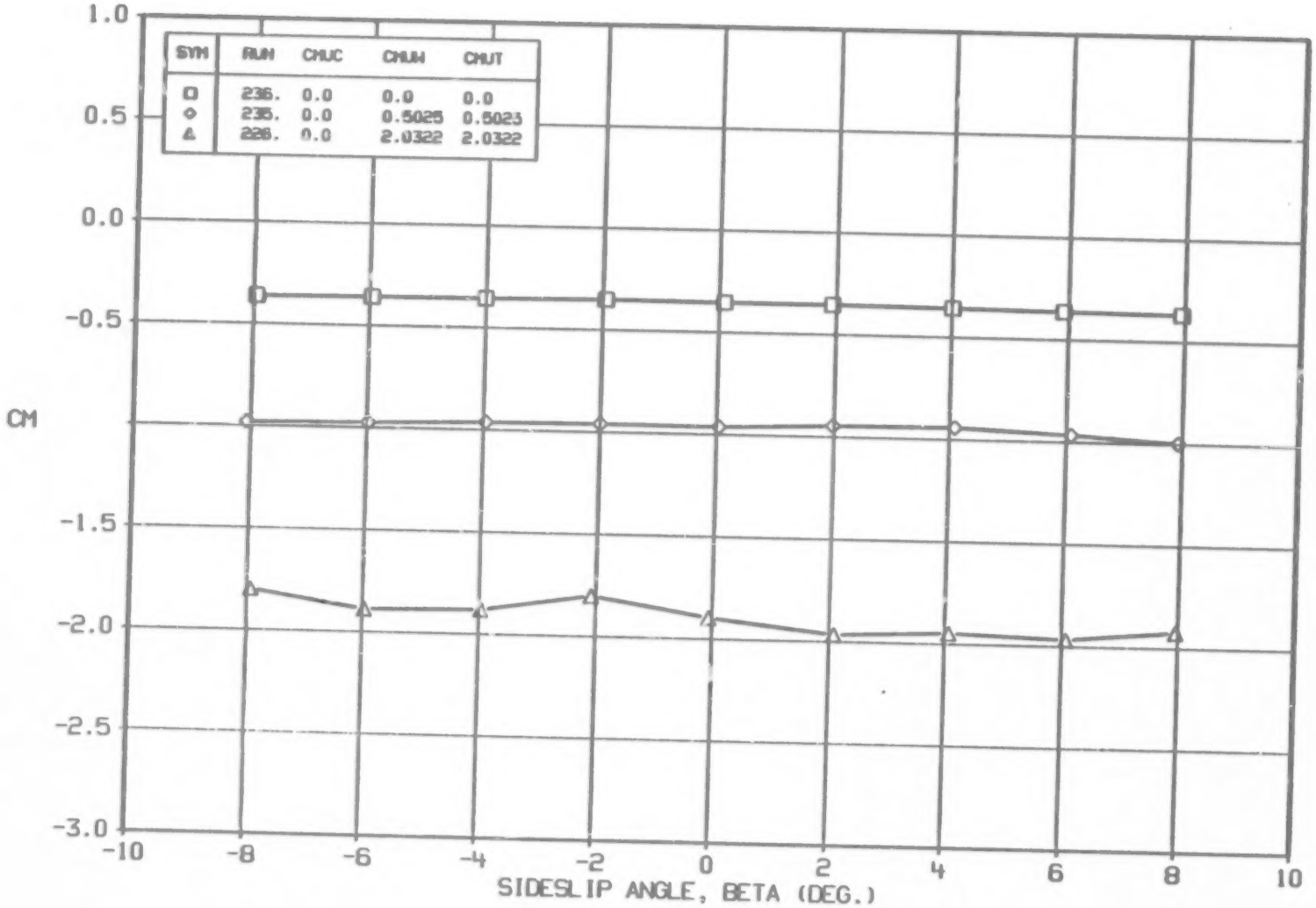

FIGURE A26c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

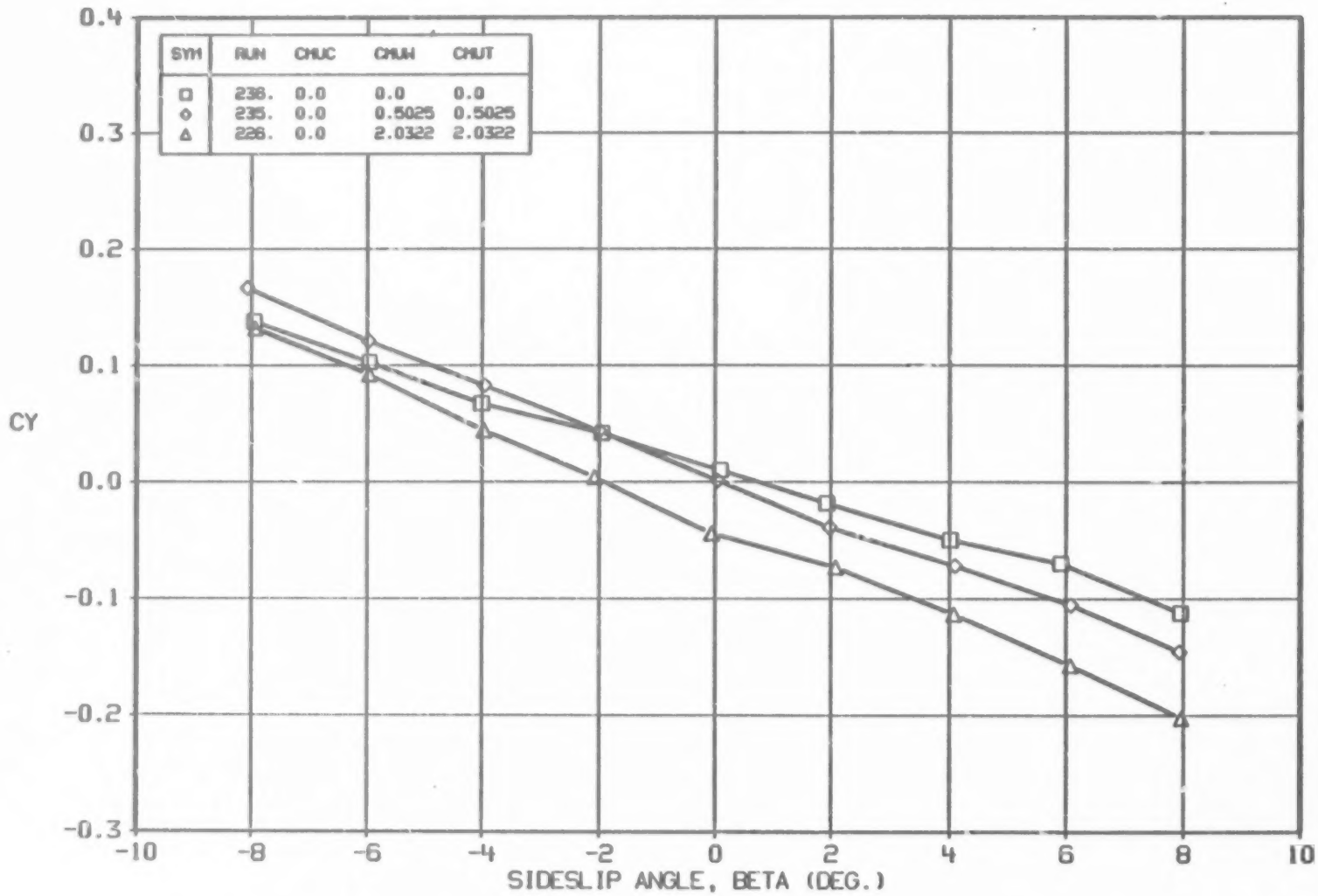

FIGURE A26d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

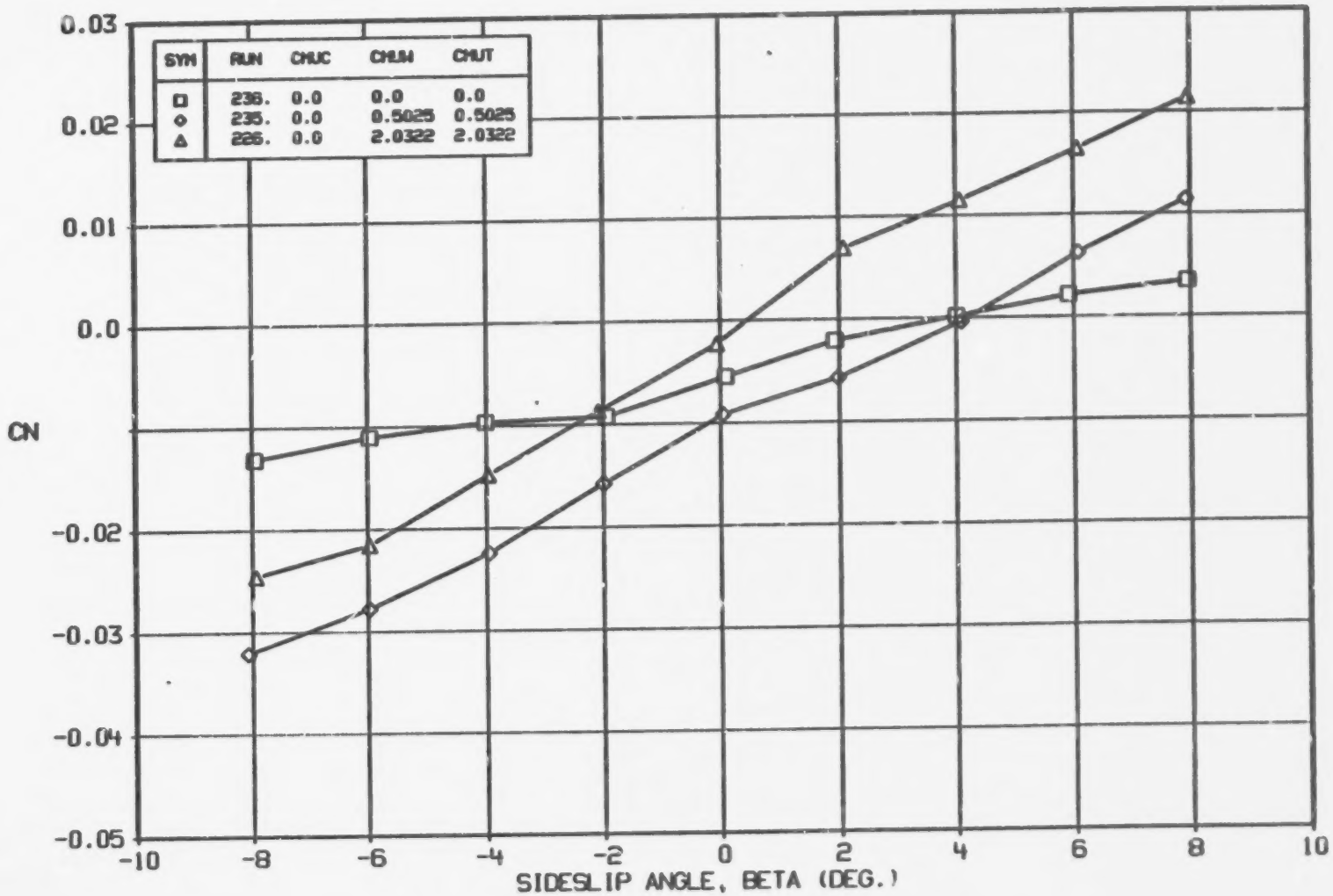

FIGURE A26e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

C-3

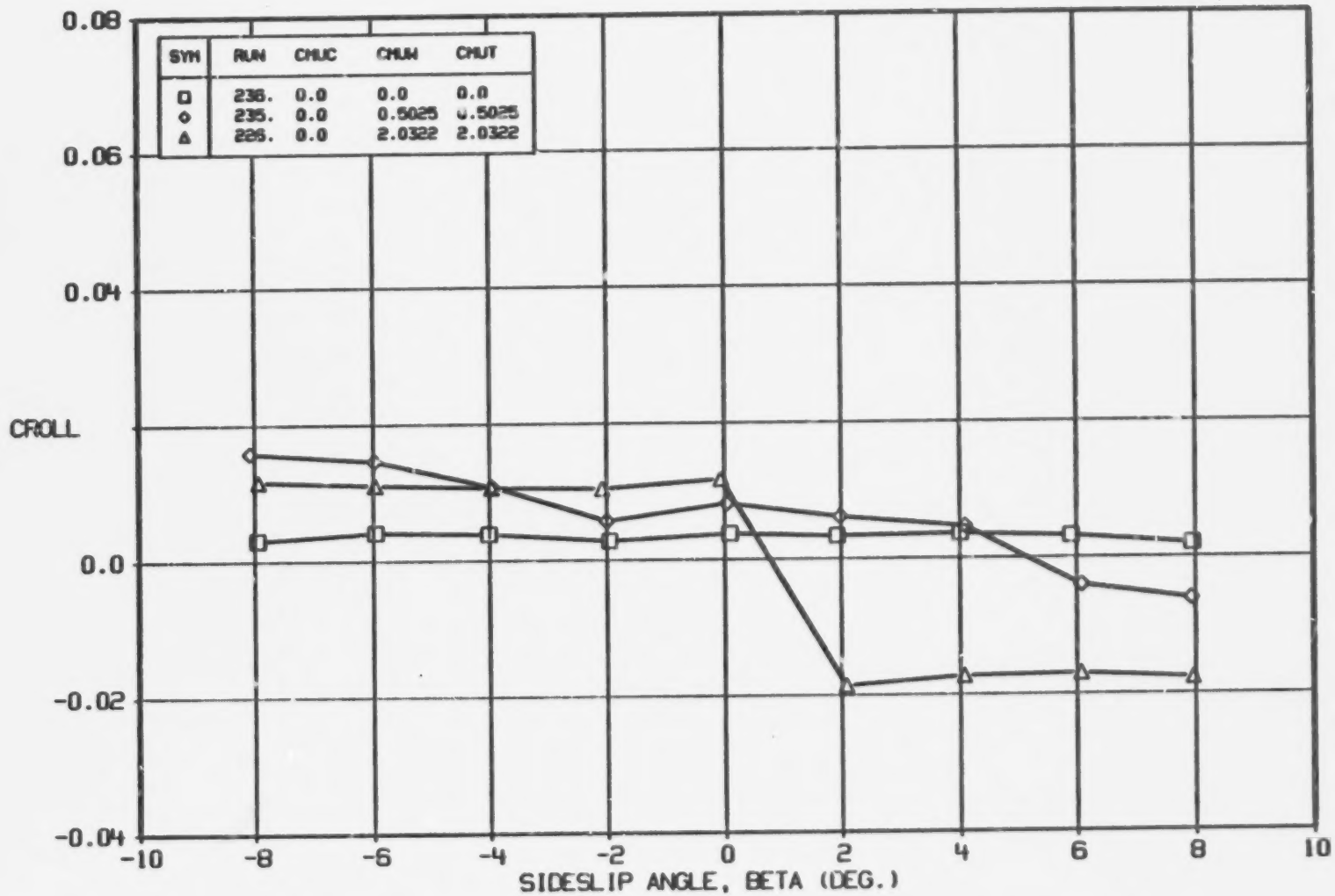

FIGURE A26f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5 DELF=45

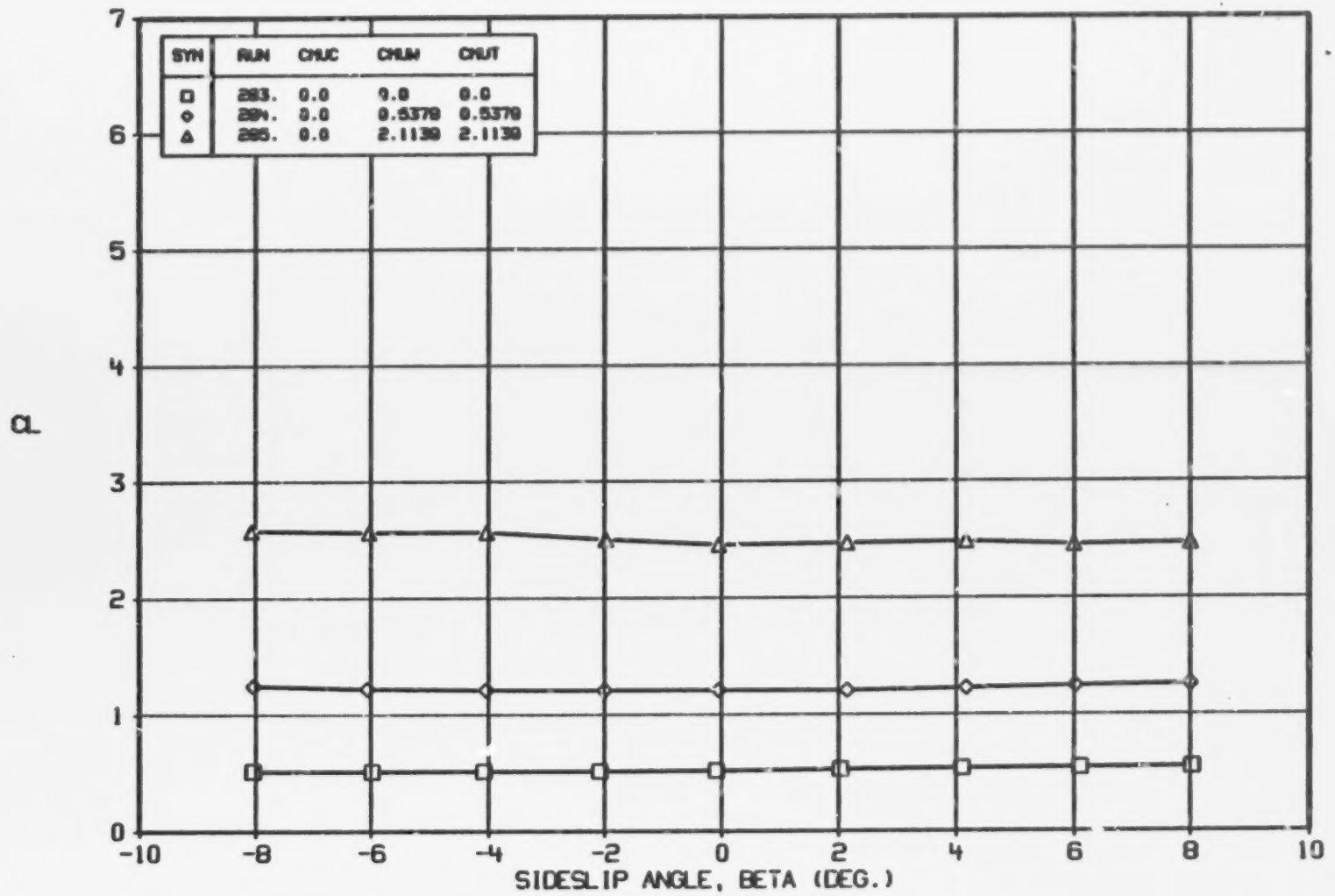

FIGURE A27a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

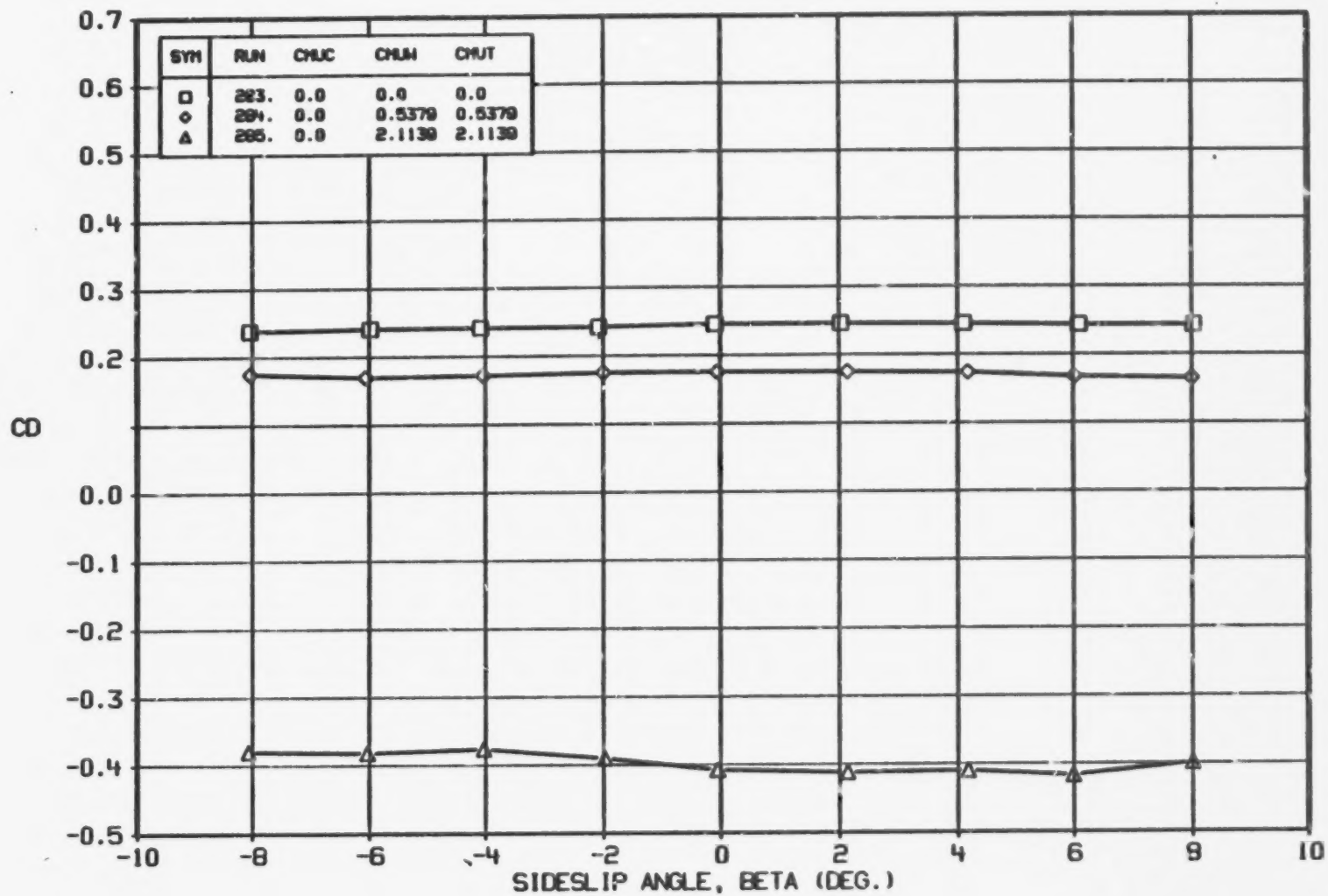

FIGURE A27b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

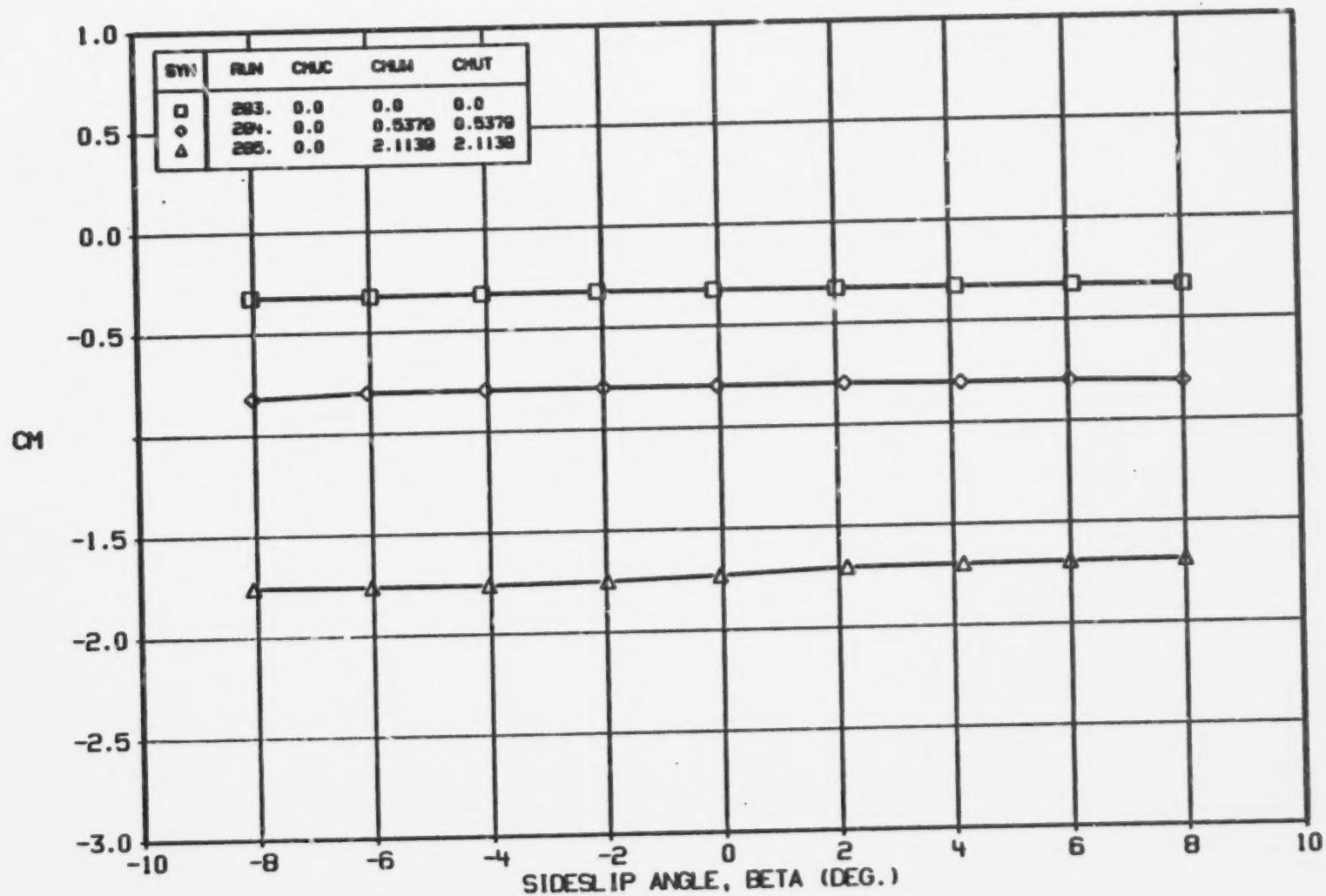

FIGURE A27c BASIC DATA EFFECT OF CMU
CONFIGURATION BH6V, DELF=. BN/B=.25

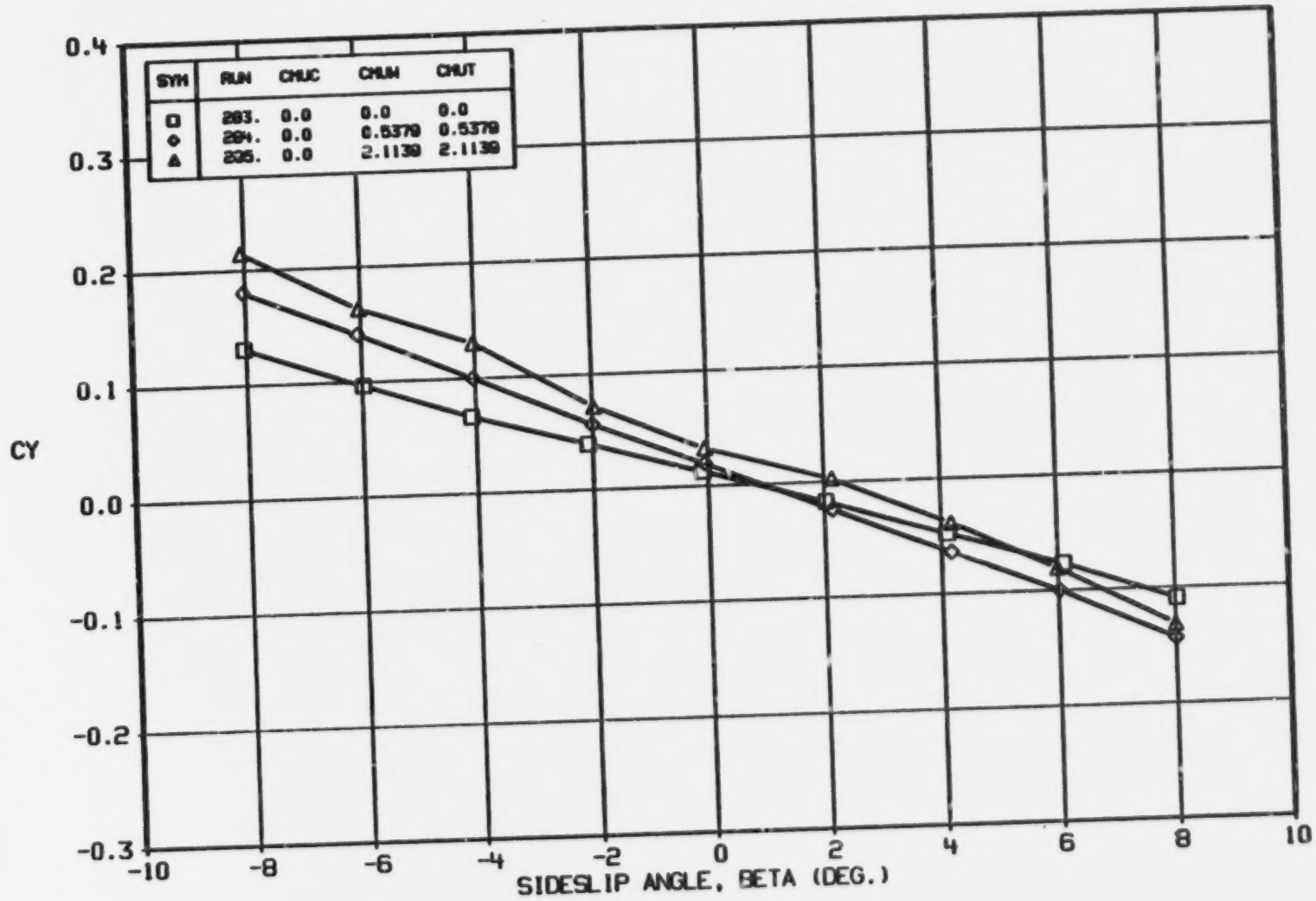

FIGURE A27d BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

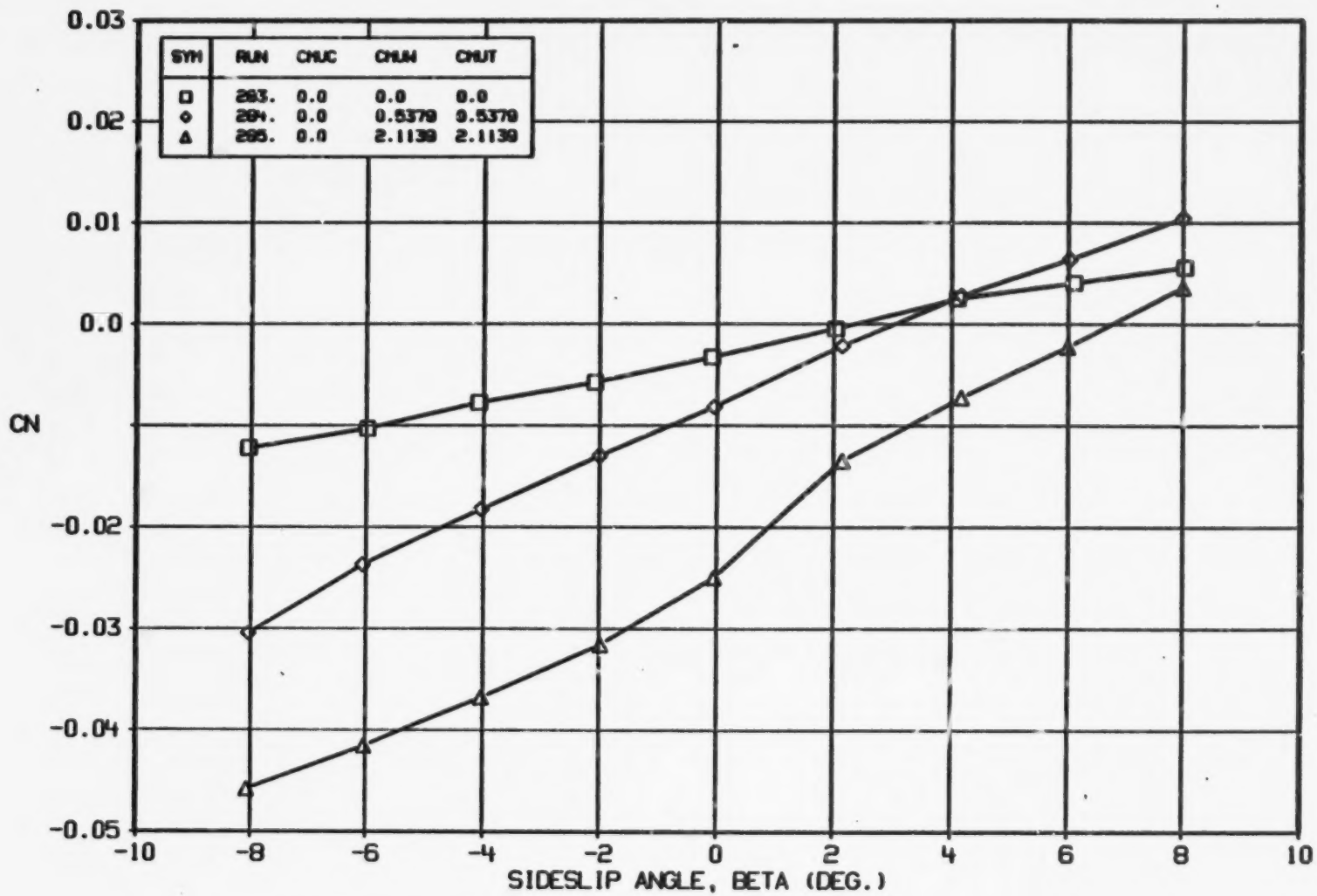

FIGURE A27e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

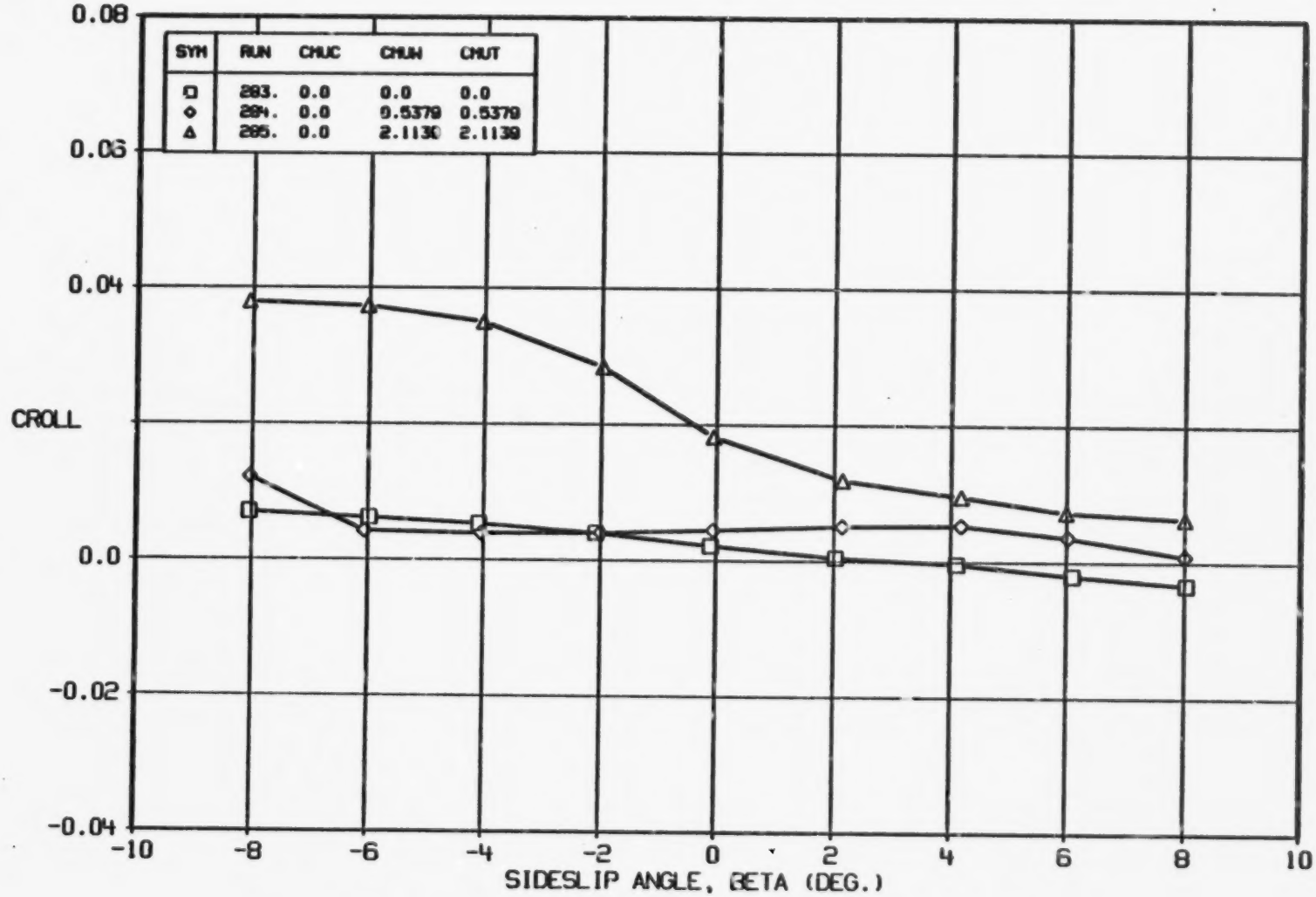

FIGURE A27f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45,BN/B=.25

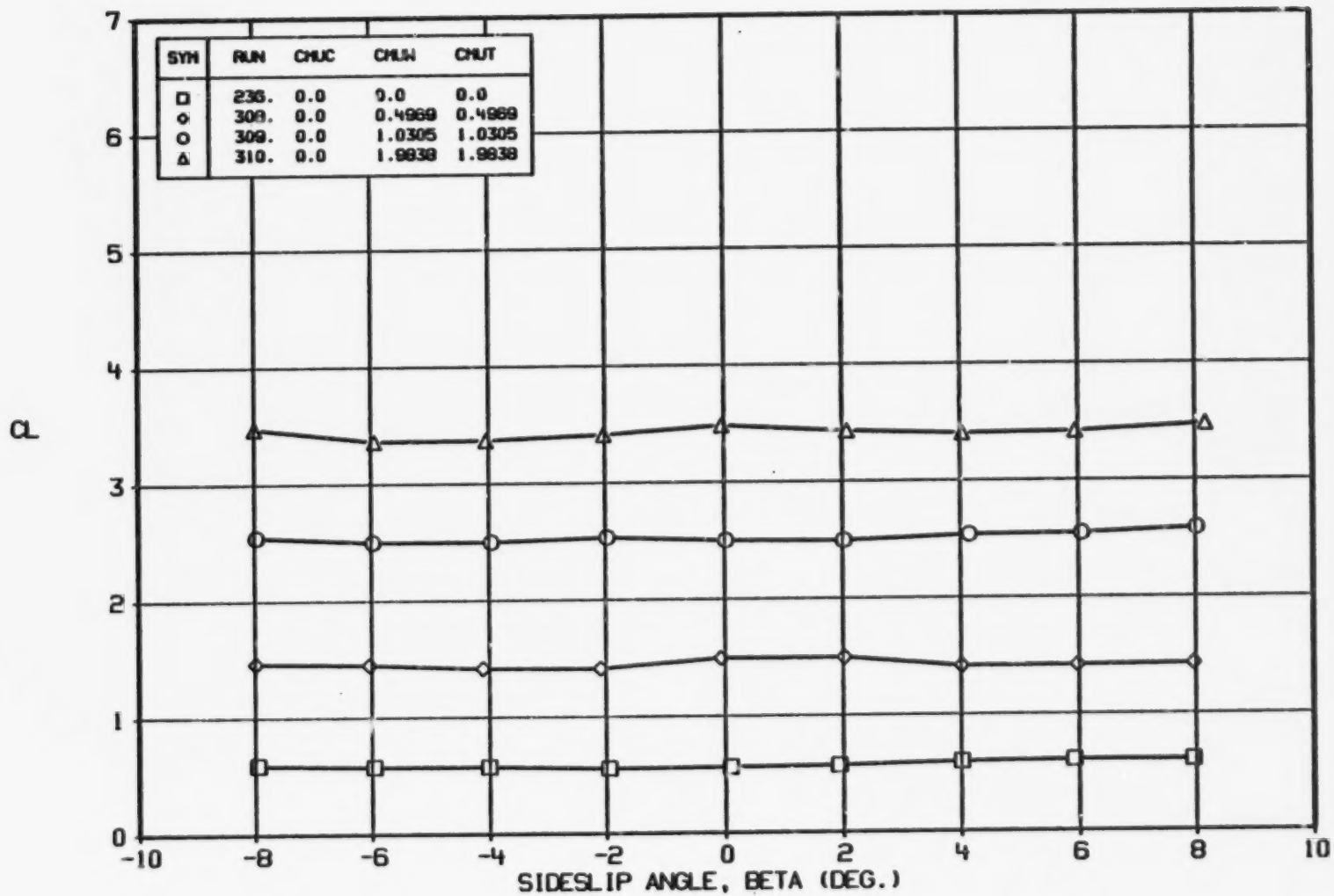

FIGURE A28a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

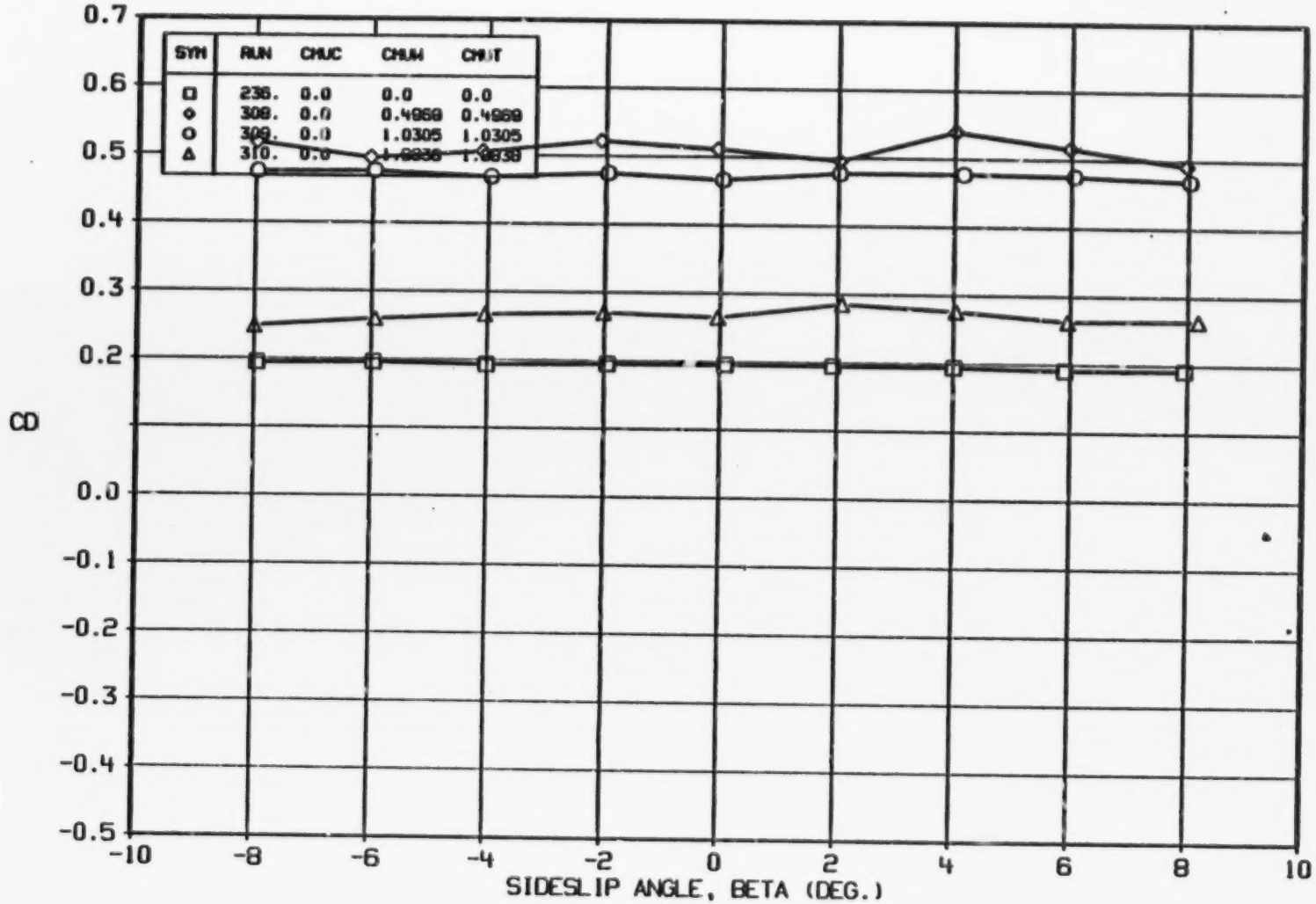

FIGURE A28b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

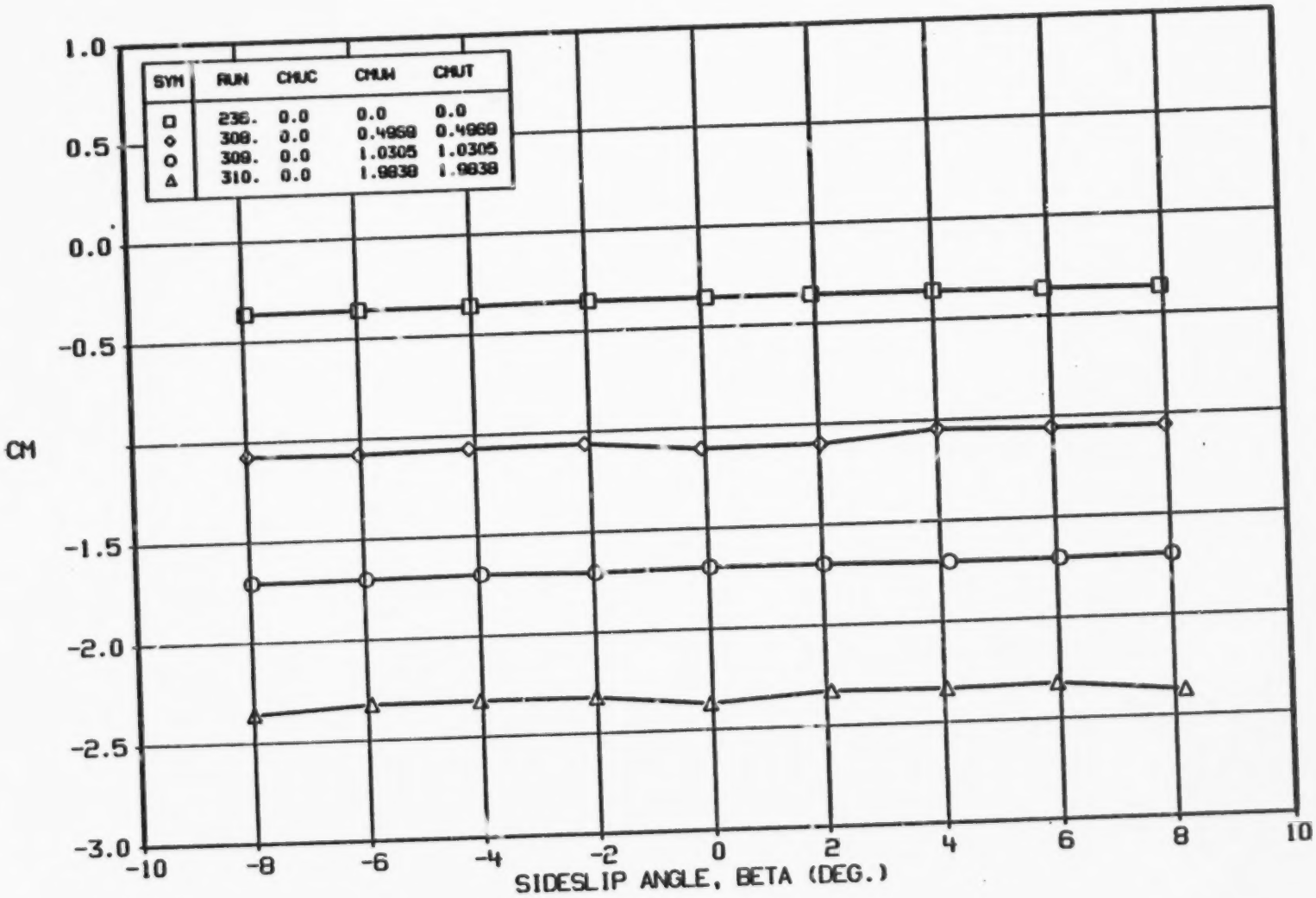

FIGURE A28c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

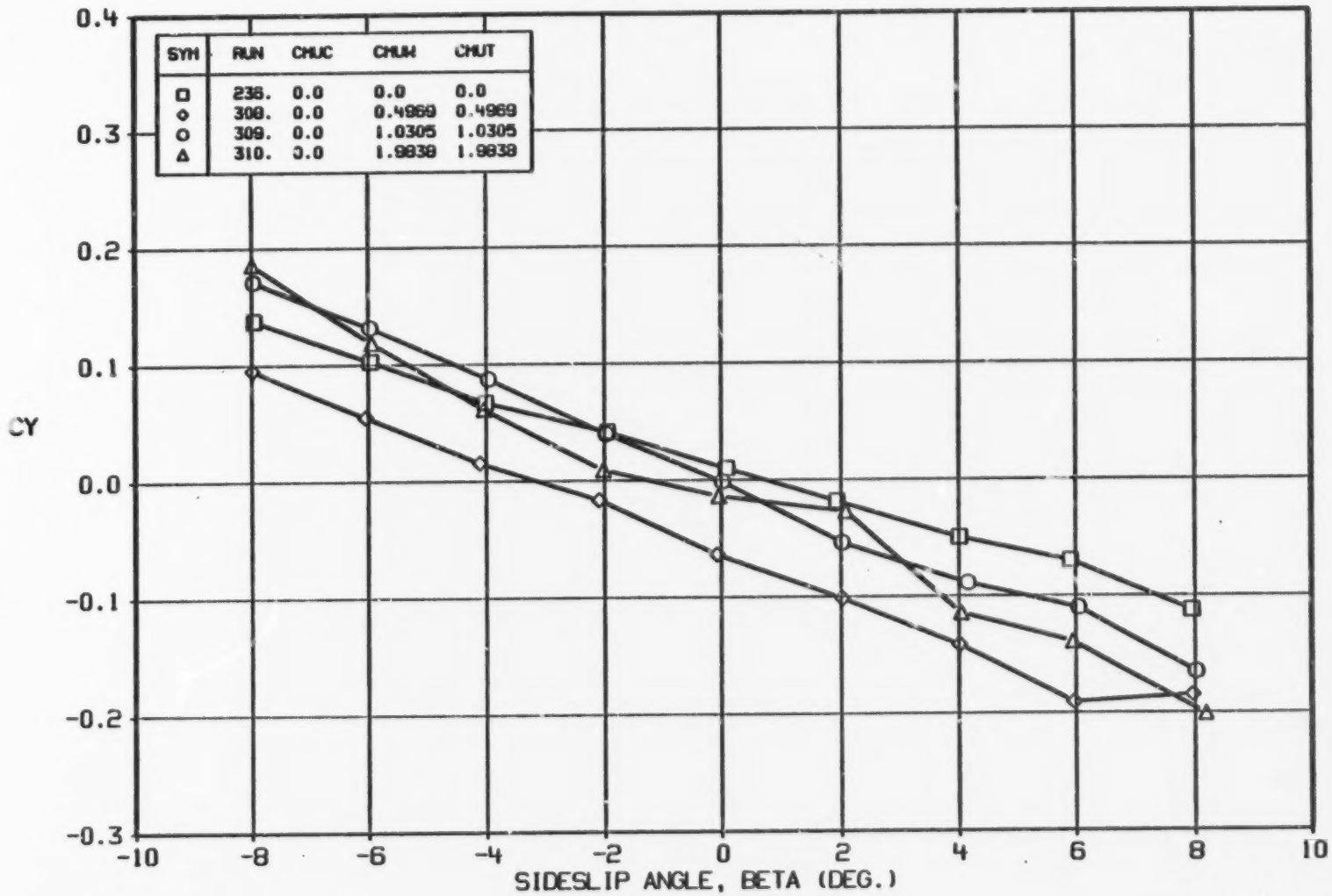

FIGURE A28d BASIC DATA EFFECT OF CMU
CONFIGURATION BN6V, DELF=45, BN/B=1

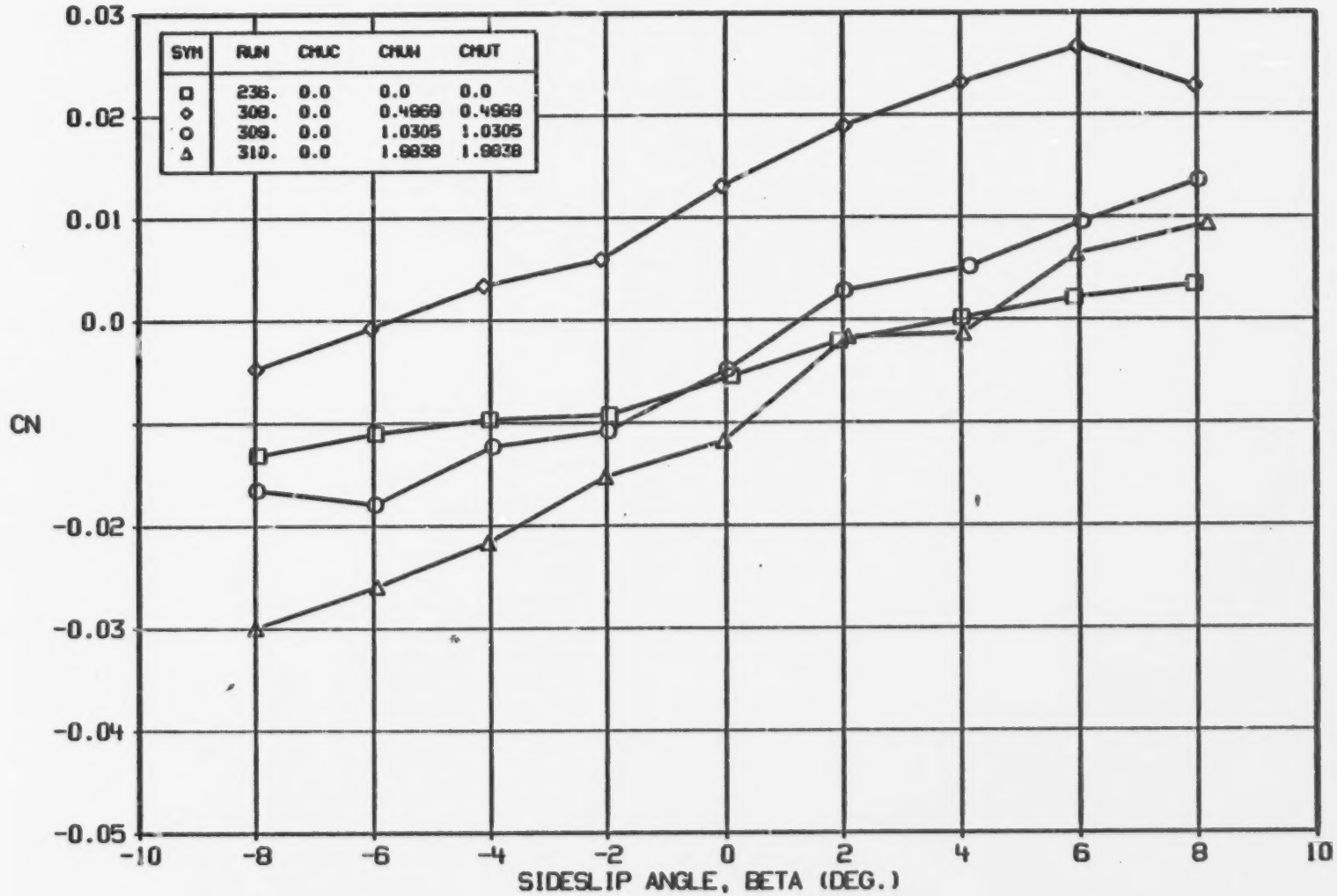

FIGURE A28e BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

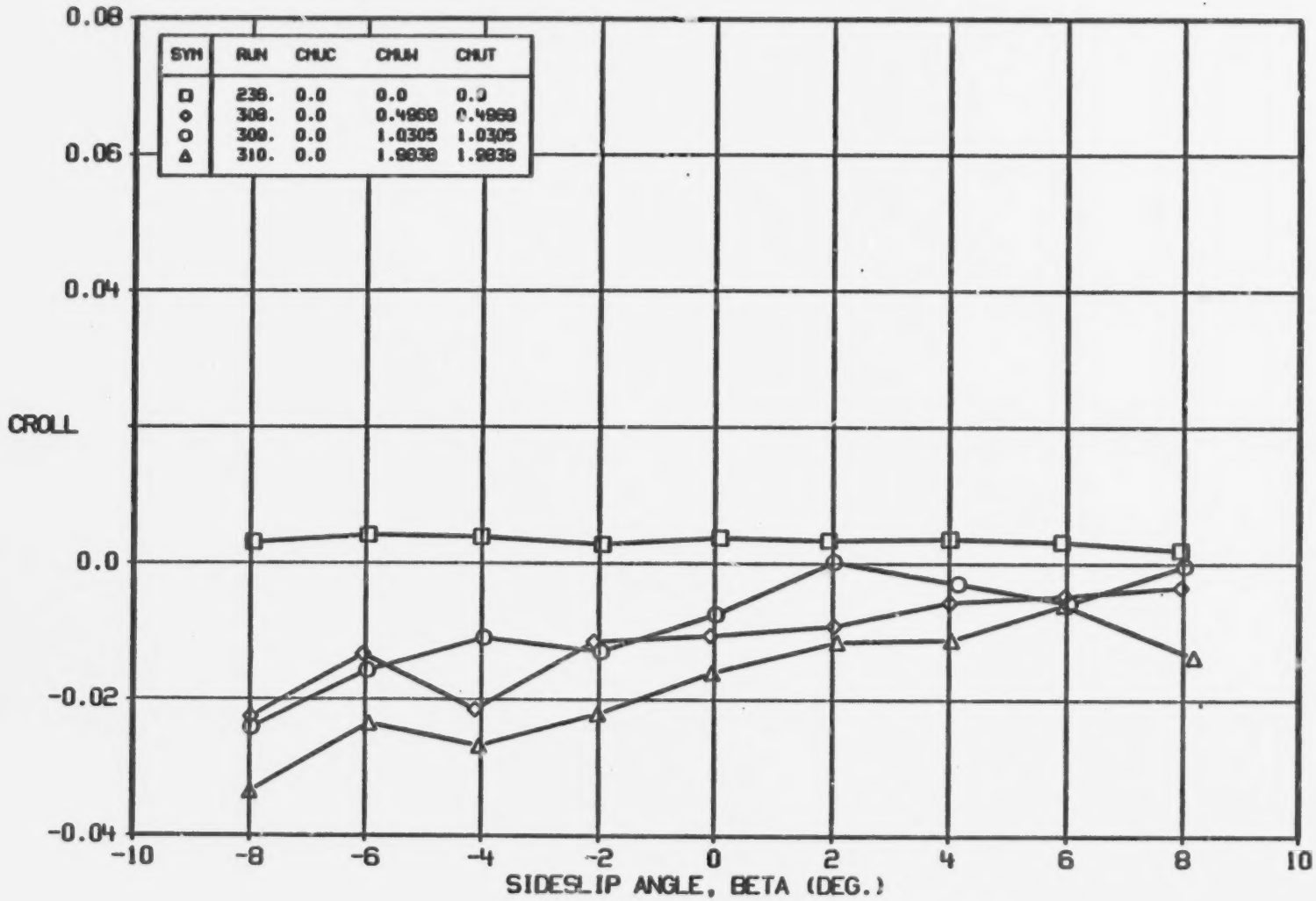

FIGURE A28f BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

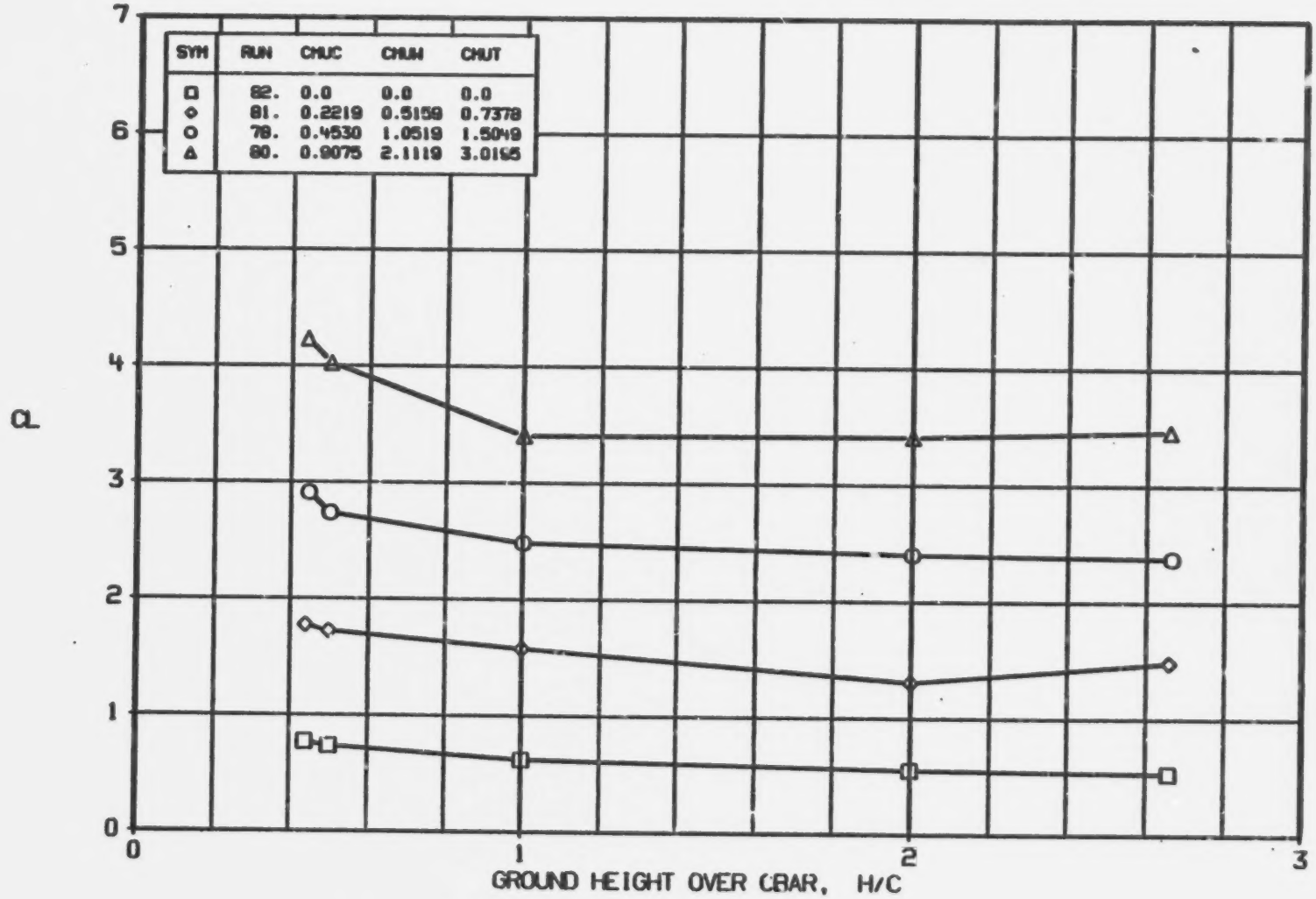

FIGURE A29a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BW/B=1

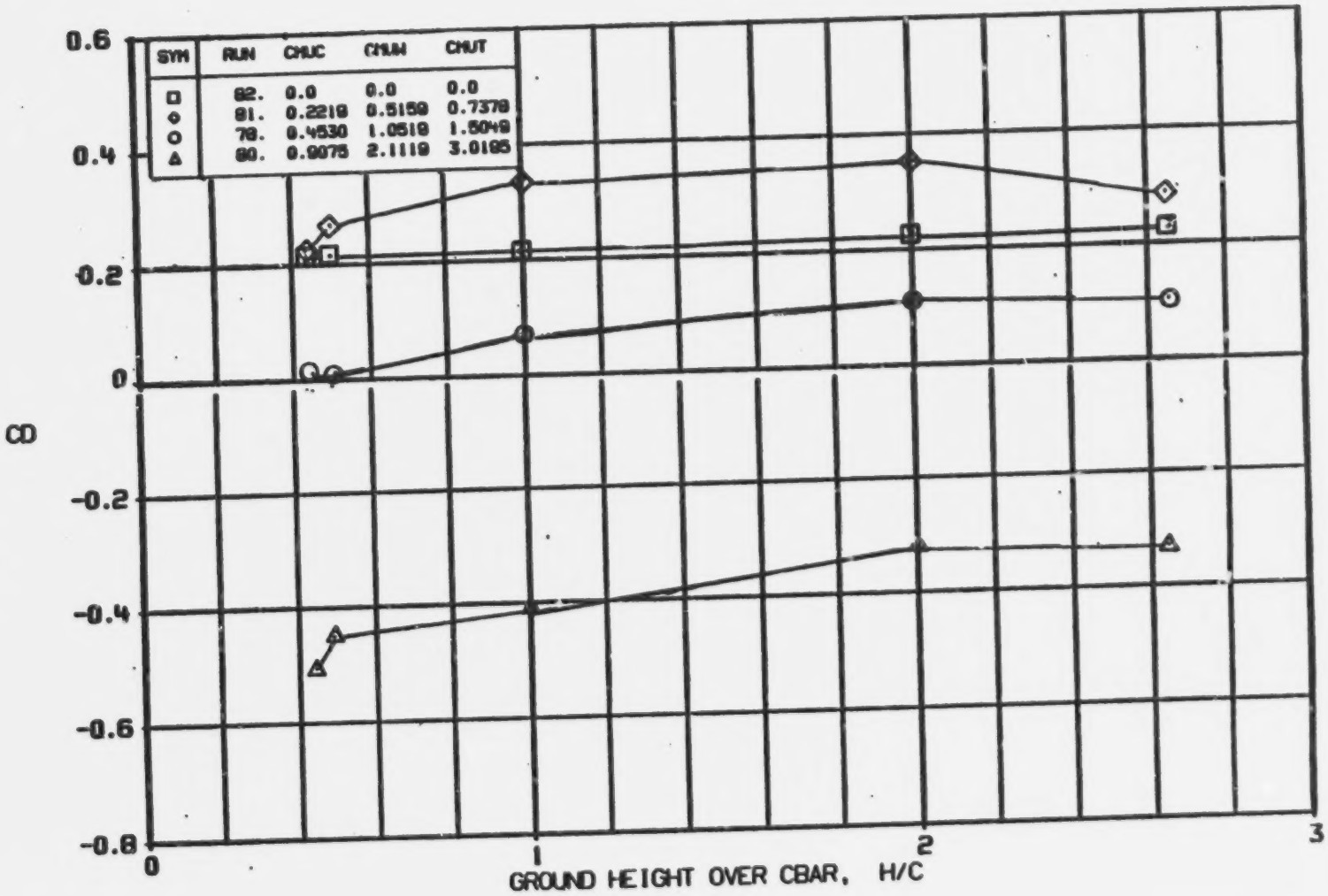

FIGURE A29b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1

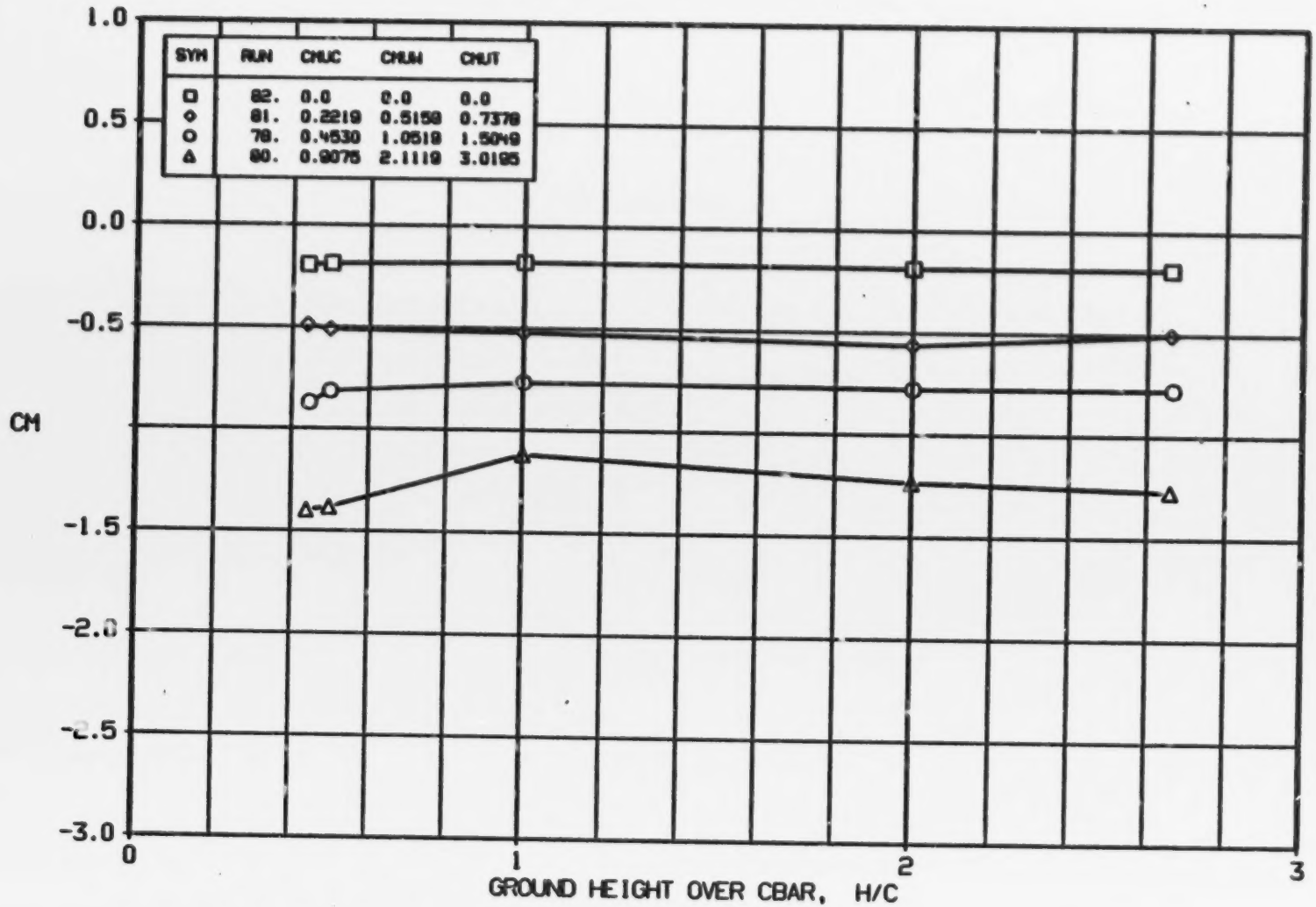

FIGURE A29c BASIC DATA EFFECT OF CMU
CONFIGURATION BCIW6V, DELF=45, BN/B=1

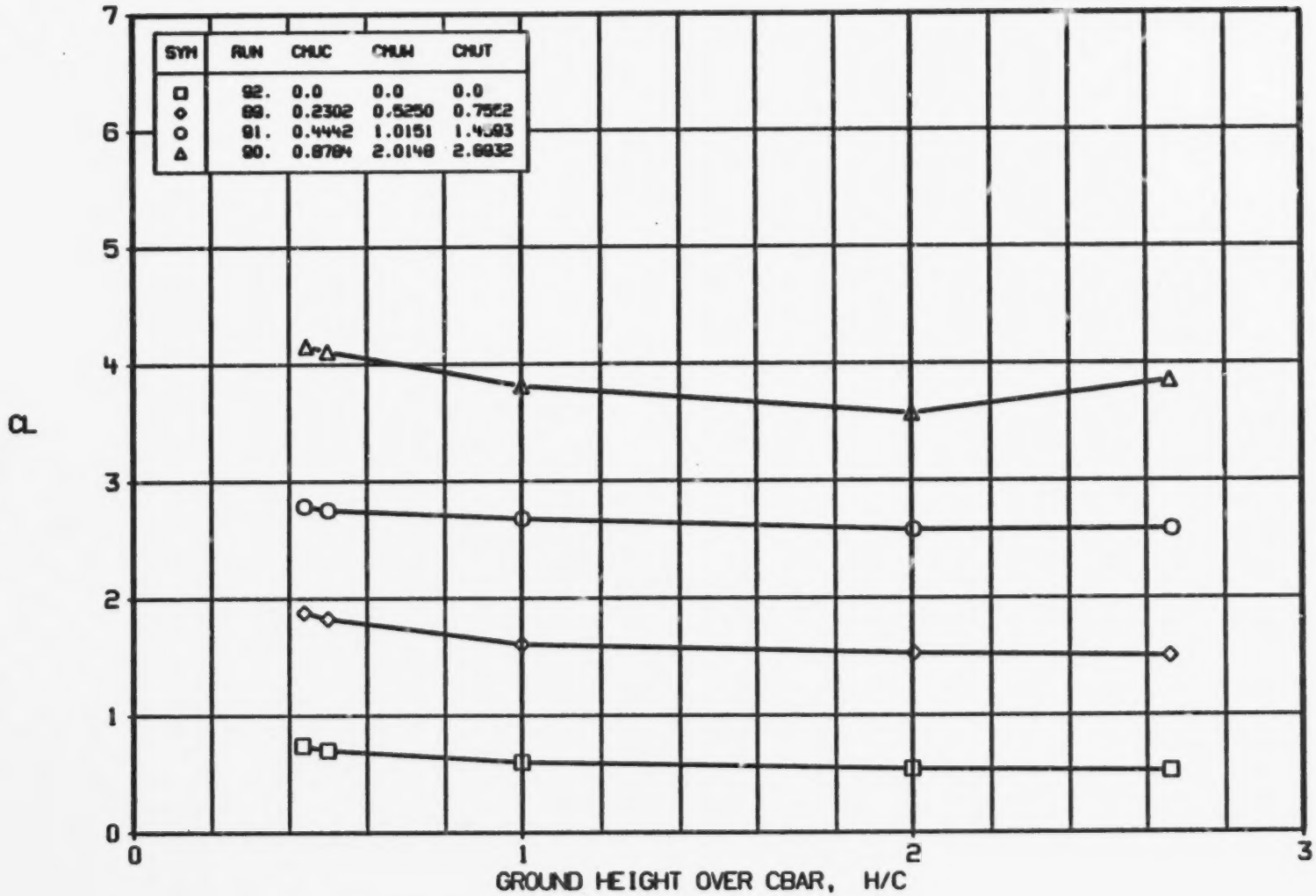

FIGURE A30a BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

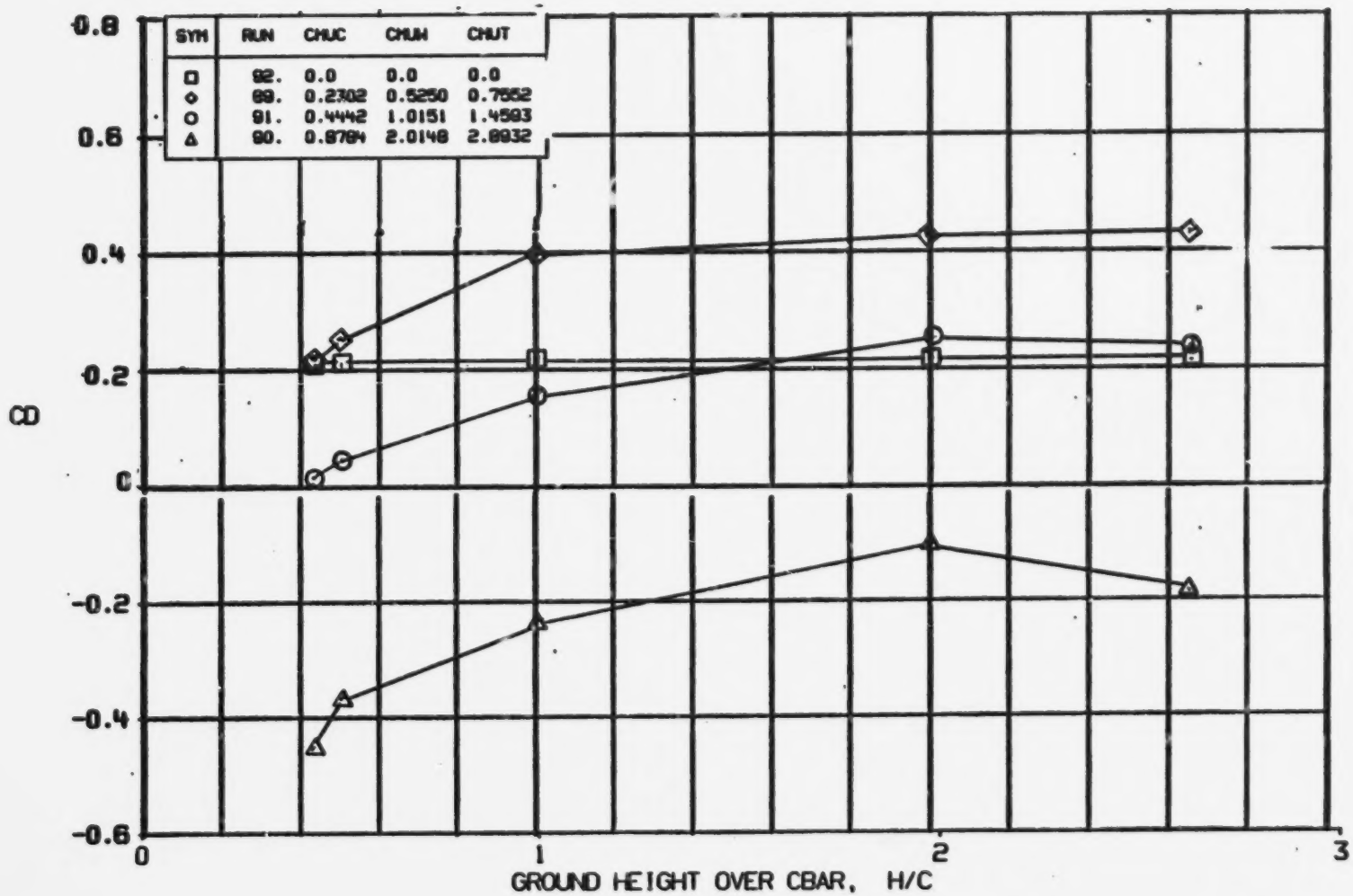

FIGURE A30b BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BN/B=1

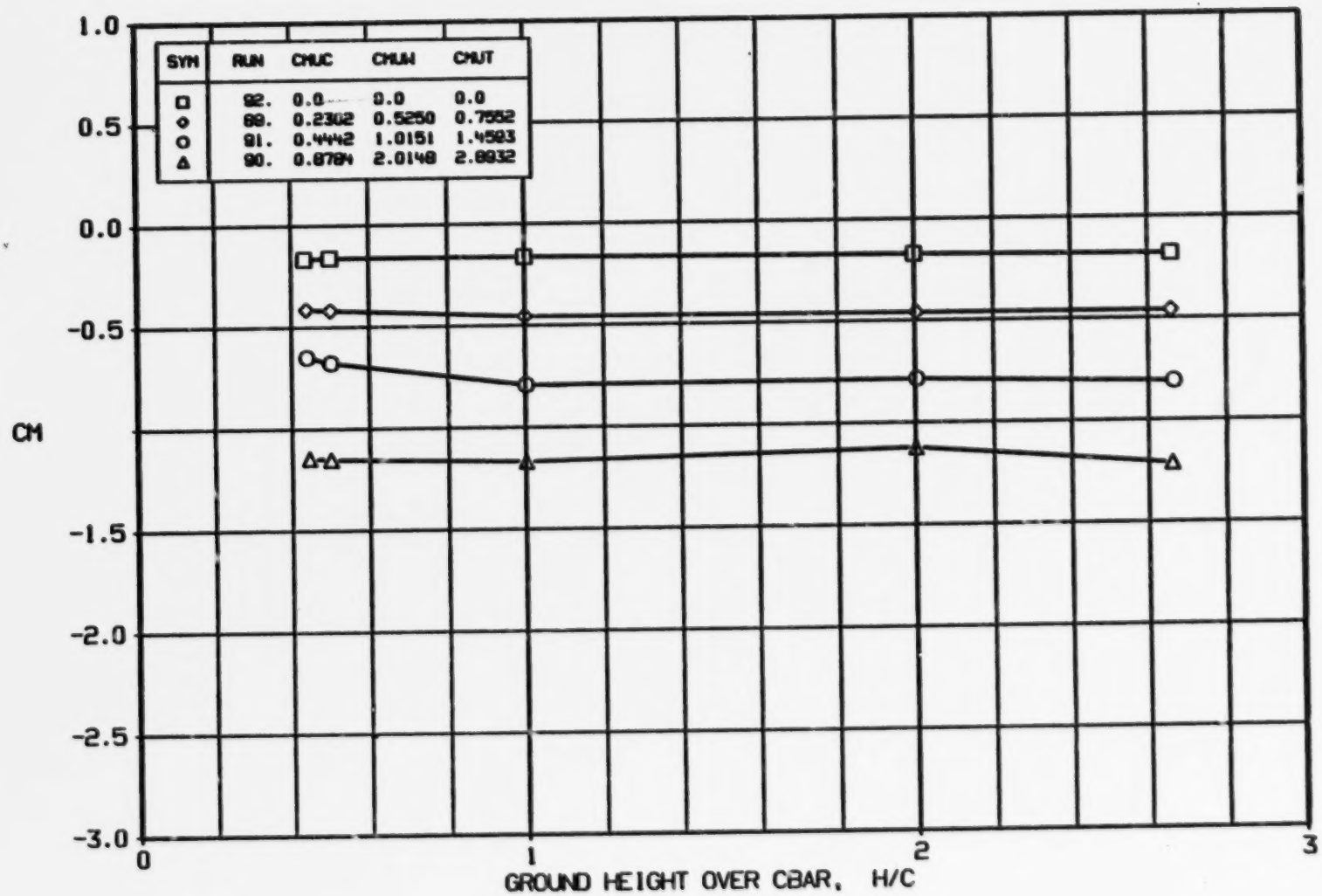

FIGURE A30c BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W6V, DELF=45, BW/B=1.

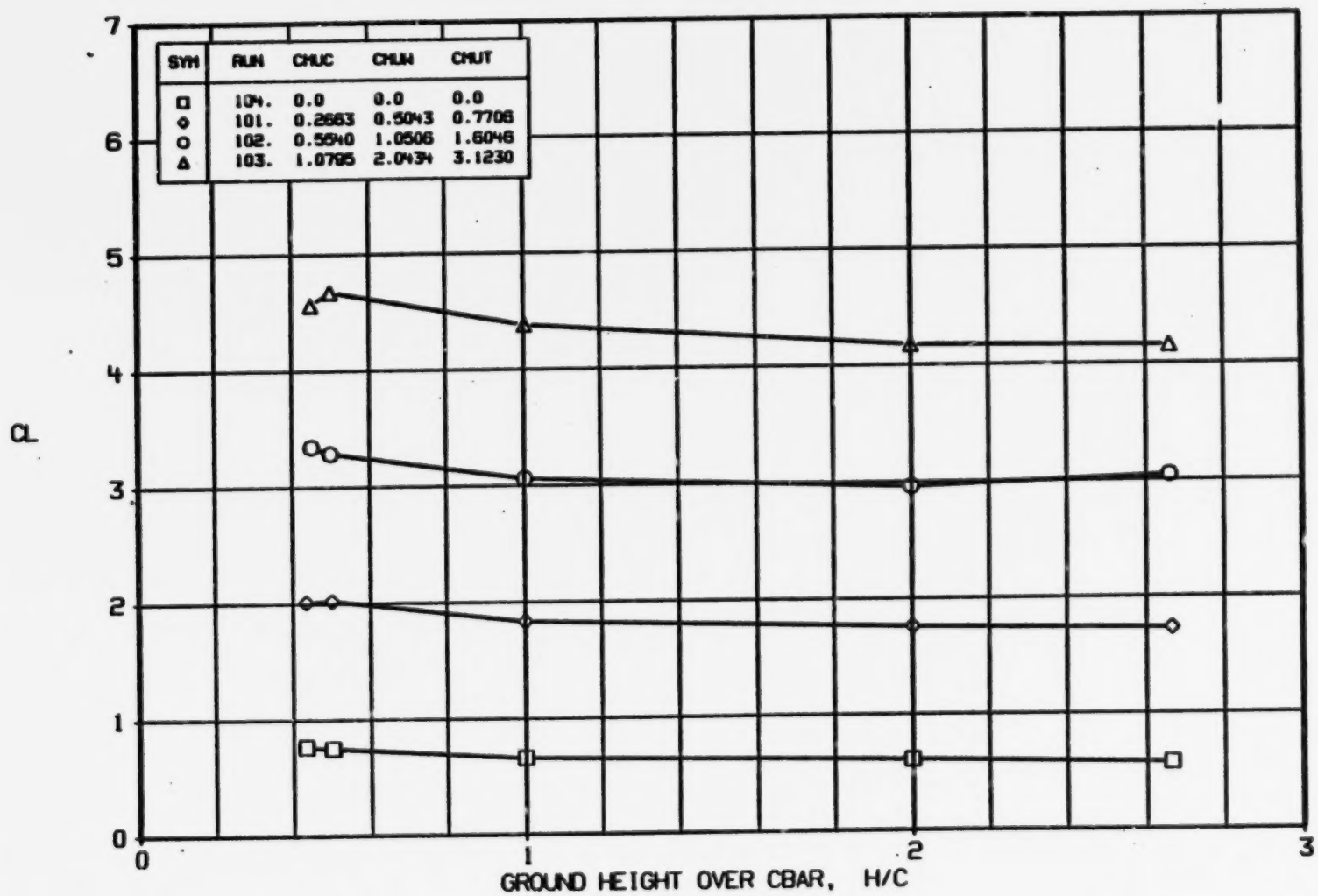

FIGURE A31a BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BW/B=1

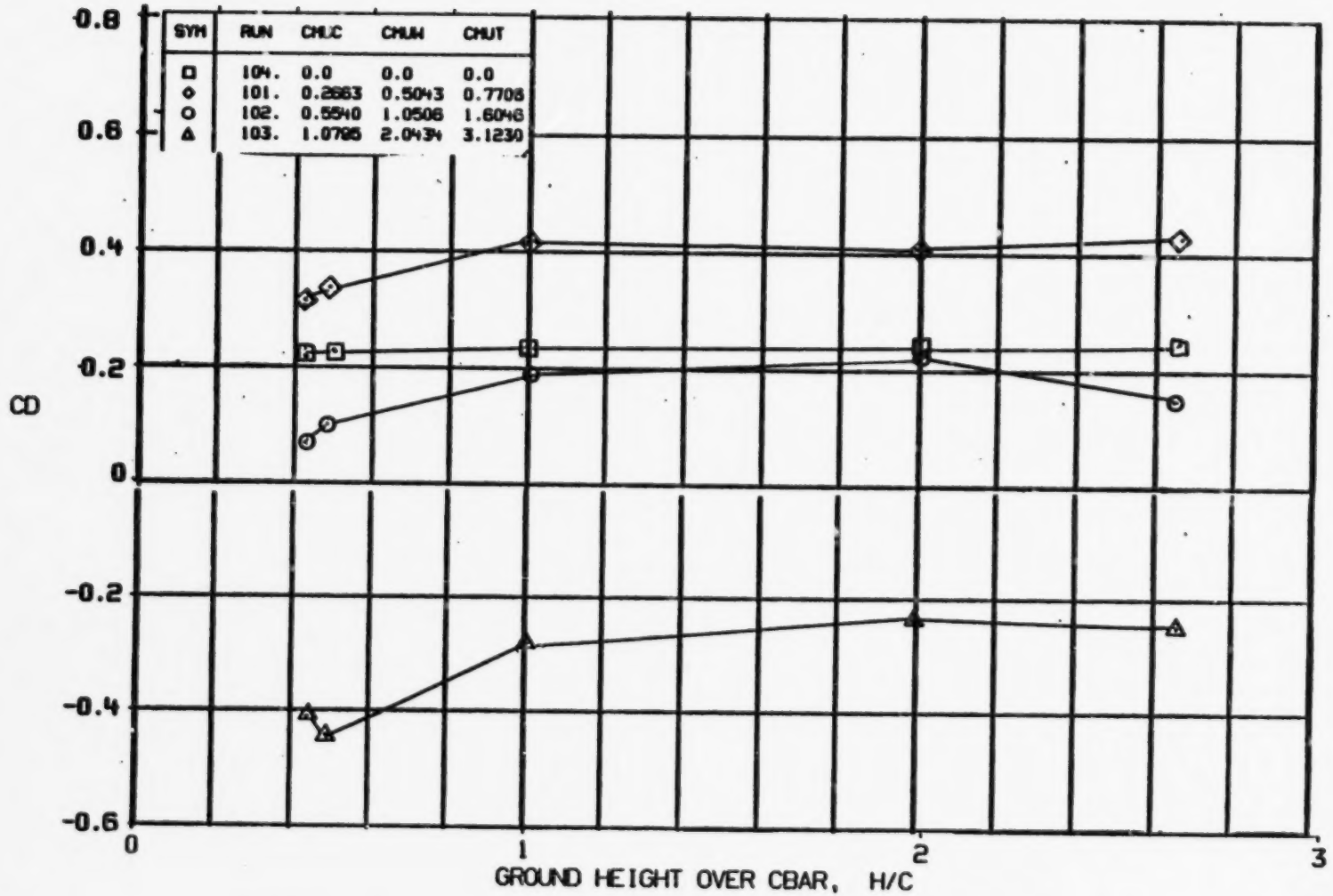

FIGURE A31b BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BN/B=1

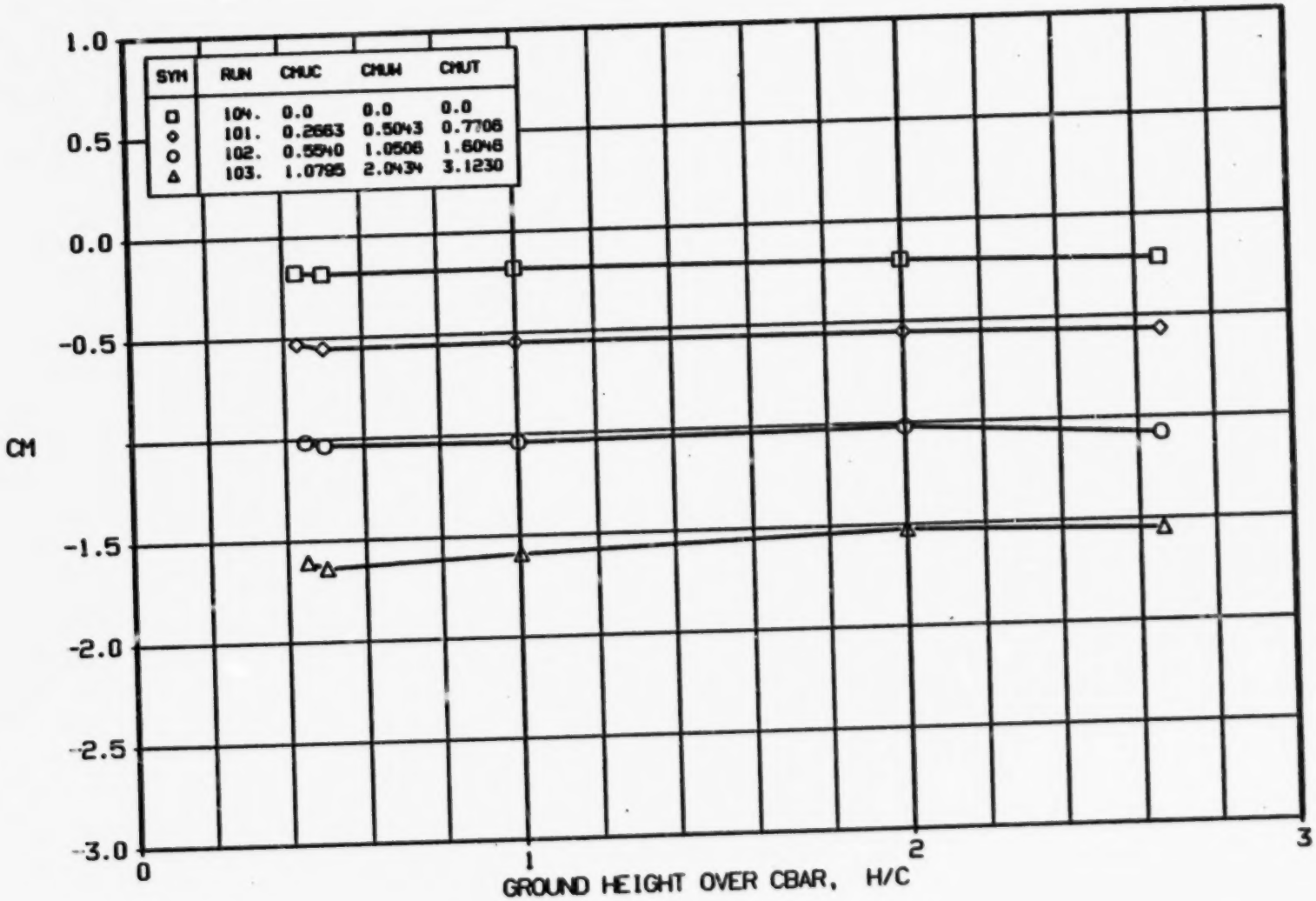

FIGURE A31c BASIC DATA EFFECT OF CMU
CONFIGURATION BC2W5V, DELF=45, BW/B=1.

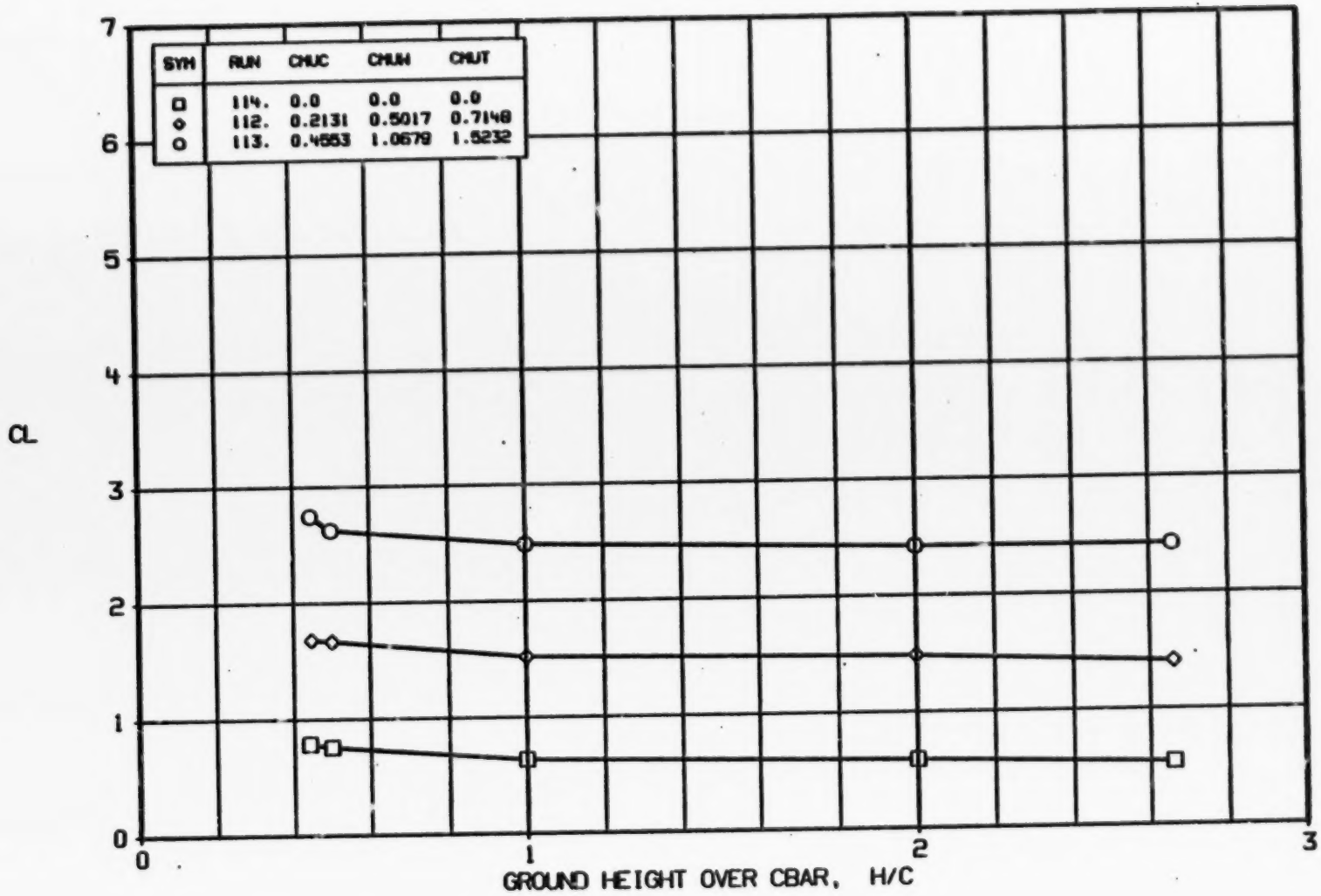

FIGURE A32a BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

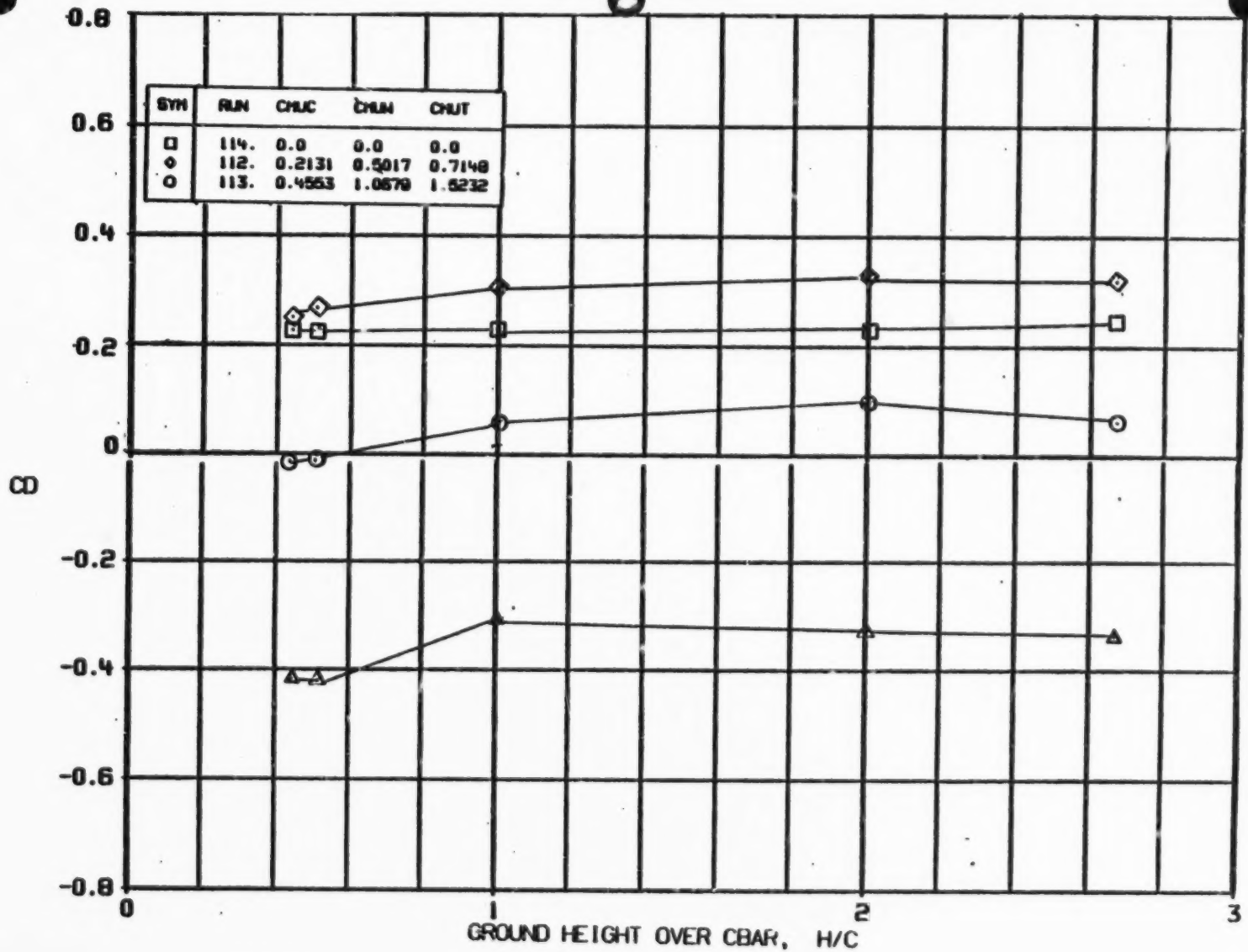

FIGURE A32b BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

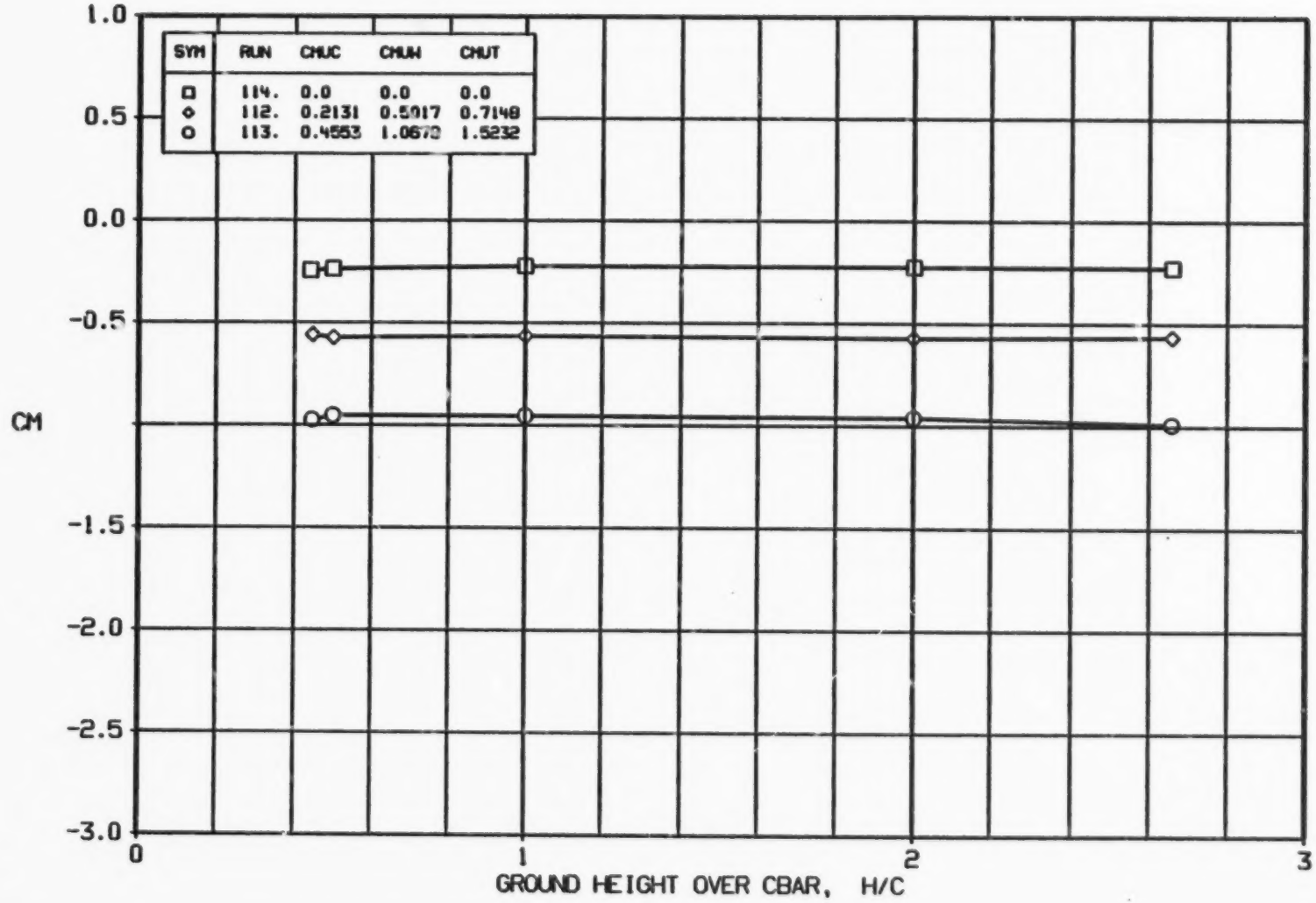

FIGURE A32c BASIC DATA EFFECT OF CMU
CONFIGURATION BC3W6V, DELF=45, BN/B=1

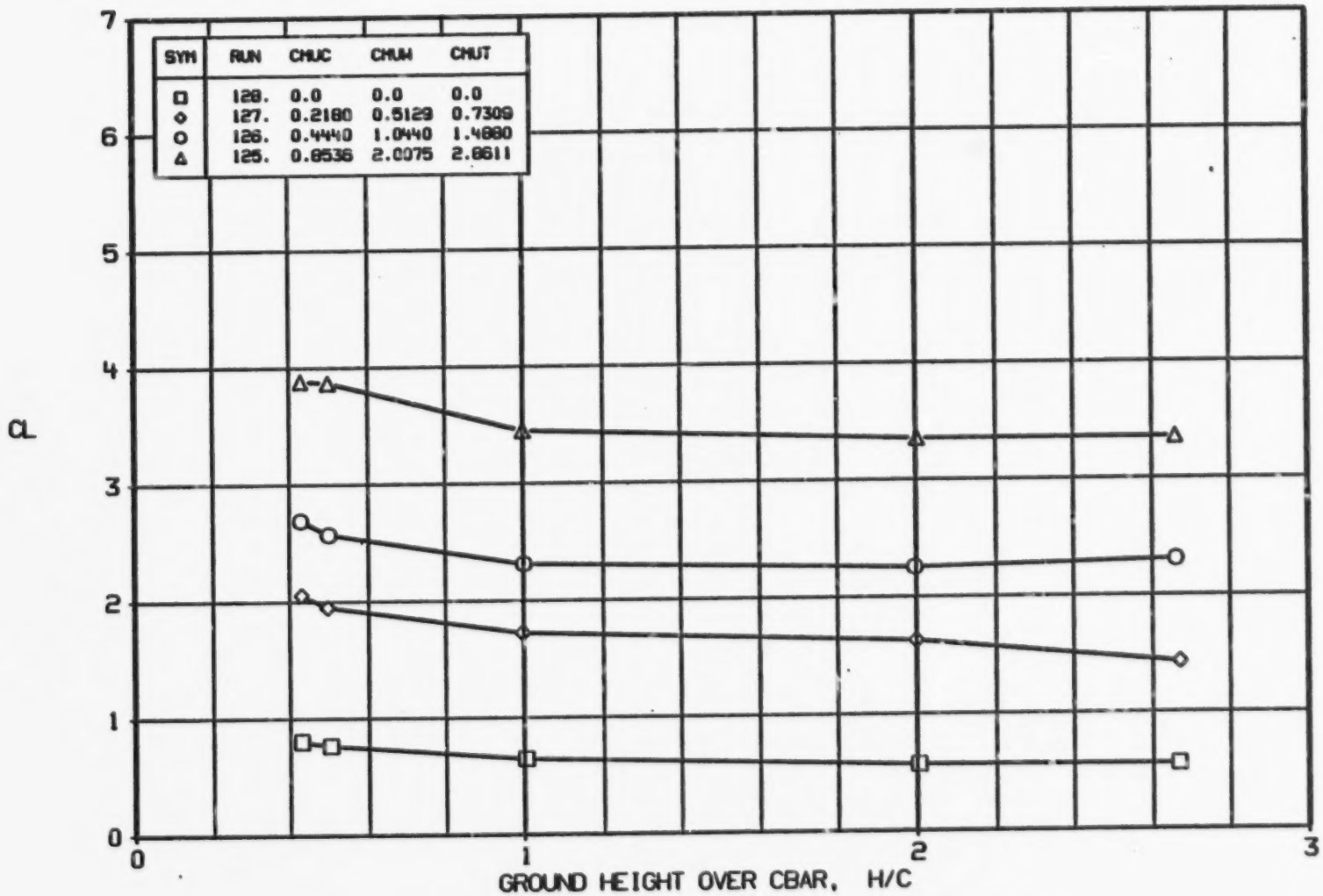

FIGURE A33a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10°

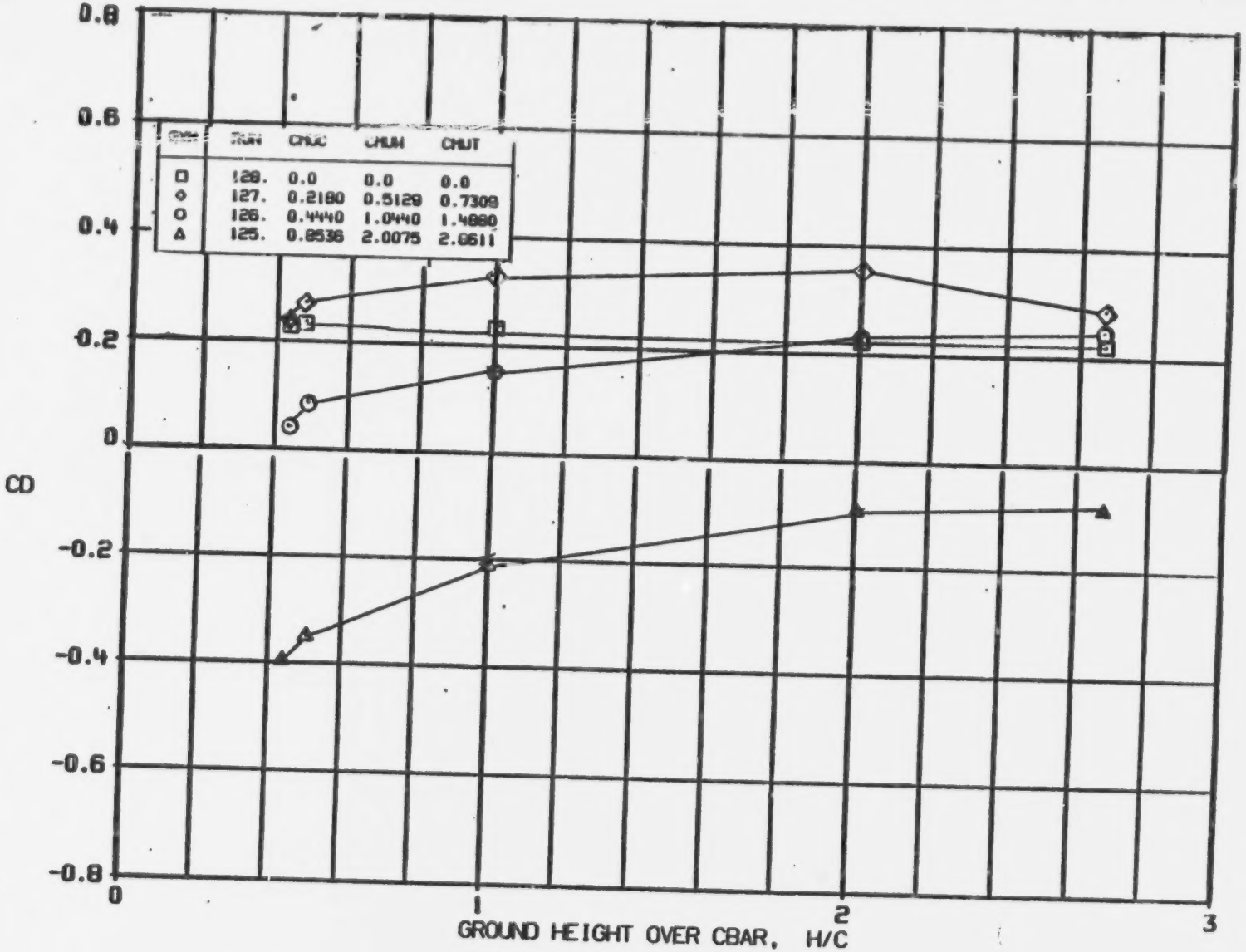

FIGURE A33b BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BN/B=1, DELC=10°

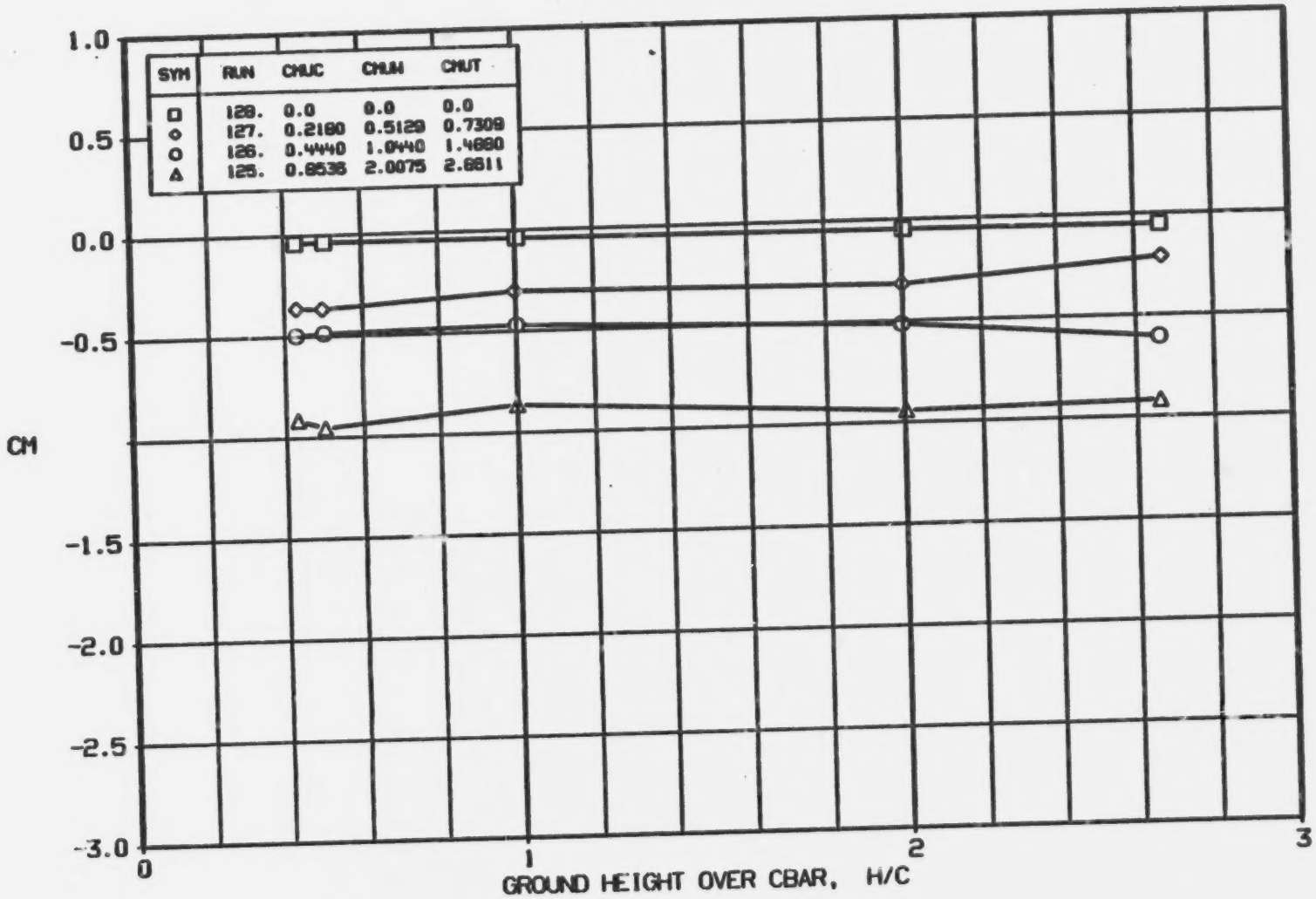

FIGURE A33c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BW/B=1, DELC=10°

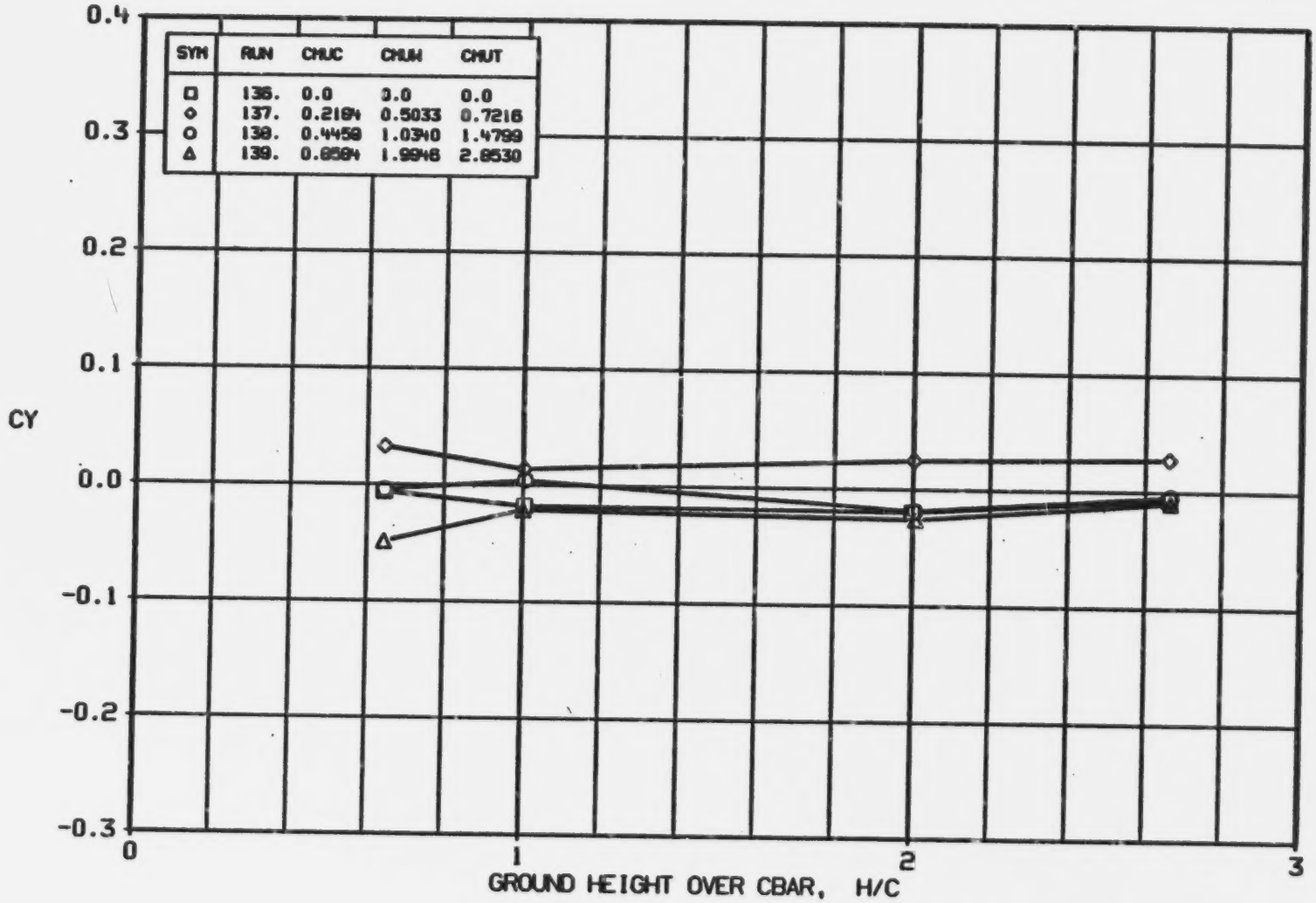

FIGURE A34a BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BH/B=1 β = 0.8, α = 2

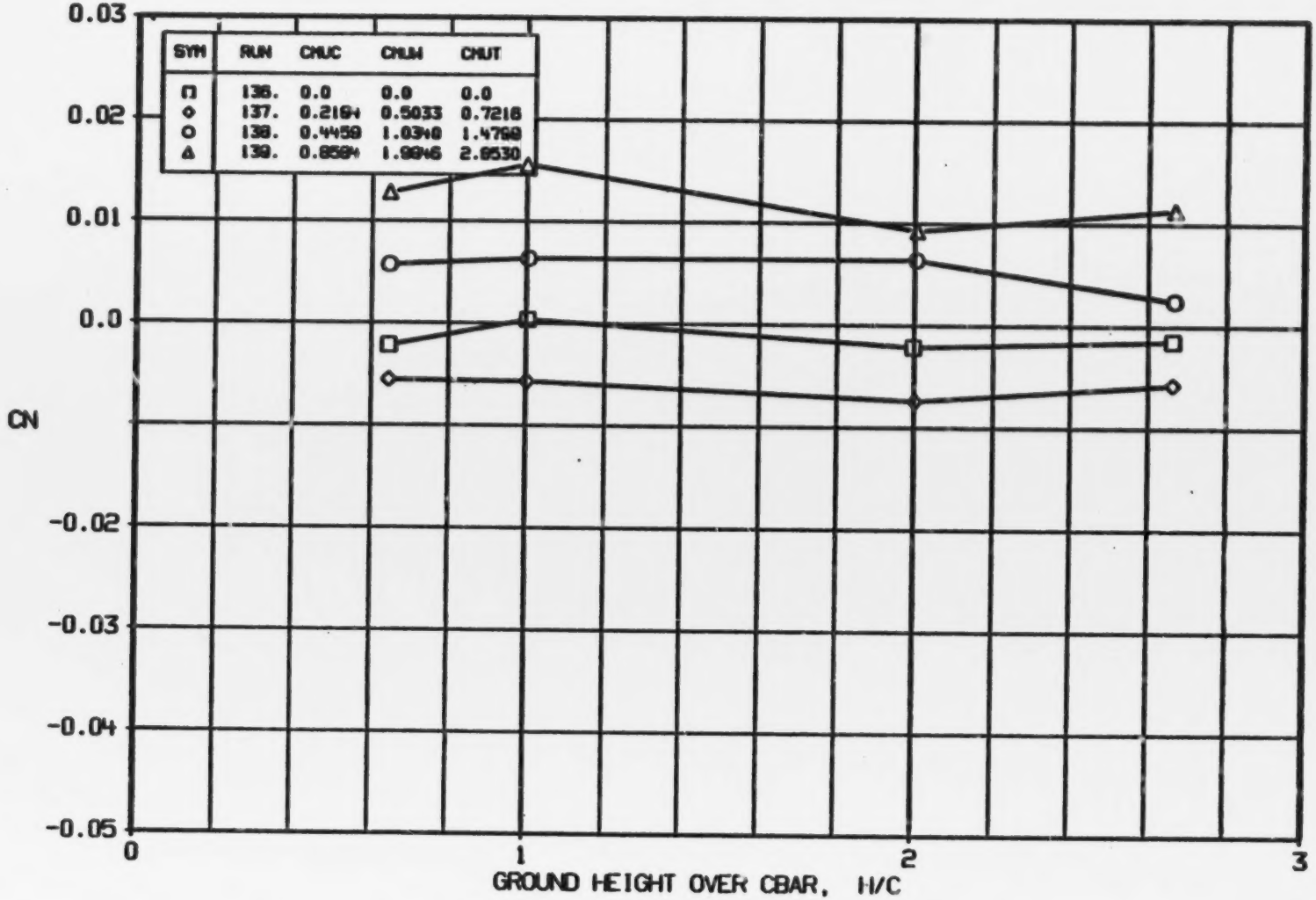

FIGURE A34b. BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BW/B=1 β = 0.8, α = 2

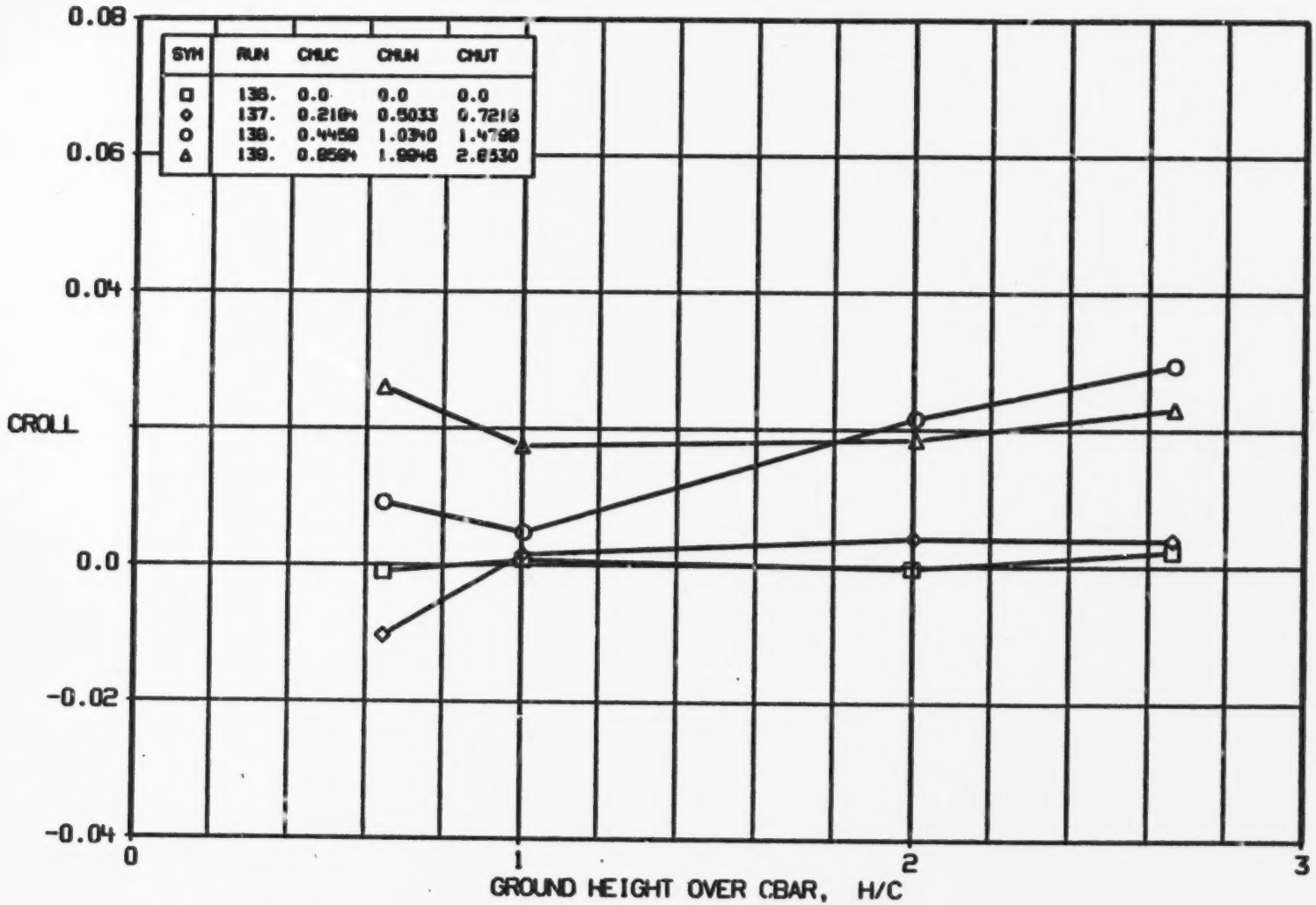

FIGURE A34c BASIC DATA EFFECT OF CMU
CONFIGURATION BC1W6V, DELF=45, BW/B=1, β = 0.8, α = 2

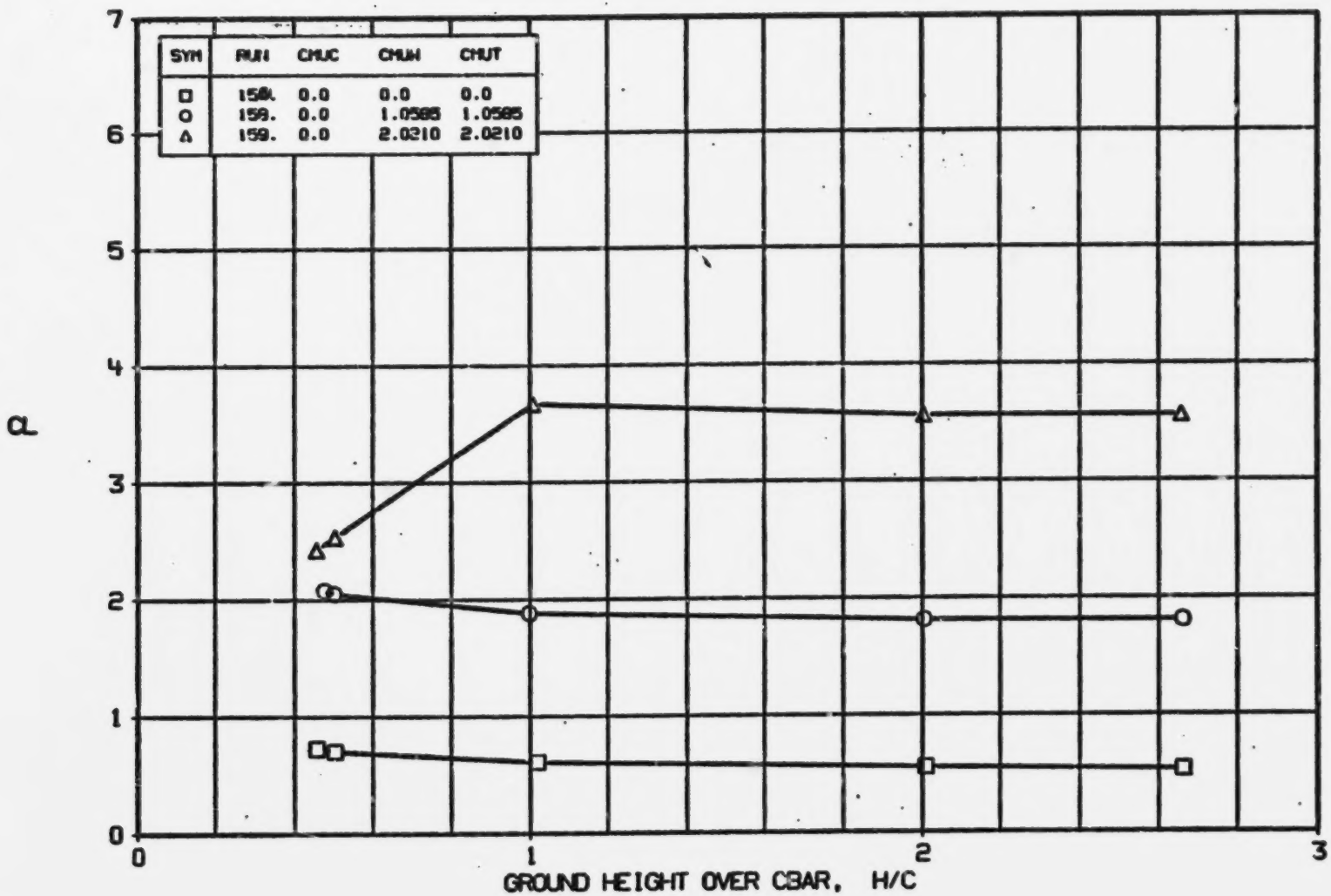

FIGURE A35a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BW/B=1

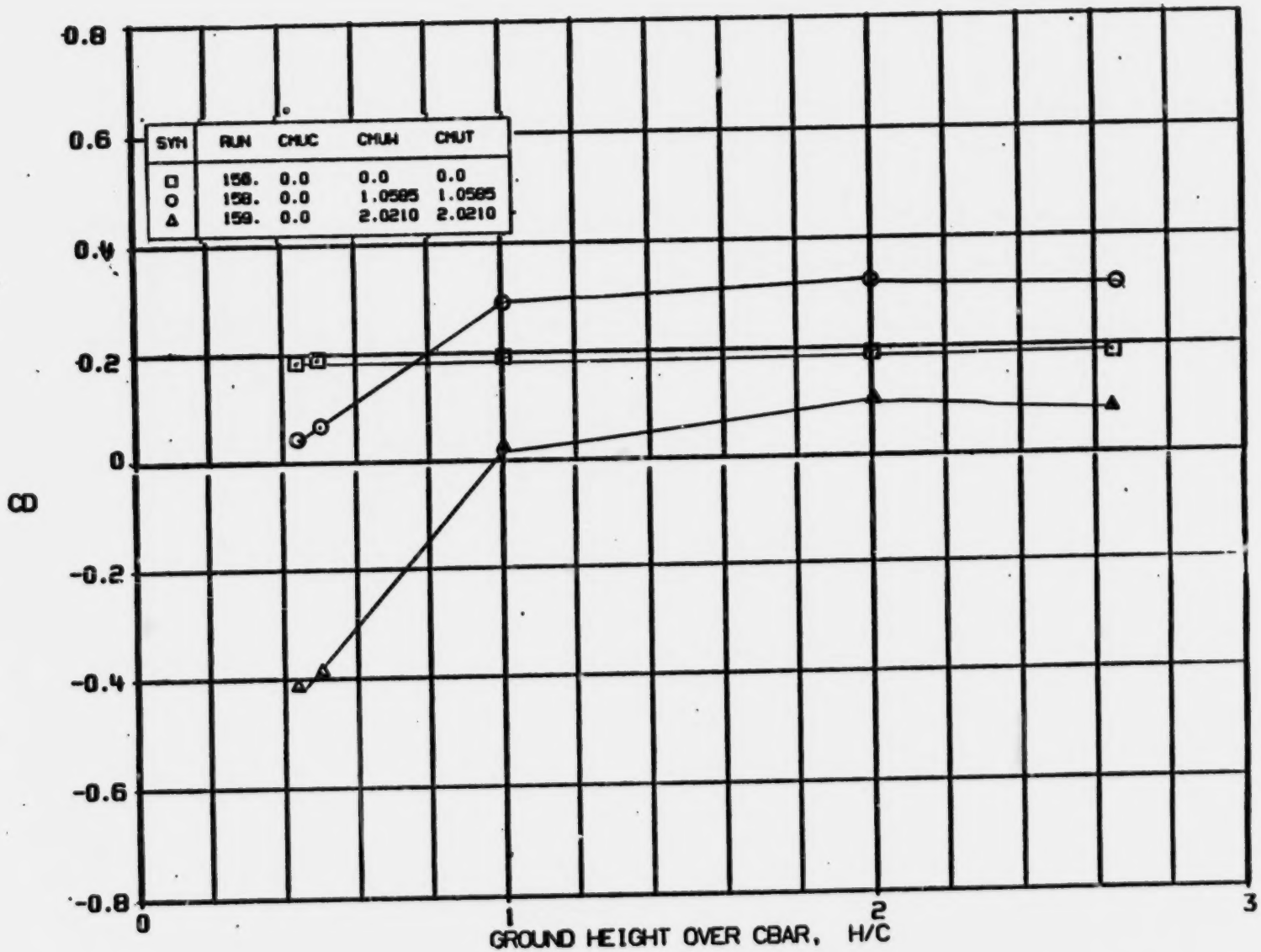

FIGURE A35b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BW/B=1

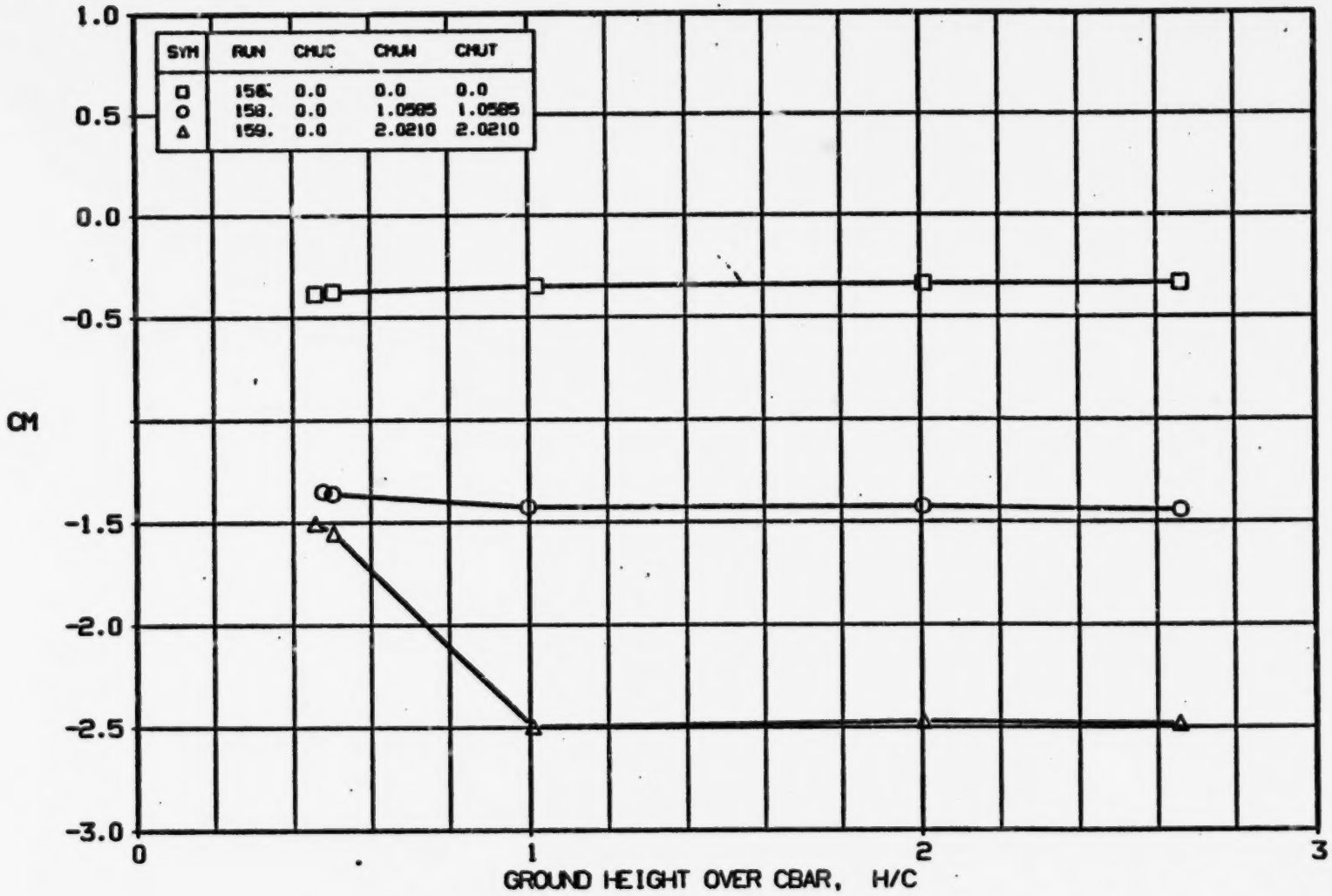

FIGURE A35c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1

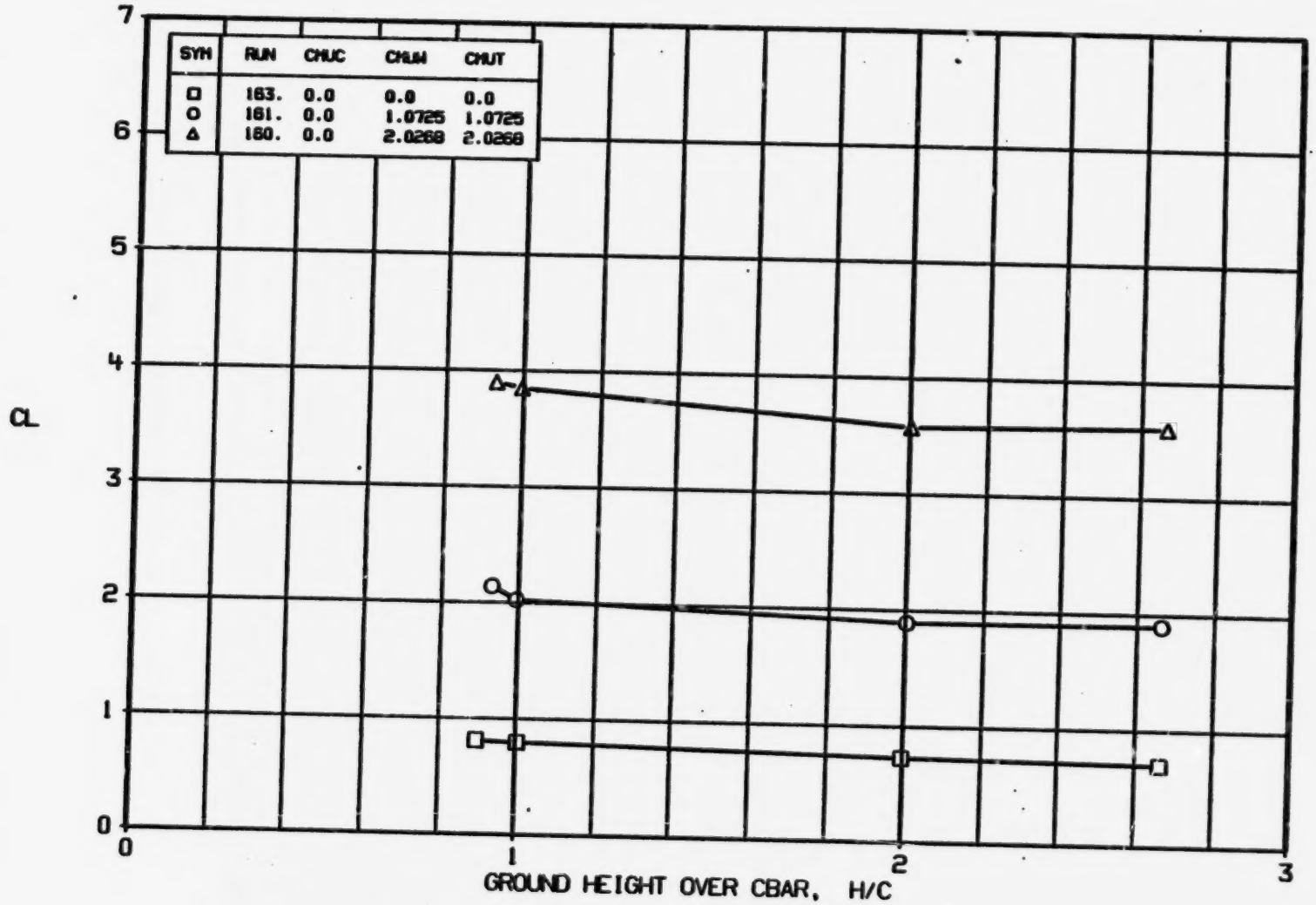

FIGURE A36a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BW/B=1 $\alpha = 4°$

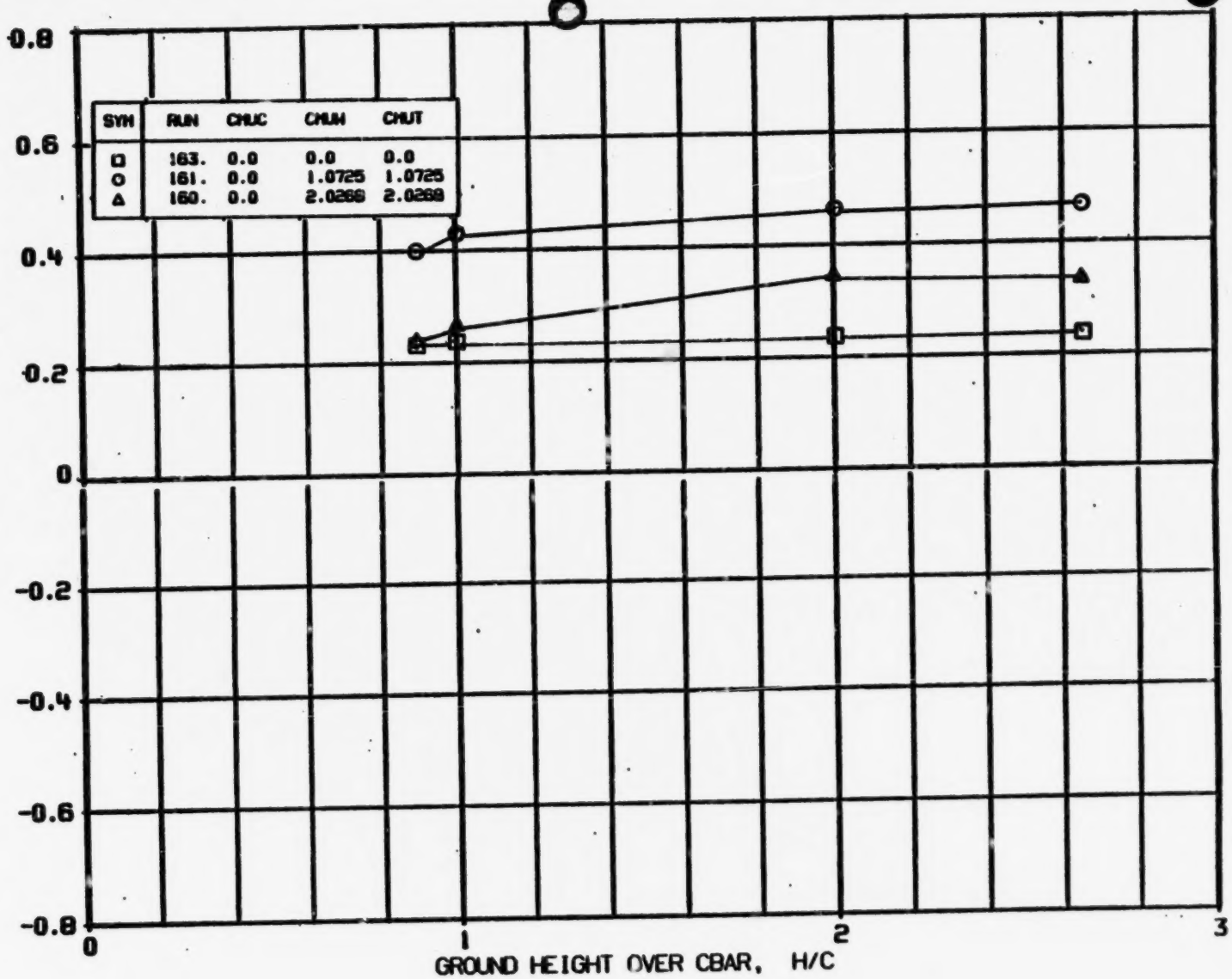

FIGURE A36b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, βN/β=1 α = 4°

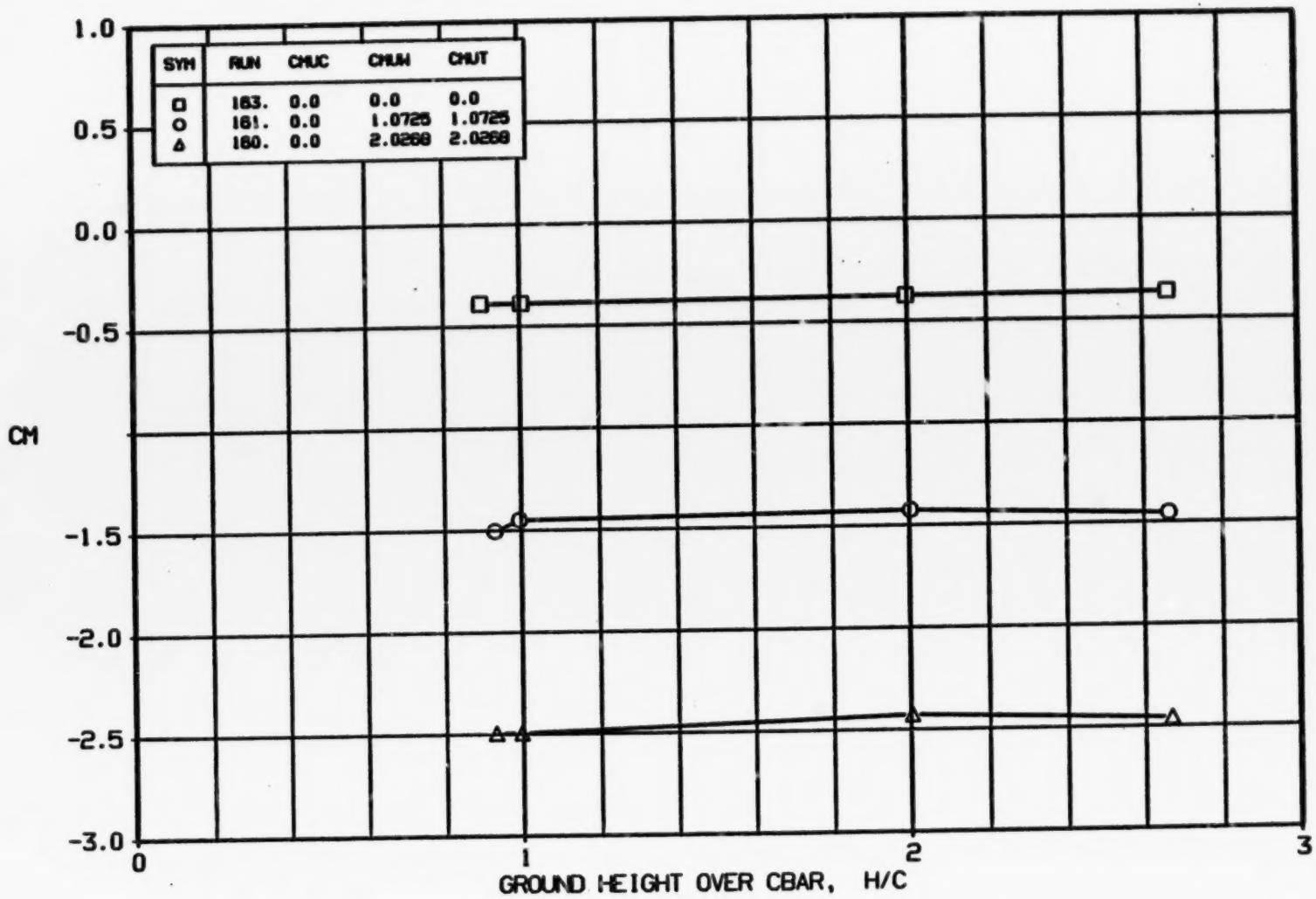

FIGURE A36c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1 $\alpha = 4°$

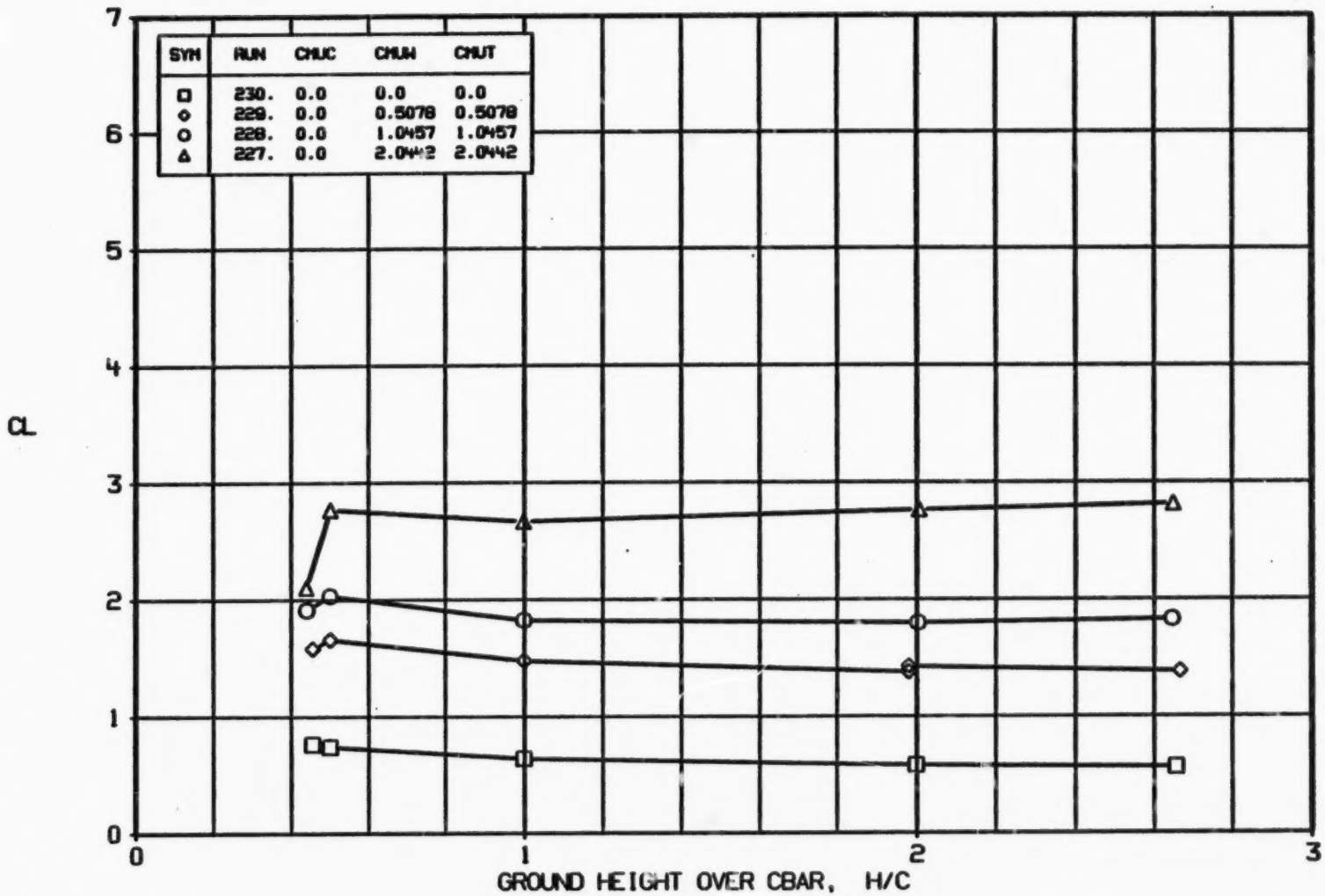

FIGURE A37a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

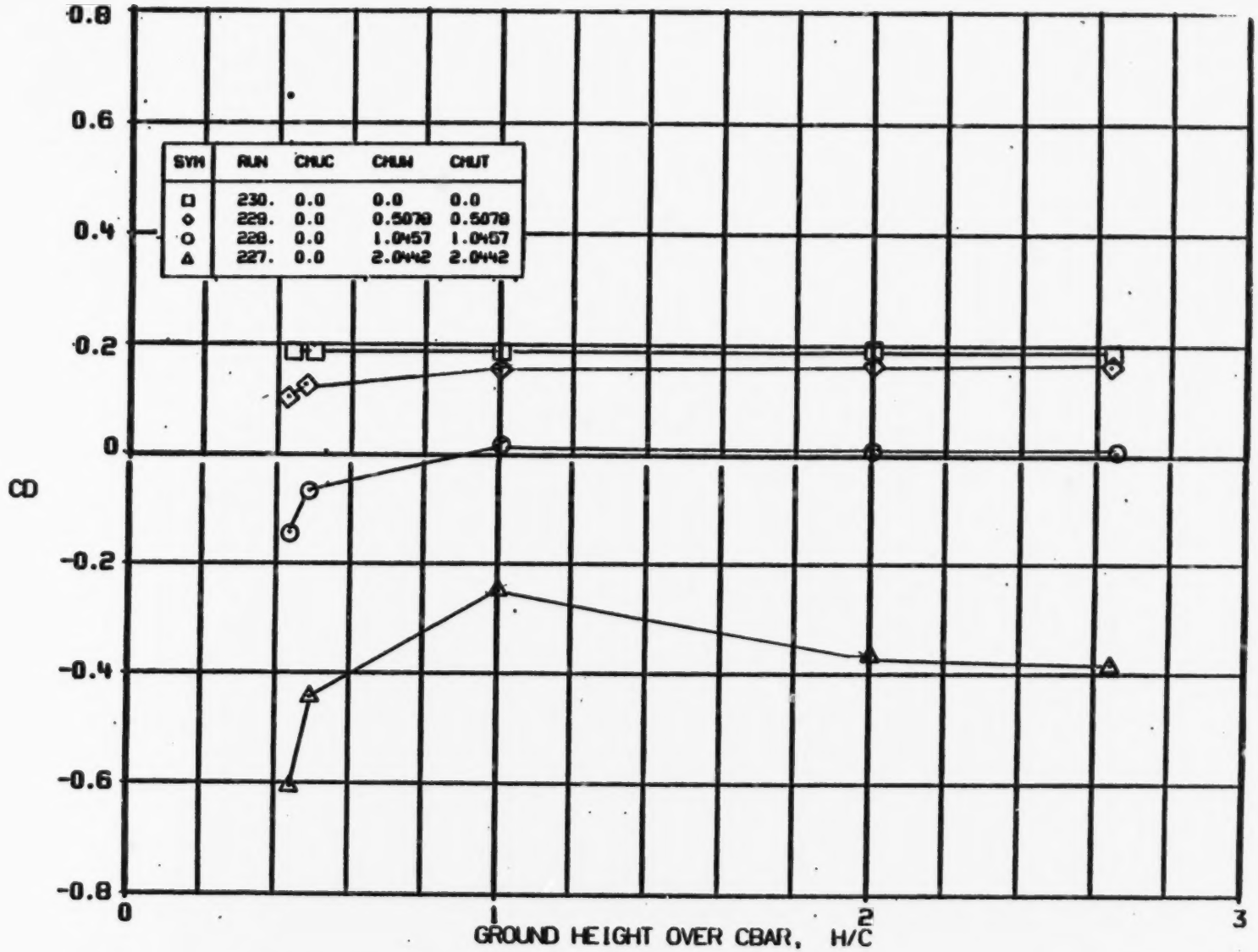

FIGURE A37b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5, DELF=45

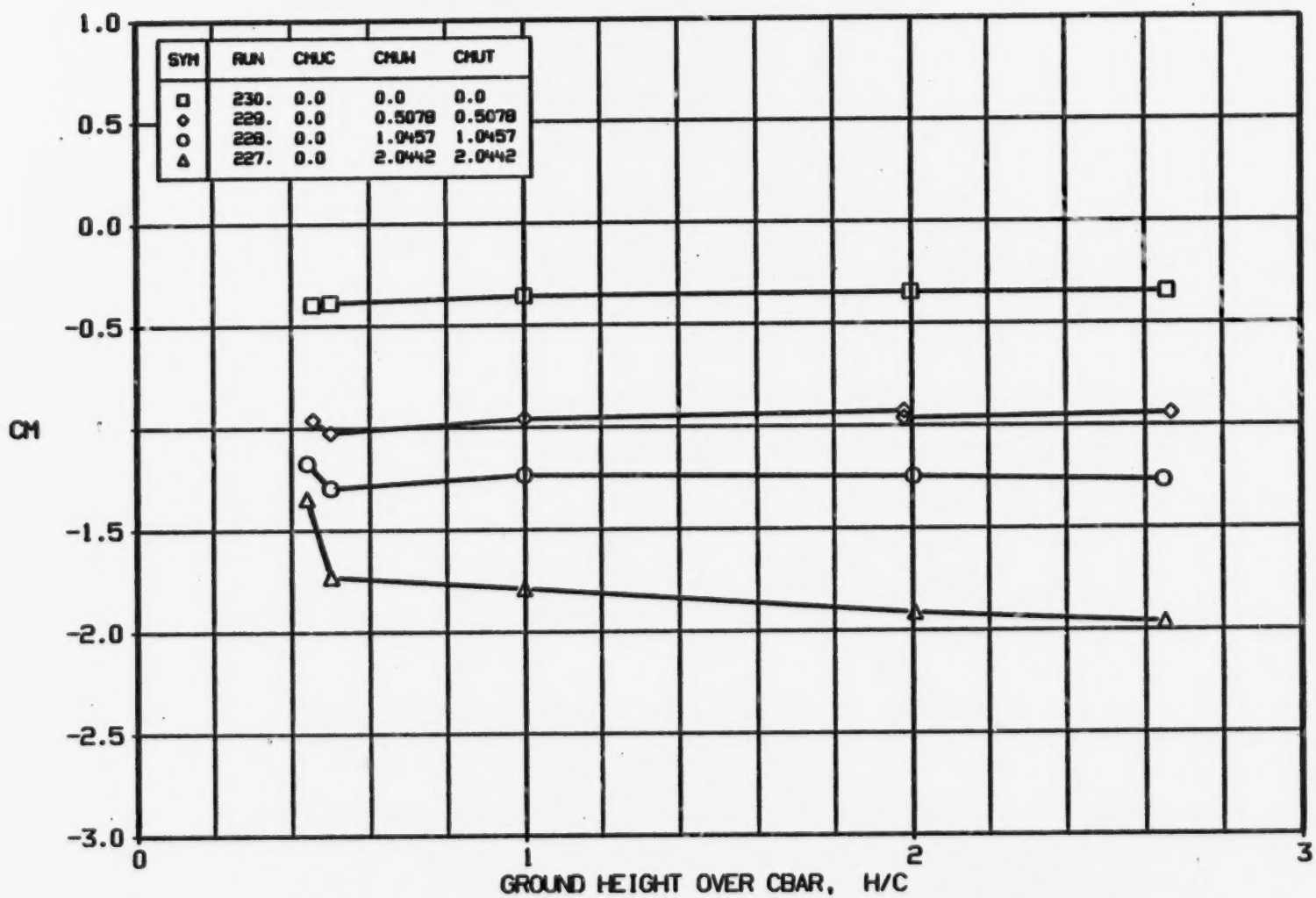

FIGURE A37c BASIC DATA EFFECT OF CMU
CONFIGURATION BWSV, BN/B=0.5, DELF=45

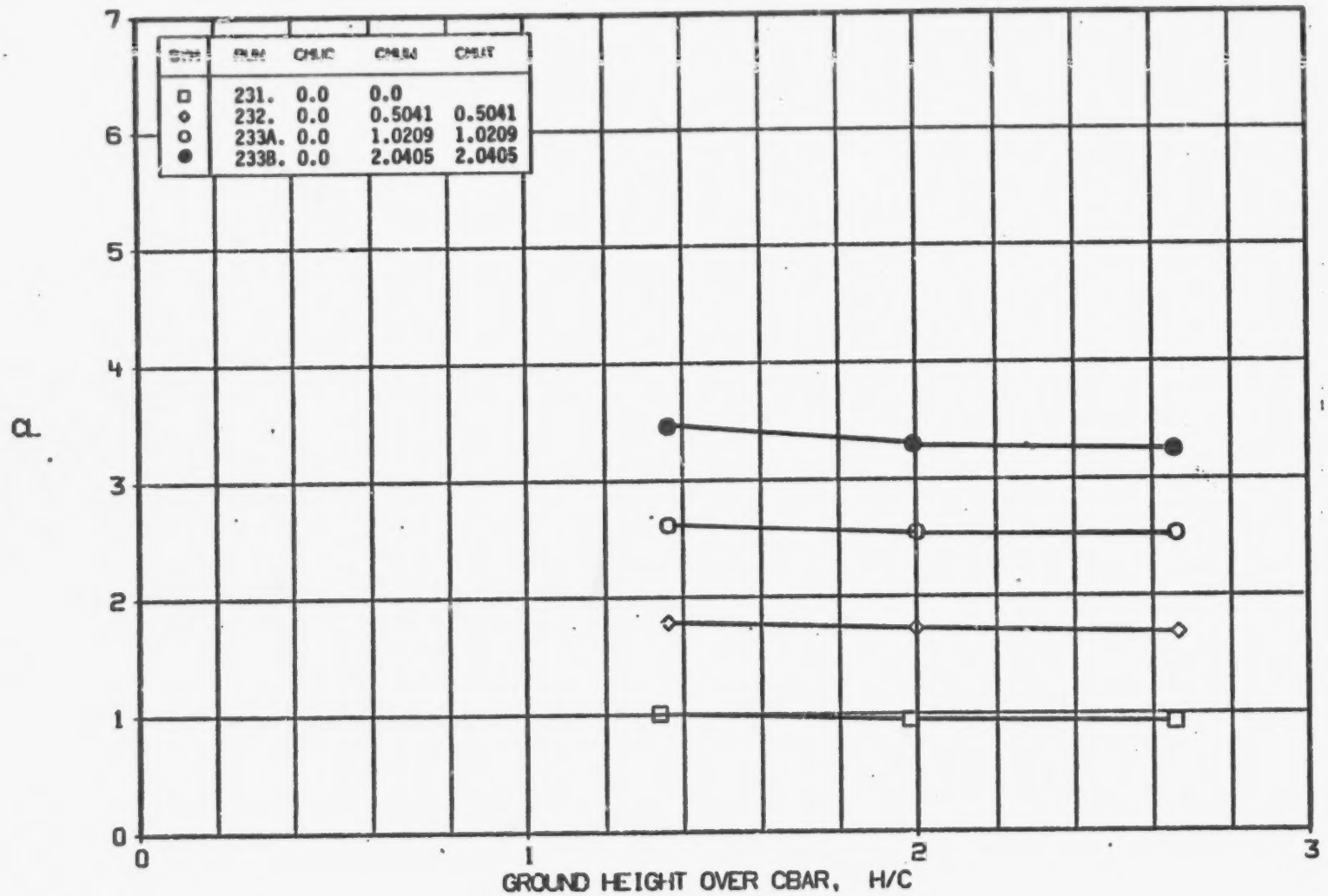

FIGURE A38a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BW/B=0.5

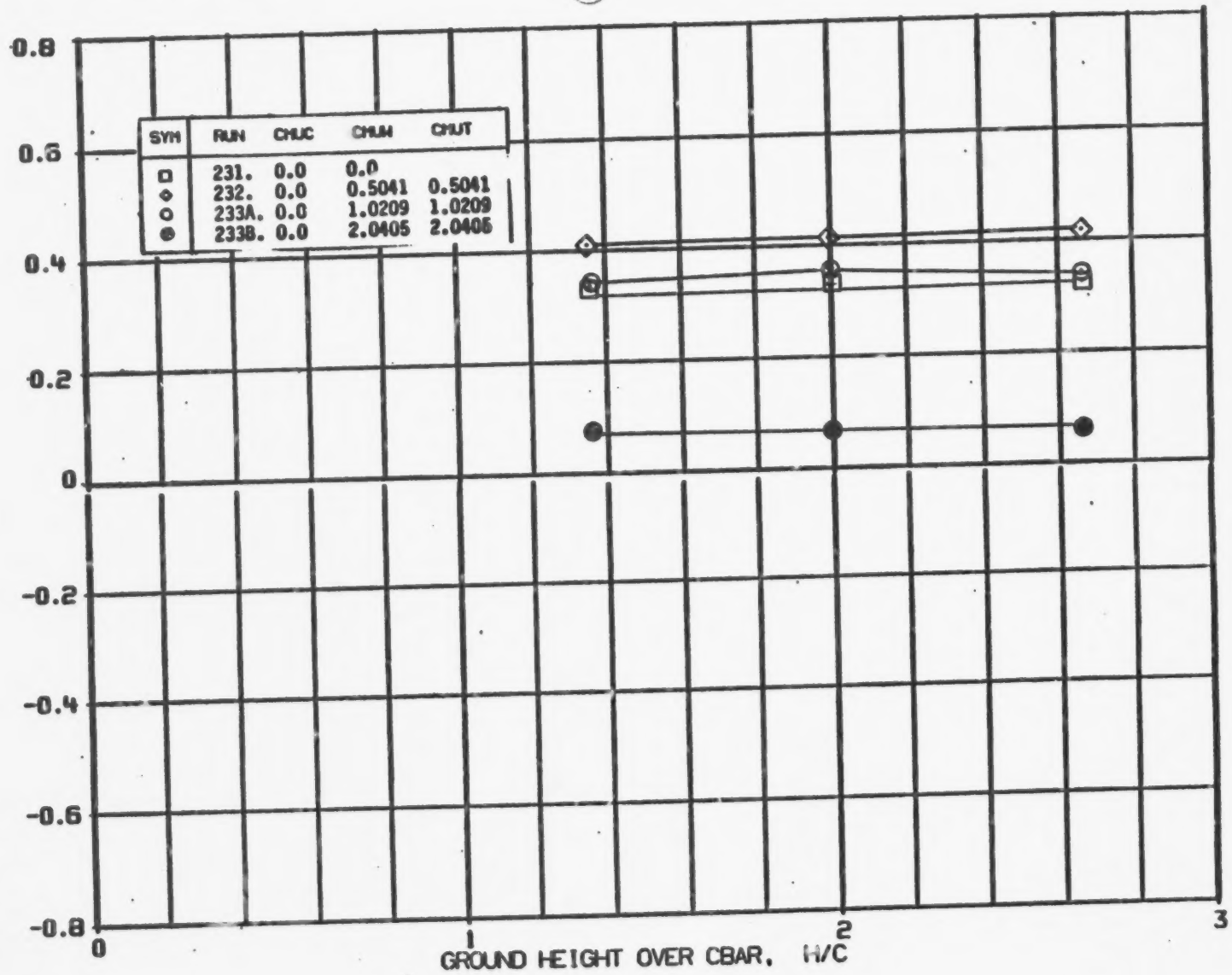

FIGURE A38b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5

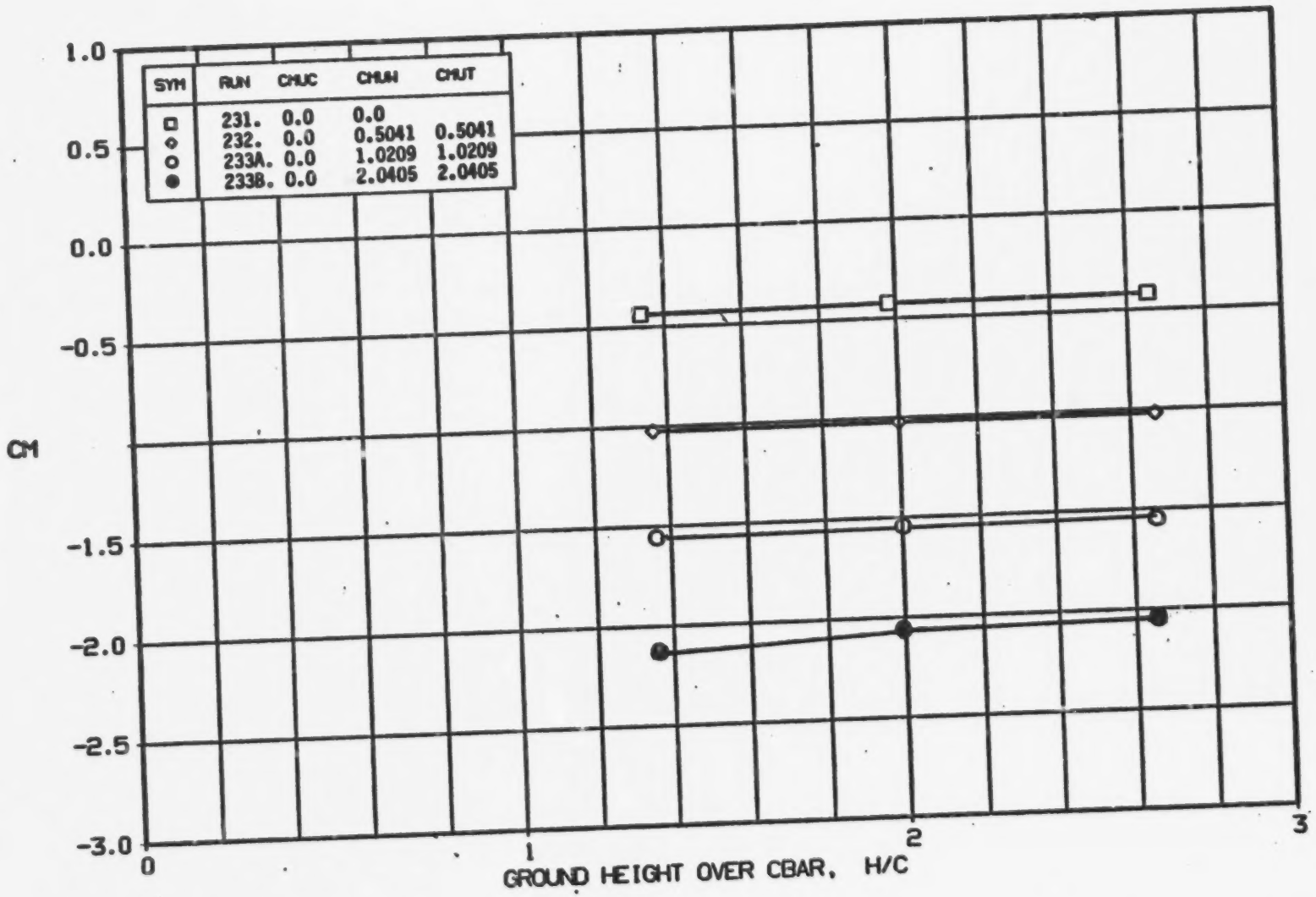

FIGURE A38c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, BN/B=0.5

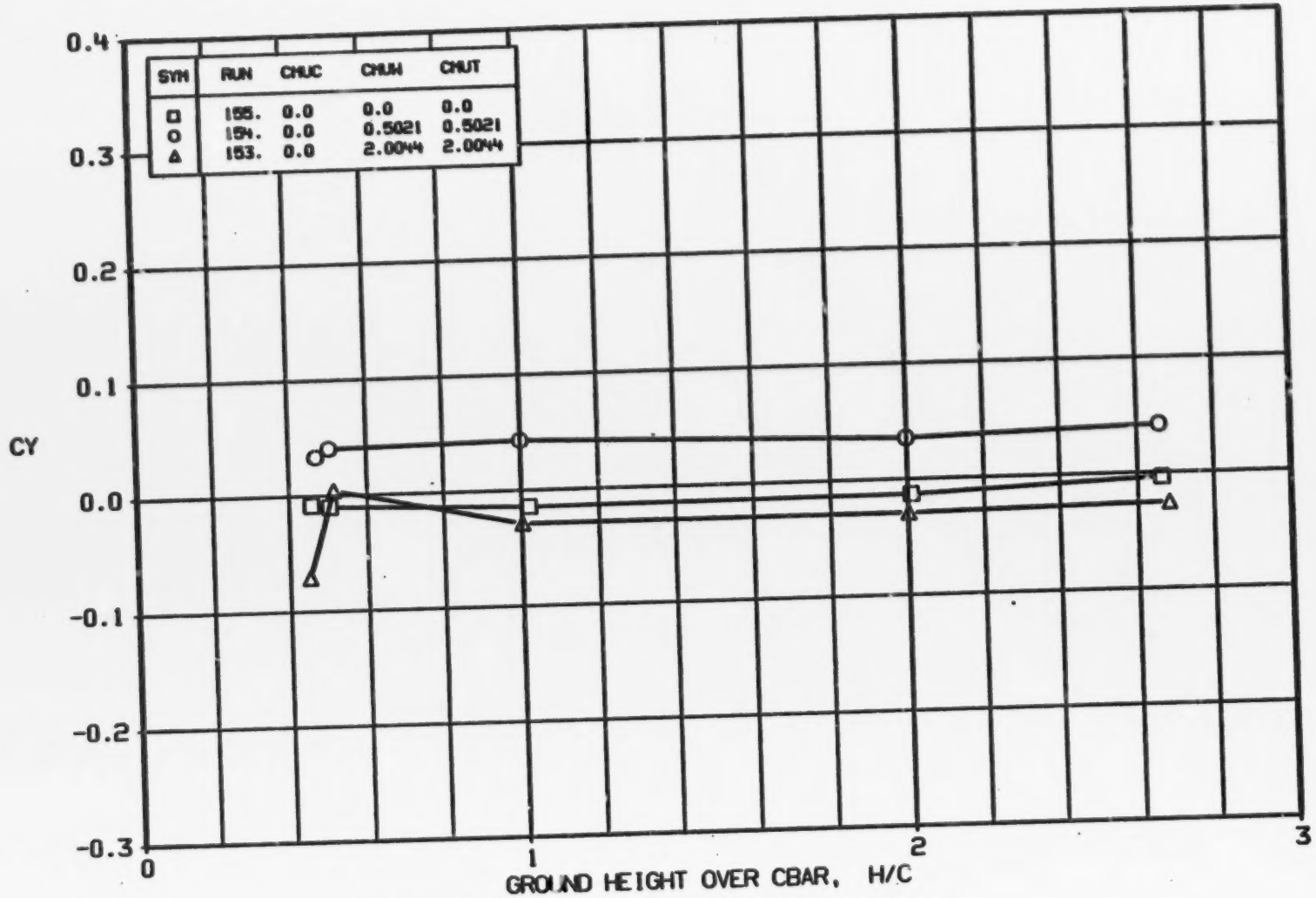

FIGURE A39a BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BN/B=1, β = 0.8

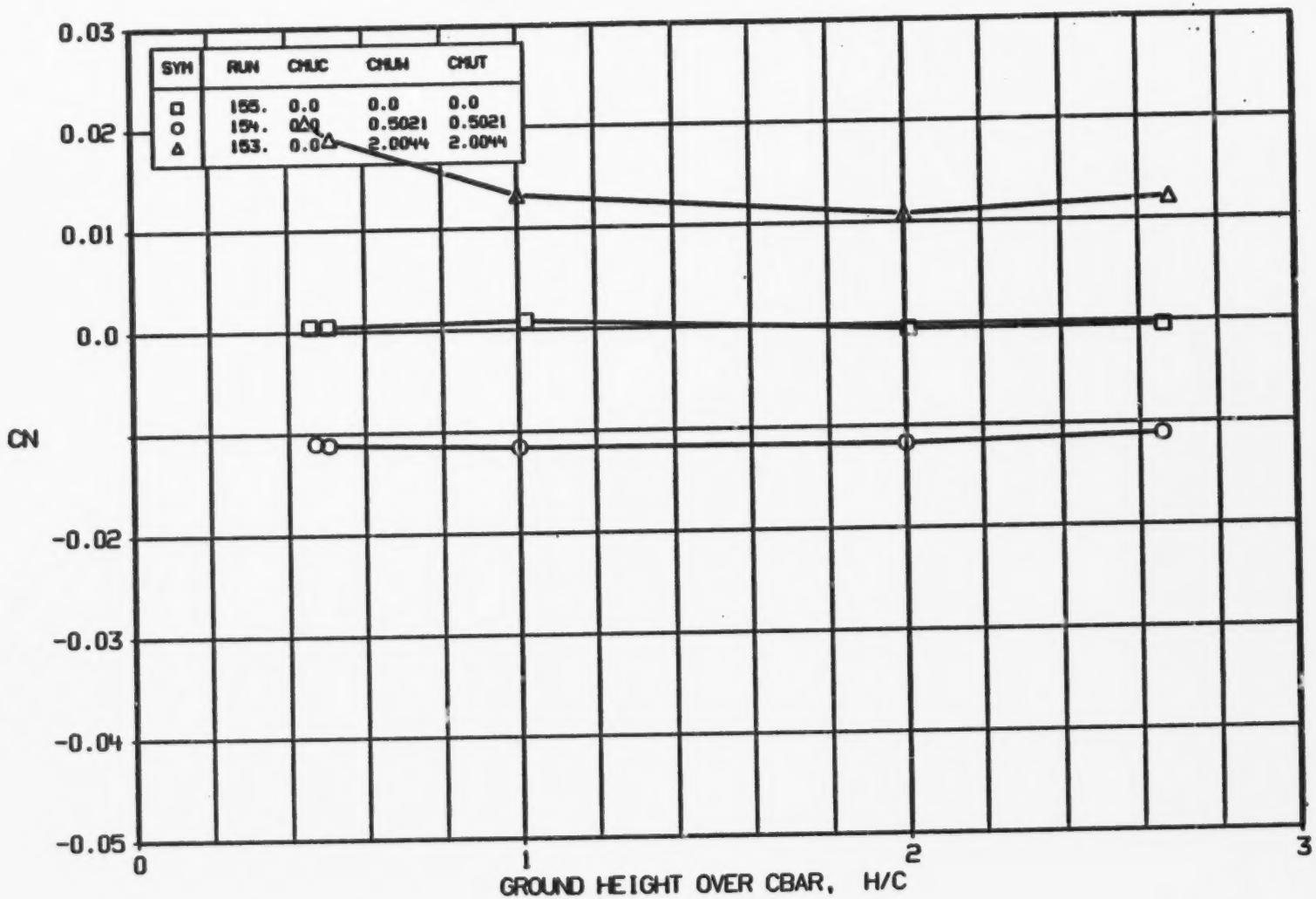

FIGURE A39b BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BW/B=1 , β = 0.8

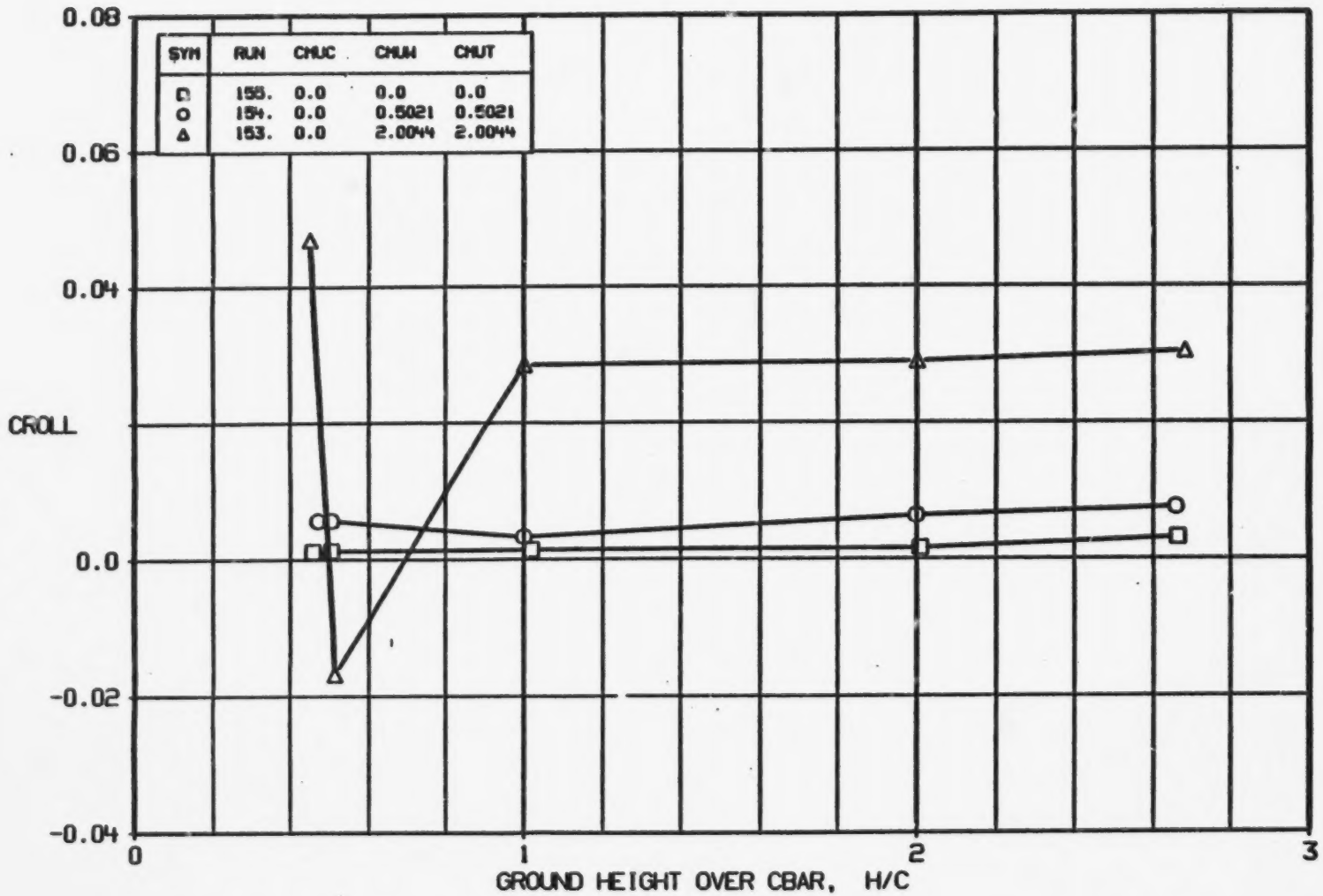

FIGURE A39c BASIC DATA EFFECT OF CMU
CONFIGURATION BW6V, DELF=45, BW/B=1, β = 0.8

A3.0 TABULATED SURFACE PRESSURE DATA

The surface pressure distribution for selected data runs is presented in Tables A1 to A50. The pressure coefficients are presented for the canard and wing with the locations described for each coefficient. The fuselage pressures are presented for various fuselage stations three inches to the left of the plane of symmetry. The wing pressures, presented as miscellaneous pressure coefficients, are as described below:

MISCELLANEOUS WING PRESSURES

	X/C	BP	
1	0.33	9	Upper
2	0.33	9	Lower
3	0.33	19	Upper
4	0.33	19	Lower
5	0.33	23.5	Upper
6	0.33	23.5	Lower
7			
8			
9	Downwash probe pressure		
10	coefficients		
11			
12			

TABLE A1. RUN61, SW6V,DELF=0, CMU=1.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
61 4	CLAERO =	0.3119	CDAERO =	0.0606	DCLC =	0.0	DWLC =	0.2681	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.06	2111.83	2099.73	12.09	101.17	0.01	2.67	0.0	1.032	1.032	87.2626	0.644190E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0.58	-0.94	-0.34	0.01	-0.01	0.02	0.30	-0.27	0.30	0.03	

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.2375	-0.2375	0.0	0.5167	0.5167	0.2890	0.3744	0.3744	38.2	-0.1259	1	-0.3161
2.5	-0.2172	-0.2375	2.5	-0.1949	-0.2802	-0.2802	-0.5060	-0.3657	43.4	-0.1043	2	-0.1515
5.0	-0.2172	-0.2375	5.0	-0.1664	-0.2233	-0.3087	-0.5060	-0.4226	50.4	-0.1043	3	-0.3983
10.0	-0.2172	-0.2375	10.0	-0.2518	-0.2802	-0.3657	-0.3941	-0.4226	56.1	-0.1043	4	-0.1515
15.0	-0.2375	-0.2375	15.0	-0.2802	-0.3372	-0.3657	-0.3941	-0.4510	62.5	-0.2343	5	-0.3161
25.0	-0.2172	-0.2375	24.0	-0.2802	-0.3372	-0.3657	-0.3941	-0.3941	69.7	-0.2126	6	-0.1515
35.0	-0.2172	-0.2172	33.0	-0.3161	-0.3161	-0.3983	-0.4807	-0.3161	76.9	-0.1693	7	0.5069
50.0	-0.2172		54.0	-0.3161	-0.3983	-0.4807	1.0006	-0.3983	83.4	-0.1476	8	-0.2161
56.0	-0.2172	-0.2375	65.0	-0.4942	-0.5375	-0.5592	-0.5808	-0.5592	89.4	-0.2126	9	-0.0221
65.0	-0.2172	-0.2375	78.5	-1.1007	-1.1440	-1.1657	-1.2307	-1.3174	95.4	-0.2559	10	0.5950
76.0	-0.2172	-0.2578	79.5	130.2903	130.2903	130.2903	130.2903	124.2206	101.1	-0.2126	11	0.2953
79.0	-0.2375	-0.2375	80.5	-54.0155	-45.9406	-58.8084	-53.8989	-40.7963	---BOTTOM---		12	0.0131
80.5	-0.2158	-0.2012	81.3	-21.4959	119.6643	-5.6529	-10.5185	-3.0683	38.2	-0.1043		
81.0	-0.2012	-0.1865	82.0	28.2891	25.7538	20.0529	2.9351	2.1437	43.4	-0.0609		
82.0	-0.2158	-0.2012	84.0	14.3513	8.2990	1.0445	1.5428	1.1032	50.4	-0.1043		
84.0	-0.2012		87.0	3.4920	-2.0624	-1.4909	1.1032	0.4290	56.1	-0.0609		
87.0	-0.2012		89.0	0.1190	-0.0750	-1.9610	-1.5914	1.1593	62.5	-0.0609		
89.0	-0.2343	-0.2343	93.0	0.3305	0.4716	-0.0045	-0.3395	-0.5864	69.7	-0.0609		
93.0	-0.2343		96.0	0.2600	-0.5887	1.2298	1.5824	-1.5737	76.9	-0.0820		
96.0	-0.2343	-0.2343	100.0	-2.3319	-2.2438	-1.7853	-1.6619	-1.6796	83.4	-0.1043		
100.0	-0.2343	-0.2126	96.0	-0.0397	-0.1455	-0.1102	-0.1455	0.0131	89.4	-0.0609		
96.0	-0.2343	-0.2126	84.0	0.0637	0.1013	0.2777	0.1013	0.0661	95.4	-0.0393		
84.0	-0.2343	-0.2126	73.5	-0.2338	-0.0692	-0.0692	-0.0692	-0.0692	101.1	-0.0609		
73.0	-0.2343	-0.2126	54.0	-0.1515	-0.1515	-0.1515	-0.1515	-0.1515				
50.0	-0.2343	-0.2126	33.0	-0.1515	-0.1515	-0.1515	-0.1515	-0.1515				
35.0	-0.2172	-0.2172	24.0	-0.1379	-0.1379	-0.1664	-0.1095	-0.1949				
25.0	-0.2172	-0.2375	10.0	-0.0810	-0.0810	-0.1095	-0.0810	-0.1515				
10.0	-0.2375	-0.2172	5.0	-0.0810	-0.1095	-0.0810	0.0044	-0.1515				
5.0	-0.2375	-0.2375	2.5	-0.0241	0.0044	-0.0526	0.0897	-0.0692				
2.5	-0.2375	-0.2375										

TABLE A1. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 76

61 6	CLAERO =	0.5431	CDAERO =	0.1122	DCLC =	0.0	DWLC =	0.3440	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.06	2112.04	2099.94	12.09	101.14	0.01	2.66	0.0	1.053	1.053	86.8984	0.644619E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.89	-0.88	-0.40	0.01	-0.01	0.01	0.54	-0.33	0.53	0.07

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.2172	-0.2172
2.5	-0.2172	-0.2172
5.0	-0.2172	-0.2172
10.0	-0.2172	-0.2172
15.0	-0.2172	-0.2172
25.0	-0.2172	-0.2172
35.0	-0.2172	-0.2172
50.0	-0.1969	
56.0	-0.2172	-0.1969
65.0	-0.1969	-0.2172
76.0	-0.1969	-0.2172
79.0	-0.2172	-0.1969
80.5	-0.2305	-0.2158
81.0	-0.2158	-0.2158
82.0	-0.2158	-0.2305
84.0	-0.2158	
87.0	-0.2012	
89.0	-0.2125	-0.2125
93.0	-0.2125	
96.0	-0.2125	-0.1909
100.0	-0.2125	-0.2125
96.0	-0.2125	-0.2125
84.0	-0.2125	-0.2125
73.0	-0.2125	-0.2125
50.0	-0.2125	-0.2125
35.0	-0.2172	-0.2172
25.0	-0.1969	-0.2172
10.0	-0.1969	-0.2172
5.0	-0.2172	-0.2172
2.5	-0.2172	-0.2172

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.0810	-0.7072	-1.1911	-1.1627	-1.3619
2.5	-0.6787	-0.8779	-1.3050	-1.2460	-1.7319
5.0	-0.5649	-0.7072	-0.9065	-1.0488	-1.0488
10.0	-0.5080	-0.6787	-0.7642	-0.8210	-0.8210
15.0	-0.4510	-0.5364	-0.6503	-0.6503	-0.7356
24.0	-0.4226	-0.4510	-0.5364	-0.5364	-0.6218
33.0	-0.4807	-0.4807	-0.4807	-0.5630	-0.5630
54.0	-0.4807	-0.4807	-0.5630	1.0006	-0.5630
65.0	-0.5592	-0.6242	-0.6025	-0.6458	-0.6892
78.5	-1.1224	-1.2091	-1.2524	-1.2957	-1.4474
79.5	130.3233	130.3233	130.3233	130.3233	125.6016
80.5	-54.7631	-47.0545	-59.2921	-54.4703	-42.1450
81.3	-22.1550	98.0485	-4.6858	-7.7193	-3.2935
82.0	28.5818	27.3216	20.1555	-0.6408	1.5428
84.0	12.5783	12.0797	2.1144	-2.5167	1.4403
87.0	3.5359	-1.6081	-1.5934	0.8980	0.3264
89.0	0.0485	-0.0750	-1.9969	-1.6090	1.0358
93.0	0.2953	0.4363	-0.1102	-0.3395	-0.7274
96.0	0.2424	-0.5864	1.1945	1.4943	-1.7148
100.0	-2.3143	-2.2614	-1.6972	-1.7325	-1.7325
96.0	-0.0045	-0.0045	-0.0926	-0.1279	-0.0397
84.0	0.1366	0.1366	0.2248	0.1366	0.0308
73.5	-0.1515	-0.0692	0.0131	-0.0692	-0.0692
54.0	-0.0692	-0.0692	-0.0692	-0.0692	-0.1515
33.0	-0.0692	-0.0692	-0.0692	-0.0692	-0.1515
24.0	0.0044	-0.0241	-0.0526	0.0044	-0.0810
10.0	0.0897	0.0897	0.0897	0.1752	0.0954
5.0	0.1182	0.2321	0.2321	0.2890	0.2599
2.5	0.2605	0.3175	0.3175	0.3744	0.3422

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1043
43.4	-0.0826
50.4	-0.1043
56.1	-0.1043
62.5	-0.2125
69.7	-0.2125
76.9	-0.2125
83.4	-0.1909
89.4	-0.2342
95.4	-0.2558
101.1	-0.2125
---BOTTOM---	
38.2	-0.0826
43.4	-0.0393
50.4	-0.0826
56.1	0.0041
62.5	0.0041
69.7	0.0041
76.9	0.0041
83.4	0.0041
89.4	-0.0176
95.4	0.0041
101.1	-0.0393

MISC.*

NO.	CP
1	-0.4807
2	-0.0692
3	-0.5630
4	-0.0692
5	-0.4807
6	-0.1515
7	0.5069
8	-0.2161
9	-0.0750
10	0.5421
11	0.3129
12	-0.0045

ORIGINAL PAGE IS OF POOR QUALITY

TABLE A1. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 79

61 12	CLAERO =	1.0220	CDAERO =	0.2656	DCLC =	0.0	DWLC =	0.4800	BASEPR =	8.1904			
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
11.99	2112.25	2100.27	11.98	100.59	0.01	2.67	0.0	1.058	1.058	87.2142	0.642746E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
-0.01	1.50	-0.68	-0.52	0.01	-0.01	0.02	1.04	-0.46	1.01	0.24			

CANARD BP=3.375 %X/C	CP	CANARD BP=10.125 CP
0.0	-0.2686	-0.2686
2.5	-0.2686	-0.2686
5.0	-0.2686	-0.2686
10.0	-0.2481	-0.2686
15.0	-0.2686	-0.2686
25.0	-0.2686	-0.2891
35.0	-0.2686	-0.2481
50.0	-0.2481	
56.0	-0.2686	-0.2686
65.0	-0.2891	-0.2891
76.0	-0.2686	-0.2891
79.0	-0.2481	-0.2686
80.5	-0.2854	-0.2558
81.0	-0.2558	-0.2707
82.0	-0.2558	-0.2707
84.0	-0.2558	
87.0	-0.2707	
89.0	-0.2449	-0.2885
93.0	-0.2667	
96.0	-0.2449	-0.2667
100.0	-0.2449	-0.2667
96.0	-0.2667	-0.2667
84.0	-0.2449	-0.2667
73.0	-0.2667	-0.2667
50.0	-0.2667	-0.2449
35.0	-0.2686	-0.2891
25.0	-0.2481	-0.2891
10.0	-0.2686	-0.2686
5.0	-0.2686	-0.2891
2.5	-0.2891	-0.2686

WING %X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.2452	-4.1417	-5.6645	-3.1072	-1.8429
2.5	-1.4981	-2.2452	-3.8543	-2.3889	-1.6131
5.0	-1.2106	-1.5556	-3.0498	-2.5038	-1.5556
10.0	-0.9809	-1.1821	-2.7337	-2.6188	-1.5556
15.0	-0.8659	-1.0384	-1.3832	-2.2452	-1.4981
24.0	-0.7223	-0.8659	-0.8947	-1.6705	-1.4119
33.0	-0.6597	-0.7428	-0.8259	-0.9089	-1.4904
54.0	-0.5767	-0.6597	-0.6597	0.9187	-1.4074
65.0	-0.6822	-0.7259	-0.7478	-0.6385	-1.5789
78.5	-1.2289	-1.2945	-1.3333	-1.0978	-2.0162
79.5	131.5552	131.5552	131.5552	131.5552	128.3079
80.5	-56.5356	-48.6201	-60.4115	-55.4111	-44.2558
81.3	-23.0846	74.2065	-3.6290	-5.3156	-3.4367
82.0	28.8163	27.7511	20.0726	-4.9014	0.2916
84.0	8.2068	14.1247	3.0286	-7.1503	0.7946
87.0	3.3246	-1.3359	-1.6021	0.6910	-0.5665
89.0	0.0406	-0.1375	-2.1489	-1.6327	0.0049
93.0	0.2541	0.3432	-0.1019	-0.4222	-1.8107
96.0	0.2185	-0.6359	1.1619	1.4646	-2.6651
100.0	-2.2557	-2.2201	-1.7039	-1.7395	-1.8997
96.0	0.0406	0.0406	-0.0307	-0.1731	-0.1553
84.0	0.2007	0.2363	0.3075	0.0939	-0.0484
73.5	-0.0782	0.0879	0.0879	0.0879	-0.0782
54.0	0.0048	0.0048	0.0048	-0.0782	-0.1613
33.0	0.1710	0.1710	0.1710	0.1710	-0.0782
24.0	0.1685	0.2259	0.1972	0.2259	0.0248
10.0	0.3696	0.3409	0.3983	0.4556	0.2541
5.0	0.4845	0.4845	0.5133	0.5133	0.4202
2.5	0.5995	0.5420	0.5133	0.5133	0.5033

FUSELAGE TOP XLOC	CP
38.2	-0.1574
43.4	-0.1574
50.4	-0.1792
56.1	-0.2230
62.5	-0.2449
69.7	-0.2667
76.9	-0.3106
83.4	-0.2449
89.4	-0.2656
95.4	-0.2667
101.1	-0.2230
BOTTOM	
38.2	-0.0480
43.4	-0.0043
50.4	-0.0480
56.1	0.0613
62.5	0.1051
69.7	0.1488
76.9	0.1270
83.4	0.1051
89.4	0.0832
95.4	0.0832
101.1	-0.0043

MISC. NO.	CP
1	-0.7428
2	0.1710
3	-1.8228
4	0.0679
5	-1.2413
6	-0.1613
7	0.4499
8	-0.3510
9	-0.1375
10	0.4499
11	0.0639
12	-0.1553

TABLE A2. RUN 62, BWGV, DELF=0, CMU=2.0, (BN/B)=1

RUN:SEQ — PROPLULSIVE — WING / CANARD PRESSURES — PAGE 80

62 2 CLAERO = 0.3651 CDAERO = 0.0117 DCLC = 0.0 DWLC = 0.5330 BASEPR = 0.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.06	2112.25	2106.37	5.88	70.40	0.01	2.65	0.0	2.051	2.051	86.7146	0.450351E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	0.90	-1.97	-0.49	0.01	-0.01	0.04	0.32	-0.35	0.32	0.01

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3325	-0.2908
2.5	-0.3325	-0.3325
5.0	-0.3325	-0.3325
10.0	-0.3325	-0.3325
15.0	-0.3325	-0.3743
25.0	-0.3325	-0.3325
35.0	-0.3325	-0.3325
50.0	-0.3325	
56.0	-0.3325	-0.3325
65.0	-0.3325	-0.3325
76.0	-0.3325	-0.3325
79.0	-0.3325	-0.3325
80.5	-0.2928	-0.2928
81.0	-0.2325	-0.2325
82.0	-0.2626	-0.2928
84.0	-0.2626	
87.0	-0.2928	
89.0	-0.2888	-0.3334
93.0	-0.2888	
96.0	-0.3334	-0.2888
100.0	-0.3334	-0.3334
96.0	-0.2888	-0.2888
84.0	-0.3334	-0.3334
73.0	-0.2888	-0.2888
50.0	-0.2888	-0.2888
35.0	-0.3325	-0.3325
25.0	-0.3325	-0.3325
10.0	-0.3325	-0.3325
5.0	-0.3325	-0.3325
2.5	-0.3325	-0.3325

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5354	0.5354	0.1839	0.3011	0.4182
2.5	-0.3431	-0.2846	-0.6360	-0.75[illegible]	-0.5774
5.0	-0.3431	-0.2846	-0.5774	-0.5189	-0.6360
10.0	-0.4017	-0.4017	-0.5774	-0.4603	-0.5774
15.0	-0.3431	-0.4603	-0.5774	-0.4603	-0.5774
24.0	-0.3431	-0.4017	-0.5189	-0.4603	-0.5189
33.0	-0.3507	-0.5201	-0.5201	-0.5201	-0.5201
54.0	-0.5201	-0.5201	-0.6894	1.0039	-0.6894
65.0	-0.6900	-0.7791	-0.7791	-0.8237	-0.7791
78.5	-1.6260	-1.7152	-1.7697	-1.8043	-1.9826
79.5	267.1089	267.1089	267.1089	267.1089	248.4031
80.5	********	-86.8113	-114.3440	-104.9345	-74.2658
81.3	-44.4410	167.0733	-12.2045	-32.5591	0.3706
82.0	59.3262	51.6063	37.6433	23.1683	7.4574
84.0	20.5751	-10.9680	-1.8006	18.3444	1.3658
87.0	6.2813	-3.4593	-1.9212	2.2705	1.3356
89.0	0.6774	0.4597	-3.4225	-2.5880	2.7092
93.0	1.0402	1.2578	0.2420	-0.5199	-0.6288
96.0	0.8225	-0.7738	2.4552	3.3623	-2.6243
100.0	-4.2932	-4.2207	-3.2411	-3.2411	-3.2411
96.0	-0.2297	-0.2297	-0.4111	-0.4836	-0.2297
84.0	-0.0120	0.0243	-0.4474	-0.0483	-0.0483
73.5	-0.3507	-0.1814	-0.1814	-0.1814	-0.1814
54.0	-0.1814	-0.3507	-0.3507	-0.1814	-0.3507
33.0	-0.1814	-0.3507	-0.3507	-0.1814	-0.3507
24.0	-0.1675	-0.2260	-0.2260	-0.1089	-0.2846
10.0	-0.1675	-0.1675	-0.1089	-0.0503	-0.1814
5.0	-0.1089	-0.1675	-0.1089	-0.1089	-0.1814
2.5	-0.1675	-0.2260	-0.0503	0.0083	-0.0120

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.1551	38.2	-0.1551
43.4	-0.1551	43.4	-0.1105
50.4	-0.1551	50.4	-0.1551
56.1	-0.1551	56.1	-0.1105
62.5	-0.2888	62.5	-0.1105
69.7	-0.3334	69.7	-0.1105
76.9	-0.2442	76.9	-0.1551
83.4	-0.1996	83.4	-0.1551
89.4	-0.2442	89.4	-0.1551
95.4	-0.3334	95.4	-0.1105
101.1	-0.2442	101.1	-0.1996

MISC.*

NO.	CP
1	-0.5201
2	-0.3507
3	-0.5201
4	-0.1814
5	-0.5201
6	-0.3507
7	0.4234
8	-0.3386
9	-0.0846
10	0.6411
11	0.2057
12	-0.0483

TABLE A2. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
62 4	CLAERO =	0.5829	CDAERO =	0.0966	DCLC =	0.0		DWLC =	0.6839	BASEPR =	8.1904
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.02	2112.32	2106.44	5.88	70.36	0.01	2.65	0.0	2.099	2.099	86.6010	0.450830E+06
BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	1.27	-1.89	-0.55	0.01	-0.01	0.03	0.55	-0.41	0.55	0.06	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.3325	-0.3325
2.5	-0.3325	-0.3325
5.0	-0.3325	-0.3325
10.0	-0.3325	-0.3325
15.0	-0.3325	-0.3325
25.0	-0.3325	-0.3325
35.0	-0.3325	-0.3325
50.0	-0.3325	
56.0	-0.3325	-0.3325
65.0	-0.3325	-0.3325
76.0	-0.3325	-0.3325
79.0	-0.3325	-0.3325
80.5	-0.2626	-0.2626
81.0	-0.2324	-0.2626
82.0	-0.2625	-0.2626
84.0	-0.2324	
87.0	-0.2626	
89.0	-0.2888	-0.2888
93.0	-0.2888	
96.0	-0.3333	-0.3333
100.0	-0.2888	-0.3333
96.0	-0.2888	-0.3333
84.0	-0.2888	-0.3333
73.0	-0.2888	-0.3333
50.0	-0.3333	-0.3333
35.0	-0.3742	-0.3325
25.0	-0.3325	-0.3325
10.0	-0.3325	-0.3325
5.0	-0.3325	-0.3325
2.5	-0.3325	-0.3325

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.0083	-0.5188	-1.1630	-1.5728	-1.4558
2.5	-0.5773	-0.9287	-1.6313	-1.6899	-1.6313
5.0	-0.5188	-0.7530	-0.9287	-1.1043	-1.1043
10.0	-0.5188	-0.7530	-0.8115	-0.8115	-0.9287
15.0	-0.5188	-0.6359	-0.8115	-0.7530	-0.8702
24.0	-0.4602	-0.5773	-0.6944	-0.6359	-0.6359
33.0	-0.5200	-0.5200	-0.6893	-0.6893	-0.6893
54.0	-0.6893	-0.6893	-0.6893	0.8344	-0.8586
65.0	-0.7344	-0.8236	-0.8236	-0.9127	-0.9127
78.5	-1.6703	-1.8040	-1.8486	-1.9377	-2.1606
79.5	267.1191	267.1191	267.1191	267.1191	249.0005
80.5	********	-88.0331	-115.1397	-105.4016	-75.3704
81.3	-44.7354	149.8324	-12.1726	-30.6249	-0.2023
82.0	60.1303	52.7740	37.9693	21.7782	7.0944
84.0	23.9490	-8.6749	-1.5290	18.4624	1.4259
87.0	6.3104	-3.4587	-1.8607	2.3304	1.2450
89.0	0.6410	0.3508	-3.5671	-2.5877	2.5636
93.0	1.0400	1.1851	0.2782	-0.4836	-0.8464
96.0	0.7861	-0.8464	2.5636	3.2891	-2.8053
100.0	-4.2563	-4.2200	-3.2044	-3.3131	-3.2768
96.0	-0.1571	-0.1933	-0.3748	-0.4836	-0.1933
84.0	0.0606	0.0606	-0.3385	-0.0483	-0.0483
73.5	-0.3507	-0.0120	-0.1814	-0.1814	-0.1814
54.0	-0.1814	-0.3507	-0.1814	-0.3507	-0.3507
33.0	-0.0120	-0.0120	-0.1814	-0.1814	-0.1814
24.0	-0.0503	-0.0503	-0.0503	-0.0503	-0.1089
10.0	0.1254	0.0668	0.1254	0.1254	-0.0120
5.0	0.1254	0.1254	0.2425	0.2425	0.1572
2.5	0.1839	0.2425	0.3596	0.4182	0.3265

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1551
43.4	-0.1551
50.4	-0.1551
56.1	-0.1551
62.5	-0.2888
69.7	-0.2888
76.9	-0.2441
83.4	-0.2441
89.4	-0.2441
95.4	-0.3333
101.1	-0.2888
---BOTTOM---	
38.2	-0.1551
43.4	-0.1551
50.4	-0.1996
56.1	-0.1104
62.5	-0.0659
69.7	-0.0659
76.9	-0.1104
83.4	-0.1104
89.4	-0.1104
95.4	-0.1104
101.1	-0.1551

MISC.*

NO.	CP
1	-0.6893
2	-0.0120
3	-0.6893
4	-0.1814
5	-0.6893
6	-0.1814
7	0.4596
8	-0.3385
9	-0.1206
10	0.6047
11	0.2419
12	-0.0483

TABLE A2. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

PAGE 84

RUN:SEQ 62 8 CLAERO = 1.1089 CDAERO = 0.3144 DCLC = 0.0 DWLC = 0.9205 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.00	2112.32	2106.22	6.10	71.67	0.01	2.67	0.0	2.028	2.028	87.2420	0.460225E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	2.03	-1.49	-0.68	0.01	-0.01	0.03	1.12	-0.55	1.10	0.24

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3679	-0.3276
2.5	-0.3679	-0.3276
5.0	-0.3276	-0.2874
10.0	-0.3679	-0.3276
15.0	-0.3276	-0.3679
25.0	-0.3679	-0.3276
35.0	-0.3679	-0.3679
50.0	-0.3679	
56.0	-0.3276	-0.3679
65.0	-0.3679	-0.3679
76.0	-0.3679	-0.3679
79.0	-0.3679	-0.3276
80.5	-0.3363	-0.3072
81.0	-0.3072	-0.3363
82.0	-0.3072	-0.3363
84.0	-0.2782	
87.0	-0.2782	
89.0	-0.2453	-0.2882
93.0	-0.2882	
96.0	-0.2882	-0.3312
100.0	-0.2882	-0.3312
96.0	-0.3312	-0.3312
84.0	-0.2882	-0.3312
73.0	-0.3312	-0.3312
50.0	-0.2882	-0.3312
35.0	-0.3679	-0.3276
25.0	-0.3679	-0.3276
10.0	-0.3276	-0.3679
5.0	-0.3679	-0.3679
2.5	-0.3276	-0.3679

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.3430	-4.8092	-3.1174	-2.6098	-1.8766
2.5	-1.9332	-4.5273	-3.1174	-2.5534	-1.7076
5.0	-1.4256	-1.5946	-3.2866	-2.3842	-1.7639
10.0	-1.2000	-1.2000	-3.1738	-2.3278	-1.7076
15.0	-1.0872	-1.0872	-2.2715	-2.3278	-1.6512
24.0	-0.8616	-0.9744	-1.2563	-2.2151	-1.7076
33.0	-0.7941	-0.7941	-0.7941	-2.0985	-1.6092
54.0	-0.7941	-0.7941	-0.7941	0.5103	-2.0985
65.0	-0.8462	-0.9750	-0.7604	-0.9321	-1.7475
78.5	-1.7046	-1.8334	-1.7475	-1.4901	-3.1212
79.5	257.1965	257.1965	257.1965	257.1965	234.8652
80.5	-93.5741	-80.9142	-107.6284	-98.1617	-69.2127
81.3	-42.2954	150.2793	-11.3993	-37.6194	2.4513
82.0	53.6151	43.2490	34.1017	30.6463	7.8232
84.0	8.2877	-19.4429	-3.6756	18.5959	0.3896
87.0	6.1681	-3.0076	-1.9915	2.1319	-0.4815
89.0	0.5103	0.8247	-2.7388	-2.2844	1.8379
93.0	0.9645	1.5933	0.8247	-0.5377	-1.4112
96.0	1.4536	-0.2932	2.8859	3.1305	-3.4374
100.0	-3.0880	-3.2627	-2.3544	-3.1578	-2.4941
96.0	0.6850	0.5103	0.3007	-0.0487	0.3356
84.0	0.8247	0.7548	-0.1884	0.6151	0.4754
73.5	-0.1419	0.0212	0.0212	0.0212	-0.3050
54.0	0.0212	0.0212	-0.1419	-0.1419	-0.3050
39.0	0.1842	0.0212	0.0212	0.0212	-0.1419
24.0	0.1535	0.1535	0.0971	0.1535	-0.0721
10.0	0.3227	0.3791	0.3227	0.3791	0.1842
5.0	0.5483	0.4919	0.4919	0.4919	0.3472
2.5	0.6047	0.4919	0.4919	0.4919	0.3472

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1595
43.4	-0.1595
50.4	-0.2024
56.1	-0.2453
62.5	-0.3312
69.7	-0.2882
76.9	-0.3741
83.4	-0.2882
89.4	-0.3312
95.4	-0.3312
101.1	-0.2453
---BOTTOM---	
38.2	-0.0736
43.4	-0.0307
50.4	-0.1166
56.1	0.0122
62.5	0.0980
69.7	0.0980
76.9	0.0552
83.4	0.0552
89.4	0.0122
95.4	-0.0307
101.1	-0.0736

MISC.*

NO.	CP
1	-0.7941
2	0.1842
3	-1.9354
4	0.0212
5	-1.6092
6	-0.3050
7	1.0344
8	0.1260
9	0.4055
10	1.1042
11	0.6850
12	0.4055

TABLE A3. RUN 64, BW6V, DELF=15, CMU=0, (BN/B)=1

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 85

64	3	CLAERO =	0.4705	CDAERO =	0.1113	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
0.02		2115.72	2085.43	30.29	159.16	0.01	2.67	0.0	0.0	0.0	87.1804	0.103854E+07
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.47	0.1[illegible]	-0.31	0.00	-0.00	0.01	0.47	-0.31	0.47	0.12	

********** CANARD ******			********** WING **********						**FUSELAGE**			
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		**MISC.***	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.2152	-0.2152	0.0	0.5345	0.5004	0.4095	0.3413	0.2617	38.2	-0.0721	1	-0.2810
2.5	-0.2071	-0.2071	2.5	-0.2043	-0.2497	-0.4202	-0.6361	-0.5679	43.4	-0.0549	2	0.0147
5.0	-0.2152	-0.2071	5.0	-0.2156	-0.2497	-0.3406	-0.4998	-0.4429	50.4	-0.0836	3	-0.3139
10.0	-0.2071	-0.2071	10.0	-0.2838	-0.3293	-0.3975	-0.4088	-0.4202	56.1	-0.0549	4	0.0147
15.0	-0.2071	-0.2152	15.0	-0.2724	-0.2952	-0.3634	-0.4656	-0.4088	62.5	-0.1932	5	-0.2481
25.0	-0.2152	-0.2071	24.0	-0.2611	-0.2838	-0.3406	-0.3747	-0.3634	69.7	-0.1932	6	-0.0510
35.0	-0.2071	-0.2071	33.0	-0.2153	-0.2481	-0.3139	-0.3139	-0.3139	76.9	-0.1154	7	0.5217
50.0	-0.2071		54.0	-0.3139	-0.3139	-0.3796	0.1133	-0.3467	83.4	-0.1154	8	-0.2035
56.0	-0.2071	-0.2071	65.0	-0.4441	-0.4787	-0.4960	-0.5133	-0.4787	89.4	-0.1759	9	-0.0274
65.0	-0.2071	-0.2071	78.5	-1.1015	-1.1448	-1.1880	-1.2140	-1.2313	95.4	-0.2106	10	0.5288
76.0	-0.2152	-0.2071	79.5	-1.3988	-1.3988	-1.3637	-1.3988	-1.4456	101.1	-0.1759	11	0.2330
79.0	-0.2071	-0.2071	80.5	-1.0536	-1.2291	-1.2116	-1.2467	-1.4164	---BOTTOM---		12	-0.0345
80.5	-0.2050	-0.1992	81.5	-0.6322	52.6168	-0.8780	-0.9248	-1.1238	38.2	-0.0462		
81.0	-0.1933	-0.1933	82.0	-0.6088	-1.2291	-0.7961	-0.8253	-0.9775	43.4	-0.0116		
82.0	-0.1933	-0.1933	84.0	-0.6205	-0.6839	-0.7083	-0.6986	-0.7785	50.4	-0.0549		
84.0	-0.1933		87.0	-0.5093	-0.5562	-0.5012	-0.5503	-0.6439	56.1	0.0057		
87.0	-0.1933		89.0	-0.4640	-0.4992	-0.5344	-0.5274	-0.5555	62.5	0.0057		
89.0	-0.1846	-0.1932	93.0	-0.3091	-0.3091	-0.3232	-0.2246	-0.5485	69.7	0.0144		
93.0	-0.1932		96.0	-0.2246	-0.2105	-0.2246	-0.2387	-0.4288	76.9	0.0144		
96.0	-0.1932	-0.1932	100.0	-0.1471	-0.0908	-0.1471	-0.1401	-0.2457	83.4	0.0490		
100.0	-0.1932	-0.1846	96.0	0.1415	0.2260	0.2260	0.1979	0.1838	89.4	0.1009		
96.0	-0.2019	-0.1846	84.0	0.3739	0.5076	-0.2457	0.4231	0.5428	95.4	0.1268		
84.0	-0.1932	-0.1846	73.5	0.2776	0.3104	0.3433	0.3433	0.3104	101.1	0.0317		
73.0	-0.2019	-0.1846	54.0	0.0804	0.0476	0.0476	0.0804	0.0147				
50.0	-0.1932	-0.1846	33.0	0.0147	0.0147	0.0147	0.0147	-0.0181				
35.0	-0.2071	-0.2152	24.0	-0.0679	-0.0451	-0.0565	-0.0565	-0.0792				
25.0	-0.2071	-0.2071	10.0	-0.0565	-0.0451	-0.0110	-0.0338	-0.0101				
10.0	-0.2152	-0.2152	5.0	-0.0110	-0.0565	0.0117	0.0230	0.0476				
5.0	-0.2152	-0.2071	2.5	-0.0110	0.0230	0.0458	0.1140	0.2119				
2.5	-0.2152	-0.1990										

TABLE A3. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 87

84 5 CLMRD = 0.0000 CDMRD = 0.1000 BCLC = 0.0 BCLC = 0.0 BASFTR = 0.1004

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.04	2115.93	2085.53	30.40	159.58	0.01	2.66	0.0	0.0	0.0	87.0821	0.103826E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.65	0.15	-0.34	0.00	-0.00	0.01	0.66	-0.34	0.65	0.16

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2030	-0.2030
2.5	-0.1950	-0.2030
5.0	-0.1950	-0.2030
10.0	-0.1950	-0.1950
15.0	-0.1950	-0.1950
25.0	-0.1950	-0.1869
35.0	-0.2030	-0.1950
50.0	-0.2030	
56.0	-0.2030	-0.1950
65.0	-0.1950	-0.1950
76.0	-0.2111	-0.1950
79.0	-0.1950	-0.1950
80.5	-0.1951	-0.1835
81.0	-0.1835	-0.1893
82.0	-0.1835	-0.1893
84.0	-0.1835	
87.0	-0.1835	
89.0	-0.1806	-0.1892
93.0	-0.1892	
96.0	-0.1892	-0.1892
100.0	-0.1806	-0.1892
96.0	-0.1892	-0.1892
84.0	-0.1806	-0.1892
73.0	-0.1892	-0.1892
50.0	-0.1806	-0.1892
35.0	-0.2030	-0.1950
25.0	-0.2111	-0.1950
10.0	-0.2030	-0.2030
5.0	-0.2030	-0.2030
2.5	-0.2030	-0.1950

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0262	-0.4040	-0.8909	-1.3212	-1.7402
2.5	-0.6078	-0.8230	-1.1967	-1.1174	-1.5590
5.0	-0.5059	-0.6305	-0.8230	-0.9136	-0.9589
10.0	-0.4606	-0.5399	-0.6984	-0.7324	-0.8117
15.0	-0.4380	-0.4946	-0.5965	-0.6418	-0.6418
24.0	-0.3927	-0.4267	-0.4946	-0.5286	-0.5172
33.0	-0.3749	-0.4076	-0.4731	-0.5058	-0.5058
54.0	-0.3749	-0.4404	-0.4404	-0.1457	-0.5386
65.0	-0.4736	-0.5167	-0.5426	-0.5857	-0.6115
78.5	-1.0769	-1.1286	-1.1631	-1.2234	-1.3613
79.5	-1.3612	-1.3495	-1.3612	-1.3962	-1.5245
80.5	-1.0347	-1.1222	-1.1746	-1.2271	-1.5186
81.3	-0.6382	46.5919	-0.8481	-0.9298	-1.2971
82.0	-0.6266	-1.1338	-0.7723	-0.8248	-1.1233
84.0	-0.6207	-0.6674	-0.7024	-0.6907	-0.9531
87.0	-0.4867	-0.5391	-0.5741	-0.5586	-0.7898
89.0	-0.4379	-0.4519	-0.4870	-0.5011	-0.7045
93.0	-0.2906	-0.2836	-0.3117	-0.2205	-0.6764
96.0	-0.2064	-0.1924	-0.2205	-0.2826	-0.5431
100.0	-0.1363	-0.1152	-0.1784	-0.2556	-0.3467
96.0	0.1723	0.2355	0.2144	0.1794	0.1443
84.0	0.4249	0.3056	-0.2415	0.4249	0.2215
73.5	0.2798	0.2798	0.3453	0.3126	0.2471
54.0	0.0834	0.0507	0.0834	0.0834	-0.0475
33.0	0.0507	0.0507	0.0507	0.0507	-0.0475
24.0	0.0376	0.0489	0.0715	0.0829	0.0149
10.0	0.1508	0.1508	0.1846	0.2188	0.1489
5.0	0.2188	0.2188	0.2641	0.3094	0.2798
2.5	0.3433	0.3207	0.3886	0.4226	0.4108

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0945
43.4	-0.0686
50.4	-0.0686
56.1	-0.0772
62.5	-0.1892
69.7	-0.1806
76.9	-0.1548
93.4	-0.1376
89.4	-0.1892
95.4	-0.2237
101.1	-0.1806
---BOTTOM---	
38.2	-0.0614
43.4	-0.0169
50.4	-0.0427
56.1	0.0176
62.5	0.0607
69.7	0.0779
76.9	0.0693
83.4	0.0952
89.4	0.1296
95.4	0.1469
101.1	0.0435

MISC.*

NO.	CP
1	-0.4404
2	0.0507
3	-0.5386
4	0.0507
5	-0.4731
6	-0.0802
7	0.0231
8	-0.1784
9	-0.0591
10	0.4740
11	0.2706
12	-0.0311

TABLE A3. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES PAGE 89

RUN:SEQ											
64 9	CLAERO = 1.0835	CDAERO = 0.3162	DCLC = 0.0	DWLC = 0.0						BASEPR = 8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.03	2116.00	2085.83	30.17	159.09	0.01	2.67	0.0	0.0	0.0	87.3045	0.103263E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		CLTR	CDTR
-0.01	1.08	0.32	-0.44	0.01	-0.00	0.00	1.12	-0.44		1.08	0.32

************ CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2600	-0.2437
2.5	-0.2437	-0.2356
5.0	-0.2437	-0.2437
10.0	-0.2356	-0.2437
15.0	-0.2519	-0.2437
25.0	-0.2519	-0.2356
35.0	-0.2193	-0.2275
50.0	-0.2275	
56.0	-0.2275	-0.2193
65.0	-0.2193	-0.2193
76.0	-0.2275	-0.2193
79.0	-0.2275	-0.2193
80.5	-0.2326	-0.2326
81.0	-0.2326	-0.2267
82.0	-0.2267	-0.2326
84.0	-0.2267	
87.0	-0.2267	
89.0	-0.2146	-0.2407
93.0	-0.2233	
96.0	-0.2320	-0.2233
100.0	-0.2320	-0.2146
96.0	-0.2320	-0.2146
84.0	-0.2320	-0.2146
73.0	-0.2320	-0.2146
50.0	-0.2233	-0.2146
35.0	-0.2275	-0.2112
25.0	-0.2275	-0.2193
10.0	-0.2275	-0.2193
5.0	-0.2437	-0.2519
2.5	-0.2437	-0.2275

************************ WING ******************

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.8436	-5.5699	-5.1860	-2.8549	-1.6001
2.5	-1.6001	-2.2845	-3.1856	-2.9119	-1.5772
5.0	-1.1780	-1.6001	-2.3186	-2.7979	-1.6001
10.0	-0.9384	-1.1668	-1.5316	-3.0375	-1.4632
15.0	-0.8015	-1.0297	-1.1437	-2.4900	-1.5316
24.0	-0.6760	-0.7673	-0.8700	-1.5088	-1.5202
33.0	-0.6152	-0.6482	-0.7801	-1.0109	-1.4067
54.0	-0.5162	-0.5822	-0.6482	-0.3513	-1.4727
65.0	-0.5967	-0.6314	-0.6835	-0.6140	-1.1002
78.5	-1.1523	-1.1697	-1.1697	-1.1957	-1.4736
79.5	-1.2840	-1.3310	-1.3427	-1.4073	-1.5248
80.5	-1.2311	-1.2370	-0.9551	-1.2840	-1.5777
81.3	-0.8258	52.8320	-0.7671	-1.0138	-1.4954
82.0	-0.7377	-1.3545	-0.7436	-0.8611	-1.3545
84.0	-0.6614	-0.7025	-0.6908	-0.7436	-1.3251
87.0	-0.5439	-0.5815	-0.5791	-0.6320	-1.2605
89.0	-0.4832	-0.4902	-0.5255	-0.6033	-1.1474
93.0	-0.3206	-0.3065	-0.3347	-0.2499	-1.1333
96.0	-0.2358	-0.2217	-0.2923	-0.3277	-1.0344
100.0	-0.1510	-0.1439	-0.2288	-0.2075	-0.8153
96.0	0.1811	0.2730	0.2165	0.1882	-0.0097
84.0	0.4568	0.1953	-0.2358	0.4285	0.0610
73.5	0.3083	0.3413	0.4072	0.3743	0.2423
54.0	0.2094	0.1764	0.2094	0.1764	0.0115
33.0	0.2423	0.2423	0.2423	0.2423	0.0774
24.0	0.2822	0:2938	0.3164	0.3164	0.1681
10.0	0.4762	0.4762	0.4762	0.4762	0.3743
5.0	0.5788	0.5332	0.5560	0.5560	0.4402
2.5	0.6701	0.5674	0.5104	0.5218	0.4402

FUSELAGE

XLOC	CP
-----TOP----	
38.2	-0.1018
43.4	-0.0931
50.4	-0.1278
56.1	-0.1625
62.5	-0.2146
69.7	-0.2146
76.9	-0.2667
83.4	-0.2146
89.4	-0.2494
95.4	-0.2146
101.1	-0.1452
---BOTTOM---	
38.2	0.0024
43.4	0.0458
50.4	0.0024
56.1	0.1066
62.5	0.1761
69.7	0.2195
76.9	0.2195
83.4	0.2021
89.4	0.2108
95.4	0.2021
101.1	0.0806

MISC.*

NO.	CP
1	-0.7141
2	0.2753
3	-1.5386
4	0.2094
5	-1.3078
6	-0.0215
7	0.5627
8	-0.1581
9	0.0186
10	0.4638
11	0.1811
12	-0.0379

TABLE A4. RUN 65, BW6V, DELF=15,CMU=0.5, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
65 4	CLAERO =	0.6761	CDAERO =	0.2135	DCLC =	0.0	DWLC =	0.2475	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
-0.03	2116.78	2088.98	27.80	152.70	0.01	2.67	0.0	0.495	0.495	87.3269	0.990272E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	0.92	-0.22	-0.61	0.01	-0.00	0.02	0.69	-0.92	0.69	0.14		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2802	-0.2714
2.5	-0.2714	-0.2714
5.0	-0.2714	-0.2714
10.0	-0.2714	-0.2714
15.0	-0.2802	-0.2626
25.0	-0.2802	-0.2626
35.0	-0.2714	-0.2714
50.0	-0.2714	
56.0	-0.2802	-0.2802
65.0	-0.2714	-0.2714
76.0	-0.2714	-0.2714
79.0	-0.2714	-0.2714
80.5	-0.2814	-0.2686
81.0	-0.2623	-0.2686
82.0	-0.2623	-0.2623
84.0	-0.2623	
87.0	-0.2559	
89.0	-0.2439	-0.2627
93.0	-0.2627	
96.0	-0.2627	-0.2627
100.0	-0.2627	-0.2627
96.0	-0.2627	-0.2627
84.0	-0.2627	-0.2627
73.0	-0.2627	-0.2627
50.0	-0.2533	-0.2627
35.0	-0.2714	-0.2714
25.0	-0.2714	-0.2714
10.0	-0.2626	-0.2714
5.0	-0.2714	-0.2714
2.5	-0.2714	-0.2714

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4924	0.3809	0.0961	0.1457	-0.1639
2.5	-0.2753	-0.3744	-0.6468	-0.7582	-0.8696
5.0	-0.3000	-0.3248	-0.5229	-0.6220	-0.7954
10.0	-0.3248	-0.3991	-0.5353	-0.5972	-0.6096
15.0	-0.3125	-0.3991	-0.4858	-0.5229	-0.5477
24.0	-0.3248	-0.3867	-0.4239	-0.4239	-0.4858
33.0	-0.3603	-0.3961	-0.4319	-0.5035	-0.5035
54.0	-0.4676	-0.5035	-0.5751	-0.3245	-0.5751
65.0	-0.6774	-0.7340	-0.7528	-0.7999	-0.7811
78.5	-1.6386	-1.7517	-1.8271	-1.9025	-1.9684
79.5	-5.2477	-12.3307	-23.2325	-18.8719	-20.9184
80.5	-8.3845	-3.1757	0.3498	5.3416	-1.6075
81.3	3.6904	42.1274	-2.1494	-6.2042	-5.9617
82.0	-3.7688	-2.4490	1.7970	-0.8424	-3.3098
84.0	-1.2568	-1.0337	-1.7414	-2.0474	-1.2058
87.0	0.4772	-1.6903	-1.5182	-0.5173	-0.9126
89.0	-0.7028	-0.7718	-1.6923	-1.5389	-0.3039
93.0	-0.4190	-0.3423	-0.6107	-0.3576	-1.1093
96.0	-0.3576	-0.6644	0.0489	0.1716	-1.3701
100.0	-0.9559	-0.8025	-0.6951	-0.7028	-0.7872
96.0	0.1793	0.2713	0.3020	0.2177	0.1716
84.0	0.4094	0.2253	0.1716	0.4478	0.2790
73.5	0.2125	0.3199	0.3557	0.3557	0.2125
54.0	0.0693	0.0693	0.0693	0.0693	-0.0381
33.0	-0.0023	-0.0023	-0.0023	0.0335	-0.0739
24.0	-0.0029	0.0095	0.0342	0.0219	-0.0153
10.0	0.0219	0.0342	0.0714	0.1209	0.0335
5.0	0.0714	0.0590	0.1085	0.1457	0.1409
2.5	0.0838	0.1457	0.1085	0.2448	0.2483

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1026
43.4	-0.0837
50.4	-0.0837
56.1	-0.0931
62.5	-0.2627
69.7	-0.2627
76.9	-0.1780
83.4	-0.1780
89.4	-0.2627
95.4	-0.3004
101.1	-0.2345
---BOTTOM---	
38.2	-0.0649
43.4	-0.0272
50.4	-0.0743
56.1	0.0011
62.5	0.0200
69.7	0.0294
76.9	0.0388
83.4	0.0765
89.4	0.1236
95.4	0.1424
101.1	0.0388

MISC.*

NO.	CP
1	-0.4319
2	-0.0023
3	-0.5035
4	-0.0023
5	-0.4319
6	-0.1097
7	0.4784
8	-0.4650
9	-0.1735
10	0.7009
11	0.2177
12	-0.0662

TABLE A4. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
65	6	CLAERO =	0.7718	CDAERO =	0.2694	DCLC =	0.0	DWLC =	0.2816	BASEPR =	8.1904	
ALPHA		PTOT		PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.04		2116.92		2088.89	28.03	153.32	0.01	2.65	0.0	0.503	0.503	86.8381	0.994294E+06
BETA		CL		CD	CN	CROLL		CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01		1.05		-0.15	-0.62	0.01		-0.01	0.03	0.80	-0.93	0.79	0.19

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2796	-0.2796
2.5	-0.2708	-0.2708
5.0	-0.2796	-0.2708
10.0	-0.2708	-0.2796
15.0	-0.2708	-0.2796
25.0	-0.2708	-0.2708
35.0	-0.2796	-0.2708
50.0	-0.2796	
56.0	-0.2796	-0.2884
65.0	-0.2796	-0.2884
76.0	-0.2796	-0.2796
79.0	-0.2796	-0.2796
80.5	-0.2720	-0.2467
81.0	-0.2593	-0.2467
82.0	-0.2657	-0.2593
84.0	-0.2657	
87.0	-0.2530	
89.0	-0.2441	-0.2628
93.0	-0.2628	
96.0	-0.2535	-0.2628
100.0	-0.2535	-0.2722
96.0	-0.2628	-0.2628
84.0	-0.2535	-0.2722
73.0	-0.2628	-0.2628
50.0	-0.2628	-0.2535
35.0	-0.2708	-0.2796
25.0	-0.2796	-0.2708
10.0	-0.2796	-0.2796
5.0	-0.2708	-0.2796
2.5	-0.2708	-0.2796

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.1186	-0.7449	-2.0469	-2.0960	-3.1276
2.5	-0.7449	-0.9537	-1.4082	-1.5433	-1.7644
5.0	-0.5853	-0.7327	-0.9906	-1.2485	-1.3713
10.0	-0.5484	-0.7081	-0.9169	-0.9414	-0.9906
15.0	-0.5116	-0.6221	-0.7572	-0.8186	-0.8678
24.0	-0.4748	-0.5362	-0.6344	-0.6958	-0.6958
33.0	-0.4923	-0.5633	-0.6343	-0.6698	-0.6698
54.0	-0.5633	-0.5988	-0.7053	-0.2437	-0.7763
63.0	-0.7302	-0.8050	-0.8424	-0.8985	-0.9452
78.5	-1.6744	-1.7866	-1.9081	-2.0202	-2.2166
79.5	-4.9517	-11.9964	-22.8928	-18.4785	-20.6098
80.5	-8.9927	-3.9336	-0.4301	4.7618	-1.1447
81.3	3.9081	56.8064	-2.2451	-6.8300	-6.7415
82.0	-4.0728	-2.8774	2.1627	0.1517	-3.0041
84.0	-1.3660	-0.9993	-2.1376	-1.3281	-1.3534
87.0	0.4955	-1.7265	-1.5494	-0.5692	-1.0056
89.0	-0.6976	-0.7813	-1.6106	-1.5954	-0.5150
93.0	-0.4237	-0.3552	-0.6443	-0.3704	-1.4052
96.0	-0.3628	-0.6748	-0.0052	0.0881	-1.6106
100.0	-0.9259	-0.7737	-0.6976	-0.6824	-0.8650
96.0	0.2231	0.2839	0.2611	0.2231	0.1241
84.0	0.4589	0.2611	0.1546	0.4437	0.2383
73.5	0.2179	0.3600	0.3600	0.3600	0.2179
54.0	0.1114	0.1114	0.1114	0.1114	-0.0307
33.0	0.0759	0.0759	0.0759	0.1114	-0.0307
24.0	0.0780	0.0902	0.1148	0.1271	0.0534
10.0	0.1762	0.1885	0.2499	0.2622	0.2179
5.0	0.2499	0.2990	0.3359	0.3481	0.3245
2.5	0.3481	0.3727	0.4587	0.4956	0.4310

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1226
43.4	-0.0946
50.4	-0.1039
56.1	-0.1133
62.5	-0.2628
69.7	-0.2722
76.9	-0.2348
83.4	-0.2255
89.4	-0.3189
95.4	-0.3189
101.1	-0.2722
---BOTTOM---	
38.2	-0.0479
43.4	-0.0105
50.4	-0.0572
56.1	0.0269
62.5	0.0643
69.7	0.0924
76.9	0.1111
83.4	0.1297
89.4	0.1671
95.4	0.1858
101.1	0.0643

MISC.*

NO.	CP
1	-0.5988
2	0.0759
3	-0.7063
4	0.0759
5	-0.6698
6	-0.1017
7	0.4665
8	-0.4998
9	-0.2182
10	0.6567
11	0.2078
12	-0.1117

TABLE A4. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 96

65 10 CLAERO = 1.0131 CDAERO = 0.4264 DCLC = 0.0 DWLC = 0.3421 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.00	2117.34	2089.77	27.57	152.03	0.01	2.67	0.0	0.511	0.511	87.3361	0.986863E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.36	0.05	-0.65	0.01	-0.01	0.06	1.09	-0.97	1.04	0.35

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3201	-0.3023
2.5	-0.3112	-0.3112
5.0	-0.3112	-0.3112
10.0	-0.3112	-0.3112
15.0	-0.3112	-0.3112
25.0	-0.3112	-0.3201
35.0	-0.3112	-0.3023
50.0	-0.3112	
56.0	-0.3112	-0.3023
65.0	-0.3023	-0.3023
76.0	-0.3112	-0.2934
79.0	-0.3112	-0.3023
80.5	-0.3139	-0.2946
81.0	-0.3074	-0.2946
82.0	-0.3074	-0.2882
84.0	-0.3010	
87.0	-0.2946	
89.0	-0.2663	-0.2948
93.0	-0.2948	
96.0	-0.2948	-0.2948
100.0	-0.2948	-0.2948
96.0	-0.2948	-0.2948
84.0	-0.2948	-0.2948
73.0	-0.2948	-0.2948
50.0	-0.2948	-0.2948
35.0	-0.3112	-0.3112
25.0	-0.3023	-0.3112
10.0	-0.3023	-0.3112
5.0	-0.3112	-0.3201
2.5	-0.3023	-0.3112

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.2469	-6.3052	-7.7907	-3.7586	-2.2482
2.5	-1.8363	-2.5229	-4.4828	-3.9457	-2.1234
5.0	-1.3620	-1.8238	-3.1718	-4.1956	-2.2233
10.0	-1.1248	-1.4369	-1.8740	-4.0085	-2.1858
15.0	-1.0000	-1.1997	-1.5866	-1.8738	-2.1609
24.0	-0.8751	-0.9625	-1.1872	-0.9001	-2.1109
33.0	-0.7350	-0.8433	-0.9876	-0.8072	-2.2507
54.0	-0.6989	-0.7711	-0.9154	-0.3380	-2.3230
65.0	-0.8838	-0.9503	-1.0358	-0.9893	-1.9954
78.5	-1.8054	-1.9479	-1.8719	-2.2424	-2.4134
79.5	-4.8451	-12.0440	-23.1566	-18.7220	-20.9843
80.5	-10.0192	-4.8836	-0.6352	3.8061	-0.9438
81.3	4.0310	57.7065	-1.5857	-7.6989	-7.5767
82.0	-4.4916	-3.3477	2.3085	0.9973	-2.8142
84.0	-2.0621	-0.9823	-2.0364	-0.5581	-2.1778
87.0	0.4574	-1.8693	-1.5929	-0.8024	-2.0428
89.0	-0.7581	-0.8277	-1.6706	-1.7324	-1.4540
93.0	-0.4642	-0.4023	-0.7504	-0.3946	-2.5057
96.0	-0.4023	-0.7039	-0.0466	-0.0234	-2.6449
100.0	-0.8586	-0.6962	-0.6498	-0.6653	-0.9050
96.0	0.2395	0.3709	0.3091	0.2627	0.0848
84.0	0.5024	0.2395	0.1390	0.4792	0.1158
73.5	0.3116	0.4199	0.4560	0.3838	0.2033
54.0	0.2394	0.2394	0.2394	0.2394	0.0229
33.0	0.2755	0.2755	0.3116	0.3116	0.0951
24.0	0.2983	0.3232	0.3482	0.3357	0.1859
10.0	0.4855	0.4855	0.4980	0.5105	0.3838
5.0	0.5854	0.5729	0.5479	0.5604	0.4560
2.5	0.6728	0.5479	0.4605	0.4605	0.3838

FUSELAGE

XLOC	-----TOP---- CP
38.2	-0.1238
43.4	-0.1238
50.4	-0.1523
56.1	-0.2093
62.5	-0.2948
69.7	-0.2948
76.9	-0.3613
83.4	-0.3043
89.4	-0.3613
95.4	-0.3138
101.1	-0.2378
---BOTTOM---	
38.2	-0.0003
43.4	0.0377
50.4	-0.0193
56.1	0.1137
62.5	0.1897
69.7	0.2468
76.9	0.2468
83.4	0.2563
89.4	0.2563
95.4	0.2658
101.1	0.1137

MISC.*

NO.	CP
1	-0.8793
2	0.3116
3	-2.2868
4	0.2394
5	-1.8538
6	-0.0493
7	0.4869
8	-0.5335
9	-0.1472
10	0.6880
11	0.0616
12	-0.1395

TABLE A5. RUN 66, BW6V, DELF=15, CMU=1.0, (BN/3)=1

PROPLULSIVE WING / CANARD PRESSUPES

RUN:SEQ													
66	3	CLAERO =	0.7803	CDAERO =	0.3068	DCLC =	0.0	DWLC =	0.5284	BASEPR =	8.1904		
ALPHA		PTOT		PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.09		2117.77		2104.43	13.34	105.36	0.01	2.66	0.0	1.054	1.054	86.8868	0.689635E+06
BETA		CL		CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		1.31		-0.60	-0.85	0.00	-0.00	0.01	0.81	-1.50	0.81	0.15	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3756	-0.3572
2.5	-0.3756	-0.3388
5.0	-0.3572	-0.3572
10.0	-0.3572	-0.3572
15.0	-0.3572	-0.3572
25.0	-0.3572	-0.3572
35.0	-0.3572	-0.3572
50.0	-0.3572	
56.0	-0.3572	-0.3572
65.0	-0.3756	-0.3756
76.0	-0.3756	-0.3756
79.0	-0.3756	-0.3572
80.5	-0.3507	-0.3242
81.0	-0.3375	-0.2976
82.0	-0.3507	-0.2976
84.0	-0.3375	
87.0	-0.3242	
89.0	-0.3114	-0.3310
93.0	-0.3310	
96.0	-0.3310	-0.3310
100.0	-0.3310	-0.3310
96.0	-0.3310	-0.3310
84.0	-0.3310	-0.3310
73.0	-0.3310	-0.3310
50.0	-0.3310	-0.3310
35.0	-0.3756	-0.3572
25.0	-0.3756	-0.3756
10.0	-0.3756	-0.3756
5.0	-0.3572	-0.3756
2.5	-0.3572	-0.3572

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4112	0.3338	0.1273	-0.1308	-0.2599
2.5	-0.3632	-0.4664	-0.6987	-0.9052	-1.0859
5.0	-0.3374	-0.4406	-0.5955	-0.7503	-0.8794
10.0	-0.4148	-0.4564	-0.6213	-0.6729	-0.7245
15.0	-0.4406	-0.4922	-0.6213	-0.6213	-0.6213
24.0	-0.4148	-0.4922	-0.5696	-0.5696	-0.5955
33.0	-0.4131	-0.4131	-0.5623	-0.5623	-0.5623
54.0	-0.5623	-0.5623	-0.7115	-1.4577	-0.7115
65.0	-0.8222	-0.9007	-0.9204	-0.9793	-0.9400
78.5	-2.0991	-2.2367	-2.3348	-2.4134	-2.5510
79.5	-10.5977	-25.5890	-48.6217	-39.3446	-43.0796
80.5	-12.1125	-1.4007	3.1048	14.7072	-3.9392
81.3	8.4209	118.3052	1.4833	-7.2088	-8.0060
82.0	-5.4410	-4.8296	3.0117	-6.2517	-4.0455
84.0	0.6726	-0.8292	-0.2577	-5.1884	-0.9355
87.0	1.9617	-2.5836	-1.9855	0.1809	-0.4571
89.0	-0.6421	-0.6901	-2.2731	-1.7614	0.4772
93.0	-0.3063	-0.1144	-0.5941	-0.3063	-1.2497
96.0	-0.2583	-0.8979	0.9249	0.9729	-1.8893
100.0	-1.8094	-1.6335	-0.7221	-1.2337	-1.3457
96.0	0.1255	0.2054	0.5731	0.1894	0.0775
84.0	0.3973	-0.4182	0.9889	0.5092	-0.3702
73.5	0.1093	0.3332	0.3332	0.3332	0.1840
54.0	0.0347	0.0347	0.0347	0.0347	-0.0399
33.0	-0.0399	-0.0399	-0.0399	-0.0399	-0.1145
24.0	-0.0793	-0.0534	-0.0534	-0.0534	-0.1308
10.0	-0.0276	-0.0276	0.0240	0.0757	-0.0399
5.0	-0.0276	-0.0276	0.0757	0.1015	0.0347
2.5	-0.0018	0.0240	0.1273	0.2047	0.1840

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1542
43.4	-0.1149
50.4	-0.1149
56.1	-0.1345
62.5	-0.3703
69.7	-0.3507
76.9	-0.2524
83.4	-0.2524
89.4	-0.3310
95.4	-0.3507
101.1	-0.2917
---BOTTOM---	
38.2	-0.1149
43.4	-0.0559
50.4	-0.1345
56.1	-0.0364
62.5	-0.0364
69.7	-0.0167
76.9	0.0029
83.4	0.0422
89.4	0.0815
95.4	0.1012
101.1	-0.0364

MISC.*

NO.	CP
1	-0.4877
2	-0.0399
3	-0.5623
4	-0.0399
5	-0.5623
6	-0.1145
7	0.4292
8	-0.5782
9	-0.2424
10	0.7330
11	0.1255
12	-0.1464

TABLE A5. (Continued)

RUN:SEQ PROPULSIVE WING / CANARD PRESSURES PAGE 104

RUN	SEQ																	
66	5	CLAERO =	0.9923	CDAERO =	0.3862	DCLC =	0.0	DWLC =	0.5921	BASEPR =	8.1904							
ALPHA		PTOT		PSTAT		Q		VEL		YAW		H/C		CMUC		CMUW		CMUT
4.02		2118.12		2104.67		13.45		105.76		0.01		2.66		0.0		1.058		1.058
BETA		CL		CD		CM		CROLL		CN		CY		CNTR		CMTR		CLTR
-0.01		1.58		-0.49		-0.91		0.00		-0.01		0.02		1.05		-1.56		1.03

HQT	RN	CDTR
87.1585	0.693369E+06	0.22

CANARD BP=3.375 %X/C	CP	BP=10.125 CP
0.0	-0.3724	-0.3541
2.5	-0.3724	-0.3724
5.0	-0.3724	-0.3541
10.0	-0.3359	-0.3724
15.0	-0.3541	-0.3724
25.0	-0.3541	-0.3541
35.0	-0.3541	-0.3359
50.0	-0.3359	
56.0	-0.3541	-0.3359
65.0	-0.3541	-0.3541
76.0	-0.3541	-0.3359
79.0	-0.3541	-0.3359
80.5	-0.3609	-0.3346
81.0	-0.3346	-0.3346
82.0	-0.3346	-0.3214
84.0	-0.3478	
87.0	-0.3478	
89.0	-0.3282	-0.3477
93.0	-0.3477	
96.0	-0.3477	-0.3477
100.0	-0.3477	-0.3282
96.0	-0.3477	-0.3477
84.0	-0.3477	-0.3282
73.0	-0.3477	-0.3477
50.0	-0.3477	-0.3477
35.0	-0.3359	-0.3541
25.0	-0.3541	-0.3724
10.0	-0.3541	-0.3541
5.0	-0.3541	-0.3541
2.5	-0.3541	-0.3359

WING %X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.1297	-0.5136	-1.9981	-2.2284	-3.1242
2.5	-0.7951	-1.0255	-1.3327	-1.4862	-1.9725
5.0	-0.6160	-0.7951	-1.0511	-1.2303	-1.4094
10.0	-0.6160	-0.7440	-0.9488	-0.9999	-1.0767
15.0	-0.5648	-0.6672	-0.8208	-0.9231	-0.9231
24.0	-0.5136	-0.5904	-0.6928	-0.7696	-0.8208
33.0	-0.5575	-0.5575	-0.7055	-0.7795	-0.7795
54.0	-0.6315	-0.7055	-0.8534	-1.2974	-0.8534
65.0	-0.8931	-0.9710	-1.0100	-1.0879	-1.1463
78.5	-2.1203	-2.2956	-2.4125	-2.5488	-2.8410
79.5	-10.4818	-25.2673	-48.1701	-38.9590	-43.0307
80.5	-12.7087	-2.0345	2.7095	14.1877	-3.9190
81.3	8.4683	117.3322	1.0622	-8.1225	-8.7813
82.0	-5.6188	-4.3011	2.9599	-5.6980	-4.6174
84.0	0.5220	-0.7958	-0.3478	-0.4871	-1.3756
87.0	1.9056	-2.5617	-2.0609	-0.0579	-0.9276
89.0	-0.6842	-0.7318	-2.4758	-2.1112	0.0768
93.0	-0.3196	-0.1811	-0.6525	-0.4464	-1.5879
96.0	-0.3037	-0.9379	0.4573	0.7903	-2.1904
100.0	-1.7624	-1.6355	-1.3818	-1.3184	-1.5087
96.0	0.1561	0.2512	0.1561	0.1085	0.0451
84.0	0.4256	-0.3354	0.6793	0.4256	-0.2403
73.5	0.1084	0.3304	0.3304	0.3304	0.1084
54.0	0.1084	0.0344	0.0344	0.1084	-0.1135
33.0	0.0344	0.0344	0.0344	0.0344	-0.0396
24.0	0.0238	0.0750	0.1006	0.1006	0.0238
10.0	0.1518	0.1773	0.2030	0.2542	0.1824
5.0	0.2286	0.2542	0.3053	0.3565	0.3304
2.5	0.3053	0.3821	0.4077	0.4589	0.4044

FUSELAGE TOP XLOC	CP
38.2	-0.1334
43.4	-0.1139
50.4	-0.1334
56.1	-0.1723
62.5	-0.3672
69.7	-0.3477
76.9	-0.3087
83.4	-0.2892
89.4	-0.3282
95.4	-0.3672
101.1	-0.3087
BOTTOM	
38.2	-0.0944
43.4	-0.0361
50.4	-0.1139
56.1	-0.0166
62.5	0.0224
69.7	0.0613
76.9	0.0613
83.4	0.1003
89.4	0.1393
95.4	0.1393
101.1	0.0029

MISC. NO.	CP
1	-0.6315
2	0.0344
3	-0.7795
4	0.0344
5	-0.7795
6	-0.1135
7	0.4414
8	-0.6208
9	-0.2086
10	0.7427
11	0.1402
12	-0.1451

TABLE A5. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
66 10	CLAERO =	1.5637	CDAERO =	0.6391	DCLC =	0.0	DWLC =	0.7155	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
11.95	2118.26	2105.04	13.22	104.78	0.01	2.66	0.0	1.070	1.070	87.1563	0.688724E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	2.28	-0.16	-1.11	0.00	-0.01	0.02	1.69	-1.77	1.62	0.49		

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.4049	-0.3678
2.5	-0.3864	-0.3678
5.0	-0.3864	-0.3864
10.0	-0.3864	-0.3864
15.0	-0.3864	-0.3864
25.0	-0.3864	-0.3678
35.0	-0.3678	-0.3678
50.0	-0.3678	
56.0	-0.3678	-0.3864
65.0	-0.3864	-0.3864
76.0	-0.3864	-0.3678
79.0	-0.3864	-0.3864
80.5	-0.3881	-0.3613
81.0	-0.3747	-0.3479
82.0	-0.3479	-0.3613
84.0	-0.3613	
87.0	-0.3613	
89.0	-0.3414	-0.3612
93.0	-0.3612	
96.0	-0.3610	-0.3612
100.0	-0.3612	-0.3612
96.0	-0.3612	-0.3612
84.0	-0.3612	-0.3612
73.0	-0.3612	-0.3612
50.0	-0.3612	-0.3612
35.0	-0.3864	-0.3678
25.0	-0.3864	-0.3678
10.0	-0.3678	-0.3678
5.0	-0.3864	-0.3864
2.5	-0.3864	-0.3678

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-3.5236	-5.9704	-4.7991	-3.2372	-2.2481
2.5	-1.9356	-2.6385	-4.6168	-3.0290	-2.1699
5.0	-1.4671	-1.9617	-4.0442	-3.1591	-2.2481
10.0	-1.2849	-1.4932	-2.5207	-3.3673	-2.1960
15.0	-1.0766	-1.2849	-2.4563	-3.4454	-2.2481
24.0	-0.8944	-1.0246	-1.3369	-2.4563	-2.1960
33.0	-0.8004	-0.8757	-0.8757	-1.4777	-2.0798
54.0	-0.8004	-0.8004	-0.9509	-1.2519	-2.7571
65.0	-1.0546	-1.1537	-1.1537	-1.1140	-2.9368
78.5	-2.3027	-2.4613	-2.5801	-2.6198	-3.4122
79.5	-10.4001	-25.4246	-48.8134	-39.3504	-43.8274
80.5	-13.8448	-2.7202	2.6141	14.4755	-3.9265
81.3	8.6992	113.3101	0.4831	-9.3012	-9.5289
82.0	-6.1112	-4.2347	3.0967	-4.1677	-4.5430
84.0	0.1078	-0.7231	-0.7634	-4.8111	-2.3583
87.0	1.8904	-2.6666	-2.1841	-0.0530	-2.4522
89.0	-0.7358	-0.8486	-2.6224	-2.2354	-1.4291
93.0	-0.3809	-0.2359	-0.7358	-0.4938	-3.3481
96.0	-0.3326	-0.9937	0.3124	0.7317	-3.7835
100.0	-1.7032	-1.5581	-1.3324	-1.3647	-1.5581
96.0	0.2157	0.3124	0.2157	0.1190	-0.0262
84.0	0.4737	-0.1069	0.6349	0.4414	-0.1230
73.5	0.3285	0.4790	0.4790	0.4790	0.1779
54.0	0.2532	0.3285	0.3285	0.2532	0.0275
33.0	0.3285	0.3285	0.3285	0.3285	0.0275
24.0	0.2770	0.3290	0.3290	0.3550	0.1468
10.0	0.4331	0.4852	0.5112	0.5112	0.4037
5.0	0.5633	0.5633	0.5373	0.5633	0.5543
2.5	0.6674	0.5373	0.4592	0.4592	0.4037

FUSELAGE

----TOP----	
XLOC	CP
38.2	-0.1829
43.4	-0.1829
50.4	-0.2027
56.1	-0.2621
62.5	-0.3612
69.7	-0.3612
76.9	-0.4206
83.4	-0.3612
89.4	-0.4206
95.4	-0.3612
101.1	-0.3017
---BOTTOM---	
38.2	-0.0442
43.4	-0.0046
50.4	-0.0840
56.1	0.0746
62.5	0.1539
69.7	0.1935
76.9	0.2133
83.4	0.2332
89.4	0.2332
95.4	0.2133
101.1	0.0548

MISC.*

NO.	CP
1	-0.8757
2	0.3285
3	-2.6066
4	0.2532
5	-2.0798
6	-0.0478
7	0.3769
8	-0.7680
9	-0.3003
10	0.7478
11	-0.0262
12	-0.2359

TABLE A6. RUN 67, BW6V, DELF=15, CMU=2.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ													
67	2	CLAERO =	0.8491	CDAERO =	0.4339	DCLC =	0.0	DWLC =	1.0115	BASEPR =	8.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
0.03		2118.76	2111.75	7.01	76.09	0.01	2.66	0.0	2.021	2.021	87.1544	0.503651E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01		1.86	-1.32	-1.19	0.00	-0.01	0.03	0.90	-2.43	0.90	0.13		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4371	-0.4371
2.5	-0.4371	-0.4371
5.0	-0.4021	-0.4371
10.0	-0.4021	-0.4371
15.0	-0.4371	-0.4371
25.0	-0.4371	-0.4371
35.0	-0.4371	-0.4371
50.0	-0.4371	
56.0	-0.4371	-0.4371
65.0	-0.4021	-0.4371
76.0	-0.4371	-0.4371
79.0	-0.4371	-0.4722
80.5	-0.4174	-0.4174
81.0	-0.4174	-0.3921
82.0	-0.3921	-0.4174
84.0	-0.4174	
87.0	-0.4174	
89.0	-0.3843	-0.4217
93.0	-0.4591	
96.0	-0.4217	-0.4591
100.0	-0.4217	-0.4217
96.0	-0.4217	-0.4217
84.0	-0.4217	-0.4217
73.0	-0.4217	-0.4217
50.0	-0.4217	-0.4217
35.0	-0.4371	-0.4371
25.0	-0.4371	-0.4371
10.0	-0.4371	-0.4371
5.0	-0.4371	-0.4371
2.5	-0.4021	-0.4021

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4586	0.3603	-0.1799	-0.2291	-0.7202
2.5	-0.3764	-0.4746	-0.8676	-1.0149	-1.2605
5.0	-0.4255	-0.4746	-0.7693	-0.9167	-0.9658
10.0	-0.4746	-0.5237	-0.7202	-0.8185	-0.8676
15.0	-0.4746	-0.5237	-0.6711	-0.7693	-0.7693
24.0	-0.4746	-0.4746	-0.6220	-0.7202	-0.7202
33.0	-0.5632	-0.5632	-0.7052	-0.7052	-0.8472
54.0	-0.8472	-0.8472	-0.9892	-3.5452	-0.9892
65.0	-0.9825	-1.1693	-1.1693	-1.2067	-1.2067
78.5	-2.7020	-2.8889	-2.9637	-3.1506	-3.3750
79.5	-20.5228	-48.6715	-92.8023	-75.0478	-81.8773
80.5	-18.5508	2.9463	8.3583	30.5131	-7.3975
81.3	16.5273	224.0789	7.7766	-6.7147	-12.3037
82.0	-8.3584	-5.5260	5.9305	-16.0978	-5.5513
84.0	2.7439	-0.9991	0.9736	-8.0550	-0.5438
87.0	4.6154	-3.8569	-2.7694	1.0747	-0.2909
89.0	-0.5530	-0.5530	-3.6565	-2.8958	1.5161
93.0	-0.0661	0.2686	-0.4313	-0.6442	-1.7397
96.0	-0.1270	-1.3441	1.3336	2.1247	-3.1392
100.0	-3.2306	-3.0784	-2.5002	-2.5916	-2.5611
96.0	-0.0053	0.0556	-0.0661	-0.1878	-0.1270
84.0	0.2991	-1.1310	1.6987	0.2686	-0.7355
73.5	0.1468	0.2888	0.2888	0.2888	0.1468
54.0	0.0048	0.0048	0.0048	0.0048	-0.2792
33.0	0.0048	-0.1372	-0.1372	0.0048	-0.2792
24.0	-0.0817	-0.0817	-0.0817	-0.0817	-0.1306
10.0	-0.0326	-0.0817	-0.0817	0.0656	0.0048
5.0	-0.0326	-0.0326	0.0165	0.1639	0.0048
2.5	-0.0326	0.1148	0.2130	0.2130	0.1468

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1974
43.4	-0.1974
50.4	-0.1974
56.1	-0.1974
62.5	-0.4591
69.7	-0.4217
76.9	-0.3095
83.4	-0.3095
89.4	-0.3843
95.4	-0.4591
101.1	-0.3469
---BOTTOM---	
38.2	-0.1226
43.4	-0.0852
50.4	-0.1974
56.1	-0.0852
62.5	-0.0852
69.7	-0.0478
76.9	-0.0478
83.4	-0.0478
89.4	0.0269
95.4	0.0269
101.1	-0.1226

MISC.*

NO.	CP
1	-0.5632
2	-0.1372
3	-0.8472
4	-0.1372
5	-0.7052
6	-0.2792
7	0.3295
8	-0.7964
9	-0.3704
10	0.5946
11	-0.0661
12	-0.2791

TABLE A6. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
67	4	CLAERO =	1.0591	CDAERO =	0.5121	DCLC =	0.0	DWLC =	1.1202	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.00		2118.83	2111.71	7.12	76.68	0.01	2.66	0.0	2.003	2.003	87.0827	0.508079E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		2.18	-1.15	-1.25	0.00	-0.01	0.02	1.15	-2.47	1.14	0.21	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4851	-0.4851
2.5	-0.4507	-0.4507
5.0	-0.4851	-0.4507
10.0	-0.4507	-0.4851
15.0	-0.4851	-0.4851
25.0	-0.4851	-0.4851
35.0	-0.4851	-0.4507
50.0	-0.4507	
56.0	-0.4851	-0.4507
65.0	-0.4507	-0.4851
76.0	-0.4851	-0.4507
79.0	-0.4507	-0.4507
80.5	-0.4465	-0.3967
81.0	-0.4465	-0.4216
82.0	-0.4465	-0.4216
84.0	-0.4465	
87.0	-0.4216	
89.0	-0.4378	-0.4378
93.0	-0.4378	
96.0	-0.4378	-0.4378
100.0	-0.4378	-0.4010
96.0	-0.4746	-0.4378
84.0	-0.4378	-0.4010
73.0	-0.4746	-0.4378
50.0	-0.4746	-0.4378
35.0	-0.4851	-0.4851
25.0	-0.4851	-0.4851
10.0	-0.4851	-0.4851
5.0	-0.4851	-0.4851
2.5	-0.4507	-0.4851

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.2597	-0.9848	-2.0484	-2.8702	-3.1119
2.5	-0.6399	-1.1299	-1.5167	-1.6616	-2.9186
5.0	-0.7916	-0.9365	-1.1782	-1.4683	-1.6133
10.0	-0.6948	-0.8399	-1.0816	-1.2265	-1.2750
15.0	-0.6465	-0.7916	-0.9365	-1.0816	-1.1299
24.0	-0.5982	-0.6948	-0.8399	-0.8882	-0.9365
33.0	-0.6801	-0.6801	-0.9595	-0.9595	-0.9595
54.0	-0.8198	-0.9595	-0.9595	-3.0560	-1.0993
65.0	-1.1000	-1.2104	-1.2104	-1.3208	-1.4312
78.5	-2.7189	-2.9029	-3.0869	-3.3077	-3.6388
79.5	-19.7623	-47.5401	-91.0998	-73.7518	-80.8205
80.5	-18.5419	0.8727	7.7924	29.4982	-7.4409
81.3	16.5784	208.0168	7.3693	-8.6357	-12.8672
82.0	-8.4366	-5.6737	5.5772	-14.3860	-6.2711
84.0	2.6648	-0.6705	0.9225	-8.5361	-0.9692
87.0	4.5068	-3.9064	-2.7864	0.9474	-0.4713
89.0	-0.5302	-0.5601	-3.6745	-2.8960	1.3565
93.0	-0.0809	0.2485	-0.5601	-0.5900	-1.8778
96.0	-0.1109	-1.3088	1.2967	2.0453	-3.2254
100.0	-3.1355	-2.9558	-2.5067	-2.4467	-2.5665
96.0	0.0688	0.1287	-0.0210	-0.0809	-0.0809
84.0	0.3982	-0.7997	1.7159	0.3683	-0.5001
73.5	0.1586	0.2983	0.2983	0.2983	0.1586
54.0	0.0188	0.0188	0.0188	0.0188	-0.1209
33.0	0.0188	0.0188	0.0188	0.0188	-0.1209
24.0	-0.0180	-0.0180	0.0303	0.0787	-0.0663
10.0	0.1270	0.0787	0.2237	0.2237	0.1586
5.0	0.2237	0.2237	0.2720	0.3687	0.2983
2.5	0.3204	0.3204	0.4171	0.4171	0.2983

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1802
43.4	-0.1802
50.4	-0.1802
56.1	-0.2170
62.5	-0.4378
69.7	-0.4378
76.9	-0.3642
83.4	-0.3274
89.4	-0.4378
95.4	-0.4378
101.1	-0.4010
---BOTTOM---	
38.2	-0.1068
43.4	-0.0698
50.4	-0.1802
56.1	-0.0330
62.5	0.0038
69.7	0.0405
76.9	0.0405
83.4	0.0774
89.4	0.0774
95.4	0.0774
101.1	-0.0698

MISC.*

NO.	CP
1	-0.8198
2	0.0188
3	-0.9595
4	0.0188
5	-0.9595
6	-0.2607
7	0.3982
8	-0.7397
9	-0.3505
10	0.7875
11	0.0688
12	-0.2007

TABLE A6. (Concluded)

RUN:SEQ 67 6 — PROPULSIVE WING / CANARD PRESSURES — PAGE 110

CLAERO = 1.3178 CDAERO = 0.6275 DCLC = 0.0 DWLC = 1.2361 BASEPR = 8.1004

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
8.01	2118.90	2111.78	7.12	76.66	0.01	2.66	0.0	2.007	2.007	87.1003	0.508532E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	2.55	-0.95	-1.33	0.00	-0.01	0.02	1.45	-2.56	1.42	0.33

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4507	-0.4161
2.5	-0.4161	-0.4507
5.0	-0.4507	-0.4507
10.0	-0.4507	-0.4507
15.0	-0.4851	-0.4507
25.0	-0.4507	-0.4507
35.0	-0.4507	-0.4507
50.0	-0.4507	
56.0	-0.4851	-0.4507
65.0	-0.4507	-0.4161
76.0	-0.4507	-0.4161
79.0	-0.4507	-0.4507
80.5	-0.4216	-0.4216
81.0	-0.4216	-0.3967
82.0	-0.4216	-0.3967
84.0	-0.4216	
87.0	-0.3967	
89.0	-0.3642	-0.4010
93.0	-0.4010	
96.0	-0.4010	-0.4010
100.0	-0.4377	-0.4010
96.0	-0.4010	-0.4377
84.0	-0.4377	-0.4010
73.0	-0.4010	-0.4377
50.0	-0.4010	-0.4010
35.0	-0.4507	-0.4507
25.0	-0.4507	-0.4161
10.0	-0.4507	-0.4507
5.0	-0.4507	-0.4507
2.5	-0.4507	-0.4507

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.6131	-3.4499	-5.2383	-3.7882	-2.2898
2.5	-1.3232	-1.8065	-2.9183	-3.4016	-1.9999
5.0	-1.0815	-1.4198	-2.0482	-3.3049	-1.9514
10.0	-0.9847	-1.2264	-1.5648	-3.2565	-1.9514
15.0	-0.8881	-0.9847	-1.3232	-2.2415	-1.8548
24.0	-0.8398	-0.8881	-1.0815	-1.3232	-1.8065
33.0	-0.9594	-0.9594	-1.0991	-0.9594	-1.6582
54.0	-0.9594	-0.9594	-1.0991	-2.9159	-1.6582
65.0	-1.1000	-1.2838	-1.3574	-1.3574	-1.6517
78.5	-2.7554	-2.9761	-3.1969	-3.1969	-3.8590
79.5	-19.6852	-47.4098	-91.0632	-73.7162	-80.8845
80.5	-19.5602	0.5988	7.4181	29.5179	-7.1662
81.3	16.8007	204.6313	6.6714	-9.0825	-13.0893
82.0	-8.7589	-5.7476	5.5515	-13.3631	-5.9217
84.0	2.5899	-0.6456	0.9472	-8.7341	-1.1931
87.0	4.4813	-3.9307	-2.8109	0.8975	-0.9691
89.0	-0.5601	-0.6199	-3.7639	-3.0153	0.8174
93.0	-0.1109	0.2484	-0.5900	-0.6199	-2.7759
96.0	-0.1109	-1.3685	1.1467	1.9552	-4.3029
100.0	-3.0454	-2.8957	-2.4165	-2.4464	-2.5363
96.0	0.1287	0.1885	0.0388	-0.6199	-0.1109
84.0	0.4281	-0.6199	1.6258	0.3682	-0.4104
73.5	0.1585	0.2983	0.2983	0.2983	0.0188
54.0	0.0188	0.0188	0.1585	0.0188	-0.2607
33.0	0.1585	0.1585	0.1585	0.0188	-0.1210
24.0	0.1270	0.1753	0.1753	0.1753	-0.0181
10.0	0.2720	0.3203	0.3686	0.3686	0.1585
5.0	0.4170	0.4170	0.4653	0.4653	0.2983
2.5	0.4653	0.4653	0.5136	0.4653	0.2983

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1434
43.4	-0.1434
50.4	-0.1802
56.1	-0.2170
62.5	-0.4010
69.7	-0.4010
76.9	-0.4010
83.4	-0.3642
89.4	-0.4377
95.4	-0.4377
101.1	-0.3642
---BOTTOM---	
38.2	-0.0699
43.4	-0.0331
50.4	-0.1434
56.1	0.0037
62.5	0.1141
69.7	0.1509
76.9	0.1141
83.4	0.1509
89.4	0.1509
95.4	0.1509
101.1	0.0037

MISC.*

NO.	CP
1	-0.9594
2	0.1585
3	-1.7979
4	0.0188
5	-1.5183
6	-0.2607
7	0.3682
8	-0.7996
9	-0.4104
10	0.7575
11	0.0388
12	-0.2606

TABLE A7. RUN 75, BC1W6V, DELF=45, CMU=0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
75 10	CLAERO =	0.5282	CDAERO =	0.2099	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.06	2133.33	2103.27	30.06	156.41	0.01	2.87	0.0	0.0	0.0	87.4624	0.106414E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0.53	0.21	-0.18	0.00	-0.00	0.01	0.52	-0.18	0.52	0.21	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.4671	0.5325
2.5	-0.0638	-0.2435
5.0	-0.1291	-0.1945
10.0	-0.0801	-0.2271
15.0	-0.2435	-0.2598
25.0	-0.2353	-0.2516
35.0	-0.2190	-0.2190
50.0	-0.2271	
56.0	-0.2353	-0.2598
65.0	-0.3251	-0.3170
76.0	-0.2108	-0.5130
79.0	-0.4885	-0.6600
80.5	-0.4524	-0.6352
81.0	-0.4406	-0.6293
82.0	-0.4524	-0.6234
84.0	-0.2107	
87.0	-0.4642	
89.0	-0.4662	-0.6666
93.0	-0.4836	
96.0	-0.4836	0.6320
100.0	0.9457	-0.6492
96.0	0.3444	0.3444
84.0	0.9457	-0.6492
73.0	0.3444	0.3444
50.0	0.6756	0.5710
35.0	0.5570	0.5488
25.0	0.1813	-0.3415
10.0	0.0506	0.0506
5.0	-0.0801	0.0016
2.5	-0.1046	-0.0393

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0132	0.0705	0.3224	0.1736	-0.3074
2.5	0.1965	0.1965	-0.2616	-0.5364	-0.6967
5.0	0.1392	0.1049	-0.2043	-0.4448	-0.6166
10.0	0.0016	-0.0325	-0.2501	-0.4104	-0.4677
15.0	-0.0440	-0.0784	-0.2845	-0.3188	-0.4334
24.0	-0.1127	-0.1470	-0.2959	-0.2959	-0.3646
33.0	-0.1276	-0.1938	-0.3262	-0.2931	-0.2931
54.0	-0.2600	-0.2931	-0.3593	-0.3262	-0.3262
65.0	-0.3442	-0.3877	-0.4488	-0.4226	-0.4139
78.5	-0.6666	-0.6928	-0.7364	-0.7364	-0.7799
79.5	-0.5821	-0.5821	-0.6410	-0.6764	-0.7236
80.5	-0.5526	-0.5585	-0.6352	-0.6646	-0.7295
81.3	-0.5762	48.8672	-0.6470	-0.6823	-0.7354
82.0	-0.5644	-1.0125	-0.6410	-0.6705	-0.7295
84.0	-0.5644	-0.5939	-0.6352	-0.6764	-0.7413
87.0	-0.5939	-0.5998	-0.6470	-0.6764	-0.7236
89.0	-0.6477	-0.6477	-0.6832	-0.6193	-0.7115
93.0	-0.7044	-0.6548	-0.6832	-0.2505	-0.7896
96.0	-0.7044	-0.6335	-0.6832	-0.7115	-0.7967
100.0	-0.5626	-0.5697	-0.5626	-0.7115	-0.7612
96.0	0.2035	0.2887	0.2957	0.3028	0.2816
84.0	0.3880	0.5299	0.5795	0.5653	0.5228
73.5	0.3690	0.4021	0.5014	0.5345	0.4683
54.0	0.1704	0.2035	0.2366	0.2366	0.2035
33.0	0.0049	0.0049	0.0711	0.1042	0.0711
24.0	-0.0440	-0.0096	0.0362	0.0705	0.0362
10.0	-0.0784	-0.0898	0.0132	0.0820	0.1042
5.0	-0.1127	-0.1585	0.0132	0.1277	0.1704
2.5	-0.1814	-0.2845	-0.0440	0.2079	0.2035

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1524
43.4	-0.1524
50.4	-0.1350
56.1	-0.0827
62.5	-0.2047
69.7	-0.2047
76.9	-0.0914
83.4	-0.0827
89.4	-0.1698
95.4	-0.2221
101.1	-0.1786
---BOTTOM---	
38.2	0.0046
43.4	0.0132
50.4	-0.0391
56.1	0.0219
62.5	0.0132
69.7	0.0132
76.9	0.0480
83.4	0.1090
89.4	0.1439
95.4	0.0567
101.1	-0.3703

MISC.*

NO.	CP
1	-0.2600
2	0.0711
3	-0.3262
4	0.0711
5	-0.2931
6	0.0049
7	0.5157
8	-0.3143
9	-0.1228
10	0.6006
11	0.2745
12	-0.0518

TABLE A7. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
75 12	CLAERO =	0.7283	CDAERO =	0.2632	DCLC =	0.0	DWLC =	0.0		BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.01	2133.26	2103.08	30.17	156.90	0.01	2.67	0.0	0.0	0.0	87.4835	0.106275E+07	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	0.73	0.26	-0.14	0.00	-0.00	0.00	0.74	-0.14	0.72	0.26		

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	0.1546	-0.6102
2.5	-0.6021	-1.1798
5.0	-0.5370	-0.6916
10.0	-0.0650	-0.5532
15.0	-0.4637	-0.5289
25.0	-0.3742	-0.3986
35.0	-0.3092	0.3417
50.0	-0.3010	
56.0	-0.3092	-0.3336
65.0	-0.3905	-0.4312
76.0	-0.2196	-0.5777
79.0	-0.5614	-0.6997
80.5	-0.5498	-0.6849
81.0	-0.5380	-0.6966
82.0	-0.5321	-0.6849
84.0	-0.2267	
87.0	-0.5380	
89.0	-0.5273	-0.7269
93.0	-0.5533	
96.0	-0.5359	0.6709
100.0	0.8012	-0.7356
96.0	0.3149	0.3236
84.0	0.8012	-0.7356
73.0	0.3149	0.3236
50.0	0.7665	0.6102
35.0	0.6266	0.5859
25.0	0.2035	-0.4068
10.0	0.1384	0.1384
5.0	0.0[illegible]26	0.1221
2.5	0.2767	0.3581

********** WING ******

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.1453	0.2480	-0.2996	-1.3833	-2.0678
2.5	0.0769	-0.0486	-0.7217	-0.9841	-1.4404
5.0	0.0084	-0.1057	-0.5277	-0.7787	-1.0411
10.0	-0.0943	-0.1655	-0.4821	-0.6304	-0.7673
15.0	-0.1513	-0.2311	-0.4251	-0.5620	-0.6418
24.0	-0.2083	-0.2768	-0.4022	-0.4365	-0.4593
33.0	-0.2524	-0.3183	-0.3843	-0.4173	-0.4503
54.0	-0.3183	-0.3513	-0.4173	-0.3183	-0.4173
65.0	-0.4231	-0.4491	-0.4838	-0.4578	-0.4752
78.5	-0.7269	-0.7356	-0.7617	-0.7617	-0.8311
79.5	-0.6144	-0.6203	-0.6673	-0.7143	-0.7083
80.5	-0.5909	-0.6026	-0.6614	-0.7201	-0.6790
81.3	-0.6261	44.3310	-0.6614	-0.6790	-0.6849
82.0	-0.6144	-1.0256	-0.6731	-0.6966	-0.6966
84.0	-0.6261	-0.6379	-0.6790	-0.6849	-0.6614
87.0	-0.6731	-0.6555	-0.6849	-0.6731	-0.5733
89.0	-0.7517	-0.6881	-0.7305	-0.7022	-0.7729
93.0	-0.7800	-0.6810	-0.7305	-0.2641	-0.8577
96.0	-0.7517	-0.6810	-0.7234	-0.7659	-0.8506
100.0	-0.6245	-0.5892	-0.5962	-0.7517	-0.8295
96.0	0.2447	0.3295	0.3578	0.2942	0.2518
84.0	0.4709	0.5981	0.6051	0.5698	0.4709
73.5	0.4072	0.4732	0.5392	0.5392	0.4402
54.0	0.2094	0.2094	0.2753	0.2753	0.2094
33.0	0.0774	0.1104	0.1434	0.1764	0.0774
24.0	0.0655	0.0769	0.1339	0.1795	0.1225
10.0	0.0198	0.0198	0.1795	0.2594	0.2423
5.0	-0.0144	0.0198	0.2366	0.3392	0.3413
2.5	-0.0144	-0.0600	0.3278	0.4305	0.4402

FUSELAGE

-----TOP----		---BOTTOM---	
XLOC	CP	XLOC	CP
38.2	-0.1886	38.2	0.0371
43.4	-0.1886	43.4	0.0545
50.4	-0.1539	50.4	-0.0063
56.1	-0.0931	56.1	0.0458
62.5	-0.1973	62.5	0.0545
69.7	-0.1973	69.7	0.0632
76.9	-0.1192	76.9	0.0892
83.4	-0.1105	83.4	0.1500
89.4	-0.1886	89.4	0.1934
95.4	-0.2060	95.4	0.0892
101.1	-0.1713	101.1	-0.3710

MISC.*

NO.	CP
1	-0.3513
2	0.1434
3	-0.4503
4	0.1764
5	-0.4173
6	0.0115
7	0.3931
8	-0.4408
9	-0.4761
10	0.4497
11	0.4497
12	-0.1439

TABLE A7. (Concluded)

RUN:SEQ 75 16 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 137

CLAERO =	1.1935	CDAERO =	0.4592	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.08	2133.54	2103.48	30.06	156.80	0.01	2.66	0.0	0.0	0.0	87.1285	0.105731E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	1.19	0.46	-0.06	0.00	-0.00	0.00	1.25	-0.06	1.18	0.46	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-5.0868	-1.7054
2.5	-2.3427	-1.5748
5.0	-1.6074	-1.5746
10.0	-0.0556	-1.5339
15.0	-1.0684	-1.5094
25.0	-0.7988	-1.4441
35.0	-0.6518	-1.3379
50.0	-0.5375	
56.0	-0.5130	-0.9050
65.0	-0.5538	-0.7335
76.0	-0.3170	-0.7417
79.0	-0.6763	-0.8805
80.5	-0.6764	-0.9181
81.0	-0.6469	-0.8887
82.0	-0.6410	-0.9004
84.0	-0.3168	
87.0	-0.6764	
89.0	-0.6753	-0.9281
93.0	-0.6928	
96.0	-0.6840	0.7278
100.0	0.6843	-0.8845
96.0	0.3356	0.2833
84.0	0.6843	-0.8845
73.0	0.3356	0.2833
50.0	0.7888	0.6494
35.0	0.6876	0.5815
25.0	0.3283	-0.5865
10.0	0.3038	0.2629
5.0	0.2629	0.3201
2.5	0.6060	0.5815

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0018	-1.0746	-4.4754	-3.6653	-1.5784
2.5	-0.4104	-0.9142	-2.0593	-3.5363	-1.5555
5.0	-0.3875	-0.7196	-1.3723	-3.5592	-1.5097
10.0	-0.4562	-0.6623	-1.0631	-1.7845	-1.5211
15.0	-0.4906	-0.6165	-0.8914	-0.9142	-1.4982
24.0	-0.4906	-0.5707	-0.7196	-0.7196	-1.4410
33.0	-0.4917	-0.5579	-0.6573	-0.6242	-1.4518
54.0	-0.4586	-0.5248	-0.5911	-0.3262	-1.1538
65.0	-0.5359	-0.5708	-0.5969	-0.5882	-0.8584
78.5	-0.7973	-0.7973	-0.8148	-0.8322	-0.9804
79.5	-0.7000	-0.6823	-0.7236	-0.7295	-0.7413
80.5	-0.6882	-0.6528	-0.7118	-0.7295	-0.6823
81.3	-0.7000	40.0000	-0.7118	-0.7118	-0.7413
82.0	-0.6882	-1.0184	-0.7118	-0.6351	-0.7825
84.0	-0.7118	-0.6587	-0.7177	-0.6351	-0.8061
87.0	-0.7236	-0.7295	-0.7059	-0.5880	-0.6764
89.0	-0.8179	-0.7896	-0.7966	-0.7612	-0.8818
93.0	-0.8534	-0.7825	-0.8108	-0.3569	-0.9669
96.0	-0.7966	-0.7470	-0.8038	-0.8250	-0.9669
100.0	-0.6548	-0.6264	-0.6264	-0.8179	-0.9385
96.0	0.2957	0.4234	0.3880	0.2745	0.2106
84.0	0.5937	0.7143	0.6433	0.5724	0.3809
73.5	0.5345	0.6007	0.6007	0.5676	0.3690
54.0	0.2697	0.3359	0.3359	0.3359	0.2035
33.0	0.2035	0.2697	0.3028	0.3028	0.2035
24.0	0.1965	0.2652	0.3224	0.3453	0.2423
10.0	0.2079	0.2995	0.4255	0.4484	0.4021
5.0	0.2308	0.3339	0.4484	0.5056	0.4352
2.5	0.2652	0.3568	0.4255	0.4598	0.4352

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3180
43.4	-0.2831
50.4	-0.2134
56.1	-0.1611
62.5	-0.3006
69.7	-0.2919
76.9	-0.2222
83.4	-0.1960
89.4	-0.2483
95.4	-0.2570
101.1	-0.1960
---BOTTOM---	
38.2	0.1090
43.4	0.1178
50.4	0.0306
56.1	0.1178
62.5	0.1526
69.7	0.1962
76.9	0.2136
83.4	0.2398
89.4	0.2659
95.4	0.1352
101.1	-0.3790

MISC.*

NO.	CP
1	-0.6242
2	0.3028
3	-1.1207
4	0.2697
5	-1.2200
6	0.0711
7	0.2816
8	-0.4845
9	0.4376
10	0.0120
11	-0.7470
12	-0.3001

TABLE A8. RUN 76,BC1W6V, DELF=45, CMU=0.5, (BN/B)=1

RUN	SEQ											
76	2	CLAERO =	0.9376	CDAERO =	0.6596	DCLC =	0.1844	DWLC =	0.4560	BASEPR =	8.1914	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
-0.01		2133.54	2106.65	26.89	146.62	0.01	2.66	0.225	0.527	0.752	87.0591	0.994008E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		1.58	0.27	-0.51	0.00	-0.01	0.03	1.05	-0.30	1.05	0.52	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.2425	-1.0811
2.5	-0.5517	-1.4006
5.0	-0.5152	-0.9442
10.0	-0.0679	-0.7708
15.0	-0.5334	-0.7982
25.0	-0.5243	-0.6612
35.0	-0.5334	-0.6612
50.0	-0.6795	
56.0	-0.8073	-0.9442
65.0	-1.1633	-1.4098
76.0	-0.3783	-3.0712
79.0	-4.4770	-4.8879
80.5	-22.1755	-19.2103
81.0	-19.4605	-18.4194
82.0	-14.4592	-16.1591
84.0	-0.3243	
87.0	0.0447	
89.0	-0.9382	-1.5032
93.0	-0.7824	
96.0	-0.3343	0.6787
100.0	1.2047	-0.0518
96.0	0.6495	0.5813
84.0	1.2047	-0.0518
73.0	0.6495	0.5813
50.0	0.7859	0.5424
35.0	0.6898	0.5802
25.0	0.3612	-1.0994
10.0	0.2425	0.2060
5.0	0.0964	0.1877
2.5	0.1786	0.4342

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4588	-0.4460	-0.8811	-0.9835	-3.0697
2.5	0.6802	0.6162	-0.8427	-0.9579	-1.8410
5.0	0.5266	0.4626	-1.0219	-0.8427	-1.4058
10.0	0.2963	0.1811	-0.9707	-0.8044	-1.0603
15.0	0.1427	0.0147	-1.0347	-0.7916	-0.9067
24.0	-0.0749	-0.2028	-0.7916	-0.7787	-0.8044
33.0	-0.2209	-0.3319	-0.8869	-0.8869	-0.8128
54.0	-0.6279	-0.7018	-0.8869	0.2602	-1.1089
65.0	-1.0162	-1.1136	-1.1428	-1.5812	-1.5812
78.5	-3.1593	-3.2370	-2.9351	-3.9870	-5.0002
79.5	-21.9119	-22.1030	-13.5233	-29.1210	-37.2448
80.5	-22.2614	-20.1788	-14.3670	-21.2991	-17.8595
81.3	-21.7538	44.1829	-14.9663	-14.2350	-6.8414
82.0	-19.0783	-17.1675	-14.6503	-11.3620	-3.5862
84.0	0.0183	-5.3062	-13.7342	-10.1759	-3.1645
87.0	0.4071	-1.0426	-2.7164	-5.2731	-2.8164
89.0	-0.7837	-0.6172	0.6910	-3.3047	-2.2108
93.0	-0.6330	-0.6172	-0.2604	-0.5300	-3.1779
96.0	-0.6330	-0.7123	-0.3873	-0.7123	-3.0987
100.0	-0.3555	-0.2287	-0.1336	+0.0701	-0.3635
96.0	0.4135	0.2391	0.3342	0.4214	0.4373
84.0	0.3580	0.2708	0.2312	0.4848	0.5086
73.5	0.1492	0.1492	0.2602	0.3712	0.3712
54.0	-0.1468	-0.0728	-0.0358	0.1492	0.0752
33.0	-0.2949	-0.3319	-0.2949	-0.0728	-0.0358
24.0	-0.3308	-0.3180	-0.3436	-0.1388	0.0019
10.0	-0.4716	-0.3948	-0.3948	-0.1900	0.1492
5.0	-0.4588	-0.3308	-0.4588	-0.2540	0.2602
2.5	-0.4332	-0.3692	-0.5228	-0.0749	0.3712

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3343
43.4	-0.3538
50.4	-0.2174
56.1	-0.1103
62.5	-0.3733
69.7	-0.3733
76.9	-0.1687
83.4	-0.2272
89.4	-0.3733
95.4	-0.3538
101.1	-0.2759
---BOTTOM---	
38.2	0.1527
43.4	0.2014
50.4	0.1138
56.1	0.1138
62.5	-0.0616
69.7	-0.1980
76.9	-0.1590
83.4	-0.0129
89.4	0.1040
95.4	0.0456
101.1	-0.2662

**MISC.

NO.	CP
1	-0.4428
2	-0.3319
3	-0.8128
4	0.0011
5	-0.8499
6	-0.1468
7	0.3025
8	-0.8233
9	-0.5220
10	0.8495
11	0.1122
12	-0.2366

TABLE A8. (Continued)

RUN:SEQ 76 4 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 140

CLAERO = 1.1080 CDAERO = 0.7649 DCLC = 0.1868 DWLC = 0.4619 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HST	RN
3.99	2133.61	2106.04	27.57	150.58	0.01	2.65	0.218	0.514	0.732	86.7386	0.100513E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.76	0.43	-0.49	-0.02	-0.01	0.06	1.26	-0.29	1.22	0.64

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-1.0737	-4.7686
2.5	-1.3853	-2.2757
5.0	-0.8867	-1.7503
10.0	-0.0675	-1.2695
15.0	-0.7976	-1.2072
25.0	-0.7175	-0.9490
35.0	-0.6730	-0.8599
50.0	-0.7887	
56.0	-0.9134	-1.1182
65.0	-1.2784	-1.6079
76.0	-0.4059	-3.2552
79.0	-4.5372	-4.9826
80.5	-21.0771	-15.9415
81.0	-19.5539	-16.4042
82.0	-15.8707	-16.0252
84.0	-0.5673	
87.0	0.2554	
89.0	-0.6522	-2.2294
93.0	-0.7473	
96.0	-0.3577	0.7824
100.0	1.5520	-0.0632
96.0	0.6779	0.5449
84.0	1.5520	-0.0632
73.0	0.6779	0.5449
50.0	0.7729	0.5734
35.0	0.7338	0.5735
25.0	0.4133	-1.2339
10.0	0.3064	0.2708
5.0	0.2263	0.2619
2.5	0.4311	0.5469

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4096	-0.4596	-1.2959	-1.6205	-5.7648
2.5	0.6265	0.5141	-1.0837	-1.3458	-2.8436
5.0	0.4766	0.3393	-1.1836	-1.0962	-2.0199
10.0	0.2145	0.1147	-1.4082	-0.9589	-1.3334
15.0	0.0772	-0.0726	-1.3583	-0.8840	-1.0837
24.0	-0.1475	-0.2598	-1.2709	-0.8965	-0.9339
33.0	-0.2983	-0.4065	-1.2727	-0.9840	-0.9118
54.0	-0.6592	-0.6953	-1.1283	0.6400	-1.1644
65.0	-1.0418	-1.0988	-1.3078	-1.5643	-1.4788
78.5	-3.1414	-3.2270	-2.9513	-3.3412	-4.7661
79.5	-20.7942	-21.0387	-12.9401	-30.3645	-36.2189
80.5	-21.5913	-19.0846	-12.9143	-19.6950	-16.6229
81.3	-21.3277	35.4838	-14.0070	-10.9991	-5.6771
82.0	-19.2003	-17.2594	-14.4310	-8.4217	-3.1638
84.0	-0.6702	-6.3520	-14.0389	-1.6407	-2.7849
87.0	0.4353	-0.9594	-2.9133	-1.2551	-2.4313
89.0	-0.6848	-0.6462	0.4828	-1.7907	-2.0072
93.0	-0.6385	-0.6594	-0.2982	-0.5457	-2.9582
96.0	-0.6771	-0.6694	-0.5920	-1.1179	-2.8345
100.0	-0.2441	-0.0121	-0.1281	-0.3369	-0.3369
96.0	0.6143	0.3050	0.2663	0.2586	0.3900
84.0	0.3978	0.3746	0.2895	0.4442	0.4442
73.5	0.2070	0.2791	0.2431	0.3153	0.2791
54.0	-0.0096	0.0265	0.0987	0.1709	0.0626
33.0	-0.2261	-0.2261	-0.0457	0.0265	0.0265
24.0	-0.2723	-0.3347	-0.1350	0.0148	0.1147
10.0	-0.3847	-0.3223	-0.3472	0.0397	0.2791
5.0	-0.6093	-0.4471	-0.4346	0.0273	0.3874
2.5	-0.5095	-0.3971	-0.5095	0.1022	0.3513

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3957
43.4	-0.3957
50.4	-0.2627
56.1	-0.1392
62.5	-0.4147
69.7	-0.4052
76.9	-0.2057
83.4	-0.2437
89.4	-0.3872
95.4	-0.3767
101.1	-0.3007
---BOTTOM---	
38.2	0.1649
43.4	0.2218
50.4	0.1174
56.1	0.1459
62.5	0.0033
69.7	-0.0917
76.9	-0.0252
83.4	0.1174
89.4	0.2313
95.4	0.1459
101.1	-0.2532

MISC.*

NO.	CP
1	-0.5870
2	-0.1539
3	-0.8757
4	0.0626
5	-0.9479
6	-0.1178
7	0.1813
8	-1.0560
9	-0.7313
10	0.8772
11	-0.0121
12	-0.3755

TABLE A8. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 76 10

CLAERO =	1.6791	CDAERO =	1.1623	DCLC =	0.2026	DWLC =	0.4902	BASEPR =	8.1914		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.03	2133.54	2106.42	27.12	149.54	0.01	2.65	0.220	0.515	0.735	86.7934	0.993237E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	2.37	0.92	-0.47	-0.02	-0.02	0.06	1.99	-0.26	1.81	1.05	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-9.7371	-3.1739
2.5	-3.7624	-3.1831
5.0	-2.3321	-3.2285
10.0	-0.0690	-3.3642
15.0	-1.5626	-3.3822
25.0	-1.2277	-3.9527
35.0	-1.1372	-3.3280
50.0	-1.1643	
56.0	-1.2368	-1.0919
65.0	-1.6169	-1.1734
76.0	-0.5126	-2.8480
79.0	-4.9843	-4.5500
80.5	-21.4542	-9.6917
81.0	-20.3757	-10.3318
82.0	-16.6314	-8.5675
84.0	-0.7521	
87.0	-0.1313	
89.0	-0.8072	-6.6513
93.0	-0.9327	
96.0	-0.5077	0.8350
100.0	1.4725	-0.6719
96.0	0.7094	0.4486
84.0	1.4725	-0.6719
73.0	0.7094	0.4486
50.0	0.8543	0.6032
35.0	0.8091	0.5738
25.0	0.5013	-1.5807
10.0	0.4651	0.3655
5.0	0.4289	0.4199
2.5	0.5919	0.3837

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.5264	-1.0568	-0.8283	-6.2730	-4.4202
2.5	0.4281	0.0220	-1.1513	-3.0622	-4.6230
5.0	0.2251	-0.1557	-1.1076	-2.1483	-4.6359
10.0	-0.0288	-0.4222	-1.2598	-1.6913	-5.3973
15.0	-0.2064	-0.4857	-1.3106	-1.4756	-4.6864
24.0	-0.3841	-0.7268	-1.4122	-1.3360	-2.3005
33.0	-0.5419	-0.7987	-1.5325	-1.3858	-1.2757
54.0	-0.8354	-1.0188	-1.7894	-0.0282	-1.6426
65.0	-1.1839	-1.3578	-2.0532	-1.9566	-2.1981
78.5	-3.4442	-3.5795	-3.9080	-4.0238	-6.5451
79.5	-23.0747	-21.9571	-18.8857	-11.7566	-34.7325
80.5	-23.2052	-20.2647	-18.5004	-8.9727	-26.6166
81.3	-22.8329	13.6241	-18.0299	-11.7566	-18.6309
82.0	-20.5390	-18.6571	-16.9319	-11.5866	-14.4423
84.0	-0.8305	-7.1756	-14.0894	-11.0575	-6.3132
87.0	-0.0529	-1.2095	-2.5753	-8.9661	-3.0522
89.0	-1.1943	-1.2808	-1.3594	-5.1646	-2.3422
93.0	-1.0921	-0.9663	-1.1078	-0.7147	-3.2620
96.0	-1.0764	-0.9663	-1.2887	-0.5496	-2.9398
100.0	0.2681	0.2445	0.1029	0.0479	-0.2430
96.0	0.8341	0.7319	0.6690	0.6297	0.2366
84.0	0.5432	0.6612	0.6061	0.5589	0.0951
73.5	0.4855	0.5956	0.5222	0.4488	0.1553
54.0	0.3755	0.4122	0.4488	0.4488	0.0819
33.0	0.1920	0.1920	0.3021	0.3388	0.1186
24.0	0.0601	0.1362	0.3139	0.4027	0.2631
10.0	-0.2445	-0.1430	0.1870	0.3900	0.3388
5.0	-0.5745	-0.9299	0.0474	0.3393	0.2654
2.5	-0.9933	-1.2726	0.0220	0.1489	-0.0282

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.5753
43.4	-0.5463
50.4	-0.3724
56.1	-0.2566
62.5	-0.4884
69.7	-0.5173
76.9	-0.3049
83.4	-0.3145
89.4	-0.4304
95.4	-0.4207
101.1	-0.3145
---BOTTOM---	
38.2	0.2458
43.4	0.2844
50.4	0.1492
56.1	0.1974
62.5	0.1299
69.7	0.1685
76.9	0.2651
83.4	0.3617
89.4	0.3907
95.4	0.1974
101.1	-0.3532

MISC.*

NO.	CP
1	-1.2757
2	0.3021
3	-1.2757
4	0.3388
5	-2.9634
6	-0.1015
7	-0.1172
8	-1.7919
9	-1.4380
10	0.8813
11	-0.3373
12	-0.8484

TABLE A9. RUN 77, BC1W6V, DELF=45, CMU=1.0, (BN/B)=1

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 143

77	2	CLAERO =	1.0790	CDAERO =	0.8478	DCLC =	0.3621	DWLC =	0.8896	BASEPR =	8.1914
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.10		2133.96	2120.29	13.67	105.96	0.0	2.78	0.442	1.026	1.468	90.9884	0.706432E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		2.33	0.06	-0.78	0.00	0.00	0.01	1.27	-0.36	1.27	0.58	

CANARD

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.1939	-1.4220
2.5	-0.6320	-1.6734
5.0	-0.5961	-1.0309
10.0	-0.0754	-0.9013
15.0	-0.6320	-0.8834
25.0	-0.5961	-0.8295
35.0	-0.6499	-0.8295
50.0	-0.8295	
56.0	-0.9552	-1.1347
65.0	-1.4220	-1.7272
76.0	-0.4524	-3.9177
79.0	-5.7132	-6.5394
80.5	-37.1033	-34.3557
81.0	-18.9447	-26.5295
82.0	-5.6341	-15.4585
84.0	-0.4238	
87.0	-0.9422	
89.0	-1.6803	-2.6575
93.0	-0.9906	
96.0	-0.1668	0.6953
100.0	1.7299	-0.5309
96.0	0.5995	0.5421
84.0	1.7299	-0.5309
73.0	0.5995	0.5421
50.0	0.7528	0.5804
35.0	0.7146	0.5889
25.0	0.3555	-1.2784
10.0	0.2298	0.1939
5.0	0.1221	0.2118
2.5	0.1939	0.4273

WING

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.5588	-0.4833	-0.6847	-0.6091	-3.2019
2.5	0.7502	0.7502	-0.7601	-1.2133	-1.9433
5.0	0.5488	0.5488	-0.7854	-1.0622	-1.6160
10.0	0.3474	0.1712	-0.9364	-0.9364	-1.2636
15.0	0.1460	-0.0301	-0.7854	-0.8860	-1.0874
24.0	-0.0805	-0.2567	-0.9364	-0.8860	-1.0119
33.0	-0.3080	-0.4536	-0.9831	-1.0358	-1.0358
54.0	-0.7447	-0.8903	-1.1814	-1.1814	-1.3269
65.0	-1.2397	-1.3739	-1.4697	-1.8911	-1.8528
78.5	-4.0369	-4.3435	-4.2094	-5.2440	-6.1636
79.5	-45.0095	-54.0822	-36.4230	-61.8878	-67.8470
80.5	-44.7111	-47.0440	-35.8074	-39.2416	-10.4300
81.3	-36.7793	115.4234	-33.8240	-15.3291	-9.3023
82.0	-21.0706	-0.5664	-30.1821	-7.1764	-4.7398
84.0	6.3289	-0.1776	-4.4288	-1.3959	-3.7807
87.0	0.1335	-3.5603	0.4984	-1.5125	-3.2881
89.0	-1.5243	-2.0077	-1.8206	-2.8966	-2.3040
93.0	-1.0253	-1.1033	-1.5555	-0.6043	-4.0349
96.0	-1.0253	-1.2280	-0.9006	-0.9313	-4.3780
100.0	-0.6823	-0.6667	-0.5575	-0.6199	-0.7758
96.0	0.0351	0.0662	0.1286	0.4717	0.4717
84.0	0.0819	0.0819	0.0662	0.6432	0.5497
73.5	-0.1625	-0.1625	-0.1625	0.3469	0.3469
54.0	-0.4536	-0.4536	-0.3808	-0.1625	-0.0897
33.0	-0.5264	-0.5992	-0.5992	-0.4536	-0.1625
24.0	-0.5084	-0.4833	-0.5336	-0.4077	-0.1308
10.0	-0.4077	-0.4833	-0.4581	-0.3071	-0.0170
5.0	-0.4330	-0.4330	-0.5084	-0.2567	0.2014
2.5	-0.4330	-0.4581	-0.5588	-0.2316	0.3469

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.4351	38.2	0.1397
43.4	-0.4351	43.4	0.2164
50.4	-0.2626	50.4	0.0631
56.1	-0.1285	56.1	-0.0135
62.5	-0.4159	62.5	-0.3010
69.7	-0.4351	69.7	-0.4351
76.9	-0.2051	76.9	-0.3776
83.4	-0.2626	83.4	-0.2626
89.4	-0.3967	89.4	-0.1093
95.4	-0.3967	95.4	-0.1093
101.1	-0.3201	101.1	-0.4925

MISC.

NO.	CP
1	-0.5264
2	-0.5992
3	-0.9631
4	-0.1625
5	-1.0358
6	-0.2353
7	0.2534
8	-0.9318
9	-0.6043
10	0.8304
11	0.0351
12	-0.2456

TABLE A9. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
77 4	CLAERO =	1.2778	CDAERO =	0.9521	DCLC =	0.3803	DWLC =	0.9274	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.02	2133.82	2120.15	13.67	105.96	0.01	2.68	0.444	1.032	1.475	87.5090	0.706407E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	2.59	0.27	-0.75	-0.00	-0.00	0.02	1.54	-0.33	1.50	0.70	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-1.7811	-4.9231
2.5	-1.6195	-3.2713
5.0	-1.0270	-2.7506
10.0	-0.0574	-1.4938
15.0	-0.9193	-1.2963
25.0	-0.8295	-1.1168
35.0	-0.7936	-1.0270
50.0	-0.9731	
56.0	-1.1168	-1.3143
65.0	-1.5656	-1.9427
76.0	-0.4884	-4.2229
79.0	-5.9107	-7.0236
80.5	-37.5962	-35.8330
81.0	-20.0468	-27.3826
82.0	-6.3603	-17.4155
84.0	-0.4498	
87.0	-1.0330	
89.0	-1.7953	-2.8874
93.0	-1.0481	
96.0	-0.2435	0.7337
100.0	1.6725	-0.5692
96.0	0.6379	0.4846
84.0	1.6725	-0.5692
73.0	0.6379	0.4846
50.0	0.7528	0.5995
35.0	0.7326	0.5889
25.0	0.3914	-1.4220
10.0	0.3016	0.2478
5.0	0.2118	0.2657
2.5	0.4632	0.5530

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.5840	-0.4581	-1.4398	-1.6412	-5.4422
2.5	0.7250	0.7250	-1.3391	-1.4398	-2.9753
5.0	0.5740	0.5236	-1.4901	-1.2630	-2.1950
10.0	0.3222	0.1712	-1.6160	-1.1629	-1.5908
15.0	0.1460	-0.0301	-1.6412	-1.1126	-1.2636
24.0	-0.1057	-0.2819	-1.4901	-1.1629	-1.1126
33.0	-0.2353	-0.4536	-1.3269	-1.1814	-1.0358
54.0	-0.7447	-0.8175	-1.2541	-0.8903	-1.3997
65.0	-1.2397	-1.3547	-1.4505	-2.1785	-1.9294
78.5	-4.0561	-4.4010	-4.0753	-5.2823	-6.2786
79.5	-45.5536	-54.3411	-38.0883	-61.6645	-68.0152
80.5	-45.2296	-47.4594	-35.0293	-33.3055	-10.7796
81.3	-36.8444	115.4746	-34.3169	-9.4186	-9.5612
82.0	-20.8118	-0.9941	-31.1540	-0.6960	-4.8565
84.0	6.0437	0.0428	-5.2971	-2.1088	-3.9492
87.0	0.0428	-3.6511	0.6131	-1.5773	-3.2881
89.0	-1.5867	-2.1169	-1.9921	-3.2397	-2.5379
93.0	-1.0409	-1.1812	-1.7894	-0.6199	-4.3312
96.0	-1.0097	-1.3372	-1.1033	-1.2436	-4.7210
100.0	-0.7602	-0.4171	-0.1833	-0.4640	-0.7758
96.0	0.1442	0.1754	0.2846	0.4717	0.4249
84.0	0.1910	0.1442	0.2378	0.4249	0.6276
73.5	-0.0170	-0.0170	-0.0170	0.2013	0.3469
54.0	-0.3080	-0.3080	-0.2353	-0.1625	-0.0170
33.0	-0.4536	-0.4536	-0.4536	-0.3080	-0.0897
24.0	-0.4077	-0.4833	-0.4581	-0.3071	-0.0554
10.0	-0.3826	-0.4581	-0.6594	-0.3574	0.2013
5.0	-0.3826	-0.3323	-0.7098	-0.3071	0.3469
2.5	-0.4077	-0.3826	-0.7601	-0.2567	0.4197

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.4925
43.4	-0.4925
50.4	-0.3010
56.1	-0.1668
62.5	-0.4542
69.7	-0.4734
76.9	-0.2435
83.4	-0.2818
89.4	-0.3776
95.4	-0.4159
101.1	-0.3393
---BOTTOM---	
38.2	0.1781
43.4	0.2356
50.4	0.0623
56.1	0.0631
62.5	-0.2435
69.7	-0.3776
76.9	-0.3393
83.4	-0.1477
89.4	-0.0135
95.4	-0.1285
101.1	-0.4734

MISC.*

NO.	CP
1	-0.5264
2	-0.5264
3	-1.1086
4	-0.1625
5	-1.1086
6	-0.2353
7	0.1754
8	-1.0565
9	-0.6978
10	0.8927
11	0.0195
12	-0.3236

TABLE A9. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
77	8	CLAERO =	1.9938	CDAERO =	1.4224	DCLC =	0.4308	DWLC =	1.0351	BASEPR =	5.1914	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11 98		2133.82	2120.94	12.88	102.86	0.01	2.62	0.468	1.088	1.557	85.5867	0.685633E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR	
-0.01		3.46	0.90	-0.79	0.01	-0.01	0.00	2.45	-0.34	2.26	1.19	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-10.3360	-3.6480
2.5	-4.2387	-3.8196
5.0	-2.3524	-3.8577
10.0	-0.0849	-3.8767
15.0	-1.7428	-4.3149
25.0	-1.4187	-4.7723
35.0	-1.3044	-3.7052
50.0	-1.3997	
56.0	-1.5140	-1.0758
65.0	-1.9713	-1.7236
76.0	-0.6375	-4.6389
79.0	-6.6777	-7.6306
80.5	-39.5196	-40.0698
81.0	-21.0749	-31.0749
82.0	-8.2612	-19.6445
84.0	-0.6632	
87.0	-1.4334	
89.0	-2.1707	-3.7159
93.0	-1.3981	
96.0	-0.4831	0.7978
100.0	1.7941	-0.6864
96.0	0.6555	0.4319
84.0	1.7941	-0.6864
73.0	0.6555	0.4319
50.0	0.8385	0.5742
35.0	0.8106	0.5248
25.0	0.5058	-1.8951
10.0	0.4676	0.3343
5.0	0.4105	0.4295
2.5	0.5629	0.3343

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.4681	-0.3078	-2.3848	-4.2347	-5.7841
2.5	0.4936	0.0127	-1.6168	-2.7655	-5.3567
5.0	0.3333	-0.1208	-1.5901	-2.3648	-5.0896
10.0	0.0662	-0.3612	-1.6168	-1.9908	-4.2080
15.0	-0.1208	-0.5215	-1.8037	-1.7770	-2.8456
24.0	-0.3880	-0.7352	-1.4572	-1.6702	-2.1777
33.0	-0.5435	-0.7752	-1.9338	-1.7020	-1.9336
54.0	-0.9296	-1.0841	-2.0681	-0.6207	-2.3198
65.0	-1.3777	-1.5404	-2.4147	-2.7807	-2.9026
78.5	-4.3259	-4.8342	-5.2205	-6.3998	-8.5751
79.5	-50.0970	-57.4971	-42.4493	-62.6550	-79.0764
80.5	-49.2309	-50.6612	-40.3865	-42.9996	-15.8347
81.3	-36.8653	106.9150	-38.2545	-16.9212	-12.1482
82.0	-16.3298	-2.9876	-34.2243	-7.3613	-6.5226
84.0	4.5912	0.4372	-0.4568	-1.7498	-5.5048
87.0	-0.3880	-4.3631	-1.5847	-2.4237	-4.1961
89.0	-1.9776	-2.6892	-3.6656	-3.7980	-3.1194
93.0	-1.3488	-1.7624	-2.5237	-0.8357	-4.8240
96.0	-1.3322	-1.9610	-1.5804	-1.6632	-5.0557
100.0	-0.5875	-0.4386	0.0082	-0.1076	-0.5213
96.0	0.6536	0.4054	0.3392	0.4385	0.4054
84.0	0.4882	0.3889	0.3061	0.3558	0.2730
73.5	0.2288	0.3061	0.3061	0.3833	0.2288
54.0	-0.0801	-0.0801	0.1516	0.1516	-0.0029
33.0	-0.3118	-0.3118	-0.1573	-0.0029	-0.0029
24.0	-0.3345	-0.3612	-0.2276	0.0127	0.1196
10.0	-0.3612	-0.3612	-0.4681	-0.1475	0.3061
5.0	-0.3612	-0.3612	-0.6016	-0.1475	0.3061
2.5	-0.3612	-0.3078	-0.7619	-0.2544	-0.0029

FUSELAGE

-----TOP----		---BOTTOM---	
XLOC	CP	XLOC	CP
38.2	-0.7270	38.2	0.2489
43.4	-0.6864	43.4	0.3099
50.4	-0.4424	50.4	0.1269
56.1	-0.2798	56.1	0.1068
62.5	-0.6458	62.5	-0.1171
69.7	-0.6255	69.7	-0.1984
76.9	-0.3408	76.9	-0.0561
83.4	-0.3815	83.4	0.1269
89.4	-0.5035	89.4	0.2285
95.4	-0.4931	95.4	0.1269
101.1	-0.4018	101.1	-0.5238

MISC.*

NO.	CP
1	-1.3930
2	-0.2345
3	-1.7020
4	0.1516
5	-2.6287
6	-0.2345
7	-0.2898
8	-1.9942
9	-1.5970
10	0.9019
11	-0.5875
12	-0.9185

TABLE A10. RUN 79, BC1W6V, DELF=45, CMU=2.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
79	3	CLAERO =	0.8181	CDAERO =	1.1952	DCLC =	0.7695	DWLC =	1.8684	BASEPR =	8.1914
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.04		2131.63	2124.85	6.78	74.75	0.01	2.67	0.939	2.157	3.096	87.2223	0.494944E+06
BETA		CL	CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01		3.46	-0.42	-1.27	-0.01		0.01	0.03	1.30	-0.41	1.30	0.60

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.1339	-1.8936
2.5	-0.7712	-2.0747
5.0	-0.7712	-1.4229
10.0	-0.1196	-1.2057
15.0	-0.7351	-1.1333
25.0	-0.7712	-1.0247
35.0	-0.8075	-0.9885
50.0	-1.0247	
56.0	-1.1696	-1.3868
65.0	-1.7125	-2.0747
76.0	-0.5540	-5.0072
79.0	-7.2519	-8.5553
80.5	-44.8645	-55.7621
81.0	-12.8487	-35.7690
82.0	-2.8914	-19.4609
84.0	-0.6437	
87.0	-0.8789	
89.0	-2.3196	-3.3628
93.0	-1.0061	
96.0	0.4619	0.7323
100.0	1.3118	-1.4312
96.0	0.5392	0.4619
84.0	1.3118	-1.4312
73.0	0.5392	0.4619
50.0	0.7323	0.6551
35.0	0.6769	0.5321
25.0	0.2787	-1.5677
10.0	0.2062	0.1339
5.0	0.0977	0.0977
2.5	0.1339	0.4235

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.7738	-0.7230	-0.7738	-0.5707	-2.7026
2.5	0.7997	0.9012	-0.5200	-1.1799	-2.3981
5.0	0.6982	0.6475	-0.5200	-0.9768	-1.7382
10.0	0.4444	0.2414	-0.5200	-1.0275	-1.3320
15.0	0.2414	-0.0124	-0.5707	-1.0275	-1.2306
24.0	-0.0632	-0.3170	-0.7230	-1.0275	-1.0783
33.0	-0.3180	-0.4647	-0.9050	-1.1984	-1.1984
54.0	-0.9050	-1.0517	-1.1984	-2.5191	-1.4919
65.0	-1.4312	-1.6243	-1.6630	-2.1652	-2.1652
78.5	-5.1012	-5.5262	-5.5262	-6.5306	-7.3420
79.5	-95.8533	-112.3708	-90.5991	-116.3685	-105.7585
80.5	-89.8161	-84.8244	-78.1591	-63.8122	-23.4852
81.3	-47.8949	154.8818	-48.7325	-15.0959	-12.5090
82.0	5.1843	26.0142	-26.6217	0.7414	-5.1651
84.0	3.6163	-3.2572	9.6272	-1.8460	-3.7539
87.0	0.9505	-5.5310	-3.1788	-1.7153	-3.3879
89.0	-2.0788	-2.3618	-4.1855	-4.3741	-1.4813
93.0	-1.1983	-0.9783	-2.0473	-0.7267	-4.5943
96.0	-1.3241	-1.5757	-0.7581	-0.6009	-5.6320
100.0	-1.1039	-1.1983	-1.3555	-1.4499	-1.5127
96.0	-0.1921	-0.0978	-0.0663	0.3110	0.4996
84.0	-0.0978	-0.2550	-0.0978	0.7198	0.8141
73.5	-0.3180	-0.4647	-0.1712	0.4158	0.5625
54.0	-0.6115	-0.6115	-0.6115	-0.3180	-0.0245
33.0	-0.6115	-0.6115	-0.6115	-0.4647	-0.1712
24.0	-0.5707	-0.6216	-0.5707	-0.6216	-0.3170
10.0	-0.5200	-0.5707	-0.5707	-0.5200	-0.0245
5.0	-0.5200	-0.6216	-0.7230	-0.4692	0.2690
2.5	-0.5707	-0.6723	-0.6216	-0.3677	0.2690

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4653
43.4	-0.5039
50.4	-0.2721
56.1	-0.1176
62.5	-0.5039
69.7	-0.5039
76.9	-0.1949
83.4	-0.2335
89.4	-0.4267
95.4	-0.4653
101.1	-0.3494
---BOTTOM---	
38.2	0.1528
43.4	0.2301
50.4	0.0369
56.1	-0.1949
62.5	-0.5812
69.7	-0.6198
76.9	-0.5425
83.4	-0.4267
89.4	-0.2335
95.4	-0.2721
101.1	-0.7357

MISC.*

NO.	CP
1	-0.6115
2	-0.6115
3	-1.1984
4	-0.3180
5	-1.1984
6	-0.3180
7	0.2481
8	-0.9467
9	-0.6009
10	0.9084
11	0.0594
12	-0.2235

TABLE A10. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 79 11

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
12.00	2131.56	2124.89	6.67	74.14	0.01	2.67	0.914	2.133	3.048

CLAERO = 1.7438 CDAERO = 1.6927 DCLC = 0.8416 DWLC = 2.0290

BASEPR = 8.1914

HGT	RN
87.3492	0.490514E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	4.61	0.68	-1.26	0.01	0.00	-0.01	2.51	-0.40	2.30	1.23

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-8.5671	-3.6916
2.5	-8.5671	-3.7813
5.0	-2.3822	-4.0758
10.0	-0.0261	-4.1125
15.0	-1.6091	-4.5175
25.0	-1.3882	-4.8857
35.0	-1.3515	-4.2230
50.0	-1.4619	
56.0	-1.6460	-1.5355
65.0	-2.1613	-1.7196
76.0	-0.6520	-4.9225
79.0	-8.4934	-9.1930
80.5	-33.1974	-52.9697
81.0	-9.8913	-33.2514
82.0	-4.3370	-19.1665
84.0	-0.9087	
87.0	-1.5200	
89.0	-2.8844	-5.0843
93.0	-1.5488	
96.0	0.0226	0.8082
100.0	1.4761	-1.5881
96.0	0.5726	0.3369
84.0	1.4761	-1.5881
73.0	0.5726	0.3369
50.0	0.8082	0.5333
35.0	0.8206	0.5261
25.0	0.4893	-2.0509
10.0	0.4893	0.3420
5.0	0.4893	0.3788
2.5	0.5261	0.2316

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.6986	-0.5953	-2.0921	-2.8147	-7.2535
2.5	0.5918	0.2305	-1.9889	-2.4017	-5.6534
5.0	0.4369	0.0756	-1.9889	-2.4017	-4.5180
10.0	0.1273	-0.2340	-2.1437	-2.1954	-3.2277
15.0	-0.0792	-0.4921	-2.2470	-2.1437	-2.6600
24.0	-0.3889	-0.7502	-2.2470	-2.2470	-2.1437
33.0	-0.4875	-0.7860	-2.1290	-2.2782	-2.1290
54.0	-1.0844	-1.2336	-2.2782	-1.3828	-2.8751
65.0	-1.5095	-1.7060	-2.4916	-3.7880	-3.9845
78.5	-5.2022	-5.6736	-6.3022	-8.4236	-11.8448
79.5	********	-118.2384	-113.0306	-120.8160	-128.6299
80.5	-87.8630	-66.7351	-75.1068	-52.9967	-24.3218
81.2	-20.5476	162.7105	-16.2695	-7.2868	-18.2624
82.0	23.1687	15.1427	7.7016	-0.5899	-9.9709
84.0	-1.7857	-4.8153	-1.7060	-5.1077	-7.6588
87.0	0.2074	-6.3566	-5.9049	-3.6991	-6.7021
89.0	-2.4485	-2.9280	-6.0296	-5.9656	-3.8234
93.0	-1.4574	-1.6491	-2.9602	-1.0417	-7.3085
96.0	-1.6491	-2.5764	-1.3294	-1.8090	-8.5555
100.0	-1.3294	-1.4574	-0.8178	-1.1055	-1.1696
96.0	0.4611	0.2053	0.1733	0.4292	0.4611
84.0	0.3652	0.1733	0.3652	0.3972	0.3972
73.5	0.1094	0.1094	0.2586	0.2586	0.2586
54.0	-0.3383	-0.3383	-0.1891	-0.0399	-0.0399
33.0	-0.6368	-0.6368	-0.6368	-0.1891	-0.0399
24.0	-0.6986	-0.7502	-0.8018	-0.4405	-0.0792
10.0	-0.5953	-0.5437	-0.9051	-0.7502	0.2586
5.0	-0.5437	-0.4921	-1.1114	-0.8534	0.2586
2.5	-0.5953	-0.5437	-1.3179	-1.0598	0.2586

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.8024
43.4	-0.7631
50.4	-0.5274
56.1	-0.3310
62.5	-0.7238
69.7	-0.7631
76.9	-0.3703
83.4	-0.4488
89.4	-0.5274
95.4	-0.5274
101.1	-0.4488
---BOTTOM---	
38.2	0.2583
43.4	0.2976
50.4	0.0619
56.1	-0.0167
62.5	-0.4095
69.7	-0.5666
76.9	-0.4488
83.4	-0.2131
89.4	0.0226
95.4	-0.0560
101.1	-0.7631

MISC.*

NO.	CP
1	-1.3828
2	-0.6368
3	-2.1290
4	-0.0399
5	-2.7258
6	-0.1891
7	-0.3062
8	-1.9688
9	-1.6172
10	0.8448
11	-0.5620
12	-0.8498

TABLE A11. RUN 142, BW6V, DELF= 45, CMU=0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
142 2	CLAERO =	0.5290	CDAERO =	0.1947	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
0.06	2107.37	2077.42	29.95	160.73	0.0	2.68	0.0	0.0	0.0	87.7309	0.997162E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.0	0.53	0.19	-0.34	0.00	-0.00	0.01	0.52	-0.33	0.52	0.20		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2198	-0.2116
2.5	-0.2198	-0.2198
5.0	-0.2116	-0.2116
10.0	-0.2198	-0.2198
15.0	-0.2280	-0.2198
25.0	-0.2280	-0.2116
35.0	-0.2280	-0.2362
50.0	-0.2198	
56.0	-0.2280	-0.2362
65.0	-0.2280	-0.2280
76.0	-0.2444	-0.2280
79.0	-0.2362	-0.2362
80.5	-0.2114	-0.2055
81.0	-0.2055	-0.2055
82.0	-0.2055	-0.1878
84.0	-0.2174	
87.0	-0.2055	
89.0	-0.2056	-0.2056
93.0	-0.2056	
96.0	-0.2056	-0.2143
100.0	-0.2056	-0.2143
96.0	-0.2056	-0.2056
84.0	-0.2056	-0.2143
73.0	-0.2056	-0.2056
50.0	-0.1968	-0.2143
35.0	-0.2280	-0.2362
25.0	-0.2280	-0.2280
10.0	-0.2198	-0.2198
5.0	-0.2198	-0.2198
2.5	-0.2198	-0.2198

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5076	0.4387	0.1973	0.2088	-0.0555
2.5	-0.2164	-0.2854	-0.4807	-0.5152	-0.6761
5.0	-0.2509	-0.3084	-0.4233	-0.5152	-0.5957
10.0	-0.2854	-0.3313	-0.4348	-0.4692	-0.4922
15.0	-0.2969	-0.3084	-0.4233	-0.4003	-0.4578
24.0	-0.2739	-0.2854	-0.3658	-0.3543	-0.3428
33.0	-0.2610	-0.2943	-0.3275	-0.3607	-0.3607
54.0	-0.2943	-0.3607	-0.3607	0.3372	-0.3607
65.0	-0.3893	-0.4243	-0.4418	-0.4506	-0.4243
78.5	-0.6693	-0.6955	-0.7218	-0.7655	-0.8180
79.5	-0.5310	-0.5369	-0.6375	-0.7027	-0.7973
80.5	-0.4837	-0.4955	-0.6080	-0.7027	-0.2233
81.3	-0.4955	54?0773	-0.6375	-0.6849	-0.8033
82.0	-0.5074	-1.0282	-0.6612	-0.6967	-0.8210
84.0	-0.4955	-0.5428	-0.6553	-0.6908	-0.8033
87.0	-0.5133	-0.5310	-0.6553	-0.7086	-0.7973
89.0	-0.5860	-0.6074	-0.6857	-0.7569	-0.8494
93.0	-0.6145	-0.6074	-0.6857	-0.2514	-0.8637
96.0	-0.6145	-0.5789	-0.6786	-0.7569	-0.8565
100.0	-0.5433	-0.5219	-0.5860	-0.7426	-0.2442
96.0	0.3254	0.3752	0.3539	0.3325	0.2542
84.0	0.6387	0.6387	0.6316	0.6316	0.5176
73.5	0.5366	-0.1946	0.6030	0.6030	0.4701
54.0	0.2042	0.2042	0.2707	0.2707	0.2042
33.0	0.0713	0.0713	0.1045	0.1378	0.0713
24.0	0.0134	0.0364	0.0364	0.1054	0.0249
10.0	0.0249	0.0364	0.0709	0.1284	0.1045
5.0	0.0594	0.0709	0.1284	0.1628	0.1710
2.5	0.0939	0.1399	0.1743	0.2663	0.2375

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.0831	38.2	-0.0481
43.4	-0.0656	43.4	-0.0218
50.4	-0.0743	50.4	-0.0568
56.1	-0.0656	56.1	0.0044
62.5	-0.0743	62.5	0.0219
69.7	-0.1268	69.7	0.0481
76.9	-0.1268	76.9	0.0831
83.4	-0.1181	83.4	0.1531
89.4	-0.1968	89.4	0.2056
95.1	-0.2581	95.4	0.1269
101.1	-0.2056	101.1	-0.3193

MISC.*

NO.	CP
1	-0.2943
2	0.0713
3	-0.3607
4	0.1045
5	-0.2943
6	0.0381
7	0.5034
8	-0.3083
9	-0,0662
10	0.5817
11	0.2044
12	-0.0591

TABLE A11. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 258

142 4 CLAERO = 0.7048 CDAERO = 0.2423 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.06	2107.23	2076.83	30.40	162.09	0.0	2.66	0.0	0.0	0.0	86.9151	0.100239E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	0.70	0.24	-0.37	0.00	-0.00	0.01	0.71	-0.37	0.70	0.24

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2437	-0.2276
2.5	-0.2276	-0.2276
5.0	-0.2357	-0.2195
10.0	-0.2195	-0.2114
15.0	-0.2276	-0.2195
25.0	-0.2276	-0.2114
35.0	-0.2276	-0.2276
50.0	-0.2195	
58.0	-0.2195	-0.2276
65.0	-0.2276	-0.2195
76.0	-0.2437	-0.2357
79.0	-0.2276	-0.2276
80.5	-0.2243	-0.2068
81.0	-0.2243	-0.2068
82.0	-0.2126	-0.2010
84.0	-0.2068	
87.0	-0.2068	
89.0	-0.1980	-0.1980
93.0	-0.2066	
96.0	-0.2066	-0.1980
100.0	-0.1980	-0.1980
96.0	-0.1980	-0.1980
84.0	-0.1980	-0.1980
73.0	-0.1980	-0.1980
50.0	-0.1807	-0.2066
35.0	-0.2276	-0.2276
25.0	-0.2276	-0.2195
10.0	-0.2276	-0.2195
5.0	-0.2357	-0.2195
2.5	-0.2276	-0.2276

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0529	-0.5510	-1.4228	-1.6379	-2.5324
2.5	-0.2114	-0.8341	-1.0945	-1.2416	-1.5360
5.0	-0.4831	-0.6303	-0.8567	-0.9926	-1.1511
10.0	-0.4831	-0.5397	-0.7322	-0.7661	-0.8227
15.0	-0.4378	-0.4718	-0.6303	-0.6529	-0.6755
24.0	-0.3585	-0.4038	-0.5057	-0.4944	-0.5057
33.0	-0.3749	-0.4077	-0.4404	-0.4732	-0.4732
54.0	-0.3749	-0.3749	-0.4404	0.2798	-0.4404
65.0	-0.4479	-0.4738	-0.4996	-0.5169	-0.5255
78.5	-0.7324	-0.7324	-0.7496	-0.7927	-0.9047
79.5	-0.6207	-0.5799	-0.6616	-0.7140	-0.8423
80.5	-0.5799	-0.5391	-0.6499	-0.7199	-0.2010
81.3	-0.5683	47.6347	-0.6499	-0.7257	-0.8598
82.0	-0.5799	-0.9881	-0.6616	-0.7082	-0.8598
84.0	-0.5741	-0.5624	-0.6616	-0.7082	-0.8831
87.0	-0.5741	-0.6033	-0.6849	-0.7140	-0.8481
89.0	-0.6693	-0.6693	-0.7325	-0.7956	-0.9148
93.0	-0.7184	-0.6693	-0.7255	-0.2485	-0.9359
96.0	-0.6974	-0.6834	-0.7255	-0.7956	-0.9429
100.0	-0.6132	-0.5922	-0.5992	-0.7886	-0.2274
96.0	0.3407	0.3758	0.3477	0.3477	0.2215
84.0	0.7055	0.7125	0.6704	0.6073	0.5161
73.5	0.5745	-0.1785	0.6072	0.6072	0.4762
54.0	0.2144	0.2798	0.3125	0.3125	0.1816
33.0	0.1489	0.1489	0.1816	0.1816	0.1161
24.0	0.1170	0.1509	0.1849	0.2415	0.1170
10.0	0.2075	0.2302	0.2528	0.3208	0.2798
5.0	0.2755	0.2981	0.3774	0.4113	0.3780
2.5	0.3660	0.4113	0.4680	0.5019	0.4762

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0773
43.4	-0.0601
50.4	-0.0601
56.1	-0.0687
62.5	-0.1032
69.7	-0.1549
76.9	-0.1635
83.4	-0.1463
89.4	-0.2066
95.4	-0.2497
101.1	-0.2066
---BOTTOM---	
38.2	-0.0342
43.4	0.0089
50.4	-0.0256
56.1	0.0347
62.5	0.0951
69.7	0.1296
76.9	0.1554
83.4	0.1985
89.4	0.2416
95.4	0.1468
101.1	-0.3186

MISC.*

NO.	CP
1	-0.4404
2	0.1816
3	-0.5059
4	0.1816
5	-0.4404
6	0.0179
7	0.5091
8	-0.2625
9	-0.0871
10	0.5161
11	0.2425
12	-0.0521

TABLE A11. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 260

RUN:SEQ												
142	6	CLAERO =	1.0686	CDAERO =	0.4181	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.98		2107.09	2076.91	30.17	161.60	0.0	2.65	0.0	0.0	0.0	86.8355	0.998939E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		1.09	0.42	-0.46	0.01	0.00	0.00	1.14	-0.45	1.08	0.42	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2766	-0.2684
2.5	-0.2603	-0.2684
5.0	-0.2684	-0.2684
10.0	-0.2603	-0.2603
15.0	-0.2603	-0.2603
25.0	-0.2684	-0.2684
35.0	-0.2684	-0.2684
50.0	-0.2766	
56.0	-0.2603	-0.2603
65.0	-0.2766	-0.2603
76.0	-0.2766	-0.2684
79.0	-0.2684	-0.2684
80.5	-0.2678	-0.2561
81.0	-0.2502	-0.2502
82.0	-0.2443	-0.2619
84.0	-0.2443	
87.0	-0.2561	
89.0	-0.2408	-0.2495
93.0	-0.2495	
96.0	-0.2495	-0.2495
100.0	-0.2495	-0.2582
96.0	-0.2495	-0.2582
84.0	-0.2495	-0.2582
73.0	-0.2495	-0.2582
50.0	-0.2495	-0.2582
35.0	-0.2684	-0.2684
25.0	-0.2684	-0.2684
10.0	-0.2603	-0.2603
5.0	-0.2684	-0.2684
2.5	-0.2603	-0.2684

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.8887	-5.6604	-6.0252	-2.3412	-1.4058
2.5	-0.2310	-2.1700	-4.4397	-2.4210	-1.4400
5.0	-1.1663	-1.5769	-2.6832	-2.3069	-1.4172
10.0	-0.9496	-1.2119	-1.3373	-2.3866	-1.3830
15.0	-0.8241	-0.9724	-1.1663	-2.2042	-1.3487
24.0	-0.6758	-0.7556	-0.9610	-1.7023	-1.2917
33.0	-0.5822	-0.6812	-0.7801	-1.2089	-1.2749
54.0	-0.5163	-0.5493	-0.5822	0.1764	-1.1100
65.0	-0.5534	-0.5795	-0.6055	-0.5447	-1.0136
78.5	-0.8139	-0.5052	-0.8226	-0.8573	-0.9962
79.5	-0.6496	-0.6437	-0.7612	-0.8728	-1.0020
80.5	-0.6261	-0.6144	-0.7612	-0.8376	-0.2502
81.3	-0.6085	43.2572	-0.7671	-0.8435	-0.9962
82.0	-0.6026	-1.0197	-0.7201	-0.8670	-0.9844
84.0	-0.6203	-0.6261	-0.7318	-0.8611	-0.9727
87.0	-0.6672	-0.6672	-0.7436	-0.8258	-0.9727
89.0	-0.7304	-0.7163	-0.8152	-0.9353	-1.0060
93.0	-0.7799	-0.7304	-0.8152	-0.3135	-1.0060
96.0	-0.7658	-0.6951	-0.8081	-0.9212	-0.9777
100.0	-0.6597	-0.5962	-0.6527	-0.8717	-0.2852
96.0	0.3367	0.4356	0.3720	0.3508	0.1883
84.0	0.7112	0.7606	0.6829	0.6264	0.4285
73.5	0.5051	-0.2524	0.6381	0.6051	0.4402
54.0	0.3413	0.3413	0.3742	0.3413	0.2093
33.0	0.3413	0.3413	0.3413	0.3413	0.1764
24.0	0.3393	0.3507	0.3735	0.3964	0.2481
10.0	0.4990	0.4876	0.5104	0.5333	0.4072
5.0	0.6131	0.5789	0.5789	0.6017	0.4732
2.5	0.6929	0.5789	0.5447	0.5447	0.4732

FUSELAGE

-----TOP----		---BOTTOM---	
XLOC	CP	XLOC	CP
38.2	-0.1019	38.2	0.0197
43.4	-0.0932	43.4	0.0544
50.4	-0.1279	50.4	0.0110
56.1	-0.1627	56.1	0.1326
62.5	-0.2234	62.5	0.2107
69.7	-0.3016	69.7	0.2602
76.9	-0.2669	76.9	0.2715
83.4	-0.2061	83.4	0.2889
89.4	-0.2755	89.4	0.3062
95.4	-0.2321	95.4	0.1673
101.1	-0.1627	101.1	-0.3190

MISC.*

NO.	CP
1	-0.6812
2	0.3413
3	-1.4068
4	0.3083
5	-1.1429
6	0.0774
7	0.5063
8	-0.2923
9	-0.0098
10	0.5274
11	0.0984
12	-0.0874

TABLE A12. RUN 143, BW6V, DELF=45, CMU=0.5, (BN/B)=1

PROPULSIVE WING / CANARD PRESSURES

RUN:SEQ											
143	1	CLAERO =	0.2820	CDAERO =	0.6253	DCLC =	0.0	DWLC =	0.4420	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.05		2107.23	2079.31	27.91	155.49	0.0	2.66	0.0	0.510	0.510	86.9365	0.957290E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		0.72	0.37	-0.80	0.00	-0.01	0.06	0.35	-0.53	0.35	0.54	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4254	-0.4166
2.5	-0.4254	-0.4078
5.0	-0.4166	-0.4166
10.0	-0.4254	-0.4166
15.0	-0.4254	-0.4078
25.0	-0.4254	-0.4166
35.0	-0.4078	-0.4166
50.0	-0.4254	
56.0	-0.4342	-0.4254
65.0	-0.4254	-0.4254
76.0	-0.4430	-0.4254
79.0	-0.4254	-0.4254
80.5	-0.4227	-0.4037
81.0	-0.4100	-0.4037
82.0	-0.4037	-0.4164
84.0	-0.4164	
87.0	-0.4100	
89.0	-0.3615	-0.4084
93.0	-0.3990	
96.0	-0.3990	-0.4084
100.0	-0.3990	-0.4084
96.0	-0.3990	-0.4084
84.0	-0.3990	-0.4084
73.0	-0.3990	-0.4084
50.0	-0.4084	-0.3990
35.0	-0.4166	-0.4254
25.0	-0.4342	-0.4254
10.0	-0.4166	-0.4254
5.0	-0.4254	-0.4254
2.5	-0.4166	-0.4342

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.1611	-0.5170	-1.6390	-2.0089	-3.4267
2.5	-0.4061	-0.8129	-1.4294	-1.7130	-1.9596
5.0	-0.4924	-0.6897	-0.9979	-1.2322	-1.6514
10.0	-0.5540	-0.6650	-0.8992	-1.0349	-1.1828
15.0	-0.5047	-0.6280	-0.8252	-0.9362	-1.0595
24.0	-0.5294	-0.6157	-0.7636	-0.8006	-0.9239
33.0	-0.5692	-0.6761	-0.7831	-0.8544	-0.9257
54.0	-0.8188	-0.9257	-1.0683	0.2152	-1.2466
65.0	-1.2344	-1.3564	-1.4221	-1.5536	-1.7319
78.5	-3.2619	-3.5059	-3.1211	-4.0505	-5.5709
79.5	-15.8768	-18.2134	-9.5719	-19.1087	-37.0007
80.5	-17.1723	-17.1404	-9.1022	-16.6958	-0.0735
81.3	-17.6928	46.1040	-10.7151	-17.3944	-7.4514
82.0	-16.9183	-16.3977	-11.2865	-15.0643	-4.3893
84.0	-7.6355	-8.6261	-11.5341	-9.2419	-3.9021
87.0	0.0725	-2.1053	-7.4324	-3.5718	-3.2734
89.0	-0.3296	-0.4365	-3.2476	-1.5060	-2.9495
93.0	-0.6733	-0.4824	-0.3296	-0.4824	-3.6908
96.0	-0.6938	-0.4977	0.3120	-0.8109	-3.5532
100.0	0.0523	0.1975	0.0600	-0.0240	-0.4289
96.0	0.5870	0.7169	0.6787	0.6176	0.2204
84.0	0.6634	0.6787	0.7474	0.6176	0.0962
73.5	0.6073	-0.4266	0.6073	0.5717	0.1795
54.0	0.3578	0.4291	0.4291	0.3934	0.1082
33.0	0.2152	0.2508	0.2865	0.2865	0.1082
24.0	0.1734	0.1981	0.2598	0.2844	0.1241
10.0	0.1858	0.2228	0.2844	0.3830	0.2865
5.0	0.2598	0.2721	0.3830	0.4200	0.3934
2.5	0.3707	0.3584	0.4693	0.5084	0.4291

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0987
43.4	-0.0893
50.4	-0.1081
56.1	-0.1174
62.5	-0.1644
69.7	-0.2676
76.9	-0.3052
83.4	-0.3239
89.4	-0.4272
95.4	-0.4178
101.1	-0.3333
---BOTTOM---	
38.2	-0.0330
43.4	-0.0142
50.4	-0.0611
56.1	0.0515
62.5	0.1078
69.7	0.1735
76.9	0.2392
83.4	0.3143
89.4	0.3513
95.4	0.2111
101.1	-0.3427

MISC.

NO.	CP
1	-0.7118
2	0.2865
3	-0.9614
4	0.2152
5	-0.9257
6	-0.0701
7	0.2510
8	-0.9102
9	-0.5740
10	0.8697
11	0.0294
12	-0.2914

Force Data -
Use Run 313

TABLE A12. (Continued)

RUN:SEQ 143 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 263

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
4.01	2107.16	2078.57	28.59	157.44	0.0	2.66	0.0	0.521	0.521

CLAERO = 0.2949 CDAERO = 0.6624 DCLC = 0.0 DWLC = 0.4681 BASEPR = 8.1904

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	0.76	0.43	-0.78	-0.01	-0.02	0.07	0.41	-0.50	0.37	0.57

HGT	RN
87.0371	0.967873E+06

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4166	-0.3822
2.5	-0.4166	-0.3564
5.0	-0.4080	-0.3479
10.0	-0.3994	-0.3651
15.0	-0.3994	-0.3736
25.0	-0.3994	-0.3736
35.0	-0.3994	-0.4080
50.0	-0.3908	
56.0	-0.3908	-0.4080
65.0	-0.4080	-0.4080
76.0	-0.4166	-0.4166
79.0	-0.4080	-0.4251
80.5	-0.4006	-0.3820
81.0	-0.3803	-0.3820
82.0	-0.3882	-0.3944
84.0	-0.3696	
87.0	-0.3882	
89.0	-0.3101	-0.3468
93.0	-0.3284	
96.0	-0.3284	-0.3468
100.0	-0.3376	-0.3743
96.0	-0.3468	-0.3651
84.0	-0.3376	-0.3743
73.0	-0.3468	-0.3651
50.0	-0.3376	-0.3651.
35.0	-0.4080	-0.4251
25.0	-0.3994	-0.4251
10.0	-0.4166	-0.4338
5.0	-0.4080	-0.3736
2.5	-0.3908	-0.3822

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8775	-1.8044	-3.9953	-5.5000	-2.4183
2.5	-0.3840	-1.2868	-2.0692	-2.5987	-2.2136
5.0	-0.7692	-0.9257	-1.4553	-1.8525	-2.2618
10.0	-0.7451	-0.8414	-1.2025	-1.4312	-2.1053
15.0	-0.7090	-0.7331	-1.0219	-1.2506	-2.0090
24.0	-0.6247	-0.6608	-0.8896	-1.0701	-1.7201
33.0	-0.7052	-0.7749	-0.7749	-0.9837	-1.0533
54.0	-0.8793	-0.9837	-0.9489	0.3042	-0.9489
65.0	-1.2356	-1.3731	-1.4556	-1.5930	-1.9779
78.5	-3.0866	-3.4259	-3.0317	-3.9114	-5.6158
79.5	-11.7443	-3.9897	-7.6409	-12.9408	-32.1566
80.5	-9.4012	-3.7603	-7.0333	-12.4511	-0.1217
81.3	-6.4692	38.3652	-6.9092	-12.8415	-13.5543
82.0	-2.3222	-3.5186	-7.5478	-12.8166	-7.7958
84.0	-2.5578	-7.4859	-6.8969	-11.3291	-3.4567
87.0	-3.6365	-3.5249	-7.4114	-8.7503	-3.6365
89.0	-0.0959	-0.6329	-6.4199	-7.2404	-3.0940
93.0	-0.3271	-0.5658	-2.9446	-0.4763	-3.8397
96.0	-0.4912	-0.4688	-0.5136	-0.5956	-3.6756
100.0	0.1427	0.1651	-0.0288	0.0085	-0.4017
96.0	0.6200	0.6796	0.6796	0.6349	0.2247
84.0	0.7318	0.7095	0.8213	0.6424	0.0606
73.5	0.6522	-0.4268	0.6522	0.5826	0.1649
54.0	0.4086	0.4434	0.4434	0.4086	0.1301
33.0	0.3042	0.3042	0.3042	0.3042	0.0953
24.0	0.2660	0.2660	0.3142	0.3382	0.1577
10.0	0.3744	0.3623	0.3984	0.4586	0.3042
5.0	0.4225	0.4346	0.4827	0.5429	0.4434
2.5	0.5309	0.5068	0.5429	0.4947	0.4434

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1177
43.4	-0.0993
50.4	-0.1268
56.1	-0.1452
62.5	-0.2093
69.7	-0.3284
76.9	-0.3651
83.4	-0.3743
89.4	-0.5209
95.4	-0.4567
101.1	-0.3743
---BOTTOM---	
38.2	-0.0260
43.4	0.0015
50.4	-0.0444
56.1	0.0748
62.5	0.1481
69.7	0.2214
76.9	0.2672
83.4	0.3497
89.4	0.3772
95.4	0.2122
101.1	-0.2643

MISC.*

NO.	CP
1	-0.8097
2	0.3042
3	-1.0185
4	0.2693
5	-1.3318
6	-0.1135
7	0.2173
8	-0.9759
9	-0.6702
10	0.8661
11	-0.0064
12	-0.3495

Force Data -
Use Run 313

TABLE A12. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 265

RUN	SEQ														
143	5	CLAERO =	0.4314	CDAERO =	0.7349	DCLC =	0.0	DWLC =	0.4717	BASEPR =	8.1904				
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN			
11.97		2107.16	2079.58	27.57	154.67	0.0	2.66	0.0	0.496	0.496	87.1425	0.949446E+06			
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR				
0.0		0.90	0.58	-0.82	0.02	-0.02	0.05	0.63	-0.54	0.50	0.67				

********** CANARD ******			********** WING **********						**FUSELAGE**			
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		**MISC.***	
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.5127	-0.4949	0.0	-4.8901	-9.3958	-5.3768	-3.5421	-2.6812	38.2	-0.1203	1	-1.1284
2.5	-0.4949	-0.4771	2.5	-0.4469	-4.0788	-5.4145	-3.7670	-2.6561	43.4	-0.1203	2	0.4957
5.0	-0.4949	-0.4771	5.0	-1.9197	-2.3690	-5.2647	-3.8544	-2.7435	50.4	-0.1868	3	-4.1501
10.0	-0.4860	-0.4682	10.0	-1.5203	-1.8448	-4.3911	-4.1412	-2.8556	56.1	-0.2438	4	0.4235
15.0	-0.4949	-0.4593	15.0	-1.3081	-1.5452	-3.7920	-4.2163	-2.8433	62.5	-0.3483	5	-2.8607
25.0	-0.4860	-0.4593	24.0	-1.1583	-1.2707	-1.1833	-3.8667	-2.9183	69.7	-0.4813	6	-0.0818
35.0	-0.4593	-0.4326	33.0	-1.0563	-1.1645	-0.8758	-1.9946	-3.3300	76.9	-0.4813	7	0.0885
50.0	-0.4593		54.0	-1.1284	-1.2728	-1.2728	0.2430	-4.5209	83.4	-0.4053	8	-1.3807
56.0	-0.4682	-0.4237	65.0	-1.5455	-1.7071	-1.6880	-1.4600	-4.8140	89.4	-0.4813	9	-0.9863
65.0	-0.4415	-0.4237	78.5	-3.7024	-3.7593	-3.8355	-4.7665	-6.3629	95.4	-0.3958	10	0.9004
76.0	-0.4415	-0.4326	79.5	-17.8055	-20.1066	-11.3973	-28.7448	-38.4057	101.1	-0.3198	11	-0.1899
79.0	-0.4415	-0.4148	80.5	-17.3685	-18.7245	-11.9116	-23.4743	-0.0531	---BOTTOM---		12	-0.5997
80.5	-0.4838	-0.4645	81.3	-18.7633	34.3011	-12.8052	-19.1811	-8.6016	38.2	0.0128		
81.0	-0.4709	-0.4581	82.0	-17.4135	-17.4969	-12.7279	-16.4492	-6.8148	43.4	0.0508		
82.0	-0.4645	-0.4517	84.0	-6.3070	-6.3712	-13.4736	-3.6589	-5.9600	50.4	-0.0063		
84.0	-0.4517		87.0	0.7631	-1.6278	-6.6991	-2.0135	-4.7131	56.1	0.1648		
87.0	-0.4645		89.0	-0.4992	-1.3885	-2.2390	-2.5715	-3.9788	62.5	0.2693		
89.0	-0.4338	-0.4528	93.0	-1.0018	-0.8781	0.2432	-0.5533	-4.3965	69.7	0.3738		
93.0	-0.4623		96.0	-0.9863	-0.6771	-0.0739	-1.3189	-3.9944	76.9	0.4118		
96.0	-0.4528	-0.4528	100.0	-0.0198	0.1561	0.0266	-0.0430	-0.4296	83.4	0.4593		
100.0	-0.4528	-0.4528	96.0	0.5602	0.7690	0.7148	0.5989	0.1349	89.4	0.4688		
96.0	-0.4433	-0.4433	84.0	0.7535	0.8849	0.7612	0.6530	0.0266	95.4	0.2978		
84.0	-0.4528	-0.4528	73.5	0.6761	-0.4427	0.6761	0.6400	0.1348	101.1	-0.2723		
73.0	-0.4433	-0.4433	54.0	0.4595	0.5318	0.5318	0.4595	0.0987				
50.0	-0.4433	-0.4528	33.0	0.4595	0.4595	0.4957	0.4595	0.1708				
35.0	-0.4415	-0.4237	24.0	0.4393	0.5142	0.5142	0.5266	0.2770				
25.0	-0.4415	-0.4148	10.0	0.6015	0.5766	0.5891	0.5891	0.4595				
10.0	-0.4504	-0.4237	5.0	0.7014	0.6265	0.5891	0.5516	0.3874				
5.0	-0.4860	-0.4593	2.5	0.7388	0.4767	0.4018	0.2770	0.2069				
2.5	-0.4860	-0.4593										

Force Data -
Use Run 313

TABLE A13. RUN 144, BW6V, DELF=45, CMU=1.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
144	1	CLAERO =	0.8342	CDAERO =	0.8265	DCLC =	0.0	DWLC =	0.8711	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.07		2107.86	2094.08	13.79	109.08	0.0	2.66	0.0	1.005	1.005	86.9286	0.673305E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		1.71	0.33	-1.38	0.02	-0.01	0.07	0.96	-0.82	0.96	0.66	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4949	-0.4949
2.5	-0.4949	-0.4949
5.0	-0.4949	-0.4949
10.0	-0.4949	-0.4771
15.0	-0.4949	-0.4593
25.0	-0.4771	-0.4771
35.0	-0.4949	-0.4771
50.0	-0.4771	
56.0	-0.5127	-0.4949
65.0	-0.4771	-0.4771
76.0	-0.4949	-0.4949
79.0	-0.4771	-0.4771
80.5	-0.5031	-0.5287
81.0	-0.5031	-0.5159
82.0	-0.5159	-0.4902
84.0	-0.4902	
87.0	-0.5031	
89.0	-0.4243	-0.4623
93.0	-0.4813	
96.0	-0.4623	-0.4623
100.0	-0.4813	-0.4623
96.0	-0.4623	-0.4623
84.0	-0.4813	-0.4623
73.0	-0.4623	-0.4623
50.0	-0.4623	-0.4623
35.0	-0.4771	-0.4949
25.0	-0.4949	-0.4771
10.0	-0.4949	-0.4771
5.0	-0.4771	-0.4949
2.5	-0.4771	-0.4949

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.1772	-0.4968	-1.6949	-3.1925	-4.3408
2.5	-0.4718	-1.0459	-1.3954	-1.8197	-2.6933
5.0	-0.5717	-0.7713	-1.1458	-1.4703	-1.8946
10.0	-0.5717	-0.7214	-0.9960	-1.1708	-1.3704
15.0	-0.5966	-0.7464	-0.9710	-1.0459	-1.2456
24.0	-0.5966	-0.6965	-0.8463	-0.9461	-1.0958
33.0	-0.6592	-0.7314	-0.9479	-1.0201	-1.0923
54.0	-0.9479	-1.0923	-1.3088	-0.2983	-1.5253
65.0	-1.4694	-1.6594	-1.7735	-1.9065	-2.0585
78.5	-4.4527	-4.9278	-5.1938	-5.7829	-6.7138
79.5	-46.6094	-56.5970	-46.5456	-60.3503	-62.0467
80.5	-44.6685	-42.8561	-40.5809	-31.6609	0.0625
81.3	-27.5471	68.0077	-28.3188	-7.3927	-9.8992
82.0	-4.7576	7.1964	-17.1098	-1.9427	-6.4027
84.0	0.9366	-3.0481	3.3788	-3.5494	-4.9119
87.0	-0.5159	-3.8836	-2.8296	-2.7525	-3.7808
89.0	-2.1383	-2.5868	-3.4992	-3.9941	-3.1281
93.0	-1.5198	-1.3961	-1.8909	-0.6074	-4.5044
96.0	-1.5043	-1.4734	-0.9786	-1.1487	-4.6900
100.0	-0.5610	-0.4837	-0.3445	-0.6538	-0.4992
96.0	0.5679	0.6761	0.6452	0.5988	0.1194
84.0	0.6297	0.7225	0.7380	0.5833	0.0420
73.5	0.5678	-0.4427	0.6400	0.5678	0.2069
54.0	0.3513	0.3513	0.4234	0.3513	0.0626
33.0	0.2069	0.2069	0.2069	0.2069	0.0626
24.0	0.1772	0.2021	0.2520	0.3020	0.1272
10.0	0.2021	0.2271	0.3020	0.4018	0.2791
5.0	0.2021	0.3269	0.4018	0.4517	0.3513
2.5	0.2770	0.3519	0.4767	0.4767	0.4234

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1013
43.4	-0.1013
50.4	-0.1393
56.1	-0.1203
62.5	-0.1773
69.7	-0.2913
73.9	-0.3483
83.4	-0.3673
89.4	-0.4813
95.4	-0.4433
101.1	-0.3673
---BOTTOM---	
38.2	-0.0443
43.4	-0.0063
50.4	-0.0823
56.1	0.0507
62.5	0.1077
69.7	0.1648
76.9	0.2408
83.4	0.3168
89.4	0.3548
95.4	0.1458
101.1	-0.5003

MISC.*

NO.	CP
1	-0.8757
2	0.2069
3	-1.0923
4	0.2069
5	-1.0923
6	-0.0818
7	0.1658
8	-1.0713
9	-0.6848
10	0.8926
11	-0.0971
12	-0.3600

Force Data -
Use Run 312

TABLE A13. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 268

RUN:SEQ													
144 3	CLAERO =	0.8837	CDAERO =	0.9060	DCLC =	0.0	DWLC =	0.9225	BASEPR =	8.1904			
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
4.04	2108.08	2094.52	13.56	108.13	0.0	2.66	0.0	1.026	1.023	86.9895	0.668554E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR			
0.0	1.81	0.46	-1.41	0.02	-0.02	0.08	1.07	-0.94	1.02	0.74			

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.5178	-0.5178	0.0	-0.9765	-2.3723	-5.1891	-4.8592	-2.4992	38.2	-0.1177	1	-0.9785
2.5	-0.4997	-0.4816	2.5	-0.5198	-1.7125	-2.4230	-4.1486	-2.3469	43.4	-0.1177	2	0.3424
5.0	-0.4997	-0.5178	5.0	-0.9258	-1.2811	-1.8901	-4.0725	-2.3723	50.4	-0.1370	3	-1.3454
10.0	-0.4997	-0.4997	10.0	-0.8496	-1.1035	-1.5349	-1.3319	-2.2707	56.1	-0.1950	4	0.1956
15.0	-0.4997	-0.5178	15.0	-0.8496	-1.0019	-1.3572	-1.1796	-2.1693	62.5	-0.2529	5	-1.8590
25.0	-0.5178	-0.5178	24.0	-0.7736	-0.9258	-1.1542	-1.1288	-2.0678	69.7	-0.3688	6	-0.2447
35.0	-0.5178	-0.4997	33.0	-0.8317	-0.9785	-1.1252	-1.2719	-2.0057	76.9	-0.4075	7	0.1538
50.0	-0.5178		54.0	-1.1252	-1.1986	-1.4187	-0.2447	-1.5655	83.4	-0.4268	8	-1.1983
56.0	-0.4997	-0.5178	65.0	-1.5859	-1.7984	-1.9143	-2.0689	-2.7257	89.4	-0.5234	9	-0.8209
65.0	-0.5178	-0.5178	78.5	-4.5996	-5.1405	-5.3917	-6.1644	-7.3817	95.4	-0.4847	10	0.8928
76.0	-0.5360	-0.4997	79.5	-47.2570	-57.2673	-44.5259	-60.5995	-63.2657	101.1	-0.4075	11	-0.0977
79.0	-0.5178	-0.5178	80.5	-45.6496	-45.2057	-40.4358	-35.3259	0.0680	---BOTTOM---		12	-0.3984
80.5	-0.4739	-0.4608	81.3	-29.2626	63.2849	-31.9938	-10.4970	-10.9940	38.2	-0.0404		
81.0	-0.4608	-0.4739	82.0	-7.0344	7.2754	-22.4804	-2.1858	-7.3477	43.4	-0.0018		
82.0	-0.4608	-0.4347	84.0	1.4079	-2.8915	3.7301	-3.4142	-5.7403	50.4	-0.0790		
84.0	-0.4608		87.0	-0.4347	-3.9108	-2.4864	-2.7999	-4.6164	56.1	0.0562		
87.0	-0.4608		89.0	-2.1573	-2.6133	-3.5094	-4.2798	-3.9339	62.5	0.1528		
89.0	-0.4268	-0.4847	93.0	-1.4970	-1.4341	-2.0001	-0.6008	-5.4274	69.7	0.2300		
93.0	-0.4654		96.0	-1.4498	-1.4655	-1.1197	-1.3240	-5.5375	76.9	0.2880		
96.0	-0.4654	-0.4847	100.0	-0.5065	-0.4908	-0.3178	-0.6952	-0.4908	83.4	0.3460		
100.0	-0.4654	-0.4847	96.0	0.6255	0.7041	0.6569	0.5783	0.0438	89.4	0.4039		
96.0	-0.4654	-0.4847	84.0	0.7198	0.8142	0.7827	0.6412	-0.0191	95.4	0.1914		
84.0	-0.4654	-0.4847	73.5	0.6359	-0.4648	0.6359	0.5625	0.0489	101.1	-0.4847		
73.0	-0.4654	-0.4847	54.0	0.4158	0.4158	0.4158	0.3424	-0.0245				
50.0	-0.4654	-0.4847	33.0	0.2690	0.3424	0.3424	0.2690	-0.0245				
35.0	-0.4997	-0.4997	24.0	0.2669	0.2923	0.3176	0.3938	0.0892				
25.0	-0.5178	-0.5178	10.0	0.3430	0.3430	0.4191	0.4953	0.3424				
10.0	-0.4997	-0.5178	5.0	0.4191	0.4191	0.4699	0.5714	0.4158				
5.0	-0.5178	-0.5178	2.5	0.4953	0.5206	0.5206	0.5206	0.3424				
2.5	-0.4997	-0.5178										

Force Data -
Use Run 312

TABLE A13. (Concluded)

RUN:SEQ 144 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 270

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC
11.99	2107.86	2094.76	13.11	106.28	0.0	2.66	0.0

CLAERO = 1.0612 CDAERO = 1.0923 DCLC = 0.0 DWLC = 1.0102 BASEPR = 8.1904

CMUW	CMUT	HGT	RN
1.062	1.062	86.9901	0.657723E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	2.07	0.76	-1.46	0.03	-0.02	0.06	1.39	-0.87	1.22	0.94

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.5551	-0.5364
2.5	-0.5364	-0.5551
5.0	-0.5364	-0.5364
10.0	-0.5739	-0.5364
15.0	-0.5551	-0.5177
25.0	-0.5364	-0.5177
35.0	-0.5551	-0.5739
50.0	-0.5551	
56.0	-0.5364	-0.5551
65.0	-0.5551	-0.5551
76.0	-0.5739	-0.5364
79.0	-0.5551	-0.5551
80.5	-0.5689	-0.5419
81.0	-0.5149	-0.5419
82.0	-0.5419	-0.5284
84.0	-0.5419	
87.0	-0.5554	
89.0	-0.4597	-0.5396
93.0	-0.5396	
96.0	-0.5396	-0.5197
100.0	-0.5396	-0.5396
96.0	-0.5197	-0.5396
84.0	-0.5396	-0.5396
73.0	-0.5197	-0.5396
50.0	-0.5197	-0.5396
35.0	-0.5551	-0.5739
25.0	-0.5551	-0.5551
10.0	-0.5551	-0.5551
5.0	-0.5551	-0.5364
2.5	-0.5364	-0.5551

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-5.3803	-8.4255	-4.9865	-3.8051	-3.0437
2.5	-0.5234	-7.4277	-5.3803	-4.0676	-3.0963
5.0	-2.0461	-2.9388	-5.2753	-4.3302	-3.2801
10.0	-1.7048	-1.5998	-6.1154	-4.4352	-3.3325
15.0	-1.4948	-1.5473	-4.5139	-5.0916	-3.2275
24.0	-1.2585	-1.3373	-1.3635	-4.7765	-3.5163
33.0	-1.2022	-1.2782	-1.0504	-2.8724	-3.6593
54.0	-1.3540	-1.4300	-1.3540	-0.4431	-5.3016
65.0	-1.7788	-1.9986	-1.9387	-1.6589	-5.7361
78.5	-4.7767	-5.4763	-5.7561	-6.1558	-8.0342
79.5	-49.7803	-58.6757	-44.4672	-63.4350	-59.6766
80.5	-47.9419	-48.7123	-41.2625	-37.6261	-0.0282
81.3	-29.3928	60.2288	-35.2872	-12.1550	-12.6959
82.0	-6.8014	6.9479	-26.9456	-2.2995	-9.0457
84.0	1.5536	-2.8538	2.3783	-3.9218	-7.7207
87.0	-0.4879	-4.0841	-2.0426	-3.5162	-5.8281
89.0	-2.2862	-2.7580	-3.5549	-4.8236	-4.7748
93.0	-1.5543	-1.6031	-2.2212	-0.6597	-5.8971
96.0	-1.4730	-1.5706	-1.2941	-1.5869	-5.6857
100.0	-0.1555	-0.3344	-0.2369	-0.6923	-0.5297
96.0	0.5927	0.7878	0.6740	0.5764	-0.0092
84.0	0.7065	0.9017	0.8529	0.6903	0.0397
73.5	0.6197	-0.5190	0.6956	0.6197	0.0124
54.0	0.3920	0.5438	0.4679	0.3920	-0.0635
33.0	0.3920	0.4679	0.3920	0.3920	0.0124
24.0	0.4480	0.4742	0.4742	0.5267	0.1854
10.0	0.6055	0.5530	0.5530	0.5267	0.3161
5.0	0.6560	0.5792	0.5005	0.4480	0.3161
2.5	0.6842	0.4480	0.2642	0.1854	0.0124

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1799
43.4	-0.1999
50.4	-0.2538
56.1	-0.3398
62.5	-0.4397
69.7	-0.5796
76.9	-0.5796
83.4	-0.5197
89.4	-0.6196
95.4	-0.5396
101.1	-0.4597
---BOTTOM---	
38.2	-0.0000
43.4	0.0200
50.4	-0.0600
56.1	0.1399
62.5	0.2598
69.7	0.3597
76.9	0.3997
83.4	0.4197
89.4	0.4397
95.4	0.1799
101.1	-0.5396

MISC.

NO.	CP
1	-1.2782
2	0.3920
3	-4.8462
4	0.3920
5	-3.4038
6	-0.2154
7	-0.0579
8	-1.5869
9	-1.1802
10	0.8855
11	-0.2369
12	-0.7085

Force Data - Use Run 312

TABLE A14. RUN 145, BW6V, DELF=45, CNU=2.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
145 1	CLAERO =	1.8098	CDAERO =	1.1250	DCLC =	0.0	DWLC =	1.7854	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CNUC	CNUW	CNUT	HGT	RN
0.04	2107.72	2101.06	6.67	75.69	0.0	2.67	0.0	2.061	2.061	87.2267	0.470101E+06
BETA	CL	CD	CM	CROLL		CN	CY	CNTR	CNTR	CLTR	CDTR
0.0	3.60	0.10	-2.54	0.03		0.00	0.02	2.03	-1.37	2.03	0.79

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6671	-0.6671
2.5	-0.6304	-0.6304
5.0	-0.6671	-0.6304
10.0	-0.6671	-0.6304
15.0	-0.5935	-0.6671
25.0	-0.6304	-0.6671
35.0	-0.6671	-0.6671
50.0	-0.6671	
56.0	-0.6671	-0.6671
65.0	-0.6671	-0.7040
76.0	-0.7040	-0.7040
79.0	-0.7040	-0.7040
80.5	-0.6581	-0.6050
81.0	-0.6316	-0.5518
82.0	-0.7113	-0.6050
84.0	-0.6050	
87.0	-0.5518	
89.0	-0.5426	-0.6605
93.0	-0.6212	
96.0	-0.6212	-0.6212
100.0	-0.6212	-0.6212
96.0	-0.6605	-0.6212
84.0	-0.6212	-0.6212
73.0	-0.6605	-0.6212
50.0	-0.6212	-0.6212
35.0	-0.7407	-0.7040
25.0	-0.6671	-0.7040
10.0	-0.6671	-0.7040
5.0	-0.5935	-0.6671
2.5	-0.6304	-0.6671

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.1457	-1.1780	-2.6232	-3.8619	-3.2941
2.5	-0.7135	-1.2812	-1.8490	-2.1070	-3.2426
5.0	-0.7135	-0.9715	-1.5393	-1.7974	-2.8812
10.0	-0.8168	-0.9715	-1.3844	-1.4877	-1.8490
15.0	-0.7650	-0.9200	-1.2296	-1.3328	-1.8490
24.0	-0.7650	-0.9200	-1.2296	-1.2296	-1.4877
33.0	-0.8012	-0.9504	-1.0996	-1.2490	-1.2490
54.0	-1.2490	-1.2490	-1.5474	-1.8459	-1.8459
65.0	-1.8393	-2.0357	-2.1536	-2.3108	-2.5072
78.5	-5.8864	-6.3972	-6.8294	-7.3402	-8.2439
79.5	*********	-144.3701	-132.3830	-128.5018	-84.8605
80.5	-11.3429	-13.0173	-15.4889	-24.9506	0.1392
81.3	0.8835	131.3587	-4.0602	-5.4955	-15.3825
82.0	7.8737	-6.7713	6.0929	-6.4789	-8.0737
84.0	-5.6550	-5.5752	-6.0803	-5.2829	-5.7878
87.0	1.0695	-6.5321	-5.4158	-3.2097	-4.3261
89.0	-2.6875	-3.8067	-6.0771	-6.0131	-3.1032
93.0	-1.5363	-1.8882	-2.3358	-0.7370	-5.9171
96.0	-1.7602	-2.4957	-0.4811	-0.8969	-6.8126
100.0	-1.6323	-1.6323	-1.7283	-1.8561	-0.6410
96.0	0.4781	0.6381	0.5421	0.5101	-0.0654
84.0	0.6381	0.7979	0.7660	0.6381	-0.0015
73.5	0.5420	-0.6519	0.6912	0.5420	-0.0550
54.0	0.3928	0.3928	0.3928	0.3928	-0.0550
33.0	0.2435	0.2435	0.2435	0.2435	-0.0550
24.0	0.2155	0.1123	0.1640	0.3188	-0.0941
10.0	0.2672	0.2155	0.2155	0.3188	0.2435
5.0	0.2672	0.2672	0.3188	0.4220	0.3928
2.5	0.3188	0.2672	0.4220	0.4220	0.5420

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1890
43.4	-0.1890
50.4	-0.2283
56.1	-0.2676
62.5	-0.3855
69.7	-0.4248
76.9	-0.5033
83.4	-0.5033
89.4	-0.6605
95.4	-0.6212
101.1	-0.5426
---BOTTOM---	
38.2	-0.1104
43.4	-0.0711
50.4	-0.1497
56.1	-0.0318
62.5	0.0455
69.7	0.1253
76.9	0.1646
83.4	0.2825
89.4	0.2825
95.4	0.0075
101.1	-0.8177

MISC.*

NO.	CP
1	-0.9504
2	0.2435
3	-1.2490
4	0.2435
5	-1.2490
6	-0.2042
7	-0.0015
8	-1.4084
9	-0.8969
10	0.8299
11	-0.1294
12	-0.4172

Force Data - Use Run 311

TABLE A14. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
145 3	CLAERO =	1.7577	CDAERO =	1.2786	DCLC =	0.0	DWLC =	1.8619	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.01	2107.51	2100.62	6.89	76.93	0.0	2.66	0.0	2.071	2.071	87.0823	0.478439E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	3.62	0.37	-2.50	0.03	-0.01	0.06	2.11	-1.37	2.06	0.93	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6730	-0.6374
2.5	-0.6730	-0.6374
5.0	-0.6374	-0.6730
10.0	-0.6730	-0.6374
15.0	-0.6730	-0.6374
25.0	-0.6374	-0.6730
35.0	-0.6374	-0.6374
50.0	-0.6374	
56.0	-0.6730	-0.6374
65.0	-0.6374	-0.6730
76.0	-0.6730	-0.6374
79.0	-0.6730	-0.6374
80.5	-0.6187	-0.5930
81.0	-0.5930	-0.6187
82.0	-0.5930	-0.6187
84.0	-0.6187	
87.0	-0.5930	
89.0	-0.5574	-0.5954
93.0	-0.5954	
96.0	-0.5954	-0.5954
100.0	-0.5954	-0.6334
96.0	-0.5954	-0.5954
84.0	-0.5954	-0.6334
73.0	-0.5954	-0.5954
50.0	-0.5954	-0.6334
35.0	-0.6730	-0.6374
25.0	-0.6374	-0.6730
10.0	-0.6374	-0.6374
5.0	-0.6730	-0.6374
2.5	-0.6730	-0.6730

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.4454	-3.6920	-5.9386	-3.8916	-2.7434
2.5	-0.6466	-1.7949	-3.8417	-3.9915	-2.5936
5.0	-1.1459	-1.5452	-2.1443	-3.9915	-2.5437
10.0	-1.0960	-1.2956	-1.7450	-3.2426	-2.5437
15.0	-1.0459	-1.1958	-1.4953	-1.7949	-2.5437
24.0	-0.9960	-1.1958	-1.3955	-1.0960	-2.4439
33.0	-1.0201	-1.1646	-1.3089	-1.1646	-2.1751
54.0	-1.3089	-1.4533	-1.5976	-2.0308	-2.3194
65.0	-1.9256	-2.1916	-2.3057	-2.4197	-3.2178
78.5	-5.9542	-6.6004	-7.1324	-7.6645	-9.4128
79.5	********	-136.6717	-129.7044	-124.0484	-85.5359
80.5	-38.9260	-16.1208	-24.4256	-26.1228	0.1268
81.3	3.1861	129.8291	-3.4210	-4.8608	-16.1730
82.0	11.6959	-5.2978	6.7597	-5.9920	-9.4112
84.0	-5.4007	-5.6064	-5.8378	-5.2207	-7.0718
87.0	0.4353	-6.4804	-5.3235	-3.4210	-5.0922
89.0	-2.7107	-3.8861	-6.1129	-6.0821	-3.9788
93.0	-1.6282	-2.0612	-2.5251	-0.7931	-6.9481
96.0	-1.9065	-2.5860	-0.6384	-1.1333	-7.8760
100.0	-1.3498	-1.4736	-1.5972	-1.7829	-0.6074
96.0	0.5679	0.5607	0.5370	0.4751	-0.1435
84.0	0.6607	0.8463	0.7844	0.6298	-0.0198
73.5	0.7121	-0.5871	0.7121	0.5678	-0.0096
54.0	0.4234	0.4234	0.4234	0.4234	-0.1540
33.0	0.2791	0.4234	0.2791	0.2791	-0.0096
24.0	0.2520	0.3020	0.2520	0.3519	-0.0475
10.0	0.3519	0.3020	0.4018	0.5017	0.4234
5.0	0.4018	0.4018	0.5017	0.5516	0.4234
2.5	0.5017	0.4517	0.4517	0.4517	0.4234

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1393
43.4	-0.1393
50.4	-0.2153
56.1	-0.2533
62.5	-0.3293
69.7	-0.4434
76.9	-0.5194
83.4	-0.5574
89.4	-0.6334
95.4	-0.5574
101.1	-0.5194
---BOTTOM---	
38.2	-0.0633
43.4	-0.0253
50.4	-0.1013
56.1	0.0506
62.5	0.1268
69.7	0.2408
76.9	0.2788
83.4	0.3548
89.4	0.3548
95.4	0.0888
101.1	-0.7474

MISC.*

NO.	CP
1	-1.1646
2	0.4234
3	-1.7419
4	0.2791
5	-2.0308
6	-0.2983
7	-0.0198
8	-1.3807
9	-1.0404
10	0.9081
11	-0.1744
12	-0.4837

Force Data -
Use Run 311

TABLE A14. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
145	5	CLAERO =	1.7972	CDAERO =	1.4909	DCLC =	0.0	DWLC =	1.8464	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.99		2107.37	2100.14	7.23	78.79	0.0	2.66	0.0	1.942	1.942	86.9550	0.490460E+06
BETA		CL	CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
0.0		3.64	0.89	-2.41	0.04		-0.03	0.09	2.29	-1.33	2.08	1.21

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6200	-0.6200
2.5	-0.6200	-0.6200
5.0	-0.6200	-0.6200
10.0	-0.6200	-0.6200
15.0	-0.6200	-0.5860
25.0	-0.6200	-0.6200
35.0	-0.6539	-0.6200
50.0	-0.6539	
56.0	-0.6200	-0.6200
65.0	-0.6879	-0.6539
76.0	-0.6200	-0.6200
79.0	-0.6200	-0.6539
80.5	-0.7303	-0.6814
81.0	-0.6814	-0.6323
82.0	-0.7058	-0.6568
84.0	-0.6568	
87.0	-0.6568	
89.0	-0.5845	-0.6931
93.0	-0.6569	
96.0	-0.6569	-0.6569
100.0	-0.6569	-0.6931
96.0	-0.6569	-0.6931
84.0	-0.6569	-0.6931
73.0	-0.6569	-0.6931
50.0	-0.6931	-0.6569
35.0	-0.6200	-0.6200
25.0	-0.6539	-0.6200
10.0	-0.6539	-0.6200
5.0	-0.6200	-0.6200
2.5	-0.6200	-0.6200

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-6.8404	-9.0287	-5.0326	-4.0337	-3.2725
2.5	-0.7036	-8.5055	-5.4608	-4.2714	-3.4152
5.0	-2.2259	-5.5085	-5.9841	-4.4618	-3.5103
10.0	-1.8453	-1.6075	-6.7929	-4.9376	-3.6054
15.0	-1.7026	-1.6551	-5.8414	-5.7939	-3.7482
24.0	-1.4647	-1.4647	-1.7026	-5.5085	-3.9861
33.0	-1.3295	-1.3295	-1.0545	-3.2553	-4.4934
54.0	-1.4671	-1.6046	-1.6046	-1.7423	-6.2817
65.0	-2.0331	-2.3591	-2.2142	-1.8883	-6.2342
78.5	-6.0169	-6.8136	-7.2482	-7.6104	-9.6746
79.5	*********	-120.0598	-119.1530	-118.2960	-80.0053
80.5	-57.2473	-32.7242	-46.4185	-32.6026	-0.0689
81.3	1.9645	125.4589	-6.1198	-4.6255	-17.0948
82.0	15.4387	0.8130	6.6190	-4.5765	-10.9459
84.0	-4.6255	-5.6299	-4.7481	-5.8993	-9.1331
87.0	-0.0444	-6.4629	-5.6055	-4.0131	-6.75[illegible]
89.0	-2.7635	-3.9719	-6.2708	-6.5655	-5.2392
93.0	-1.8498	-2.1445	-2.7635	-0.8184	-7.9508
96.0	-1.9677	-2.5867	-0.9951	-1.4667	-8.0391
100.0	-0.5531	-1.0540	-1.2604	-1.5846	-0.6120
96.0	0.5964	0.7732	0.6259	0.5375	-0.0815
84.0	0.6849	0.8912	0.8617	0.6849	-0.1404
73.5	0.5963	-0.5041	0.7339	0.5963	-0.0915
54.0	0.5963	0.5963	0.5963	0.3212	-0.0915
33.0	0.4588	0.5963	0.4588	0.4588	0.0461
24.0	0.3906	0.4381	0.4381	0.4857	0.1052
10.0	0.5809	0.5333	0.5333	0.6284	0.3212
5.0	0.7236	0.5809	0.4857	0.4857	0.1837
2.5	0.6760	0.3430	0.1052	0.057[illegible]	-0.0915

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2223
43.4	-0.2585
50.4	-0.2585
56.1	-0.4034
62.5	-0.5120
69.7	-0.6569
76.9	-0.6931
83.4	-0.5845
89.4	-0.6569
95.4	-0.5845
101.1	-0.5120
---BOTTOM---	
38.2	-0.0050
43.4	0.0312
50.4	-0.1137
56.1	0.1398
62.5	0.2847
69.7	0.3934
76.9	0.4296
83.4	0.4296
89.4	0.3934
95.4	0.1398
101.1	-0.8018

MISC.*

NO.	CP
1	-1.3295
2	0.4588
3	-5.4563
4	0.3212
5	-3.8056
6	-0.2290
7	-0.2288
8	-1.7909
9	-1.4667
10	0.8617
11	-0.2878
12	-0.7888

Force Data -
Use Run 311

TABLE A15. RUN 146, BW6V, DELF=45, CMU=3.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
146	1	CLAERO =	1.9718	CDAERO =	1.3476	DCLC =	0.0	DWLC =	2.5273	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.05		2107.51	2102.77	4.75	63.79	0.0	2.66	0.0	2.917	2.917	86.9998	0.397616E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		4.50	-0.11	-3.20	0.04	0.01	-0.02	2.28	-1.55	2.28	0.87	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.8718	-0.8201
2.5	-0.7684	-0.8201
5.0	-0.8201	-0.8201
10.0	-0.8201	-0.8201
15.0	-0.7684	-0.7684
25.0	-0.8201	-0.8201
35.0	-0.7684	-0.8201
50.0	-0.7684	
56.0	-0.7684	-0.7684
65.0	-0.8201	-0.7684
76.0	-0.7684	-0.8201
79.0	-0.7684	-0.7684
80.5	-0.8393	-0.7647
81.0	-0.7647	-0.8393
82.0	-0.8393	-0.8020
84.0	-0.7647	
87.0	-0.7647	
89.0	-0.6657	-0.7761
93.0	-0.7209	
96.0	-0.7761	-0.7209
100.0	-0.7761	-0.7761
96.0	-0.7761	-0.7761
84.0	-0.7761	-0.7761
73.0	-0.7761	-0.7761
50.0	-0.7761	-0.7761
35.0	-0.7167	-0.7167
25.0	-0.7684	-0.7684
10.0	-0.7684	-0.7684
5.0	-0.7684	-0.8718
2.5	-0.8201	-0.8201

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.3874	-1.1124	-3.1424	-4.2298	-3.5049
2.5	-0.8949	-1.4748	-2.0550	-3.0699	-3.0699
5.0	-0.9674	-1.2574	-1.7649	-2.1274	-3.0699
10.0	-0.9674	-1.1850	-1.5475	-1.6200	-2.5624
15.0	-0.8949	-1.1124	-1.4024	-1.5475	-1.9098
24.0	-0.9674	-1.1124	-1.4024	-1.2574	-1.6924
33.0	-1.1063	-1.1063	-1.3160	-1.5256	-1.5256
54.0	-1.3160	-1.5256	-1.9448	-3.8316	-1.9448
65.0	-2.1007	-2.4870	-2.5422	-2.7078	-2.9285
78.5	-6.9023	-7.5095	-8.1166	-8.6133	-9.6619
79.5	*********	-194.1544	-176.2718	-177.3167	-99.8863
80.5	4.9101	-15.6237	-15.9970	-26.5994	0.1314
81.3	-1.4740	212.2559	-5.6180	-7.8581	-19.3944
82.0	6.8888	-8.7167	9.6516	-9.1275	-9.1648
84.0	-7.3355	-6.9994	-7.2607	-6.1781	-6.3275
87.0	2.6327	-8.6049	-6.5141	-3.3781	-4.3488
89.0	-3.0524	-4.7592	-7.9034	-7.6786	-2.8278
93.0	-1.6600	-2.2888	-2.6931	-0.9413	-6.7804
96.0	-2.0642	-3.2770	-0.2676	-0.6719	-8.0380
100.0	-2.4236	-2.4236	-2.6482	-2.7379	-0.6719
96.0	0.4960	0.5409	0.4061	0.4061	-0.2226
84.0	0.6757	0.7655	0.7655	0.6307	-0.0430
73.5	0.5708	-0.8967	0.5708	0.5708	0.1515
54.0	0.3611	0.3611	0.3611	0.3611	-0.0581
33.0	0.1515	0.1515	0.1515	0.1515	-0.0581
24.0	0.1200	0.0476	0.1200	0.3376	-0.2424
10.0	0.1926	0.1200	0.2651	0.3376	0.1515
5.0	0.1926	0.1926	0.2651	0.3376	0.3611
2.5	0.3376	0.2651	0.4826	0.4826	0.3611

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2242
43.4	-0.2242
50.4	-0.2794
56.1	-0.2794
62.5	-0.3345
69.7	-0.4449
76.9	-0.5002
83.4	-0.5553
89.4	-0.7209
95.4	-0.6105
101.1	-0.5553
---BOTTOM---	
38.2	-0.1690
43.4	-0.1138
50.4	-0.2242
56.1	-0.0034
62.5	0.0518
69.7	0.1070
76.9	0.1622
83.4	0.2725
89.4	0.2173
95.4	-0.0586
101.1	-0.9417

MISC.*

NO.	CP
1	-1.1063
2	0.1515
3	-1.5256
4	0.1515
5	-1.5256
6	-0.2678
7	-0.0430
8	-1.4354
9	-1.0312
10	0.9003
11	-0.2226
12	-0.4472

TABLE A15. (Continued)

RUN:SEQ 146 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 278

CLAERO = 2.0941 CDAERO = 1.5607 DCLC = 0.0 DWLC = 2.6791 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.01	2107.51	2102.65	4.86	64.52	0.0	2.66	0.0	2.980	2.980	87.0072	0.402709E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	4.77	0.25	-3.27	0.04	0.01	-0.02	2.60	-1.66	2.53	1.06

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.7804	-0.7298
2.5	-0.7804	-0.7298
5.0	-0.7298	-0.6794
10.0	-0.7804	-0.7298
15.0	-0.7298	-0.7298
25.0	-0.7804	-0.7804
35.0	-0.7298	-0.7804
50.0	-0.7804	
56.0	-0.7804	-0.7804
65.0	-0.7804	-0.7298
76.0	-0.7804	-0.8309
79.0	-0.7804	-0.7804
80.5	-0.7262	-0.7992
81.0	-0.6533	-0.8356
82.0	-0.7992	-0.7627
84.0	-0.7262	
87.0	-0.6898	
89.0	-0.6835	-0.7913
93.0	-0.7374	
96.0	-0.7374	-0.7374
100.0	-0.7374	-0.7913
96.0	-0.7374	-0.7913
84.0	-0.7374	-0.7913
73.0	-0.7374	-0.7913
50.0	-0.7913	-0.7913
35.0	-0.7298	-0.7804
25.0	-0.7804	-0.7804
10.0	-0.7804	-0.7298
5.0	-0.7298	-0.7804
2.5	-0.7804	-0.7804

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.6324	-4.1106	-5.9516	-4.1106	-2.9070
2.5	-0.7826	-2.1280	-4.6772	-3.7567	-2.6944
5.0	-1.2783	-1.7032	-4.0398	-3.7567	-2.6944
10.0	-1.2783	-1.4908	-1.7739	-3.8982	-2.7652
15.0	-1.2075	-1.4199	-1.4908	-3.5442	-2.6236
24.0	-1.0659	-1.2075	-1.4199	-1.4908	-2.5529
33.0	-1.0599	-1.0599	-1.4694	-1.2647	-2.2885
54.0	-1.4694	-1.4694	-1.8789	-3.7217	-2.9027
65.0	-2.1928	-2.5162	-2.6241	-2.5702	-3.9178
78.5	-6.8826	-7.5294	-8.1763	-8.4997	-10.9793
79.5	*********	-189.8630	-177.6109	-174.6214	-103.6083
80.5	-10.3527	-15.6035	-18.8487	-27.0163	0.1854
81.3	1.4616	212.6985	-4.8101	-7.9085	-20.3440
82.0	8.8637	-8.4201	9.5566	-8.6388	-10.4985
84.0	-7.1439	-6.8157	-7.3262	-6.3417	-7.3990
87.0	2.3003	-8.4565	-6.7062	-3.5339	-5.3936
89.0	-3.1362	-4.7593	-7.9179	-7.4790	-3.7063
93.0	-1.6883	-2.3465	-2.8728	-0.9426	-8.0932
96.0	-1.9655	-3.3117	-0.4601	-0.8549	-9.3655
100.0	-2.1710	-2.2587	-2.4780	-2.6535	-0.7233
96.0	0.5050	0.5928	0.4173	0.4173	-0.2846
84.0	0.6805	0.7682	0.7682	0.5928	-0.1969
73.5	0.7828	-0.6504	0.7828	0.5781	-0.0362
54.0	0.5781	0.5781	0.3733	0.3733	-0.2410
33.0	0.3733	0.3733	0.3733	0.1686	-0.0362
24.0	0.2795	0.2795	0.2795	0.3503	-0.1454
10.0	0.4211	0.3503	0.4211	0.4919	0.3733
5.0	0.4919	0.4211	0.4919	0.4919	0.5781
2.5	0.6335	0.4211	0.4211	0.4211	0.3733

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2523
43.4	-0.2523
50.4	-0.2523
56.1	-0.3062
62.5	-0.3601
69.7	-0.5757
76.9	-0.6296
83.4	-0.6835
89.4	-0.7913
95.4	-0.7374
101.1	-0.6296
---BOTTOM---	
38.2	-0.0906
43.4	-0.0906
50.4	-0.1984
56.1	0.0173
62.5	0.0712
69.7	0.1790
76.9	0.2329
83.4	0.2868
89.4	0.2868
95.4	-0.0366
101.1	-1.0069

MISC.

NO.	CP
1	-1.2647
2	0.3733
3	-2.4932
4	0.1686
5	-2.2885
6	-0.4456
7	-0.0653
8	-1.5130
9	-1.1620
10	0.8560
11	-0.3285
12	-0.5917

TABLE A15. (Concluded)

RUN:SEQ 146 5 PROPLULSIVE WING / CANARD PRESSURES PAGE 280

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
12.04	2107.51	2102.77	4.75	63.76	0.0	2.66	0.0	2.979	2.979

CLAERO = 2.6435 CDAERO = 1.9917 DCLC = 0.0 DWLC = 2.8341 BASEPR = 8.1904

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	5.48	1.07	-3.53	0.05	-0.01	0.01	3.35	-1.87	3.09	1.57

HQT	RN
86.9569	0.398030E+06

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.8201	-0.7684
2.5	-0.8201	-0.7684
5.0	-0.8201	-0.7684
10.0	-0.8201	-0.7684
15.0	-0.8201	-0.7167
25.0	-0.8201	-0.7684
35.0	-0.7167	-0.7167
50.0	-0.7167	
56.0	-0.7167	-0.7684
65.0	-0.7167	-0.7167
76.0	-0.7684	-0.7684
79.0	-0.7167	-0.7684
80.5	-0.8393	-0.8020
81.0	-0.7273	-0.7647
82.0	-0.7647	-0.8393
84.0	-0.7273	
87.0	-0.8393	
89.0	-0.7209	-0.7209
93.0	-0.7209	
96.0	-0.7761	-0.7761
100.0	-0.7761	-0.7761
96.0	-0.7209	-0.7209
84.0	-0.7761	-0.7761
73.0	-0.7209	-0.7209
50.0	-0.7761	-0.7209
35.0	-0.7167	-0.7167
25.0	-0.7684	-0.7167
10.0	-0.7167	-0.7167
5.0	-0.6718	-0.7167
2.5	-0.8718	-0.7167

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-7.0572	-8.5798	-5.8248	-4.5198	-3.7223
2.5	-0.8224	-8.5798	-6.1874	-4.5198	-3.7223
5.0	-2.5624	-4.5198	-6.3323	-4.5923	-3.7223
10.0	-2.1274	-2.0550	-8.0723	-5.3174	-3.7948
15.0	-1.9098	-1.6924	-5.9698	-6.2598	-4.0848
24.0	-1.6924	-1.5475	-1.7649	-5.8248	-4.2298
33.0	-1.5256	-1.5256	-1.3160	-4.6703	-4.8799
54.0	-1.7352	-1.7352	-1.9448	-3.6220	-1.1063
65.0	-2.4318	-2.8182	-2.6526	-1.9903	-7.3991
78.5	-7.2335	-8.1166	-8.7237	-8.6685	-11.2073
79.5	*********	-194.9759	-186.0525	-180.1163	-103.9561
80.5	-14.0556	-16.1463	-25.4429	-31.6762	0.1687
81.3	3.7901	220.9559	-4.0501	-7.0740	-22.0451
82.0	10.8462	-7.9329	9.6888	-8.8861	-13.1969
84.0	-7.1861	-6.9994	-7.7835	-6.8128	-9.9862
87.0	2.3714	-8.6421	-7.1116	-4.0874	-7.4475
89.0	-3.2321	-5.0737	-8.4422	-8.2177	-5.2982
93.0	-1.7497	-2.5582	-3.2770	-0.8964	-9.8347
96.0	-2.1540	-3.5464	-0.7617	-1.1658	-10.6432
100.0	-1.5702	-1.9743	-2.2439	-2.4685	-0.7168
96.0	0.5409	0.6757	0.5409	0.5409	-0.1328
84.0	0.6307	0.9451	0.9451	0.7206	-0.2226
73.5	0.5708	-0.8967	0.5708	0.5703	-0.2678
54.0	0.5708	0.5708	0.3611	0.3611	-0.2678
33.0	0.3611	0.5708	0.3611	0.3611	-0.0581
24.0	0.4100	0.6275	0.6275	0.7001	0.1926
10.0	0.6275	0.8450	0.7001	0.7725	0.1515
5.0	0.6275	0.7725	0.5551	0.6275	0.1515
2.5	0.6275	0.4100	0.3376	0.2651	-0.2678

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1690
43.4	-0.2242
50.4	-0.2794
56.1	-0.3897
62.5	-0.5553
69.7	-0.6657
76.9	-0.7209
83.4	-0.6105
89.4	-0.6657
95.4	-0.6105
101.1	-0.5002
---BOTTOM---	
38.2	0.0518
43.4	0.0518
50.4	-0.0586
56.1	0.1622
62.5	0.2725
69.7	0.4381
76.9	0.4933
83.4	0.4933
89.4	0.4933
95.4	0.1070
101.1	-0.9417

MISC.*

NO.	CP
1	-1.5256
2	0.3611
3	-6.1377
4	0.1515
5	-4.2506
6	-0.4774
7	-0.2676
8	-1.9743
9	-1.6151
10	0.9003
11	-0.3574
12	-0.8066

TABLE A16. RUN 147, BW6V, DELF=45, CMU=4.0, (BN/B)=1

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 281

RUN	SEQ												
147	1	CLAERO =	1.9880	CDAERO =	1.6250	DCLC =	0.0	DWLC =	3.5098	BASEPR =	8.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
0.04		2107.37	2103.87	3.50	54.75	0.0	2.67	0.0	4.051	4.051	87.1901	0.342321E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.0		5.50	-0.40	-3.88	0.06	0.02	0.00	2.47	-1.63	2.47	0.93		

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.8930	-0.8930
2.5	-0.8930	-0.8930
5.0	-0.9630	-0.8930
10.0	-0.8930	-0.8930
15.0	-0.8930	-0.9630
25.0	-0.9630	-0.8930
35.0	-0.8930	-0.8930
50.0	-0.8930	
56.0	-0.8930	-0.8930
65.0	-0.8930	-0.9630
76.0	-0.9630	-0.3930
79.0	-0.9630	-0.9630
80.5	-0.8220	-0.7715
81.0	-0.9232	-0.7715
82.0	-0.8726	-0.8220
84.0	-0.8726	
87.0	-0.7715	
89.0	-0.6461	-0.7209
93.0	-0.7957	
96.0	-0.7209	-0.7957
100.0	-0.7209	-0.7209
96.0	-0.7209	-0.7957
84.0	-0.7209	-0.7209
73.0	-0.7209	-0.7957
50.0	-0.7209	-0.7209
35.0	-0.8930	-0.9630
25.0	-0.8930	-0.8930
10.0	-0.8930	-0.8930
5.0	-0.8930	-0.8930
2.5	-0.8930	-0.8930

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.3763	-1.3585	-4.0104	-4.4034	-4.8943
2.5	-0.7691	-1.2602	-2.2424	-2.9299	-3.6174
5.0	-0.8673	-1.1620	-1.9479	-2.1442	-3.1265
10.0	-0.8673	-1.1620	-1.6532	-1.5549	-2.1442
15.0	-0.8673	-1.0638	-1.4567	-1.4567	-2.4390
24.0	-0.8673	-1.0638	-1.3585	-1.2602	-1.7513
33.0	-0.9892	-1.2733	-1.5573	-1.5573	-1.8413
54.0	-1.5573	-1.5573	-1.8413	-5.5336	2.9870
65.0	-2.2164	-2.6651	-2.6651	-2.8894	-3.0390
78.5	-7.6001	-8.2731	-8.8713	-9.4695	-10.8902
79.5	*********	-248.6658	-227.8775	-235.4120	-125.3004
80.5	19.2077	-20.0425	-17.0581	-30.0580	0.1896
81.3	-3.7556	328.7192	-6.3857	-9.8253	-24.3417
82.0	9.1929	-9.8759	14.8577	-10.9887	-10.4829
84.0	-8.6621	-7.9032	-8.4089	-6.9423	-6.8918
87.0	4.6913	-10.7358	-7.8527	-3.1994	-4.5649
89.0	-3.1999	-5.6948	-9.6502	-9.1027	-2.2264
93.0	-1.6178	-2.5307	-3.1391	-1.1309	-7.3986
96.0	-2.1653	-3.9302	0.0861	-0.1573	-9.2851
100.0	-3.3215	-3.3825	-3.6868	-3.8083	-0.8875
96.0	0.4512	0.5121	0.3296	0.3296	-0.3399
84.0	0.5729	0.6947	0.7554	0.5121	-0.2182
73.5	0.4309	-0.7053	0.7149	0.4309	-0.1372
54.0	0.4309	0.1468	0.4309	0.1468	-0.1372
33.0	0.1468	0.1468	0.1468	0.1468	-0.1372
24.0	0.3113	0.3113	0.2131	0.4096	-0.0816
10.0	0.3113	0.3113	0.3113	0.4096	0.1468
5.0	0.4096	0.3113	0.4096	0.6050	0.4309
2.5	0.4096	0.4096	0.4096	0.5077	0.4309

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.2723
43.4	-0.1975
50.4	-0.1975
56.1	-0.2723
62.5	-0.2723
69.7	-0.4218
76.9	-0.4966
83.4	-0.5714
89.4	-0.7209
95.4	-0.5714
101.1	-0.4966
---BOTTOM---	
38.2	-0.0480
43.4	-0.0480
50.4	-0.2723
56.1	0.0268
62.5	0.1016
69.7	0.1016
76.9	0.2511
83.4	0.2511
89.4	0.3259
95.4	-0.1227
101.1	-1.0201

MISC.*

NO.	CP
1	-1.5573
2	0.1468
3	-1.8413
4	0.1468
5	-1.8413
6	-0.4212
7	-0.1573
8	-1.5569
9	-1.1918
10	0.8163
11	-0.2791
12	-0.5833

TABLE A16. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 147 3

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.01	2107.44	2103.82	3.62	55.62	0.0	2.66	0.0	3.929	3.929	87.0204	0.347961E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	5.69	0.03	-3.88	0.06	0.01	-0.00	2.74	-1.70	2.67	1.11

CLAERO = 2.1577 CDAERO = 1.7551 DCLC = 0.0 DWLC = 3.5317 BASEPR = 8.1904

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.7696	-0.8374
2.5	-0.8374	-0.8374
5.0	-0.8374	-0.8374
10.0	-0.8374	-0.8374
15.0	-0.8374	-0.8374
25.0	-0.8374	-0.8374
35.0	-0.8374	-0.8374
50.0	-0.9053	
56.0	-0.8374	-0.8374
65.0	-0.8374	-0.8374
76.0	-0.8374	-0.8374
79.0	-0.8374	-0.8374
80.5	-0.7687	-0.8667
81.0	-0.8667	-0.7197
82.0	-0.8177	-0.7687
84.0	-0.8177	
87.0	-0.8177	
89.0	-0.5983	-0.7432
93.0	-0.7432	
96.0	-0.7432	-0.6708
100.0	-0.6708	-0.6708
96.0	-0.7432	-0.7432
84.0	-0.6708	-0.6708
73.0	-0.7432	-0.7432
50.0	-0.6708	-0.6708
35.0	-0.7696	-0.8374
25.0	-0.8374	-0.7696
10.0	-0.9053	-0.8374
5.0	-0.8374	-0.8374
2.5	-0.8374	-0.8374

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.6689	-3.9523	-4.6183	-4.0474	-2.9056
2.5	-0.7174	-2.0495	-4.5232	-3.3814	-2.7154
5.0	-1.2883	-1.5738	-4.0474	-3.4765	-2.5252
10.0	-1.1932	-1.3834	-2.9056	-3.6670	-2.5252
15.0	-1.0980	-1.1932	-1.5738	-3.4765	-2.7154
24.0	-1.0028	-1.1932	-1.2883	-1.8593	-2.5252
33.0	-1.2058	-1.4809	-1.4809	-1.4809	-2.3063
54.0	-1.7560	-1.7560	-2.0312	-5.3327	-2.8564
65.0	-2.1918	-2.6988	-2.6988	-2.6988	-4.0750
78.5	-7.4069	-8.1312	-8.7831	-9.2177	-11.6080
79.5	*********	-242.1238	-219.3879	-229.0888	-125.0222
80.5	19.9565	-19.9752	-18.4071	-29.7247	0.2602
81.3	-4.6882	311.3733	-6.5013	-9.9799	-24.6296
82.0	8.7855	-9.6860	14.0771	-11.0578	-11.6458
84.0	-8.4610	-7.7751	-8.5100	-6.9422	-7.5791
87.0	4.3267	-10.5190	-7.6770	-3.4146	-5.0313
89.0	-3.1309	-5.5478	-9.6152	-8.9667	-2.8362
93.0	-1.4216	-2.4826	-3.1309	-1.0090	-8.5539
96.0	-2.0699	-3.7795	-0.0658	-0.3016	-10.7352
100.0	-2.9543	-3.1309	-3.3668	-3.5437	-0.7731
96.0	0.4647	0.5826	0.3469	0.3469	-0.4795
84.0	0.7005	0.8184	0.8184	0.6415	-0.3016
73.5	0.4450	-0.9306	0.7201	0.7201	-0.1053
54.0	0.4450	0.4450	0.4450	0.4450	-0.1053
33.0	0.1698	0.1698	0.4450	0.4450	-0.1053
24.0	0.4243	0.5194	0.5194	0.6146	0.0437
10.0	0.5194	0.6146	0.5194	0.7097	0.1698
5.0	0.7097	0.6146	0.6146	0.7097	0.1698
2.5	0.7097	0.6146	0.6146	0.5194	0.1698

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1637
43.4	-0.2362
50.4	-0.1637
56.1	-0.2362
62.5	-0.3810
69.7	-0.4634
76.9	-0.5983
83.4	-0.5983
89.4	-0.7432
95.4	-0.5983
101.1	-0.5269
---BOTTOM---	
38.2	-0.0913
43.4	-0.0189
50.4	-0.1637
56.1	0.1260
62.5	0.1984
69.7	0.2709
76.9	0.3433
83.4	0.4157
89.4	0.2709
95.4	-0.0189
101.1	-1.0329

MISC.*

NO.	CP
1	-1.4809
2	0.4450
3	-2.5813
4	0.1698
5	-2.3063
6	-0.3804
7	-0.1248
8	-1.5395
9	-1.2447
10	0.8773
11	-0.2427
12	-0.5963

TABLE A16. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 285

147 5 CLAERO = 2.7187 CDAERO = 2.2061 DCLC = 0.0 DWLC = 3.7018 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.00	2107.30	2103.68	3.62	55.62	0.0	2.66	0.0	3.892	3.892	87.1712	0.348040E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	6.42	1.00	-4.15	0.07	0.00	-0.01	3.54	-1.96	3.27	1.65

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.9052	-0.9052	0.0	-6.9018	-9.5657	-5.7600	-4.6182	-3.8571	38.2	-0.3085	1	-1.7559
2.5	-0.8373	-0.9052	2.5	-0.8125	-9.5657	-6.2357	-4.9989	-3.9522	43.4	-0.3095	2	0.4451
5.0	-0.8373	-0.9052	5.0	-2.5252	-3.9522	-6.5211	-4.9038	-4.0474	50.4	-0.3809	3	-6.1578
10.0	-0.8373	-0.8373	10.0	-2.1445	-2.1445	-6.3288	-5.2842	-4.1427	56.1	-0.4534	4	0.1699
15.0	-0.9052	-0.9052	15.0	-1.9543	-1.9543	-6.5211	-6.6165	-4.3329	62.5	-0.5983	5	-4.5071
25.0	-0.9052	-0.8373	24.0	-1.5737	-1.7640	-1.6688	-6.1406	-4.7133	69.7	-0.7431	6	-0.3802
35.0	-0.9052	-0.9052	33.0	-1.4808	-1.7559	-1.4808	-3.9569	-5.0575	76.9	-0.7431	7	-0.3605
50.0	-0.9052		54.0	-1.7559	-2.0311	-2.0311	-5.0575	-6.9833	83.4	-0.6707	8	-2.0698
56.0	-0.9052	-0.9052	65.0	-2.5539	-2.9161	-2.7712	-2.2642	-6.7550	89.4	-0.7431	9	-1.7751
66.0	-0.8731	-0.9052	78.5	-7.8414	-8.7106	-9.3625	-9.5798	-11.8976	95.4	-0.7431	10	0.8774
76.0	-0.9052	-0.8373	79.5	*********	-245.4550	-227.0817	-232.3730	-125.1188	101.1	-0.5983	11	-0.4194
79.0	-0.8373	-0.8373	80.5	16.3801	-20.1712	-20.6121	-32.4698	0.1133	---BOTTOM---		12	-0.8320
80.5	-0.9646	-0.9156	81.3	-3.7084	299.0435	-5.9620	-9.9309	-26.2465	38.2	-0.0613		
81.0	-0.8666	-1.1606	82.0	8.4917	-10.0288	13.4402	-11.4008	-14.2425	43.4	-0.0188		
82.0	-0.8666	-1.0136	84.0	-8.5099	-8.3631	-9.0490	-7.8240	-10.6657	50.4	-0.1637		
84.0	-0.9156		87.0	4.2779	-10.7639	-8.5099	-4.2472	-7.0892	56.1	0.1261		
87.0	-0.9646		89.0	-7.3078	-5.7835	-10.0866	-9.8507	-5.0763	62.5	0.1984		
89.0	-0.7431	-0.8155	93.0	-1.5983	-2.7773	-3.4847	-1.1857	-11.0298	69.7	0.3433		
93.0	-0.8155		96.0	-2.1879	-4.0742	-0.2426	-0.7731	-12.4444	76.9	0.4158		
96.0	-0.8880	-0.8880	100.0	-2.2468	-2.7184	-3.1309	-3.4256	-0.8320	83.4	0.4158		
100.0	-0.8155	-0.8155	96.0	0.5237	0.6416	0.3469	0.4058	-0.3016	89.4	0.4158		
96.0	-0.8880	-0.8880	84.0	0.6416	0.8774	0.8184	0.5827	-0.3605	95.4	-0.0913		
84.0	-0.8155	-0.8155	73.5	0.7202	-0.9305	0.7202	0.4451	-0.3803	101.1	-1.1777		
73.0	-0.8880	-0.8880	54.0	0.4451	0.4451	0.4451	0.4451	-0.1052				
50.0	-0.8880	-0.8155	33.0	0.4451	0.4451	0.4451	0.4451	-0.1052				
35.0	-0.8373	-0.9052	24.0	0.4244	0.5195	0.5195	0.6147	0.0438				
25.0	-0.8373	-0.9052	10.0	0.8049	0.7098	0.6147	0.6147	0.1699				
10.0	-0.8373	-0.8373	5.0	0.6147	0.5195	0.4244	0.5195	0.1699				
5.0	-0.8373	-0.9052	2.5	0.1390	0.3292	0.1390	0.0438	-0.1052				
2.5	-0.9052	-0.9052										

TABLE A17. RUN 148, BW6V, DELF=45, CMU=2.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
148	1	CLAERO =	1.7538	CDAERO =	1.1491	DCLC =	0.0	DWLC =	1.7430	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.05		2107.16	2101.85	5.31	67.40	0.0	2.66	0.0	2.012	2.012	86.9592	0.422128E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		3.50	0.14	-2.42	0.04	0.01	0.01	2.08	-1.37	2.08	0.81	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6009	-0.5548
2.5	-0.5548	-0.5548
5.0	-0.6009	-0.6009
10.0	-0.5548	-0.6009
15.0	-0.5548	-0.6009
25.0	-0.5548	-0.5548
35.0	-0.6009	-0.6009
50.0	-0.5548	
56.0	-0.5548	-0.6009
65.0	-0.6009	-0.6009
76.0	-0.6009	-0.6009
79.0	-0.6009	-0.6472
80.5	-0.5514	-0.5514
81.0	-0.6181	-0.5514
82.0	-0.5848	-0.4847
84.0	-0.5514	
87.0	-0.5514	
89.0	-0.5123	-0.6109
93.0	-0.6109	
96.0	-0.6109	-0.6109
100.0	-0.6109	-0.6109
96.0	-0.6109	-0.6109
84.0	-0.6109	-0.6109
73.0	-0.6109	-0.6109
50.0	-0.6109	-0.6109
35.0	-0.5548	-0.6009
25.0	-0.5548	-0.6472
10.0	-0.6009	-0.6009
3.0	-0.6009	-0.5548
2.5	-0.5548	-0.6009

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0201	-0.9917	-2.3518	-3.1938	-3.1938
2.5	-0.6030	-1.1859	-1.6393	-2.2223	-3.4529
5.0	-0.6678	-0.9269	-1.3803	-1.5745	-2.3518
10.0	-0.6678	-0.8621	-1.2508	-1.3156	-2.0279
15.0	-0.7326	-0.8621	-1.0564	-1.1212	-1.5745
24.0	-0.6678	-0.7973	-1.0564	-0.9917	-1.5098
33.0	-0.8566	-0.8566	-1.2312	-1.2312	-1.2312
54.0	-1.2312	-1.2312	-1.6058	-2.3550	-1.7932
65.0	-1.7450	-2.0409	-2.1395	-2.3367	-2.4846
78.5	-5.9362	-6.4293	6.9717	-7.5634	-8.4017
79.5	*********	-102.3826	-79.5013	-67.9270	-77.4330
80.5	-11.3916	-11.6585	-17.8282	-26.8355	0.0156
81.3	-6.2551	231.4670	-8.3230	-11.6250	-15.9944
82.0	0.2824	-6.4884	3.1843	-10.0574	-9.6238
84.0	-5.7881	-5.4211	-6.4218	-6.1216	-6.4218
87.0	0.7494	-5.9882	-5.5212	-3.3199	-4.3539
89.0	-2.5150	-3.4379	-5.9259	-5.8054	-3.3176
93.0	-1.3112	-1.7926	-2.3947	-0.7092	-5.8455
96.0	-1.5920	-2.3947	-0.7895	-1.1507	-6.2871
100.0	-1.6321	-1.5519	-1.6321	-1.7926	-0.5889
96.0	0.5749	0.6551	0.5749	0.5347	0.0130
84.0	0.6952	0.8156	0.8156	0.6551	0.0532
73.5	0.5417	-0.6694	0.6417	0.6417	0.0798
54.0	0.2671	0.2671	0.2671	0.2671	-0.1075
33.0	0.2671	0.2671	0.2671	0.2671	-0.1075
24.0	0.2390	0.2390	0.2390	0.4333	0.1094
10.0	0.3038	0.2390	0.4333	0.4981	0.0798
5.0	0.3685	0.3685	0.4981	0.6276	0.2671
2.5	0.4333	0.4333	0.6276	0.6276	0.4544

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1672
43.4	-0.1672
50.4	-0.1672
56.1	-0.2165
62.5	-0.2658
69.7	-0.4137
76.9	-0.4630
83.4	-0.4630
89.4	-0.6109
95.4	-0.5616
101.1	-0.4630
---BOTTOM---	
38.2	-0.0686
43.4	-0.0193
50.4	-0.1672
56.1	0.0301
62.5	0.0794
69.7	0.1780
76.9	0.1780
83.4	0.2766
89.4	0.3259
95.4	0.0301
101.1	-0.7589

MISC.*

NO.	CP
1	-1.0440
2	0.2671
3	-1.2312
4	0.2671
5	-1.2312
6	-0.2948
7	0.0933
8	-1.2309
9	-0.8296
10	0.9761
11	-0.1876
12	-0.3481

TABLE A17. (Continued)

RUN:SEQ 148 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 288

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	DWLC = 1.7088	CMUT	BASEPR = 8.1904	RN
4.01	2106.95	2101.63	8.31	67.38	0.0	2.66	0.0	1.901	1.901	HGT 86.9275	0.422321E+06

CLAERO = 2.0914 CDAERO = 1.2509 DCLC = 0.0

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	3.80	0.42	-2.53	0.04	0.01	0.01	2.41	-1.48	2.34	0.98

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.7124	-0.6662
2.5	-0.6662	-0.6662
5.0	-0.6662	-0.6662
10.0	-0.7124	-0.6662
15.0	-0.6662	-0.6662
25.0	-0.6662	-0.6662
35.0	-0.6662	-0.6662
50.0	-0.6662	
56.0	-0.6662	-0.6662
65.0	-0.6662	-0.6662
76.0	-0.6662	-0.6662
79.0	-0.6662	-0.6662
80.5	-0.7705	-0.6371
81.0	-0.7038	-0.6037
82.0	-0.6371	-0.7372
84.0	-0.6705	
87.0	-0.7038	
89.0	-0.5313	-0.6299
93.0	-0.6299	
96.0	-0.5806	-0.6299
100.0	-0.5806	-0.6299
96.0	-0.6299	-0.6299
84.0	-0.5806	-0.6299
73.0	-0.6299	-0.6299
50.0	-0.5806	-0.5806
35.0	-0.6662	-0.6662
25.0	-0.6662	-0.7124
10.0	-0.6662	-0.7124
5.0	-0.6662	-0.6199
2.5	-0.6662	-0.6662

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.2699	-3.4724	-4.7033	-4.1851	-2.6303
2.5	-0.6868	-1.7881	-4.1851	-3.5372	-2.5655
5.0	-1.2050	-1.4642	-3.5372	-3.3429	-2.3710
10.0	-1.1402	-1.2699	-1.4642	-3.4077	-2.5655
15.0	-1.0107	-1.2050	-1.3347	-2.7598	-2.4358
24.0	-1.0755	-1.1402	-1.3347	-1.1402	-2.3063
33.0	-0.8756	-1.0630	-1.2503	-1.2503	-2.1870
54.0	-1.2502	-1.2503	-1.6249	-2.5616	-2.3743
65.0	-1.8628	-2.1587	-2.3067	-2.3067	-3.1451
78.5	-6.1535	-6.6466	-7.4357	-7.5837	-9.2112
79.5	*********	-104.2221	-79.7026	-68.9261	-77.4325
80.5	-11.5460	-11.9797	-18.2852	-27.0928	-0.0366
81.3	-6.4419	224.0753	-8.7102	-12.0131	-16.8170
82.0	0.0301	-6.6086	2.8991	-10.3116	-10.2783
84.0	-5.9746	-5.0078	-6.6420	-6.4085	-7.1425
87.0	0.5306	-6.2082	-5.8080	-3.6062	-5.0406
89.0	-2.5744	-3.4975	-6.1866	-6.1064	-4.1798
93.0	-1.4506	-1.8519	-2.6147	-0.7683	-7.1495
96.0	-1.6512	-2.4541	-0.9288	-1.2900	-7.5513
100.0	-1.4909	-1.5310	-1.6112	-1.8519	-0.5677
96.0	0.5963	0.6765	0.5562	0.4759	-0.1262
84.0	0.7167	0.8371	0.7970	0.6364	-0.0459
73.5	0.6230	-0.6883	0.6230	0.6230	0.0610
54.0	0.4356	0.4356	0.4356	0.2483	-0.1264
33.0	0.2483	0.2483	0.2483	0.2483	0.0610
24.0	0.2850	0.2850	0.2202	0.4146	-0.0389
10.0	0.4146	0.3498	0.3498	0.5441	0.2483
5.0	0.5441	0.4146	0.4793	0.5441	0.4356
2.5	0.5441	0.4793	0.5441	0.4793	0.4356

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1367
43.4	-0.1367
50.4	-0.1860
56.1	-0.2353
62.5	-0.3340
69.7	-0.3833
76.9	-0.4819
83.4	-0.5313
89.4	-0.6299
95.4	-0.5806
101.1	-0.4326
---BOTTOM---	
38.2	-0.0381
43.4	0.0113
50.4	-0.1367
56.1	0.0606
62.5	0.1592
69.7	0.2085
76.9	0.3071
83.4	0.3565
89.4	0.4058
95.4	0.1099
101.1	-0.7285

MISC.*

NO.	CP
1	-1.2503
2	0.2483
3	-2.1870
4	0.2483
5	-1.9997
6	-0.3136
7	-0.0057
8	-1.3303
9	-0.9288
10	0.8772
11	-0.1262
12	-0.4473

TABLE A17. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
148 5	CLAERO =	2.5817	CDAERO =	1.7315	DCLC =	0.0	DMLC =	1.9531	BASEPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.02	2107.02	2101.70	5.31	67.38	0.0	2.67	0.0	2.053	2.053	87.1829	0.422239E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	4.53	1.10	-2.83	0.06	-0.00	-0.00	3.17	-1.72	2.93	1.47	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6662	-0.7125
2.5	-0.6200	-0.7125
5.0	-0.6200	-0.7125
10.0	-0.6662	-0.7586
15.0	-0.7125	-0.7125
25.0	-0.7125	-0.7125
35.0	-0.7125	-0.7125
50.0	-0.7125	
56.0	-0.7125	-0.7586
65.0	-0.7586	-0.7586
76.0	-0.7125	-0.7125
79.0	-0.6662	-0.7125
80.5	-0.6038	-0.6038
81.0	-0.6705	-0.7039
82.0	-0.6040	-0.6038
84.0	-0.6705	
87.0	-0.7039	
89.0	-0.6703	-0.7286
93.0	-0.7286	
96.0	-0.7779	-0.7286
100.0	-0.7286	-0.7779
96.0	-0.7286	-0.7286
84.0	-0.7286	-0.7779
73.0	-0.7286	-0.7286
50.0	-0.7286	-0.7779
35.0	-0.7586	-0.7586
25.0	-0.7125	-0.7586
10.0	-0.7125	-0.7586
5.0	-0.6662	-0.7125
2.5	-0.7586	-0.7125

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-6.4538	-7.3606	-5.2872	-3.9266	-3.4730
2.5	-0.5573	-7.8791	-5.5434	-4.1209	-3.5378
5.0	-2.2419	-6.1843	-5.4169	-4.4449	-3.6026
10.0	-1.8532	-1.3996	-7.3606	-5.2224	-3.6026
15.0	-1.5939	-1.5939	-6.3240	-5.6112	-3.7970
24.0	-1.3996	-1.4644	-1.5939	-5.5484	-4.0561
33.0	-1.2506	-1.4378	-1.2506	-4.2484	-4.4357
54.0	-1.4378	-1.6252	-1.6252	-2.5620	-6.4967
65.0	-2.0605	-2.3564	-2.2578	-1.9124	-6.3519
78.5	-6.3519	-7.0918	-7.8317	-8.0793	-10.2980
79.5	*********	-106.9434	-73.7099	-74.7792	-78.9833
80.5	-12.2822	-12.6492	-19.0229	-29.1330	-0.1367
81.3	-6.7100	217.8481	-9.5794	-13.2164	-19.2883
82.0	0.1970	-7.0769	3.0232	-11.0811	-13.1498
84.0	-6.1427	-6.2762	-7.4106	-6.9101	-9.6463
87.0	0.7643	-6.7433	-6.4764	-4.2408	-7.1103
89.0	-2.5347	-3.8595	-6.9103	-7.0709	-8.9068
93.0	-1.4107	-2.1333	-3.0165	-0.8487	-8.5562
96.0	-1.6114	-2.5953	-1.1298	-1.6918	-8.3555
100.0	-0.9289	-1.3706	-1.5713	-1.8522	-0.6882
96.0	0.5161	0.7169	0.5563	0.4760	-0.1663
84.0	0.6366	0.8373	0.7972	0.5964	-0.1262
73.5	0.6231	-0.6884	0.6231	0.6231	-0.1264
54.0	0.4357	0.4357	0.4357	0.4357	-0.1264
33.0	0.4357	0.4357	0.4357	0.4357	0.0610
24.0	0.4795	0.4795	0.4795	0.6738	0.1555
10.0	0.6091	0.6091	0.6738	0.6738	0.2484
5.0	0.7387	0.6091	0.5443	0.5443	0.2484
2.5	0.7387	0.4147	0.2203	0.0907	-0.1264

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2847
43.4	-0.3340
50.4	-0.3833
56.1	-0.4820
62.5	-0.6299
69.7	-0.7286
76.9	-0.7779
83.4	-0.6793
89.4	-0.7286
95.4	-0.6793
101.1	-0.5806
---BOTTOM---	
38.2	-0.0874
43.4	-0.0381
50.4	-0.0874
56.1	0.0306
62.5	0.2086
69.7	0.3072
73.0	0.3586
83.4	0.3586
89.4	0.3072
95.4	0.0806
101.1	-0.8766

**MISC.

NO.	CP
1	-1.2506
2	0.4357
3	-5.7472
4	0.2484
5	-3.8735
6	-0.3137
7	-0.2486
8	-1.8926
9	-1.5713
10	0.7972
11	-0.4072
12	-0.8086

C-4

TABLE A18. RUN 149, BWGV, DELF=45, CMU=1.0, (BN/B)=1

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ										
149 1	CLAERO = 1.5773	CDAERO = 0.8964		DCLC = 0.0		DWLC = 0.9131		BASEPR = 8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.12	2106.52	2101.21	5.31	67.39	0.0	2.66	0.0	1.053	1.053	86.9863	0.422279E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	2.49	0.37	-1.66	0.03	0.01	0.02	1.91	-1.24	1.91	0.67	

************ CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.5274	-0.5274
2.5	-0.5274	-0.5274
5.0	-0.5737	-0.5737
10.0	-0.5274	-0.5274
15.0	-0.5737	-0.5274
25.0	-0.5274	-0.5737
35.0	-0.5274	-0.5274
50.0	-0.5737	
56.0	-0.5737	-0.5274
65.0	-0.5737	-0.5737
76.0	-0.5274	-0.5737
79.0	-0.5737	-0.5274
80.5	-0.6037	-0.5369
81.0	-0.5703	-0.6037
82.0	-0.6370	-0.6037
84.0	-0.5703	
87.0	-0.5369	
89.0	-0.4326	-0.5312
93.0	-0.5312	
96.0	-0.5312	-0.5312
100.0	-0.5312	-0.5312
96.0	-0.5312	-0.5312
84.0	-0.5312	-0.5312
73.0	-0.5312	-0.5312
50.0	-0.5312	-0.5312
35.0	-0.5737	-0.5274
25.0	-0.4812	-0.5274
10.0	-0.5274	-0.5274
5.0	-0.5737	-0.5274
2.5	-0.5274	-C.5737

************************ WING ******************

%X/C	BP = 2 CP	BP = 6 CP	BP = 12 CP	BP = 16 CP	BP = 22 CP
0.0	-0.0389	-0.2980	-2.2415	-2.9842	-3.3429
2.5	-0.4924	-1.0107	-1.5936	-1.9823	-3.0189
5.0	-0.5572	-0.7515	-1.1402	-1.3994	-1.9823
10.0	-0.6219	-0.7515	-1.0754	-1.0754	-1.5289
15.0	-0.6219	-0.6867	-0.9459	-0.8811	-1.2049
24.0	-0.6219	-0.7515	-0.8811	-0.8811	-1.0754
33.0	-0.8756	-0.8756	-1.2503	-1.3803	-1.2503
54.0	-1.2503	-1.2503	-1.4376	-2.3742	-1.6249
65.0	-1.5669	-1.7143	-1.8628	-2.0601	-2.1094
78.5	-5.0684	-5.4630	-6.0055	-6.4000	-6.9916
79.5	-62.4207	-58.9515	-49.4776	-44.5064	-41.4701
80.5	-10.0446	-10.1781	-14.5150	-17.7510	-0.2700
81.3	-5.8413	233.4118	-7.0090	-9.3442	-12.2799
82.0	-2.6054	-6.8087	-2.1049	-7.9430	-9.4442
84.0	-4.8405	-4.4734	-5.0741	-5.0741	-6.6754
87.0	-1.2708	-4.1399	-4.3735	-3.3726	-4.8072
89.0	-2.4139	-2.8553	-4.1798	-4.3403	-3.8186
93.0	-1.5710	-1.7316	-2.0928	-0.6479	-4.9423
96.0	-1.5710	-1.8118	-1.0893	-1.6915	-4.7417
100.0	-0.7683	-0.6479	-0.6881	-0.7683	-0.5676
96.0	0.5160	0.6766	0.5363	0.5562	0.0344
84.0	0.6364	0.7568	0.7568	0.6364	0.0745
73.5	0.6230	-0.6883	0.6230	0.6230	0.0610
54.0	0.2483	0.2483	0.2483	0.2483	-0.1263
33.0	0.0610	0.2483	0.0610	0.2483	-0.1263
24.0	0.2202	0.2202	0.2202	0.4794	0.0907
10.0	0.2202	0.2880	0.4146	0.4794	0.2483
5.0	0.3498	0.3498	0.4146	0.6090	0.2483
2.5	0.3498	0.3498	0.5442	0.6737	0.2483

FUSELAGE

-----TOP----

XLOC	CP
38.2	-0.0873
43.4	-0.1367
50.4	-0.1367
56.1	-0.1860
62.5	-0.1860
69.7	-0.2846
76.9	-0.3340
83.4	-0.3832
89.4	-0.4819
95.4	-0.4326
101.1	-0.3832

---BOTTOM---

XLOC	CP
38.2	-0.0381
43.4	-0.0381
50.4	-0.1367
56.1	0.0606
62.5	0.1099
69.7	0.1592
76.9	0.2579
83.4	0.2579
89.4	0.3072
95.4	0.1592
101.1	-0.5806

MISC.*

NO.	CP
1	-1.0630
2	0.2483
3	-1.2503
4	0.2483
5	-1.2503
6	-0.3136
7	0.1147
8	-1.0893
9	-0.6881
10	0.6773
11	-0.1261
12	-0.3670

TABLE A18. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
149	3	CLAERO =	1.7507	CDAERO =	1.0034	DCLC =	0.0	DWLC =	0.9353	BASEPR =	8.1904
ALPHA		PTOT		PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.04		2106.31		2100.88	5.43	68.09	0.0	2.66	0.0	1.040	1.040	87.0458	0.426984E+06
BETA		CL		CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
0.0		2.69		0.55	-1.74	0.03		0.01	0.02	2.16	-1.31	2.10	0.80

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4527	-0.4980
2.5	-0.4527	-0.4980
5.0	-0.4980	-0.4527
10.0	-0.4527	-0.4527
15.0	-0.4527	-0.4980
25.0	-0.4980	-0.4980
35.0	-0.4527	-0.4980
50.0	-0.4980	
56.0	-0.4527	-0.5432
65.0	-0.4980	-0.4980
76.0	-0.4980	-0.4980
79.0	-0.4980	-0.4980
80.5	-0.4746	-0.4420
81.0	-0.5073	-0.5073
82.0	-0.5073	-0.5073
84.0	-0.4746	
87.0	-0.4746	
89.0	-0.3568	-0.4051
93.0	-0.4534	
96.0	-0.4534	-0.4534
100.0	-0.4534	-0.4051
96.0	-0.4534	-0.4051
84.0	-0.4534	-0.4051
73.0	-0.4534	-0.4051
50.0	-0.4534	-0.4534
35.0	-0.4980	-0.4980
25.0	-0.4980	-0.4980
10.0	-0.4980	-0.4527
5.0	-0.4980	-0.4980
2.5	-0.4980	-0.4980

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8442	-2.4301	-4.5866	-4.0156	-2.3665
2.5	-0.5271	-1.3517	-2.7471	-3.0008	-2.1128
5.0	-1.0345	-1.1613	-1.7323	-3.6448	-2.1128
10.0	-0.8442	-1.0979	-1.4151	-2.8105	-2.1128
15.0	-0.7174	-0.9077	-1.2249	-1.0979	-1.9659
24.0	-0.7174	-0.9077	-1.0979	-0.8442	-1.9225
33.0	-0.8389	-1.0223	-1.2057	-1.2057	-1.9394
54.0	-1.2057	-1.2057	-1.5725	-1.9394	-1.9394
65.0	-1.5640	-1.7571	-1.9020	-1.9503	-2.5298
78.5	-4.9924	-5.4270	-6.0065	-6.6342	-7.2137
79.5	-61.7844	-58.2900	-49.2738	-44.5704	-41.6971
80.5	-9.8492	-10.0125	-14.4874	-17.6232	-0.1807
81.3	-5.6028	216.9450	-7.0074	-9.2939	-13.1453
82.0	-2.5978	-6.5174	-2.1406	-7.8241	-9.9471
84.0	-4.7210	-4.4270	-5.1130	-5.1782	-6.9420
87.0	-1.1605	-4.1004	-4.3814	-3.3816	-5.1782
89.0	-2.3057	-2.8558	-4.2313	-4.5064	-4.5849
93.0	-1.6377	-1.7555	-2.1879	-0.6159	-5.7245
96.0	-1.5590	-1.7948	-1.1661	-1.7555	-5.3708
100.0	-0.6945	-0.5766	-0.6945	-0.7731	-0.4980
96.0	0.5630	0.6809	0.6023	0.5237	-0.0265
84.0	0.6416	0.7595	0.7202	0.6416	0.0522
73.5	0.6285	-0.6555	0.6285	0.6285	-0.1052
54.0	0.2616	0.2316	0.4450	0.2616	-0.1052
33.0	0.2616	0.2616	0.2316	0.2616	-0.1052
24.0	0.4244	0.4244	0.3610	0.5512	0.1072
10.0	0.4244	0.4878	0.5512	0.6781	0.2616
5.0	0.5512	0.5512	0.6147	0.6781	0.2616
2.5	0.6147	0.6147	0.6781	0.6781	0.2616

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.0670	38.2	0.0298
43.4	-0.1154	43.4	0.0298
50.4	-0.1154	50.4	-0.1154
56.1	-0.1154	56.1	0.1261
62.5	-0.2119	62.5	0.1744
69.7	-0.3085	69.7	0.2710
76.9	-0.3568	76.9	0.3192
83.4	-0.3568	83.4	0.3675
89.4	-0.4534	89.4	0.4158
95.4	-0.4534	95.4	0.2226
101.1	-0.3568	101.1	-0.4534

MISC.

NO.	CP
1	-1.2057
2	0.2616
3	-1.5725
4	0.0782
5	-1.9394
6	-0.2886
7	0.0914
8	-1.1661
9	-0.7731
10	0.8774
11	-0.1443
12	-0.4196

TABLE A18. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 255

149 5	CLAERO =	2.3183	CDAERO =	1.4012	DCLC =	0.0	DWLC =	1.0406	BASIPR =	8.1904	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.03	2106.24	2100.93	5.31	67.37	0.0	2.66	0.0	1.094	1.094	87.1279	0.422648E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	3.36	1.06	-2.00	0.05	-0.00	-0.00	2.87	-1.52	2.87	1.25	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.6199	-0.5737
2.5	-0.5737	-0.5737
5.0	-0.5737	-0.5737
10.0	-0.5737	-0.5737
15.0	-0.5737	-0.5737
25.0	-0.5737	-0.5737
35.0	-0.6199	-0.5274
50.0	-0.5737	
56.0	-0.5737	-0.5737
65.0	-0.5737	-0.5737
76.0	-0.5737	-0.5737
79.0	-0.5737	-0.6199
80.5	-0.6370	-0.6370
81.0	-0.6037	-0.6704
82.0	-0.6370	-0.5703
84.0	-0.5703	
87.0	-0.6370	
89.0	-0.5312	-0.5805
93.0	-0.5805	
96.0	-0.5805	-0.6298
100.0	-0.5805	-0.5805
96.0	-0.5805	-0.5805
84.0	-0.5805	-0.5805
73.0	-0.5805	-0.5805
50.0	-0.6298	-0.5805
35.0	-0.5737	-0.6199
25.0	-0.5737	-0.5737
10.0	-0.5737	-0.5737
5.0	-0.5737	-0.5737
2.5	-0.5737	-0.5274

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-5.7397	-6.5171	-4.7033	-3.6666	-3.083[illegible]
2.5	-0.4924	-7.1002	-4.8975	-3.7316	-3.0159
5.0	-1.9823	-5.9341	-5.3510	-4.1201	-3.2779
10.0	-1.6584	-1.2049	-6.3876	-4.1201	-3.0837
15.0	-1.3994	-1.3994	-5.2215	-5.0920	-3.2132
24.0	-1.2699	-1.2049	-1.5936	-4.7033	-3.3429
33.0	-1.0630	-1.2503	-0.8756	-3.4982	-4.0502
54.0	-1.2503	-1.4375	-1.2503	-2.1869	-5.3715
65.0	-1.8628	-2.0601	-2.1094	-1.7148	-5.3643
78.5	-5.5123	-6.1534	-6.8439	-7.2384	-8.3234
79.5	-71.5953	-68.2259	-56.8828	-47.0758	-41.8033
80.5	-10.6786	-11.3124	-16.5167	-19.1850	-0.2701
81.3	-6.1415	213.0636	-8.1097	-10.6118	-14.7819
82.0	-2.4719	-7.0756	-2.7388	-8.8104	-11.4792
84.0	-5.1407	-5.1407	-5.9413	-5.9413	-8.1431
87.0	-1.1375	-4.6737	-5.0073	-4.1066	-5.9748
89.0	-2.4941	-3.2166	-4.9826	-5.3839	-5.1833
93.0	-1.6111	-1.9723	-2.5743	-0.6479	-6.0663
96.0	-1.6111	-2.0928	-1.4506	-2.0527	-5.3839
100.0	-0.2867	-0.5275	-0.7282	-0.8486	-0.5275
96.0	0.5562	0.7970	0.6766	0.5562	-0.0459
84.0	0.7167	0.9174	0.8371	0.7167	0.0344
73.5	0.6229	-0.5010	0.8103	0.6229	0.0610
54.0	0.4357	0.6229	0.6229	0.4357	0.0610
33.0	0.4357	0.4357	0.6229	0.4357	0.2483
24.0	0.5442	0.6737	0.6090	0.6737	0.2850
10.0	0.7385	0.7385	0.6737	0.7385	0.4357
5.0	0.8681	0.7385	0.5442	0.6090	0.4357
2.5	0.8033	0.5442	0.3498	0.4146	0.0610

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1860
43.4	-0.2359
50.4	-0.3353
56.1	-0.3833
62.5	-0.4819
69.7	-0.5298
76.9	-0.5298
83.4	-0.5312
89.4	-0.5805
95.4	-0.4819
101.1	-0.4326
---BOTTOM---	
38.2	0.0113
43.4	0.0113
50.4	-0.0674
56.1	0.1592
62.5	0.3072
69.7	0.4058
76.9	0.4058
83.4	0.4551
89.4	0.4551
95.4	0.2085
101.1	-0.5805

MISC.*

NO.	CP
1	-1.0830
2	0.6229
3	-4.0989
4	0.4357
5	-3.3109
6	-0.1263
7	-0.0459
8	-1.6111
9	-1.1697
10	0.9174
11	-0.2466
12	-0.6881

TABLE A19. RUN 187, BW6V, DELF=45,CMU=0, (BN/B)=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
187 2	CLAERO =	0.5462	CDAERO =	0.1996	DCLC =	0.0	DWLC =	0.0		BASEPR =	8.1916	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
0.06	2118.12	2088.18	29.95	156.24	0.0	2.68	0.0	0.0	0.0	87.6349	0.106678E+07	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR		
0.0	0.55	0.20	-0.35	0.00	-0.00	0.01	0.54	-0.34	0.54	0.20		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2310	-0.2146
2.5	-0.2146	-0.2228
5.0	-0.2064	-0.2310
10.0	-0.2228	-0.2392
15.0	-0.2146	-0.2310
25.0	-0.2146	-0.2310
35.0	-0.2228	-0.2228
50.0	-0.2228	
56.0	-0.2310	-0.2223
65.0	-0.2228	-0.2310
76.0	-0.2228	-0.2310
79.0	-0.2146	-0.2310
80.5	-0.2209	-0.2150
81.0	-0.2209	-0.2150
82.0	-0.2090	-0.2268
84.0	-0.2150	
87.0	-0.2150	
89.0	-0.2091	-0.2353
93.0	-0.2266	
96.0	-0.2353	-0.2266
100.0	-0.2353	-0.2266
96.0	-0.2268	-0.2266
84.0	-0.2353	-0.2266
73.0	-0.2266	-0.2266
50.0	-0.2353	-0.2178
35.0	-0.2228	-0.2228
25.0	-0.2228	-0.2310
10.0	-0.2228	-0.2310
5.0	-0.2146	-0.2310
2.5	-0.2064	-0.2228

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4927	0.4582	0.2743	0.2053	-0.0131
2.5	-0.1970	-0.3119	-0.4843	-0.6798	-0.6108
5.0	-0.1740	-0.3234	-0.3694	-0.4950	-0.6223
10.0	-0.2545	-0.3234	-0.3924	-0.4843	-0.4958
15.0	-0.2545	-0.3464	-0.3924	-0.4154	-0.4613
24.0	-0.2314	-0.2774	-0.3234	-0.3809	-0.3464
33.0	-0.2311	-0.2643	-0.3308	-0.3308	-0.3308
54.0	-0.2643	-0.2976	-0.3641	0.4003	4.1223
65.0	-0.3316	-0.3491	-0.4891	-0.5503	-0.4891
78.5	-0.3316	-0.6203	-1.1541	-1.0141	-0.8041
79.5	-0.4812	-0.5348	-1.0850	-0.8958	-0.8128
80.5	-0.4676	-0.5287	-0.9548	-0.8660	-0.2623
81.3	-0.4895	52.3943	-0.9430	-0.8838	-0.8009
82.0	-0.4813	-1.0850	-0.9076	-0.8779	-0.8187
84.0	-0.4813	-0.5582	-0.7832	-0.8720	-0.8128
87.0	-0.5050	-0.5819	-0.6885	-0.9016	-0.8068
89.0	-0.5538	-0.5965	-0.7033	-0.9169	-0.8101
93.0	-0.5467	-0.6036	-0.7247	-0.2761	-0.8101
96.0	-0.5467	-0.6036	-0.7318	-0.8600	-0.7959
100.0	-0.4541	-0.5609	-0.6820	-0.8528	-0.2476
96.0	0.3292	0.3648	0.3292	0.3363	0.2650
84.0	0.6567	0.6567	0.6211	0.6140	0.5000
73.5	0.5332	-0.1979	0.5997	0.5997	0.4667
54.0	0.2009	0.2341	0.2674	0.3006	0.2009
33.0	0.0680	0.0680	0.1012	0.1344	0.0680
24.0	0.0214	0.0329	0.0559	0.1249	0.0559
10.0	0.0444	0.0329	0.0789	0.1364	0.1344
5.0	0.0789	0.0559	0.1134	0.1479	0.2009
2.5	0.0904	0.1249	0.1708	0.2513	0.2674

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0866
43.4	-0.0778
50.4	-0.0691
56.1	-0.0691
62.5	-0.0778
69.7	-0.1303
76.9	-0.1303
83.4	-0.1215
89.4	-0.2091
95.4	-0.2703
101.1	-0.2266
---BOTTOM---	
38.2	-0.0516
43.4	-0.0166
50.4	-0.0516
56.1	0.0097
62.5	0.0447
69.7	0.0622
76.9	0.0884
83.4	0.1585
89.4	0.2110
95.4	0.1497
101.1	-0.2966

MISC.*

NO.	CP
1	-0.2643
2	0.1012
3	-0.3306
4	0.1344
5	-0.2643
6	0.0347
7	0.5071
8	-0.3331
9	-0.0625
10	0.5784
11	0.1725
12	-0.0696

TABLE A19. (Continued)

RUN:SEQ 187 4 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 444

CLAERO = 0.7208 CDAERO = 0.2474 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1916

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.01	2118.26	2088.09	30.17	157.01	0.0	2.64	0.0	0.0	0.0	86.3496	0.106787E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	0.72	0.25	-0.38	0.00	-0.00	0.01	0.73	-0.38	0.72	0.25

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2275	-0.2193
2.5	-0.2193	-0.2193
5.0	-0.2112	-0.2112
10.0	-0.2275	-0.2112
15.0	-0.2275	-0.2193
25.0	-0.2193	-0.2193
35.0	-0.2193	-0.2275
50.0	-0.2193	
56.0	-0.2193	-0.2193
65.0	-0.2193	-0.2193
76.0	-0.2193	-0.2193
79.0	-0.2112	-0.2193
80.5	-0.2210	-0.2034
81.0	-0.1375	-0.1975
82.0	-0.2151	-0.2093
84.0	-0.2034	
87.0	-0.2093	
89.0	-0.2062	-0.2149
93.0	-0.2149	
96.0	-0.2149	-0.2149
100.0	-0.2149	-0.2149
96.0	-0.2149	-0.2149
84.0	-0.2149	-0.2149
73.0	-0.2149	-0.2149
50.0	-0.2149	-0.2236
35.0	-0.2112	-0.2112
25.0	-0.2193	-0.2112
10.0	-0.2275	-0.2030
5.0	-0.2193	-0.2193
2.5	-0.2112	-0.2193

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0715	-0.3681	-1.1894	-1.6800	-2.6156
2.5	-0.2198	-0.8472	-1.0982	-1.3693	-1.4176
5.0	-0.5164	-0.6761	-0.8472	-0.9841	-1.1552
10.0	-0.4708	-0.5734	-0.7673	-0.7673	-0.8016
15.0	-0.4251	-0.5050	-0.6191	-0.6847	-0.6761
24.0	-0.4023	-0.4479	-0.4936	-0.5278	-0.5392
33.0	-0.3514	-0.4173	-0.4503	-0.4833	-0.4833
54.0	-0.3514	-0.3514	-0.4503	0.0444	3.2767
65.0	-0.3626	-0.3886	-0.5623	-0.6318	-0.6067
76.5	-0.3452	-0.6144	-1.1529	-1.2050	-0.9184
79.5	-0.4619	-0.4971	-1.0435	-1.0904	-0.8907
80.5	-0.4619	-0.4912	-0.9436	-1.0849	-0.2151
81.3	-0.4560	49.8194	-0.8790	-1.0904	-0.8731
82.0	-0.4619	-0.9730	-0.8143	-1.1139	-0.8731
84.0	-0.4677	-0.5265	-0.7203	-1.0611	-0.8731
87.0	-0.4971	-0.5500	-0.6675	-1.0552	-0.8966
89.0	-0.5609	-0.6033	-0.7163	-1.1545	-0.9142
93.0	-0.5538	-0.5962	-0.7234	-0.2711	-0.9142
96.0	-0.5609	-0.5821	-0.7376	-0.9637	-0.9072
100.0	-0.4549	-0.5185	-0.6740	-0.9213	-0.2358
96.0	0.3649	0.3961	0.3507	0.3437	0.2306
84.0	0.6829	0.6546	0.6829	0.5910	0.4921
73.5	0.5721	-0.2194	0.6051	0.5721	0.4402
54.0	0.2423	0.2423	0.3082	0.3082	0.1763
33.0	0.1434	0.1763	0.1763	0.1763	0.1104
24.0	0.1225	0.1225	0.1909	0.2023	0.1225
10.0	0.2137	0.2023	0.2366	0.3050	0.2753
5.0	0.2822	0.3164	0.3392	0.4077	0.3742
2.5	0.3734	0.4077	0.4305	0.4533	0.4402

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0933
43.4	-0.0673
50.4	-0.0760
56.1	-0.0760
62.5	-0.1020
69.7	-0.1715
76.9	-0.1628
83.4	-0.1368
89.4	-0.2149
95.4	-0.2487
101.1	-0.2236
---BOTTOM---	
38.2	-0.0499
43.4	-0.0152
50.4	-0.0326
56.1	0.0456
62.5	0.0690
69.7	0.1325
76.9	0.1498
83.4	0.1932
89.4	0.2387
95.4	0.1585
101.1	-0.3018

MISC.

NO.	CP
1	-0.3844
2	0.1763
3	-0.5163
4	0.1763
5	-0.4503
6	0.0114
7	0.5203
8	-0.2853
9	-0.0450
10	0.5557
11	0.1811
12	-0.0521

TABLE A19. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 446

187 6 CLAERO = 1.1278 CDAERO = 0.4261 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1916

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.03	2118.48	2088.64	20.83	156.26	0.0	2.66	0.0	0.0	0.0	87.1149	0.105907E+07

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	1.13	0.43	-0.47	0.01	-0.00	0.00	1.19	-0.47	1.12	0.43

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2483	-0.2483
2.5	-0.2401	-0.2319
5.0	-0.2319	-0.2483
10.0	-0.2319	-0.2483
15.0	-0.2319	-0.2401
25.0	-0.2236	-0.2401
35.0	-0.2319	-0.2483
50.0	-0.2401	
53.0	-0.2319	-0.2401
65.0	-0.2401	-0.2401
76.0	-0.2483	-0.2401
79.0	-0.2401	-0.2401
80.5	-0.2573	-0.2395
81.0	-0.2395	-0.2276
82.0	-0.2455	-0.2455
84.0	-0.2336	
87.0	-0.2276	
89.0	-0.2274	-0.2274
93.0	-0.2274	
96.0	-0.2274	-0.2274
100.0	-0.2362	-0.2186
96.0	-0.2274	-0.2274
84.0	-0.2362	-0.2186
73.0	-0.2274	-0.2274
50.0	-0.2362	-0.2186
35.0	-0.2401	-0.2401
25.0	-0.2401	-0.2483
10.0	-0.2401	-0.2565
5.0	-0.2319	-0.2483
2.5	-0.2401	-0.2483

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.0244	-5.3667	-7.1204	-2.8628	-1.5361
2.5	-0.1746	-2.1822	-3.9589	-2.5977	-1.5245
5.0	-1.2361	-1.7092	-2.4244	-2.5283	-1.5361
10.0	-0.9361	-1.3169	-1.4553	-2.7244	-1.4784
15.0	-0.7977	-1.0054	-1.2015	-2.5629	-1.4899
24.0	-0.6592	-0.7746	-0.9582	-1.5938	-1.4323
33.0	-0.5989	-0.6657	-0.7657	-0.9992	-1.4328
54.0	-0.4321	-0.4989	-0.5656	-0.1986	2.5700
65.0	-0.4470	-0.4821	-0.6490	-0.6666	-1.2550
78.5	-0.3680	-0.6051	-1.0179	-1.5888	-1.3780
79.5	-0.4831	-0.5306	-0.9109	-1.4753	-1.3684
80.5	-0.5009	-0.5128	-0.8158	-1.2793	-0.2455
81.3	-0.5128	45.3174	-0.7327	-1.2080	-1.3505
82.0	-0.5128	-0.9228	-0.7148	-1.1961	-1.2793
84.0	-0.5247	-0.5128	-0.6316	-1.3149	-1.2496
87.0	-0.5128	-0.5425	-0.6554	-1.2852	-1.1961
89.0	-0.5845	-0.6059	-0.7060	-1.3063	-1.1991
93.0	-0.6345	-0.5845	-0.7417	-0.2914	-1.1634
96.0	-0.6345	-0.5702	-0.7560	-0.2632	-1.1348
100.0	-0.5416	-0.5273	-0.6774	-0.7917	-0.2557
96.0	0.3661	0.4376	0.3947	0.3661	0.1732
84.0	0.7235	0.7878	0.6591	0.6091	0.3947
73.5	0.6353	-0.2320	0.6350	0.6019	0.4018
54.0	0.3351	0.3684	0.3684	0.3684	0.2016
33.0	0.3017	0.3351	0.3684	0.3684	0.1683
24.0	0.3330	0.3907	0.3907	0.4253	0.2638
10.0	0.5176	0.5176	0.5292	0.5407	0.4018
5.0	0.6330	0.5984	0.5753	0.5984	0.5018
2.5	0.6907	0.5984	0.5061	0.5176	0.4351

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.1045	38.2	0.0185
43.4	-0.1045	43.4	0.0448
50.4	-0.1396	50.4	0.0185
56.1	-0.1747	56.1	0.1239
62.5	-0.2450	62.5	0.2029
69.7	-0.3240	69.7	0.2556
76.9	-0.2889	76.9	0.2732
83.4	-0.2450	83.4	0.2908
89.4	-0.2889	89.4	0.2995
95.4	-0.2538	95.4	0.1854
101.1	-0.1923	101.1	-0.3153

MISC.*

NO.	CP
1	-0.6990
2	0.3351
3	-1.6997
4	0.3017
5	-1.2661
6	0.0349
7	0.5090
8	-0.2700
9	0.0445
10	0.5234
11	0.0945
12	-0.0770

TABLE A20. RUN 188, BW6V, DELF=4R ...J=0.5, (BN/B)=0.5

RUN:SEQ 188 1 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 447

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	BASEPR = 8.1916	
-0.00	2118.76	2089.15	29.61	155.87	0.0	2.68	0.0	0.496	0.496	HGT	RN
										87.7661	0.105144E+07

CLAERO = 0.4560 CDAERO = 0.3093 DCLC = 0.0 DWLC = 0.4294

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	0.89	0.06	-0.61	0.04	0.00	-0.05	0.63	-0.40	0.63	0.22

********** CANARD ******

XX/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3398	-0.3398
2.5	-0.3233	-0.3398
5.0	-0.3398	-0.3398
10.0	-0.3398	-0.3316
15.0	-0.3316	-0.3316
25.0	-0.3398	-0.3233
35.0	-0.3233	-0.3067
50.0	-0.3150	
56.0	-0.3150	-0.3067
65.0	-0.3067	-0.3067
76.0	-0.3233	-0.3067
79.0	-0.3067	-0.3067
80.5	-0.3439	-0.3439
81.0	-0.3439	-0.3439
82.0	-0.3379	-0.3379
84.0	-0.3559	
87.0	-0.3499	
89.0	-0.3156	-0.3156
93.0	-0.3244	
96.0	-0.3244	-0.3156
100.0	-0.3156	-0.3156
96.0	-0.3244	-0.3156
84.0	-0.3156	-0.3156
73.0	-0.3244	-0.3156
50.0	-0.3156	-0.3156
35.0	-0.2984	-0.3067
25.0	-0.3067	-0.3067
10.0	-0.3067	-0.3067
5.0	-0.3482	-0.3233
2.5	-0.3398	-0.3233

********** WING **********

XX/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =18 CP	BP = 22 CP
0.0	0.4567	0.3753	0.0033	-0.3687	-0.3571
2.5	-0.3106	-0.4850	-0.7756	-0.7873	-0.9965
5.0	-0.2641	-0.4385	-0.5431	-0.6361	-0.7640
10.0	-0.3571	-0.4734	-0.5664	-0.5431	-0.6245
15.0	-0.3571	-0.4385	-0.4966	-0.5315	-0.5548
24.0	-0.3687	-0.4152	-0.4385	-0.4850	-0.4734
33.0	-0.3414	-0.4086	-0.4758	-0.5094	-0.4422
54.0	-0.5094	-0.5430	-0.6102	-0.1397	2.7509
65.0	-0.6873	-0.8112	-0.8023	-0.8377	-0.7935
78.5	-0.8112	-1.9706	-1.7316	-1.4042	-1.4661
79.5	-7.8810	-11.4611	-1.2958	-1.1820	-1.5532
80.5	-7.6415	-11.1379	-1.3137	-1.1640	-0.2960
81.3	-7.6476	46.7475	-1.3916	-1.1581	-1.4514
82.0	-7.6297	-12.2153	-1.3616	-1.1640	-1.4155
84.0	-8.1025	-11.5148	-1.3556	-1.1581	-1.3916
87.0	-8.3062	-11.0241	-1.3856	-1.1820	-1.4035
89.0	-8.4457	-9.9794	-1.4527	-1.3374	-1.4527
93.0	-7.1133	-2.6915	-1.3519	-0.3940	-1.5247
96.0	-5.3776	-0.7685	-1.2654	-1.3014	-1.5175
100.0	-2.6050	-1.0782	-1.0062	-1.2006	-0.3148
96.0	0.4064	0.4630	0.2974	0.2757	0.1389
84.0	0.6214	0.6574	0.7006	0.5710	0.4414
73.5	0.5326	-0.3078	0.5662	0.5662	0.3981
54.0	0.1964	0.2300	0.2972	0.2972	0.1628
33.0	0.0956	0.0956	0.0956	0.1628	0.0620
24.0	0.0149	0.0265	0.0963	0.1312	0.0498
10.0	0.0614	0.0730	0.1196	0.1428	0.1628
5.0	0.0614	0.0847	0.1428	0.2242	0.1964
2.5	0.1196	0.1428	0.2939	0.3288	0.3309

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0766
43.4	-0.0589
50.4	-0.0678
56.1	-0.0678
62.5	-0.0855
69.7	-0.1474
76.9	-0.1740
83.4	-0.1651
89.4	-0.2802
95.4	-0.3067
101.1	-0.2536
---BOTTOM---	
38.2	-0.0412
43.4	-0.0147
50.4	-0.0589
56.1	0.0207
62.5	0.0561
69.7	0.0738
76.9	0.1093
83.4	0.1624
89.4	0.2066
95.4	0.0207
101.1	-0.4749

**MISC.

NO.	CP
1	-0.4422
2	0.0956
3	-0.5094
4	0.1628
5	-0.4086
6	0.0284
7	0.4846
8	-0.4012
9	-0.1420
10	0.6358
11	0.2325
12	-0.0328

Force Data - Use Run 222

TABLE A20. (Continued)

RUN:SEQ 18L 3 CLAERO = 0.5531 CDAERO = 0.3391 PROPLULSIVE WING / CANARD PRESSURES DWLC = 0.4604 BASEFR = 8.1916

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.06	2118.90	2089.97	28.93	154.13	0.0	2.67	0.0	0.512	0.512	87.3370	0.103796E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
0.0	1.01	0.12	-0.62	0.04	-0.00	-0.04	0.75	-0.40	0.74	0.26

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3076	-0.3076
2.5	-0.2991	-0.2991
5.0	-0.3076	-0.2906
10.0	-0.2906	-0.2991
15.0	-0.2991	-0.2906
25.0	-0.2821	-0.3076
35.0	-0.3161	-0.3161
50.0	-0.3076	
56.0	-0.3161	-0.3246
65.0	-0.3076	-0.3076
76.0	-0.3076	-0.3161
79.0	-0.3076	-0.3246
80.5	-0.2877	-0.2571
81.0	-0.2693	-0.2693
82.0	-0.2077	-0.2754
84.0	-0.2754	
87.0	-0.2816	
89.0	-0.2782	-0.2782
93.0	-0.2872	
96.0	-0.2782	-0.2963
100.0	-0.2872	-0.2872
96.0	-0.2872	-0.2963
84.0	-0.2872	-0.2872
73.0	-0.2872	-0.2963
50.0	-0.2782	-0.3053
35.0	-0.3161	-0.3246
25.0	-0.3076	-0.3161
10.0	-0.3076	-0.3331
5.0	-0.2906	-0.3076
2.5	-0.2906	-0.2991

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.1790	-0.7620	-1.7496	-2.2969	-3.7367
2.5	-0.2742	-0.9524	-1.2142	-1.7496	-1.8686
5.0	-0.5716	-0.7858	-1.0238	-1.3451	-1.4759
10.0	-0.5121	-0.6668	-0.8453	-1.0238	-1.0952
15.0	-0.4883	-0.5716	-0.7739	-0.7739	-0.9048
24.0	-0.4407	-0.4883	-0.6549	-0.6787	-0.6906
33.0	-0.4458	-0.5146	-0.6178	-0.6178	-0.6522
54.0	-0.5490	-0.5834	-0.6867	0.0358	2.6502
65.0	-0.6949	-0.8579	-0.8398	-0.8579	-0.8579
78.5	-0.8126	-1.8905	-1.8995	-1.3470	-1.8633
79.5	-7.7016	-10.9613	-1.6602	-1.0965	-2.3893
80.5	-7.3954	-10.5752	-1.6050	-1.1026	-0.2877
81.3	-7.4258	41.1732	-1.5805	-1.0904	-1.6786
82.0	-7.5115	-10.9980	-1.5928	-1.1271	-1.6663
84.0	-7.9159	-10.9551	-1.6357	-1.1678	-1.4519
87.0	-8.0017	-10.6302	-1.4641	-1.1762	-1.5070
89.0	-8.3249	-9.9392	-1.4702	-1.3228	-1.5734
93.0	-7.2564	-3.2836	-1.3228	-0.3720	-1.6250
96.0	-5.6051	-1.3670	-1.2122	-1.2712	-1.6176
100.0	-3.4972	-1.1975	-1.0280	-1.1828	-0.3277
96.0	0.4462	0.4757	0.2914	0.2767	0.0924
84.0	0.6673	0.6747	0.6894	0.5715	0.3946
73.5	0.5518	-0.2738	0.5862	0.5518	0.3454
54.0	0.2766	0.2766	0.3110	0.2766	0.1390
33.0	0.1734	0.2078	0.2078	0.1734	0.0702
24.0	0.1661	0.1661	0.2137	0.2375	0.1185
10.0	0.2613	0.2375	0.2851	0.3565	0.2766
5.0	0.3208	0.3208	0.4040	0.4516	0.4142
2.5	0.4159	0.4398	0.4993	0.5350	0.4486

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0370
43.4	-0.0879
50.4	-0.0879
56.1	-0.1061
62.5	-0.1423
69.7	-0.2148
76.9	-0.2238
83.4	-0.2238
89.4	-0.3416
95.4	-0.3325
101.1	-0.2963
---BOTTOM---	
38.2	-0.0336
43.4	0.0026
50.4	-0.0336
56.1	0.0570
62.5	0.0932
69.7	0.1385
76.9	0.1657
83.4	0.2110
89.4	0.2291
95.4	0.0660
101.1	-0.4050

MISC.*

NO.	CP
1	-0.5490
2	0.2078
3	-0.6867
4	0.1734
5	-0.6522
6	-0.0330
7	0.4683
8	-0.4162
9	-0.2024
10	0.6157
11	0.2177
12	-0.1066

Force Data - Use
Run 222

TABLE A20. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 451

188	5	CLAERO =	0.7127	CDAERO =	0.4814	DCLC =	0.0	DWLC =	0.4882	BASEPR =	8.1916	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
11.97		2118.83	2089.56	29.27	155.13	0.0	2.66	0.0	0.513	0.513	87.1025	0.1042[illegible]6E+07
BETA		CL	CD	CN	CROLL		CN	CY	CNTR	CNTR	CLTR	CDTR
0.0		1.20	0.32	-0.63	0.03		-0.01	-0.01	0.98	-0.42	0.91	0.42

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3693	-0.3693
2.5	-0.3525	-0.3776
5.0	-0.3525	-0.3609
10.0	-0.3609	-0.3776
15.0	-0.3776	-0.3609
25.0	-0.3609	-0.3860
35.0	-0.3609	-0.3860
50.0	-0.3776	
56.0	-0.3776	-0.3776
65.0	-0.3776	-0.3693
76.0	-0.3776	-0.3776
79.0	-0.3525	-0.3776
80.5	-0.3468	-0.3407
81.0	-0.3407	-0.3347
82.0	-0.3468	-0.3589
84.0	-0.3347	
87.0	-0.3589	
89.0	-0.3363	-0.3453
93.0	-0.3542	
96.0	-0.3453	-0.3542
100.0	-0.3542	-0.3542
96.0	-0.3453	-0.3453
84.0	-0.3542	-0.3542
73.0	-0.3453	-0.3453
50.0	-0.3721	-0.3542
35.0	-0.3525	-0.3776
25.0	-0.3609	-0.3609
10.0	-0.3609	-0.3693
5.0	-0.3525	-0.3609
2.5	-0.3776	-0.3693

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.1890	-5.8704	-9.2576	-4.1652	-2.0954
2.5	-0.3078	-2.5071	-3.6947	-3.8594	-2.0130
5.0	-1.2839	-1.7308	-3.2362	-3.7067	-2.0248
10.0	-1.0957	-1.3309	-1.9190	-3.7067	-1.9895
15.0	-0.9899	-1.0722	-1.4720	-3.2362	-1.9190
24.0	-0.8135	-0.8958	-1.1428	-1.6349	-1.8601
33.0	-0.7024	-0.8384	-0.9404	-0.8044	-1.9264
54.0	-0.7024	-0.7364	-0.8724	0.1136	2.3577
55.0	-0.8018	-0.9899	-1.0436	-0.9988	-1.6702
78.5	-0.7660	-1.8224	-2.4760	-2.1895	-2.0731
79.5	-6.0757	-9.6730	-1.4853	-2.0970	-2.6298
80.5	-6.6630	-9.2243	-1.4914	-1.7275	-0.3528
81.3	-6.1362	36.8308	-1.8729	-1.6730	-2.1515
82.0	-6.0214	-10.3208	-1.7094	-1.5883	-2.0788
84.0	-6.2937	-9.2186	-1.6004	-1.7215	-1.9537
87.0	-6.6934	-9.3761	-1.6367	-1.6367	-1.8184
89.0	-6.8048	-8.8517	-1.7270	-1.7270	-1.7853
93.0	-5.9015	-4.8888	-1.5449	-0.4375	-1.7124
96.0	-5.1802	-2.6666	-1.3773	-1.5230	-1.6250
100.0	-5.8068	-1.3700	-1.1661	-1.1879	-0.3720
96.0	0.4002	0.4877	0.3201	0.3420	0.0724
84.0	0.7062	0.8082	0.7062	0.6188	0.3420
73.5	0.6236	-0.3624	0.6236	0.6236	0.3516
54.0	0.3516	0.3856	0.3856	0.3516	0.1476
33.0	0.3516	0.3516	0.3516	0.3516	0.1816
24.0	0.3273	0.3861	0.4214	0.4214	0.2567
10.0	0.5155	0.5155	0.5390	0.5507	0.4196
5.0	0.6213	0.5978	0.5625	0.5978	0.4876
2.5	0.7036	0.5743	0.4096	0.4684	0.3856

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1125
43.4	-0.1125
50.4	-0.1483
56.1	-0.1931
62.5	-0.2647
69.7	-0.3453
76.9	-0.3453
83.4	-0.5094
89.4	-0.3811
95.4	-0.3542
101.1	-0.2826
---BOTTOM---	
38.2	0.0128
43.4	0.0486
50.4	0.0039
56.1	0.1382
62.5	0.2188
69.7	0.2814
76.9	0.2904
83.4	0.3083
89.4	0.3262
95.4	0.1829
101.1	-0.2468

MISC.*

NO.	CP
1	-0.8724
2	0.3856
3	-2.3345
4	0.3176
5	-1.6544
6	0.0116
7	0.3274
8	-0.8455
9	-0.4157
10	0.7499
11	-0.0308
12	-0.2991

Force Data - Use
Run 222

TABLE A21. RUN 189, BW6V, DELF=45, CMU=1.0, (BN/B)N=0.5

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 452

RUN:SEQ											
189	1	CLAERO =	0.6210	CDAERO =	0.5816	DCLC =	0.0	DWLC =	0.9068	BASEPR =	8.1916
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUM	CMUT	HGT	RN
0.09		2119.11	2105.10	14.01	107.02	0.0	2.65	0.0	1.046	1.046	56.6633	0.723852E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		1.53	0.06	-1.08	0.01	0.01	-0.01	0.98	-0.64	0.98	0.39	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4372	-0.4197
2.5	-0.4372	-0.4022
5.0	-0.4197	-0.4022
10.0	-0.4197	-0.4197
15.0	-0.4372	-0.4197
25.0	-0.4372	-0.4022
35.0	-0.4372	-0.4547
50.0	-0.4372	
56.0	-0.4197	-0.4372
65.0	-0.4372	-0.4547
76.0	-0.4022	-0.4372
79.0	-0.4372	-0.4547
80.5	-0.4303	-0.3923
81.0	-0.4682	-0.4176
82.0	-0.4303	-0.4050
84.0	-0.4176	
87.0	-0.4303	
89.0	-0.3847	-0.3660
93.0	-0.3847	
96.0	-0.3847	-0.3847
100.0	-0.4034	-0.3847
96.0	-0.4034	-0.4034
84.0	-0.4034	-0.3847
73.0	-0.4034	-0.4034
50.0	-0.3660	-0.4034
35.0	-0.4372	-0.4372
25.0	-0.4372	-0.4372
10.0	-0.4197	-0.4547
5.0	-0.4197	-0.4022
2.5	-0.4197	-0.4372

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4341	0.3604	-0.0818	-0.4257	-0.9660
2.5	-0.4257	-0.4010	-0.7204	-1.1134	-1.3098
5.0	-0.3274	-0.4010	-0.5975	-0.7204	-0.8677
10.0	-0.3519	-0.4747	-0.5730	-0.5975	-0.6713
15.0	-0.3519	-0.4010	-0.5485	-0.5485	-0.5975
24.0	-0.3519	-0.4010	-0.4747	-0.4747	-0.5238
33.0	-0.4212	-0.4923	-0.4923	-0.4923	-0.4923
54.0	-0.4923	-0.6343	-0.7053	-0.4923	4.1235
65.0	-0.7773	-1.0952	-0.9456	-0.8895	-0.7773
78.5	-0.9082	-2.4414	-3.0707	-1.3382	-1.3943
79.5	-13.5971	-19.2379	-4.3891	-1.1512	-1.7204
80.5	-12.6866	-19.0609	-3.0611	-1.1512	-0.4559
81.3	-12.9014	77.8507	-2.7449	-1.1259	-1.5560
82.0	-12.9014	-20.0478	-2.7089	-1.1386	-1.5054
84.0	-13.7363	-19.6432	-2.3529	-1.1512	-1.4421
87.0	-14.2044	-18.2011	-1.2904	-1.2271	-1.3916
89.0	-14.5819	-17.5332	-1.2377	-1.2529	-1.3898
93.0	-13.2125	-7.8564	-1.1616	-0.4769	-1.4354
96.0	-10.5188	-1.7702	-1.1312	-1.2377	-1.4659
100.0	-6.6997	-1.5724	-0.9334	-1.1616	-0.4312
96.0	0.3600	0.4665	0.3143	0.2687	0.1165
84.0	0.6186	0.6795	0.6947	0.6186	0.4665
73.5	0.5019	-0.4212	0.6439	0.5729	0.3599
54.0	0.2179	0.2179	0.2889	0.2179	0.1468
33.0	0.0759	0.0759	0.0759	0.0759	0.0759
24.0	0.0656	0.0902	0.0902	0.1639	0.0411
10.0	0.1147	0.1147	0.1639	0.2375	0.0759
5.0	0.1393	0.1393	0.1884	0.2867	0.2179
2.5	0.1884	0.1884	0.2867	0.4341	0.3599

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.1042	38.2	-0.0481
43.4	-0.0855	43.4	-0.0294
50.4	-0.0855	50.4	-0.1042
56.1	-0.1042	56.1	0.0080
62.5	-0.1229	62.5	0.0454
69.7	-0.1977	69.7	0.0828
76.9	-0.2351	76.9	0.1015
83.4	-0.2351	83.4	0.1576
89.4	-0.3098	89.4	0.1950
95.4	-0.3847	95.4	-0.0294
101.1	-0.3473	101.1	-0.5529

MISC.

NO.	CP
1	-0.4923
2	0.1498
3	-0.5633
4	0.0759
5	-0.4923
6	0.0048
7	0.2991
8	-0.7356
9	-0.4921
10	0.7860
11	0.1926
12	-0.2030

Force Data - Use
Run 223

Table A21. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

PAGE 455

RUN:SEQ											
189 4	CLAERO =	1.0191	CDAERO =	0.7481	DCLC =	0.0	DWLC =	0.9518	BASEPR =	8.1916	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.96	2119.11	2104.19	14.92	110.38	0.0	2.66	0.0	1.001	1.001	86.9015	0.747689E+06
BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0	1.97	0.44	-1.13	0.04	-0.01	-0.01	1.50	-0.71	1.40	0.64	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.4820	-0.4985
2.5	-0.4655	-0.4820
5.0	-0.4655	-0.4820
10.0	-0.4655	-0.4820
15.0	-0.4655	-0.4820
25.0	-0.4655	-0.4820
35.0	-0.4985	-0.4985
50.0	-0.4820	
56.0	-0.4985	-0.4985
65.0	-0.4985	-0.4820
76.0	-0.4985	-0.4985
79.0	-0.4985	-0.4655
80.5	-0.4508	-0.4389
81.0	-0.4508	-0.4508
82.0	-0.4508	-0.4627
84.0	-0.4508	
87.0	-0.4270	
89.0	-0.4415	-0.4767
93.0	-0.4767	
96.0	-0.4767	-0.4591
100.0	-0.4767	-0.4767
96.0	-0.4767	-0.4767
84.0	-0.4767	-0.4767
73.0	-0.4767	-0.4767
50.0	-0.4767	-0.4942
35.0	-0.4985	-0.4820
25.0	-0.4985	-0.4820
10.0	-0.4985	-0.4655
5.0	-0.4985	-0.4985
2.5	-0.4820	-0.4985

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-3.1776	-5.9928	-7.2845	-3.6622	-2.1624
2.5	-0.4318	-2.5084	-4.6543	-3.4545	-2.1392
5.0	-1.3317	-1.9085	-3.5006	-3.7313	-2.0700
10.0	-1.1701	-1.4701	-2.0931	-3.8698	-2.0931
15.0	-1.0317	-1.2394	-1.4701	-3.3622	-2.0470
24.0	-0.8471	-1.0066	-1.2624	-1.7470	-2.0008
33.0	-0.8024	-0.8691	-1.1359	-0.9358	-2.1365
54.0	-0.8024	-0.8691	-1.0692	-0.3354	3.6004
65.0	-0.9861	-1.3549	-1.2847	-1.1617	-2.0751
78.5	-0.7401	-2.5318	-3.0938	-2.3210	-2.2158
79.5	-12.7008	-18.2256	-3.1836	-3.7539	-2.8390
80.5	-11.8809	-18.3087	-2.4350	-2.6489	-0.4270
81.3	-11.7268	67.3582	-2.5063	-2.1281	-2.3994
82.0	-12.0827	-18.4281	-2.8628	-1.9835	-2.1374
84.0	-12.1068	-18.2142	-2.4112	-2.1974	-2.0310
87.0	-12.6418	-17.5607	-2.2687	-2.1024	-1.9360
89.0	-12.8375	-15.9673	-2.0743	-2.0600	-1.9742
93.0	-11.9366	-7.8060	-1.6597	-0.5735	-1.9027
96.0	-9.9645	-2.7604	-1.5311	-1.4740	-1.8170
100.0	-7.7631	-1.7741	-0.9736	-1.1881	-0.5020
96.0	0.4556	0.5271	0.3270	0.2842	-0.0160
84.0	0.7129	0.8416	0.7415	0.6415	0.3842
73.5	0.6653	-0.4021	0.6653	0.6653	0.3317
54.0	0.3984	0.3984	0.3984	0.3984	0.1316
33.0	0.3984	0.3984	0.3984	0.3984	0.1316
24.0	0.3527	0.3758	0.3989	0.4450	0.2143
10.0	0.5143	0.5143	0.5373	0.5835	0.3984
5.0	0.6065	0.5835	0.5835	0.5604	0.4651
2.5	0.6988	0.5604	0.4450	0.3989	0.3984

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1429
43.4	-0.1429
50.4	-0.1780
56.1	-0.2308
62.5	-0.2835
69.7	-0.4064
76.9	-0.3713
83.4	-0.3537
89.4	-0.4415
95.4	-0.3889
101.1	-0.3186
---BOTTOM---	
38.2	-0.0200
43.4	0.0152
50.4	-0.0651
56.1	0.1030
62.5	0.1908
69.7	0.2811
76.9	0.2611
83.4	0.2962
89.4	0.2962
95.4	0.0854
101.1	-0.4942

MISC.*

NO.	CP
1	-1.0025
2	0.3934
3	-2.5368
4	0.3317
5	-1.8030
6	-0.0019
7	0.1983
8	-0.9880
9	-0.6449
10	0.7272
11	0.0412
12	-0.4305

Force Data - Use
Run 223

TABLE A22. RUN 190, BW6V, DELF=45, CMU=2.0, (BN/B)W=0.5

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 456

RUN:SEQ											
190	1	CLAERO =	0.3952	CDAERO =	0.8036	DCLC =	0.0	DWLC =	1.7470	BASEPR =	8.1916
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.05		2119.25	2111.91	7.35	77.35	0.0	2.64	0.0	2.016	2.016	86.2393	0.525544E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.0		2.14	-0.20	-1.54	0.02	0.01	-0.02	1.09	-0.70	1.09	0.43	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.5630	-0.5630
2.5	-0.5296	-0.5630
5.0	-0.5296	-0.5630
10.0	-0.5296	-0.5296
15.0	-0.5296	-0.5630
25.0	-0.5630	-0.5630
35.0	-0.5296	-0.5630
50.0	-0.5630	
56.0	-0.5296	-0.5630
65.0	-0.5296	-0.5630
76.0	-0.5630	-0.5630
79.0	-0.5296	-0.5630
80.5	-0.4886	-0.5368
81.0	-0.5127	-0.4886
82.0	-0.5609	-0.5852
84.0	-0.4644	
87.0	-0.5368	
89.0	-0.4907	-0.5265
93.0	-0.5265	
96.0	-0.5265	-0.4907
100.0	-0.5265	-0.5265
96.0	-0.4907	-0.5265
84.0	-0.5265	-0.5265
73.0	-0.4907	-0.5265
50.0	-0.4907	-0.5265
35.0	-0.5630	-0.5630
25.0	-0.5296	-0.5630
10.0	-0.5630	-0.5296
5.0	-0.5630	-0.5296
2.5	-0.5630	-0.5630

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.3982	0.0702	-0.3046	-0.6327	-0.7731
2.5	-0.4921	-0.5389	-0.9137	-1.1480	-1.2417
5.0	-0.3983	-0.4921	-0.7731	-0.8199	-0.9606
10.0	-0.3983	-0.5389	-0.6795	-0.6795	-0.7731
15.0	-0.3983	-0.4921	-0.6327	-0.6327	-0.7263
24.0	-0.4452	-0.4921	-0.5857	-0.5389	-0.6327
33.0	-0.4828	-0.4828	-0.4828	-0.6182	-0.4828
54.0	-0.6182	-0.7537	-0.7537	-1.8374	6.4256
65.0	-0.9544	-1.4895	-1.1328	-1.0614	-0.9188
78.5	-0.3837	-3.5225	-3.5580	-1.5608	-1.8105
79.5	-24.2531	-38.0062	-5.0728	-1.3572	-2.1052
80.5	-22.4196	-37.3542	-3.7940	-1.2607	-0.5609
81.3	-23.0947	139.4912	-3.1425	-1.3089	-2.0087
82.0	-23.0716	-38.9467	-2.9254	-1.3572	-1.7914
84.0	-24.8560	-37.6917	-2.4670	-1.2848	-1.6708
87.0	-25.7013	-35.2552	-1.4537	-1.4054	-1.5984
89.0	-26.0430	-31.0343	-1.3727	-1.4597	-1.8661
93.0	-23.0825	-8.5705	-1.1985	-0.5600	-1.8951
96.0	-17.2486	-0.9083	-1.0824	-1.4307	-1.7790
100.0	-9.0929	-2.8238	-0.9954	-1.3436	-0.5020
96.0	0.2527	0.3687	0.2236	0.2236	0.0495
84.0	0.5139	0.6009	0.6380	0.5719	0.4268
73.5	0.6009	-0.4828	0.6009	0.6009	0.4654
54.0	0.3300	0.1945	0.3300	0.3300	0.1945
33.0	0.0591	0.0591	0.1945	0.1945	0.0591
24.0	0.0234	0.0702	0.0234	0.1170	-0.0235
10.0	0.1170	0.0234	0.1170	0.2108	0.1945
5.0	0.0702	0.0702	0.2108	0.3045	0.3300
2.5	0.1170	0.1170	0.2576	0.3513	0.3300

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1697
43.4	-0.1341
50.4	-0.1341
56.1	-0.1697
62.5	-0.1597
69.7	-0.2411
76.9	-0.2768
83.4	-0.2768
89.4	-0.3837
95.4	-0.4550
101.1	-0.4194
---BOTTOM---	
38.2	-0.0628
43.4	-0.0628
50.4	-0.1341
56.1	-0.0271
62.5	0.0442
69.7	0.0442
76.9	0.0442
83.4	0.0799
89.4	0.0799
95.4	-0.2768
101.1	-0.5118

MISC.*

NO.	CP
1	-0.4828
2	0.1945
3	-0.6182
4	0.1945
5	-0.4828
6	-0.0764
7	0.2236
8	-0.8212
9	-0.6181
10	0.7460
11	0.1656
12	-0.2408

Force Data - Use
Run 223

TABLE A22. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 458

190 3 CLAERO = 0.5313 CDAERO = 0.8265 DCLC = 0.0 DWLC = 1.7992 BASEPR = 8.1916

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.09	2119.25	2111.79	7.46	77.93	0.0	2.66	0.0	2.000	2.000	88.0241	0.529777E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	2.33	-0.05	-1.57	0.02	0.01	-0.03	1.28	-0.74	1.25	0.50

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----		ND.	CP
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	%LOC	CP		
0.0	-0.4424	-0.4753	0.0	-0.2406	-0.9788	-2.2710	-2.7787	-4.2091	38.2	-0.1166	1	-0.5055
2.5	-0.5082	-0.5082	2.5	-0.4251	-0.9327	-1.5327	-1.7173	-2.5479	43.4	-0.0836	2	0.3384
5.0	-0.4753	-0.4753	5.0	-0.5535	-0.7020	-1.2096	-1.2558	-1.5788	50.4	-0.1186	3	-0.7289
10.0	-0.4424	-0.4753	10.0	-0.5174	-0.6559	-0.9327	-0.9788	-1.2096	56.1	-0.1186	4	0.2050
15.0	-0.4753	-0.4753	15.0	-0.5174	-0.6559	-0.8405	-0.8405	-0.9788	62.5	-0.1538	5	-0.7289
25.0	-0.4753	-0.4753	24.0	-0.4713	-0.6097	-0.7481	-0.7481	-0.8405	69.7	-0.2592	6	-0.0619
35.0	-0.5082	-0.5082	33.0	-0.4621	-0.5956	-0.7289	-0.7289	-0.7289	76.9	-0.2943	7	0.1765
50.0	-0.4753		54.0	-0.7289	-0.7289	-0.7289	-1.5295	6.3423	83.4	-0.3295	8	-0.8241
56.0	-0.4753	-0.5082	65.0	-0.9969	-1.5240	-1.1374	-1.1023	-1.0320	89.4	-0.3296	9	-0.6811
65.0	-0.4753	-0.4753	78.5	2.0242	-3.3857	-3.6317	-1.5942	-1.8401	95.4	-0.3997	10	0.7482
76.0	-0.4753	-0.5082	79.5	-24.1842	-36.5411	-5.1968	-1.4422	-2.3214	101.1	-0.3645	11	0.2050
79.0	-0.4753	-0.5082	80.5	-22.5915	-35.8520	-3.8661	-1.4184	-0.5630	---BOTTOM---		12	-0.2824
80.5	-0.5391	-0.5153	81.3	-22.6867	126.9044	-3.3195	-1.3947	-2.0125	38.2	-0.0133		
81.0	-0.5153	-0.5153	82.0	-22.8771	-39.0602	-3.2482	-1.3471	-1.9888	43.4	-0.0193		
82.0	-0.5391	-0.5391	84.0	-23.9697	-37.3254	-2.5828	-1.4659	-1.8937	50.4	-0.0836		
84.0	-0.5153		87.0	-24.9913	-34.9725	-1.6323	-1.4422	-1.8461	56.1	0.0570		
87.0	-0.4679		89.0	-26.1228	-29.0106	-1.5397	-1.5959	-1.8247	62.5	0.0921		
89.0	-0.4699	-0.4699	93.0	-22.5500	-6.7701	-1.3387	-0.5668	-1.8532	69.7	0.1272		
93.0	-0.4699		96.0	-17.0613	-1.6530	-1.2814	-1.5101	-1.9389	76.9	0.1623		
96.0	-0.4699	-0.4699	100.0	-8.5139	-3.5398	-1.1671	-1.4530	-0.5382	83.4	0.1623		
100.0	-0.5050	-0.5050	96.0	0.2908	0.3766	0.2622	0.2050	0.0335	89.4	0.1623		
96.0	-0.4699	-0.5050	84.0	0.5767	0.6910	0.7768	0.5767	0.4823	95.4	-0.1538		
84.0	-0.5050	-0.5050	73.5	0.4718	-0.4621	0.6052	0.6052	0.4718	101.1	-0.7510		
73.0	-0.4699	-0.5050	54.0	0.3384	0.3384	0.3384	0.3384	0.0715				
50.0	-0.4699	-0.5050	33.0	0.2050	0.2050	0.2050	0.2050	0.0715				
35.0	-0.4424	-0.4753	24.0	0.2210	0.2210	0.3133	0.3594	0.1287				
25.0	-0.5082	-0.4753	10.0	0.3133	0.3133	0.3133	0.4517	0.3384				
10.0	-0.4753	-0.4753	5.0	0.3594	0.4056	0.4978	0.4978	0.3384				
5.0	-0.4753	-0.5082	2.5	0.4978	0.4517	0.5440	0.5440	0.4718				
2.5	-0.4753	-0.5082										

Force Data - Use
Run 223

TABLE A22. (Concluded)

RUN:SEQ 190 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 460

CLAERO = 0.8367 CDAERO = 0.9784 DCLC = 0.0 DWLC = 1.9641 BASEPR = 8.1916

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HMT	RN
12.02	2119.40	2112.16	7.23	76.73	0.0	2.68	0.0	2.065	2.065	87.5690	0.521881E+06

BETA	CL	CD	CM	CRULL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	2.80	0.34	-1.68	0.04	0.00	-0.05	1.74	-0.82	1.62	0.75

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.5655	-0.5994
2.5	-0.5994	-0.5655
5.0	-0.5655	-0.5655
10.0	-0.5655	-0.5994
15.0	-0.5315	-0.5655
25.0	-0.5655	-0.5994
35.0	-0.5655	-0.5994
50.0	-0.5994	
56.0	-0.5655	-0.5655
65.0	-0.5655	-0.6332
76.0	-0.5655	-0.5655
79.0	-0.5315	-0.5655
80.5	-0.5239	-0.5239
81.0	-0.5728	-0.5239
82.0	-0.5484	-0.5484
84.0	-0.5728	
87.0	-0.4993	
89.0	-0.5622	-0.5622
93.0	-0.5622	
96.0	-0.5622	-0.5622
100.0	-0.5985	-0.5260
96.0	-0.5985	-0.5622
84.0	-0.5985	-0.5260
73.0	-0.5985	-0.5622
50.0	-0.5622	-0.5260
35.0	-0.5655	-0.5655
25.0	-0.5994	-0.5655
10.0	-0.5994	-0.5994
5.0	-0.5655	-0.5655
2.5	-0.5994	-0.5315

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.2869	-6.3797	-3.9055	-2.2403	-1.7645
2.5	-0.4798	-2.7636	-3.6676	-2.2879	-1.7168
5.0	-1.3838	-2.0024	-3.5250	-2.0975	-1.7168
10.0	-1.1459	-1.4314	-3.0491	-2.1450	-1.6217
15.0	-1.0534	-1.2411	-2.6208	-2.3830	-1.7645
24.0	-0.9080	-1.0507	-1.3838	-2.1450	-1.6693
33.0	-0.7930	-1.0683	-0.9306	-2.1687	-1.6184
54.0	-0.9306	-0.9306	-1.0683	-1.2058	6.4977
65.0	-1.2142	-1.9387	-1.4677	-1.1418	-1.4315
78.5	11.7889	-3.5323	-5.1261	-1.8663	-1.2866
79.5	-25.0249	-37.9617	-5.8897	-2.0429	-1.3324
80.5	-23.3841	-36.4426	-4.2070	-1.6999	-0.5484
81.3	-23.5058	122.7427	-3.5620	-1.5529	-1.3813
82.0	-23.8251	-37.7903	-2.8025	-1.5039	-1.3324
84.0	-24.9033	-37.1530	-2.7289	-1.2833	-1.3079
87.0	-25.8087	-34.4341	-1.9204	-1.2833	-1.2833
89.0	-26.7693	-30.1891	-1.9817	-1.3921	-1.3038
93.0	-24.0677	-8.4954	-1.3627	-0.6554	-1.3627
96.0	-18.9884	-1.8932	-1.2448	-1.3038	-1.3038
100.0	-9.7628	-3.1606	-1.0680	-1.2448	-0.5963
96.0	0.4352	0.4647	0.2584	0.2289	0.0226
84.0	0.6710	0.8184	0.7594	0.6710	0.4352
73.5	0.5825	-0.5180	0.5825	0.7201	0.4450
54.0	0.3074	0.3074	0.3074	0.3074	0.1698
33.0	0.3074	0.3074	0.3074	0.3074	0.1698
24.0	0.3767	0.4243	0.3767	0.4718	0.1864
10.0	0.5194	0.5194	0.5670	0.6146	0.4450
5.0	0.6622	0.6146	0.5670	0.6622	0.4450
2.5	0.7573	0.5194	0.5194	0.5670	0.3074

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2001
43.4	-0.1638
50.4	-0.2001
56.1	-0.2363
62.5	-0.3449
69.7	-0.4536
76.9	-0.4173
83.4	-0.4173
89.4	-0.4536
95.4	-0.4173
101.1	-0.3812
---BOTTOM---	
38.2	0.0173
43.4	0.0536
50.4	-0.0914
56.1	0.0897
62.5	0.1984
69.7	0.2708
76.9	0.2708
83.4	0.2346
89.4	0.1984
95.4	-0.1638
101.1	-0.8158

MISC.*

NO.	CP
1	-0.9306
2	0.3074
3	-1.8936
4	0.3074
5	-1.4809
6	0.0323
7	0.1110
8	-1.0680
9	-0.8616
10	0.7005
11	0.1405
12	-0.4784

Force Data - Use
Run 223

TABLE A23. RUN 216, BC1W6V, DELF=45, CMU=0, (BN/B)W=0.5

RUN:SEQ			PROPLULSIVE		WING / CANARD		PRESSURES			
216 2	CLAERO =	0.5263	CDAERO =	0.2154	DCLC =	0.0	DWLC =	0.0		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	
-0.12	2146.27	2126.04	20.23	128.42	0.0	2.66	0.0	0.0	0.0	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR		
0.0	0.53	0.22	-0.20	0.00	-0.00	0.01	0.43	-0.16		

BASEPR = 8.1924

HGT	RN
86.9964	0.8[illegible]5035E+06
CLTR	CDTR
0.43	0.24

************ CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.4818	0.5304
2.5	-0.0883	-0.1975
5.0	-0.1369	-0.2946
10.0	-0.1612	-0.2946
15.0	-0.2582	-0.2946
25.0	-0.2461	-0.2703
35.0	-0.2339	-0.2582
50.0	-0.2339	
56.0	-0.2582	-0.2946
65.0	-0.3310	-0.3674
76.0	-0.2461	-0.5373
79.0	-0.4887	-0.6828
80.5	-0.4765	-0.6341
81.0	-0.4415	-0.6429
82.0	-0.4765	-0.6341
84.0	-0.2839	
87.0	-0.4765	
89.0	-0.4988	-0.6672
93.0	-0.4988	
96.0	-0.4988	0.6538
100.0	2.0266	-0.6542
96.0	0.3171	0.3430
84.0	2.0266	-0.6542
73.0	0.3171	0.3430
50.0	0.6927	0.6020
35.0	0.5546	0.5546
25.0	0.1664	-0.3310
10.0	0.0087	0.0209
5.0	-0.1126	-0.0156
2.5	-0.2218	-0.0641

*********************** WING ********************

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0004	-0.0004	0.3058	0.2548	-0.0004
2.5	-0.2047	0.2888	-0.1536	-0.4429	-0.6301
5.0	0.1697	0.1186	-0.1877	-0.3578	-0.5790
10.0	-0.2047	-0.0004	-0.2727	-0.3408	-0.4769
15.0	-0.0175	-0.0515	-0.2897	-0.3238	-0.4089
24.0	-0.0856	-0.1536	-0.2727	-0.2727	-0.3578
33.0	-0.1280	-0.1772	-0.3248	-0.3248	-0.3248
54.0	-0.2264	-0.2264	-0.3740	1.9382	-0.3740
65.0	-0.3045	-0.3175	-0.5117	-0.5636	-0.4988
78.5	-0.3304	-0.5765	-1.1981	-1.3404	-0.8873
79.5	-0.5028	-0.5115	-1.2472	-1.2034	-0.8180
80.5	-0.5028	-0.5290	-0.9844	-1.0720	-0.2050
81.3	-0.5203	76.9934	-0.7567	-1.1596	-0.8531
82.0	-0.5028	-1.3172	-0.6692	-1.1506	-0.8180
84.0	-0.5028	-0.5466	-0.6166	-1.1596	-0.8531
87.0	-0.5028	-0.5641	-0.6254	-1.0895	-0.8268
89.0	-0.5987	-0.6093	-0.6831	-1.0836	-0.8833
93.0	-0.5987	-0.5987	-0.7041	-0.2825	-0.9045
96.0	-0.6093	-0.5987	-0.7253	-0.8412	-0.8728
100.0	-0.5355	-0.5461	-0.6620	-0.7885	-0.2509
96.0	0.1813	0.2551	0.3184	0.3816	0.2551
84.0	0.3816	0.4659	0.5924	0.5713	0.5292
73.5	0.3640	-0.2264	0.5115	0.5606	0.5115
54.0	0.1672	0.1672	0.2164	0.2656	0.1672
33.0	0.0196	-0.2264	0.0688	0.0688	0.0688
24.0	-0.0345	-0.0004	0.0166	0.0875	0.0166
10.0	-0.0856	-0.1026	-0.0004	0.0506	0.0688
5.0	-0.1106	-0.2047	-0.0345	0.1186	0.1672
2.5	-0.1366	-0.3408	-0.0345	0.1697	0.2656

FUSELAGE

XLOC	CP
-----TOP-----	
38.2	-0.1620
43.4	-0.1620
50.4	-0.1361
56.1	-0.0973
62.5	-0.0714
69.7	-0.0844
76.9	-0.0973
83.4	-0.0844
89.4	-0.1620
95.4	-0.2130
101.1	-0.1679
---BOTTOM---	
38.2	0.0063
43.4	0.0192
50.4	-0.0326
56.1	0.0192
62.5	0.0063
69.7	-0.0067
76.9	0.0451
83.4	0.0969
89.4	0.1487
95.4	0.0710
101.1	-0.2916

MISC.*

NO.	CP
1	-0.2264
2	0.0688
3	-0.3248
4	0.1180
5	-0.2756
6	0.0196
7	0.4764
8	-0.3863
9	-0.0822
10	0.6029
11	0.2024
12	-0.0717

TABLE A23. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 493

216 4 CLAERO = 0.7596 CDAERO = 0.2796 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1924

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
3.98	2146.20	2125.86	20.34	128.86	0.0	2.64	0.0	0.0	0.0	86.4115	0.866091E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	0.76	0.28	-0.17	0.00	-0.00	0.01	0.68	-0.13	0.67	0.30

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	0.0690	-0.8359
2.5	-0.7152	-1.1736
5.0	-0.5825	-0.8117
10.0	-0.1482	-0.6670
15.0	-0.5222	-0.5945
25.0	-0.4015	-0.4257
35.0	-0.3412	-0.3774
50.0	-0.3291	
56.0	-0.3412	-0.3654
65.0	-0.4136	-0.4619
76.0	-0.2447	-0.6086
79.0	-0.5463	-0.7031
80.5	-0.4826	-0.7177
81.0	-0.5174	-0.7003
82.0	-0.5261	-0.7090
84.0	-0.3345	
87.0	-0.5436	
89.0	-0.5475	-0.7279
93.0	-0.5347	
96.0	-0.5475	0.6631
100.0	1.6419	-0.7279
96.0	0.3154	0.3154
84.0	1.6419	-0.7279
73.0	0.3154	0.3154
50.0	0.7661	0.6244
35.0	0.6239	0.5636
25.0	0.1896	-0.4257
10.0	0.1293	0.1414
5.0	-0.0034	0.1031
2.5	0.2620	0.3947

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.1518	0.2365	-0.3220	-1.2189	-2.1326
2.5	-0.1066	-0.0174	-0.7450	-0.9600	-1.9634
5.0	0.0334	-0.1189	-0.5031	-0.7958	-1.1173
10.0	-0.2204	-0.2035	-0.5081	-0.6097	-0.8127
15.0	-0.1527	-0.2374	-0.4743	-0.5251	-0.6774
24.0	-0.2035	-0.2712	-0.4235	-0.4404	-0.5251
33.0	-0.2740	-0.3230	-0.4697	-0.4697	-0.5186
54.0	-0.3230	-0.3719	-0.4697	1.3893	-0.5186
65.0	-0.3415	-0.3801	-0.5991	-0.6635	-0.6506
78.5	-0.3543	-0.6377	-1.1786	-1.0027	-1.0627
79.5	-0.5523	-0.5523	-1.1619	-1.1189	-1.0400
80.5	-0.5348	-0.5523	-0.9180	-1.4784	-0.2300
81.3	-0.5523	74.2749	-0.7177	-1.4319	-1.0312
82.0	-0.5261	-1.2402	-0.6829	-1.4406	-1.0661
84.0	-0.5523	-0.6045	-0.6568	-1.5973	-1.0661
87.0	-0.5784	-0.5958	-0.6568	-1.3970	-1.0661
89.0	-0.6374	-0.6688	-0.7212	-1.3187	-1.1824
93.0	-0.6583	-0.6478	-0.7422	-0.2810	-1.1720
96.0	-0.6583	-0.6269	-0.7946	-0.8260	-1.1720
100.0	-0.6164	-0.5640	-0.7002	-0.7841	-0.2600
96.0	0.2327	0.3270	0.3270	0.3690	0.1698
84.0	0.4423	0.5786	0.5996	0.5891	0.4738
73.5	0.4109	-0.2740	0.5576	0.5087	0.4109
54.0	0.1662	0.2152	0.2152	0.2641	0.1173
33.0	0.0684	-0.2740	0.1173	0.1173	0.0684
24.0	0.0503	0.0503	0.1180	0.1688	0.1011
10.0	0.0165	0.0165	0.1518	0.2703	0.2152
5.0	-0.0004	-0.0004	0.2026	0.3718	0.2641
2.5	-0.0681	-0.0343	0.3041	0.4395	0.4109

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1996
43.4	-0.2127
50.4	-0.1611
56.1	-0.1226
62.5	-0.0967
69.7	-0.1734
76.9	-0.1483
83.4	-0.1229
89.4	-0.1996
95.4	-0.2384
101.1	-0.2127
---BOTTOM---	
38.2	0.0320
43.4	0.0449
50.4	-0.0324
56.1	0.0320
62.5	0.0578
69.7	0.0578
76.9	0.0964
83.4	0.1350
89.4	0.1866
95.4	0.0964
101.1	-0.3028

MISC.

NO.	CP
1	-0.4208
2	0.1173
3	-0.5186
4	0.1173
5	-0.4697
6	-0.0285
7	0.4214
8	-0.4906
9	-0.4068
10	0.5472
11	0.3505
12	-0.1237

TABLE A23. (Concluded)

RUN:SEQ 216 7 — PROPULSIVE WING / CANARD PRESSURES — PAGE 495

CLAERO = 1.2267 CDAERO = 0.4785 DCLC = 0.0 DWLC = 0.0 BASEPR = 0.1924

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.99	2146.20	2125.97	20.23	128.57	0.0	2.68	0.0	0.0	0.0	57.6630	0.862648E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	1.23	0.48	-0.08	0.00	-0.00	0.01	1.21	-0.04	1.13	0.48

********** CANARD ******

%X/C	BP=2.375 CP	BP=10.125 CP
0.0	-5.6256	-1.7068
2.5	-2.5682	-1.5491
5.0	-1.7311	-1.5370
10.0	-0.1782	-1.4521
15.0	-1.1124	-1.4278
25.0	-0.8091	-1.4521
35.0	-0.6878	-1.3793
50.0	-0.5543	
56.0	-0.5422	-0.9668
65.0	-0.5786	-0.8334
76.0	-0.6238	-0.6576
79.0	-0.6635	-0.9061
80.5	-0.6654	-0.9193
81.0	-0.6478	-0.9193
82.0	-0.6391	-0.9281
84.0	-0.3939	
87.0	-0.6654	
89.0	-0.6980	-0.9440
93.0	-0.6980	
96.0	-0.6980	0.7007
100.0	1.2576	-0.8922
96.0	0.3122	0.2733
84.0	1.2576	-0.8922
73.0	0.3122	0.2733
50.0	0.7784	0.6489
35.0	0.6831	0.5861
25.0	0.3192	-0.6029
10.0	0.3192	0.2706
5.0	0.2585	0.2828
2.5	0.6103	0.5739

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0286	-0.8903	-4.3448	-3.4768	-1.7411
2.5	-0.3287	-0.8052	-2.3877	-3.4598	-1.6560
5.0	-0.4138	-0.7031	-1.2817	-3.8682	-1.6220
10.0	-0.2947	-0.6861	-1.0264	-2.1155	-1.6050
15.0	-0.4819	-0.6520	-0.9073	-0.7031	-1.6901
24.0	-0.4819	-0.6010	-0.7371	-0.6690	-1.6731
33.0	-0.4773	-0.5265	-0.6741	-0.6249	-1.6580
54.0	-0.4773	-0.5265	-0.5757	0.8510	-1.2102
65.0	-0.4908	-0.5296	-0.6980	-0.7886	-0.3311
78.5	-0.4390	-0.6980	-1.0365	-1.7341	-1.2678
79.5	-0.5953	-0.6303	-0.9894	-1.5849	-1.4183
80.5	-0.5778	-0.5690	-0.8142	-1.3309	-0.2829
81.3	-0.5953	63.9675	-0.7004	-1.3747	-1.2784
82.0	-0.5953	-1.2063	-0.6741	-1.4536	-1.2784
84.0	-0.6303	-0.6303	-0.6654	-1.4711	-1.2063
87.0	-0.6478	-0.6478	-0.7004	-1.4273	-1.2063
89.0	-0.7513	-0.7513	-0.7829	-1.2994	-1.2678
93.0	-0.7513	-0.7091	-0.8145	-0.3718	-1.2678
96.0	-0.7618	-0.6880	-0.8251	-0.3516	-1.2046
100.0	-0.6669	-0.6037	-0.6459	-0.8989	-0.3402
96.0	0.3028	0.3977	0.3766	0.3661	0.1764
84.0	0.5769	0.7139	0.6402	0.5684	0.3450
73.5	0.5066	-0.2806	0.6050	0.5558	0.3590
54.0	0.2606	0.3098	0.3590	0.3098	0.1622
33.0	0.2114	-0.2806	0.2606	0.2606	0.1622
24.0	0.1818	0.2158	0.3009	0.3349	0.2158
10.0	0.1988	0.2838	0.3860	0.4710	0.4082
5.0	0.2158	0.3179	0.4540	0.5051	0.4574
2.5	0.2498	0.3690	0.4370	0.4540	0.4082

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3483
43.4	-0.3124
50.4	-0.2318
56.1	-0.1799
62.5	-0.1929
69.7	-0.2447
76.9	-0.2577
83.4	-0.2058
89.4	-0.3977
95.4	-0.2706
101.1	-0.2058
---BOTTOM---	
38.2	0.0920
43.4	0.0920
50.4	-0.0116
56.1	0.0790
62.5	0.1179
69.7	0.1697
76.9	0.1826
83.4	0.2216
89.4	0.2474
95.4	0.1179
101.1	-0.3613

MISC.

NO.	CP
1	-0.5757
2	0.2606
3	-1.0184
4	0.2606
5	-1.3628
6	0.0839
7	0.1242
8	-0.6142
9	0.4083
10	-0.0766
11	-0.8461
12	-0.3718

TABLE A24. RUN 217, BC1W6V, DELF=45, CMU=1.0, (BN/B)W=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
217	1	CLAERO =	0.5873	CDAERO =	0.7072	DCLC =	0.5351	DWLC =	0.9015	BASEPR =	8.1924	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.03		2146.70	2138.78	7.91	80.19	0.0	2.65	0.653	1.041	1.695	86.8000	0.541223E+06
BETA		CL	CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
0.0		2.02	-0.19	-0.44	-0.01		0.01	0.06	0.93	-0.10	0.93	0.46

********** CANARD ******

%X/C	BP=3:375 CP	BP=10.125 CP
0.0	0.1920	-1.2969
2.5	-0.7696	-2.0416
5.0	-0.6766	-1.1109
10.0	-0.2113	-1.0799
15.0	-0.7386	-1.0178
25.0	-0.6456	-0.8937
35.0	-0.7076	-0.8937
50.0	-0.8937	
56.0	-1.0486	-1.2350
65.0	-1.4831	-1.8863
76.0	-0.4284	-4.4612
79.0	-6.6637	-7.8426
80.5	-13.3559	-28.4259
81.0	-7.8916	-15.8638
82.0	-5.6971	-9.5711
84.0	-1.2406	
87.0	-1.5543	
89.0	-2.1474	-3.4390
93.0	-1.0877	
96.0	-0.0611	0.7005
100.0	1.2304	-1.0546
96.0	0.5680	0.5018
84.0	1.2304	-1.0546
73.0	0.5680	0.5018
50.0	0.7658	0.6674
35.0	0.6884	0.5953
25.0	0.2541	-1.4211
10.0	0.2230	0.1920
5.0	0.0369	0.1300
2.5	0.0990	0.4091

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8290	-0.8290	-0.8725	0.6940	-0.5244
2.5	-0.3938	0.9116	0.2153	-0.8290	-1.2641
5.0	0.7375	0.7375	-0.0457	-0.6113	-1.0029
10.0	-0.3503	0.3469	-0.2198	-0.6549	-0.7854
15.0	0.3894	0.2153	-0.3068	-0.5678	-0.6549
24.0	0.0848	-0.0457	-0.3938	-0.4809	-0.5678
33.0	-0.0078	-0.2594	-0.3852	-0.6367	-0.5110
54.0	-0.5110	-0.6367	-0.7625	0.4954	-0.6367
65.0	-0.8890	-1.2865	-0.9884	-0.9884	-0.8559
78.5	-0.3923	-4.6311	-3.5052	-1.4189	-1.4520
79.5	********	-115.8499	-5.2940	-1.1288	-1.6438
80.5	-57.2916	-38.1224	-3.4130	-1.1064	-0.3675
81.3	-20.1414	170.9370	-2.8531	-1.1288	-1.4199
82.0	-6.8167	-1.1064	-2.7635	-1.0841	-1.4423
84.0	-1.4647	-5.8538	-2.5844	-1.1512	-1.4199
87.0	-0.5690	-4.5998	-1.5094	-1.1288	-1.3975
89.0	-2.0742	-2.5133	-1.2656	-1.3734	-1.4812
93.0	-0.8074	-1.0769	-1.1338	-0.4570	-1.5620
96.0	-1.0230	-1.0230	-1.0499	-1.2387	-1.6160
100.0	-1.2387	-1.5620	-0.9152	-1.2117	-0.4030
96.0	0.0552	0.0282	0.1899	0.4056	0.2169
84.0	0.0552	-0.0526	0.4056	0.8368	0.5943
73.5	-0.1336	-0.3852	0.1180	0.4954	0.4954
54.0	-0.5110	-0.5110	-0.3852	-0.1336	-0.0078
33.0	-0.7625	-0.3852	-0.6367	-0.3852	-0.1336
24.0	-0.6984	-0.7854	-0.6984	-0.3938	-0.2198
10.0	-0.6113	-0.6984	-0.7854	-0.3068	-0.0078
5.0	-0.6549	-0.6984	-0.8290	-0.3068	0.1180
2.5	-0.6984	-0.7854	-0.9159	-0.2198	0.2438

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4585
43.4	-0.4585
50.4	-0.2929
56.1	-0.0942
62.5	-0.0611
69.7	-0.0942
76.9	-0.1606
83.4	-0.1936
89.4	-0.2598
95.4	-0.3592
101.1	-0.2929
---BOTTOM---	
38.2	0.1375
43.4	0.1707
50.4	0.0051
56.1	-0.1936
62.5	-0.5910
69.7	-0.6903
76.9	-0.5910
83.4	-0.4254
89.4	-0.1936
95.4	-0.1274
101.1	-0.4585

MISC.*

NO.	CP
1	-0.2594
2	-0.7625
3	-0.6367
4	-0.1336
5	-0.5110
6	-0.1336
7	0.2708
8	-0.7285
9	-0.4300
10	0.8368
11	0.1830
12	-0.1604

TABLE A24. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ														
217 4	CLAERO =	0.6140	CDAERO =	0.6680	DCLC =	0.5436	DWLC =	0.9192						PAGE 500
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	BASEPR =	8.1934			
4.09	2146.84	2128.93	7.91	80.16	0.0	2.67	0.634	1.022	1.655	HQT	RN			
BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CNTR	CNTR	87.4453	0.541883E+06			
0.0	2.08	-0.10	-0.25	0.02	0.00	0.03	0.97	0.09	0.94	CLTR	CDTR			
										0.94	0.47			

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-1.2969	-3.7783
2.5	-1.5759	-3.2508
5.0	-1.1418	-3.2508
10.0	-0.2112	-2.4415
15.0	-0.9556	-1.5139
25.0	-0.8936	-1.0486
35.0	-0.8936	-1.0797
50.0	-1.0486	
56.0	-1.1418	-1.3899
65.0	-1.5759	-2.0412
76.0	-0.3663	-4.5846
79.0	-6.8490	-8.0587
80.5	-13.4882	-28.6687
81.0	-8.1816	-15.9288
82.0	-5.9426	-9.8610
84.0	-1.3301	
87.0	-1.6659	
89.0	-2.2133	-3.7033
93.0	-1.1207	
96.0	-0.0611	0.7998
100.0	1.7269	-1.0876
96.0	0.6011	0.4687
84.0	1.7269	-1.0876
73.0	0.6011	0.4687
50.0	0.8329	0.7006
35.0	0.7193	0.5953
25.0	0.2230	-1.5139
10.0	0.3161	0.2230
5.0	0.1920	0.2230
2.5	0.4402	0.5332

********** WING ********** ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8723	-0.6983	-0.9158	-0.6548	-2.0470
2.5	-0.3067	0.7810	-0.9158	-0.7418	-1.9599
5.0	0.6939	0.6504	-0.9158	-0.6548	-1.2639
10.0	-0.3502	0.3459	-0.8288	-0.6983	-0.9158
15.0	0.3024	0.1718	-0.7418	-0.6983	-0.8288
24.0	-0.0022	-0.1327	-0.7418	-0.6983	-0.7852
33.0	-0.1335	-0.2593	-0.8883	-0.8883	-0.6366
54.0	-0.5109	-0.5109	-0.8263	0.5985	-0.6366
65.0	-0.8558	-1.2531	-0.9882	-1.2300	-0.8558
78.5	-0.1604	-4.6636	-3.4385	-1.8159	-1.4518
79.5	-89.7144	-115.4320	-5.9202	-1.3973	-1.5764
80.5	-58.7846	-41.6552	-4.0617	-1.2406	-0.3449
81.3	-21.0108	142.4148	-3.2656	-1.2406	-1.4421
82.0	-6.8383	-0.7256	-2.9871	-1.2630	-1.3750
84.0	-1.4421	-5.9650	-2.4049	-1.3077	-1.3750
87.0	-0.8152	-4.8902	-1.9571	-1.3973	-1.3750
89.0	-2.2086	-2.7746	-1.6427	-1.5079	-1.4271
93.0	-0.9151	-1.1576	-1.1036	-0.4029	-1.4810
96.0	-1.2116	-1.0767	-0.8881	-1.2385	-1.4810
100.0	-1.0767	-1.1307	-0.5647	-0.9689	-0.3760
96.0	-0.0795	-0.1065	0.1630	0.4325	0.2708
84.0	0.0283	-0.1065	0.3516	0.8098	0.7020
73.5	-0.1335	-0.3851	-0.0077	0.4954	0.4954
54.0	-0.5109	-0.6366	-0.5109	-0.2593	-0.0077
33.0	-0.7624	-0.2593	-0.8883	-0.6366	-0.1335
24.0	-0.6548	-0.6983	-0.7418	-0.5677	-0.1327
10.0	-0.6112	-0.7418	-0.7852	-0.5677	-0.0077
5.0	-0.5677	-0.6983	-0.7418	-0.5677	0.1180
2.5	-0.6548	-0.6548	-0.7418	-0.5243	0.3696

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4915
43.4	-0.4584
50.4	-0.2598
56.1	-0.0942
62.5	-0.0611
69.7	-0.0611
76.9	-0.1273
83.4	-0.1604
89.4	-0.2598
95.4	-0.3259
101.1	-0.2266
---BOTTOM---	
38.2	0.1707
43.4	0.2369
50.4	0.0714
56.1	-0.1273
62.5	-0.4915
69.7	-0.6571
76.9	-0.6240
83.4	-0.4253
90.4	-0.1604
95.4	-0.1604
101.1	-0.5246

MISC.*

NO.	CP
1	-0.3651
2	-0.8883
3	-0.8883
4	-0.2593
5	-0.6366
6	-0.2593
7	0.3247
8	-0.6455
9	-0.4838
10	0.7290
11	0.2978
12	-0.1604

TABLE A24. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 217 6 CLAERO = 1.2688 CDAERO = 1.0045 DCLC = 0.6058 DWLC = 1.0191 BASEPR = 8.1924

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.98	2146.70	2139.12	7.57	78.41	0.0	2.66	0.658	1.072	1.730	87.1714	0.530367E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	2.89	0.42	-0.23	0.02	-0.01	0.02	1.77	0.13	1.63	0.85

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-7.5192	-3.4361
2.5	-8.0052	-3.6953
5.0	-3.4361	-3.5981
10.0	-0.3251	-3.8573
15.0	-1.6861	-4.0842
25.0	-1.3946	-4.6999
35.0	-1.2649	-3.8573
50.0	-1.3621	
56.0	-1.4918	-1.5566
65.0	-1.9454	-1.6214
76.0	-0.5843	-4.5054
79.0	-7.6812	-8.6858
80.5	-14.6464	-31.6987
81.0	-8.8915	-17.9673
82.0	-6.6926	-11.5349
84.0	-1.5930	
87.0	-2.0843	
89.0	-2.6288	-4.8428
93.0	-1.4872	
96.0	-0.3110	0.8997
100.0	2.1797	-1.1759
96.0	0.6576	0.4154
84.0	2.1797	-1.1759
73.0	0.6576	0.4154
50.0	0.8305	0.6576
35.0	0.8091	0.5174
25.0	0.3878	-1.9130
10.0	0.4526	0.3230
5.0	0.3878	0.4202
2.5	0.5498	0.2258

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.6782	-0.4965	-1.5419	-1.5873	-4.9055
2.5	-0.4965	0.3672	-1.3601	-1.3147	-3.6781
5.0	0.5944	0.2762	-1.4509	-1.1783	-3.1328
10.0	-0.4965	0.0035	-1.6328	-1.1329	-1.8600
15.0	0.0944	-0.1783	-1.5873	-1.0873	-1.2237
24.0	-0.0874	-0.3601	-1.5419	-1.1329	-0.9964
33.0	-0.3105	-0.4420	-1.4932	-1.3618	-0.9677
54.0	-0.5734	-0.7047	-1.3618	1.3978	-0.9677
65.0	-1.0029	-1.4872	-1.6256	-1.6948	-1.2451
78.5	-0.4148	-4.4622	-4.0471	-3.0785	-1.7640
79.5	-98.6954	-120.4735	-6.9499	-3.9090	-1.8737
80.5	-64.5659	-51.0683	-4.6809	-2.9733	-0.5170
81.3	-30.1551	139.2278	-3.8621	-2.3182	-1.6866
82.0	-14.9501	0.5590	-3.4879	-2.3182	-1.6164
84.0	-0.2363	-5.9207	-2.8094	-1.8971	-1.7334
87.0	-1.0784	-5.8973	-2.2247	-1.6866	-1.7802
89.0	-2.6569	-3.3608	-2.0092	-1.7840	-1.8403
93.0	-1.4462	-1.4462	-1.4179	-0.3577	-1.8122
96.0	-1.6713	-1.5306	-1.1083	-1.3336	-1.6713
100.0	-0.8830	-1.3053	-0.7140	-0.9674	-0.5732
96.0	0.3278	0.1583	0.2151	0.4122	0.2714
84.0	0.1870	0.2433	0.2714	0.6938	0.7220
73.5	0.0837	-0.5734	0.3465	0.4779	0.7407
54.0	-0.3105	-0.3105	-0.1791	0.0837	0.2151
33.0	-0.5734	-0.4420	-0.5734	-0.3105	0.0837
24.0	-0.5874	-0.6328	-0.6328	-0.1783	0.1399
10.0	-0.4965	-0.5419	-0.8600	-0.3601	0.3465
5.0	-0.4965	-0.5419	-0.8600	-0.4510	0.4779
2.5	-0.5419	-0.5419	-0.9510	-0.3601	0.4779

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.6916
43.4	-0.6916
50.4	-0.3802
56.1	-0.2765
62.5	-0.1381
69.7	-0.1727
76.9	-0.2418
83.4	-0.2765
89.4	-0.3456
95.4	-0.4148
101.1	-0.3110
---BOTTOM---	
38.2	0.2425
43.4	0.2770
50.4	0.1041
56.1	0.0003
62.5	-0.3456
69.7	-0.5186
76.9	-0.4148
83.4	-0.2073
89.4	0.0695
95.4	-0.0343
101.1	-0.5878

MISC.*

NO.	CP
1	-0.9677
2	-0.5734
3	-0.9677
4	0.0837
5	-1.0990
6	0.0837
7	-0.2072
8	-1.7277
9	-1.6151
10	0.5812
11	0.0743
12	-0.7423

TABLE A25. RUN 218, BC1W6V, DELF=45,CMU=2.0, (BN/B)W=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
218 1	CLAERO =	0.5608	CDAERO =	0.9208	DCLC =	0.9986	DWLC =	1.7329	BASEPR =	6.1924		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
-0.07	2146.98	2142.91	4.07	57.45	0.0	2.66	1.220	2.002	3.222	86.9677	0.388510E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.0	3.29	-0.78	-0.84	0.00	0.01	0.03	1.16	-0.17	1.16	0.49		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1384	-3.2750
2.5	-0.9829	-2.6116
5.0	-0.8622	-1.6463
10.0	-0.2591	-1.4051
15.0	-0.8622	-1.2845
25.0	-0.9019	-1.1035
35.0	-0.8622	-1.0432
50.0	-1.1036	
56.0	-1.2845	-1.4654
65.0	-1.8275	-2.3100
76.0	-0.4400	-5.6880
79.0	-8.5834	-10.3328
80.5	-21.6353	-47.7190
81.0	-12.4909	-25.7275
82.0	-8.8332	-14.8860
84.0	-1.8661	
87.0	-1.5613	
89.0	-2.9203	-4.5300
93.0	-1.2460	
96.0	0.4926	0.7502
100.0	5.9662	-2.1475
96.0	0.4282	0.4282
84.0	5.9662	-2.1475
73.0	0.4282	0.4282
50.0	0.7502	0.6858
35.0	0.7062	0.5856
25.0	0.1029	-1.5862
10.0	0.2236	0.1029
5.0	-0.0177	0.0425
2.5	0.1632	0.4045

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.3613	-1.1075	-0.3460	1.0076	-0.1768
2.5	-0.4306	1.2616	0.5001	-0.3460	-0.8536
5.0	0.8386	0.9232	0.2463	-0.3460	-0.8536
10.0	-0.5152	0.5001	-0.0922	-0.3460	-0.5998
15.0	0.3309	0.3309	-0.2614	-0.3460	-0.5998
24.0	0.1617	-0.0076	-0.3460	-0.3460	-0.5153
33.0	-0.2153	-0.4599	-0.7045	-0.7045	-0.9491
54.0	-0.7045	-0.9491	-0.9491	4.1876	-0.9491
65.0	-1.1816	-1.8899	-1.1816	-1.1816	-1.1816
78.5	1.2010	-6.9128	-3.8863	-1.7611	-2.5339
79.5	*********	-221.9840	-5.3933	-1.4307	-2.5194
80.5	*********	-61.6089	-3.6951	-1.4743	-0.5598
81.3	-25.1622	285.2717	-3.1290	-1.3436	-2.3452
82.0	-3.4338	-1.1259	-2.9982	-1.3872	-2.0404
84.0	-3.8691	-10.5314	-2.6499	-1.3872	-2.2146
87.0	-0.4728	-8.1366	-1.6921	-1.4307	-2.3016
89.0	-3.6046	-4.3383	-1.4556	-1.5606	-2.2943
93.0	-1.4556	-1.6129	-1.2460	-0.4598	-2.2419
96.0	-1.9799	-1.4556	-1.1411	-1.4032	-1.9799
100.0	-2.9766	-3.1329	-0.9839	-1.3506	-0.4074
96.0	-0.0405	0.0643	0.5885	0.5361	0.2216
84.0	0.2740	0.4836	0.8505	0.9030	0.5885
73.5	-0.2153	-0.7045	0.5185	0.5185	0.5185
54.0	-0.7045	-0.7045	-0.4599	-0.2153	0.0293
33.0	-1.4383	-0.7045	-0.9491	-0.4599	-0.4599
24.0	-1.1921	-1.3613	-0.8536	-0.2614	-0.1768
10.0	-1.0229	-1.2767	-1.1921	-0.2614	-0.2153
5.0	-0.9382	-1.1075	-1.4459	-0.2614	0.0293
2.5	-1.1075	-1.2767	-1.9536	-0.2614	0.0293

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4733
43.4	-0.6021
50.4	-0.3445
56.1	-0.1513
62.5	-0.0869
69.7	-0.0869
76.9	-0.1513
83.4	-0.2157
89.4	-0.3445
95.4	-0.4089
101.1	-0.2801
---BOTTOM---	
38.2	0.0418
43.4	0.0418
50.4	-0.0869
56.1	-0.5377
62.5	-1.0628
69.7	-1.1172
76.9	-0.7308
83.4	-0.3445
89.4	-0.0225
95.4	-0.2157
101.1	-0.7308

MISC.*

NO.	CP
1	-0.7045
2	-0.9491
3	-0.9491
4	-0.4599
5	-0.7045
6	-0.2153
7	0.3264
8	-0.6695
9	-0.5122
10	0.8505
11	0.3264
12	-0.0929

TABLE A25. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 507

RUN:SEQ												
218 3	CLAERO =	0.4739	CDAERO =	0.9202	DCLC =	1.0782	DWLC =	1.8600	BASEPR =	8.1924		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.06	2147.33	2143.38	3.96	56.62	0.0	2.66	1.257	2.068	3.325	87.0961	0.383628E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.0	3.41	-0.63	-0.66	-0.02	0.02	0.03	1.16	0.03	1.12	0.52		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-2.4881	-3.7909
2.5	-2.0538	-3.2946
5.0	-1.4335	-3.7287
10.0	-0.3169	-3.1705
15.0	-1.1854	-2.9223
25.0	-0.9992	-1.3715
35.0	-1.0613	-1.3094
50.0	-1.3094	
56.0	-1.4335	-1.7436
65.0	-1.9297	-2.5502
76.0	-0.5030	-5.7759
79.0	-8.8155	-10.4906
80.5	-21.3523	-48.6767
81.0	-12.5379	-26.0180
82.0	-8.7763	-14.8664
84.0	-1.6114	
87.0	-1.3875	
89.0	-2.8551	-4.8416
93.0	-1.1332	
96.0	0.5885	0.7872
100.0	7.0782	-2.2590
96.0	0.3898	0.3898
84.0	7.0782	-2.2590
73.0	0.3898	0.3898
50.0	0.6547	0.7209
35.0	0.7376	0.4895
25.0	0.1794	-1.8677
10.0	0.3034	0.0554
5.0	0.1174	0.1174
2.5	0.3655	0.3655

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.0154	-0.8414	-1.0154	-0.0583	-1.7985
2.5	-0.4064	0.9859	0.0287	-0.5804	-1.8857
5.0	0.6988	0.8988	-0.1453	-0.4934	-1.3635
10.0	-0.3194	0.5508	-0.1453	-0.4934	-1.0154
15.0	0.5508	0.2897	-0.2323	-0.4934	-0.8414
24.0	0.2027	-0.0583	-0.4934	-0.5804	-0.6674
33.0	-0.5235	-0.5235	-1.0266	-1.0266	-1.0266
54.0	-0.7750	-1.0266	-1.0266	4.5077	-1.0266
65.0	-1.0008	-1.7955	-1.1332	-1.2657	-1.0671
78.5	1.6480	-6.6930	-3.9809	-1.6630	-1.7955
79.5	********	-228.7833	-6.0446	-1.3875	-2.0144
80.5	********	-68.2439	-4.2534	-1.3875	-0.6711
81.3	-23.0166	281.0090	-3.4026	-1.4323	-1.7009
82.0	-0.1337	-0.0441	-3.2236	-1.3875	-1.7009
84.0	-3.7610	-10.2988	-2.7310	-1.4323	-1.7906
87.0	-0.2232	-8.1942	-1.8354	-1.4771	-1.7906
89.0	-3.3801	-4.3502	-1.6013	-1.7092	-1.8710
93.0	-1.0624	-1.4397	-1.3319	-0.5773	-1.9249
96.0	-1.6554	-1.3319	-1.1163	-1.4397	-1.9787
100.0	-2.4638	-2.5717	-1.0624	-1.3858	-0.4156
96.0	-0.4695	-0.4695	-0.1461	0.3390	0.2851
84.0	-0.2000	-0.4156	0.1234	0.8242	0.8242
73.5	-0.7750	-0.5235	-0.5235	0.2312	0.4827
54.0	-1.2781	-1.2781	-1.0266	-0.7750	-0.2719
33.0	-1.2781	-0.7750	-1.2781	-1.0266	-0.5235
24.0	-0.7544	-1.0154	-0.9284	-0.6674	-0.1453
10.0	-0.7544	-0.9284	-0.9284	-0.5804	-0.5235
5.0	-0.6674	-0.8414	-1.0154	-0.7544	-0.0204
2.5	-0.7544	-0.8414	-0.9284	-0.7544	0.2312

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.6035	38.2	0.1250
43.4	-0.6035	43.4	0.1912
50.4	-0.4048	50.4	-0.0737
56.1	-0.2062	56.1	-0.4711
62.5	-0.0737	62.5	-1.0008
69.7	-0.2062	69.7	-1.1995
76.9	-0.2062	76.9	-0.9346
83.4	-0.2724	83.4	-0.7359
89.4	-0.3386	89.4	-0.4048
95.4	-0.3386	95.4	-0.3386
101.1	-0.3386	101.1	-0.6697

MISC.*

NO.	CP
1	-0.7750
2	-1.5297
3	-1.0266
4	-0.7750
5	-1.0266
6	-0.5235
7	0.1774
8	-0.7929
9	-0.6312
10	0.6625
11	0.2851
12	-0.3078

TABLE A25. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 509

218 5 CLAERO = 0.8733 CDAERO = 1.0941 DCLC = 1.1664 DWLC = 1.9785 BASEPR = 8.1924

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.02	2147.26	2143.31	3.96	56.61	0.0	2.68	1.267	2.060	3.347	87.6580	0.383861E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.0	4.02	-0.04	-0.46	0.03	-0.00	0.01	1.70	0.24	1.57	0.79

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-7.4255	-3.9516
2.5	-7.3636	-4.0757
5.0	-7.2395	-4.0757
10.0	-0.2917	-4.3240
15.0	-1.6565	-4.3859
25.0	-1.5323	-4.9443
35.0	-1.4083	-4.5100
50.0	-1.5323	
56.0	-1.7806	-2.0907
65.0	-2.3387.	-2.0907
76.0	-0.6018	-5.6265
79.0	-9.7209	-10.3413
80.5	-22.8121	-48.9194
81.0	-13.3187	-26.8876
82.0	-9.4674	-16.2741
84.0	-2.0341	
87.0	-2.0341	
89.0	-3.4259	-6.0747
93.0	-1.6377	
96.0	0.2164	0.8124
100.0	7.7657	-2.3662
96.0	0.4151	0.2826
84.0	7.7657	-2.3662
73.0	0.4151	0.2826
50.0	0.7462	0.7462
35.0	0.6249	0.4528
25.0	0.3287	-2.2149
10.0	0.5147	0.2666
5.0	0.3907	0.3287
2.5	0.5147	0.2046

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.2512	-0.9902	-1.6863	-1.4252	-5.4279
2.5	-0.5552	0.4020	-1.5993	-1.2512	-3.6875
5.0	0.4020	0.3150	-1.7732	-1.2512	-2.4694
10.0	-0.7291	0.0539	-1.7732	-1.1642	-1.5993
15.0	0.1410	-0.2071	-1.6863	-1.2512	-1.3382
24.0	-0.2071	-0.4682	-1.5123	-1.3382	-1.2512
33.0	-0.2467	-0.4983	-1.5045	-1.5045	-1.0013
54.0	-0.7498	-0.7498	-1.2529	5.2875	-0.7498
65.0	-1.1743	-1.9027	-1.7040	-2.1013	-1.5715
78.5	1.5408	-5.3482	-4.2869	-3.4259	-2.4325
79.5	*********	-229.9664	-7.8555	-4.6313	-2.6162
80.5	*********	-76.2797	-5.4822	-3.6460	-0.6458
81.3	-31.5437	281.6157	-4.3179	-2.8849	-2.4370
82.0	-7.2734	1.4141	-3.9595	-2.7952	-2.2578
84.0	-2.4370	-9.3779	-3.2879	-2.1683	-2.3475
87.0	-0.8249	-8.7511	-2.2131	-1.7654	-2.3923
89.0	-3.8940	-4.4867	-2.0614	-1.7917	-2.4386
93.0	-1.6840	-1.7378	-1.4684	-0.6060	-2.1153
96.0	-2.1691	-1.8458	-1.0371	-1.1449	-1.7378
100.0	-1.8458	-2.2229	-0.6599	-0.8215	-0.5520
96.0	-0.1208	-0.1748	0.3104	0.5799	0.3643
84.0	0.1487	-0.0670	0.4182	0.8494	0.9033
73.5	-0.2467	-0.4983	0.2564	0.7595	0.7595
54.0	-0.7498	-0.7498	-0.4983	-0.2467	0.2564
33.0	-1.2529	-0.4983	-1.2529	-0.4983	0.0048
24.0	-1.1642	-1.1642	-1.2512	-0.6422	-0.0331
10.0	-0.9902	-1.1642	-1.2512	-0.8162	0.2564
5.0	-0.9032	-1.0773	-1.3382	-0.9902	0.2564
2.5	-0.9902	-1.0773	-1.3382	-1.1642	0.5080

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.9094
43.4	-0.8432
50.4	-0.5782
56.1	-0.3796
62.5	-0.2472
69.7	-0.2472
76.9	-0.3134
83.4	-0.3796
89.4	-0.5120
95.4	-0.4458
101.1	-0.3796
---BOTTOM---	
38.2	0.1502
43.4	0.2164
50.4	-0.1147
56.1	-0.3796
62.5	-0.9094
69.7	-1.0418
76.9	-0.9756
83.4	-0.6445
89.4	-0.1809
95.4	-0.3796
101.1	-0.9756

MISC.*

NO.	CP
1	-0.7408
2	-1.2529
3	-1.2529
4	0.0048
5	-1.2529
6	-0.2467
7	-0.0670
8	-1.4144
9	-1.3606
10	0.5799
11	0.3104
12	-0.6080

TABLE 26. RUN 240, BC1W6V, DELF=0, CMU=0, (BN/B)W=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
240	2	CLAERO =	0.1475	CDAERO =	0.0409	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914	
ALPHA		PTOT		PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.03		2139.20		2109.14	30.06	157.73	0.01	2.66	0.0	0.0	0.0	86.9818	0.103952E+07
BETA		CL		CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.15		0.04	-0.12	0.00	-0.00	0.01	0.12	-0.13	0.12	0.07	

********** CANARD ******

XX/C	BP=3.375 CP	BP=10.125 CP
0.0	0.2661	0.2007
2.5	0.0455	0.0537
5.0	0.0128	-0.0117
10.0	-0.1588	-0.1261
15.0	-0.1343	-0.1424
25.0	-0.1343	-0.1343
35.0	-0.1261	-0.1261
50.0	-0.1261	
56.0	-0.1261	-0.1261
65.0	-0.1915	-0.1188
76.0	-0.1179	-0.2731
79.0	-0.3794	-0.3794
80.5	-0.3787	-0.3787
81.0	-0.3787	-0.3669
82.0	-0.3787	-0.3610
84.0	-0.1015	
87.0	-0.0720	
89.0	-0.1143	-0.1405
93.0	-0.1143	
96.0	-0.0707	-0.0969
100.0	2.1605	-0.0794
96.0	0.0774	0.1123
84.0	2.1605	-0.0794
73.0	0.0774	0.1123
50.0	0.1472	0.1384
35.0	-0.0035	-0.0035
25.0	-0.1751	-0.1915
10.0	-0.1751	-0.2078
5.0	-0.2323	-0.2405
2.5	-0.3957	-0.4284

********** WING **********

XX/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5662	0.5776	0.5204	0.5204	0.4975
2.5	-0.1209	-0.1095	-0.1896	-0.2354	-0.1667
5.0	-0.0751	-0.1209	-0.1553	-0.1896	-0.2354
10.0	-0.0980	-0.2011	-0.2240	-0.2240	-0.2240
15.0	-0.1438	-0.1898	-0.2240	-0.1896	-0.2011
24.0	-0.1438	-0.1667	-0.2125	-0.1553	-0.2011
33.0	-0.1242	-0.1242	-0.1573	-0.1904	-0.1904
54.0	-0.1242	-0.1573	-0.1904	1.7627	1.4979
65.0	-0.1666	-0.1840	-0.2363	-0.2625	-0.2451
78.5	-0.2363	-0.4629	-0.3584	-0.3235	-0.3148
79.5	-0.4023	-0.4200	-0.4200	22.8606	52.4750
80.5	-0.4082	-0.4200	-0.1782	-0.5379	-0.9330
81.3	-0.4200	52.4750	-0.1900	-0.1428	-0.2666
82.0	-0.3669	-0.8858	-0.1953	-0.2135	-0.1664
84.0	-0.1546	-0.1900	-0.2725	-0.2548	-0.2017
87.0	-0.1958	-0.2135	-0.3079	-0.2902	-0.2312
89.0	-0.2258	-0.2471	-0.3109	-0.3464	-0.2542
93.0	-0.1832	-0.1832	-0.2258	-0.1549	-0.2825
96.0	-0.1407	-0.1478	-0.1619	-0.1762	-0.2329
100.0	-0.0981	-0.0981	-0.0556	-0.0697	-0.1478
96.0	0.0863	0.1289	0.1502	0.1502	0.1360
84.0	0.1430	0.1643	0.2140	0.1643	0.1218
73.5	0.0413	-0.0911	0.0744	0.0744	0.1075
54.0	-0.0580	-0.0580	-0.0580	-0.0580	-0.0580
33.0	-0.0580	-0.0911	-0.0911	-0.0911	-0.1242
24.0	-0.1095	-0.1324	-0.1438	-0.1209	-0.1667
10.0	-0.0751	-0.1209	-0.1209	-0.1095	-0.1573
5.0	-0.0522	-0.1209	-0.1438	-0.1095	-0.1242
2.5	-0.0637	-0.0866	-0.1209	-0.1209	-0.1242

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0882
43.4	-0.0794
50.4	-0.0446
56.1	-0.0446
62.5	-0.0446
69.7	-0.0707
76.9	-0.0707
83.4	-0.0446
89.4	-0.1056
95.4	-0.1579
101.1	-0.1230
---BOTTOM---	
38.2	-0.0446
43.4	-0.0359
50.4	-0.0533
56.1	-0.0097
62.5	-0.0184
69.7	-0.0446
76.9	-0.0446
83.4	-0.0446
89.4	-0.0272
95.4	-0.0010
101.1	-0.0620

MISC.*

NO.	CP
1	-0.1573
2	-0.0580
3	-0.1904
4	-0.0911
5	-0.1242
6	-0.0911
7	0.5332
8	-0.0201
9	0.0509
10	0.3629
11	0.2282
12	-0.0059

TABLE A26. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
240	4	CLAERO =	0.3606	CDAERO =	0.0643	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03		2139.13	2109.07	30.06	157.88	0.01	2.67	0.0	0.0	0.0	87.2797	0.103690E+07
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.36	0.06	-0.08	0.00	-0.00	0.01	0.34	-0.09	0.33	0.09	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3608	0.1810
2.5	-0.5461	-0.6360
5.0	-0.4317	-0.5951
10.0	-0.1294	-0.4154
15.0	-0.3582	-0.3908
25.0	-0.2846	-0.2928
35.0	-0.2111	-0.2193
50.0	-0.2029	
56.0	-0.1866	-0.1784
65.0	-0.2356	-0.2275
76.0	-0.1458	-0.3092
79.0	-0.4154	-0.4072
80.5	-0.4292	-0.4233
81.0	-0.4056	-0.4351
82.0	-0.3702	-0.3702
84.0	-0.1402	
87.0	-0.1579	
89.0	-0.1699	-0.1787
93.0	-0.1525	
96.0	-0.1089	-0.1264
100.0	2.0004	-0.0915
96.0	0.0828	0.1003
84.0	2.0004	-0.0915
73.0	0.0828	0.1003
50.0	0.1700	0.1526
35.0	0.0095	-0.0395
25.0	-0.1131	-0.2601
10.0	-0.0886	-0.0967
5.0	-0.1213	-0.1131
2.5	0.0421	0.1075

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4255	0.2651	0.0819	-0.0784	-0.2388
2.5	-0.1357	-0.4105	-0.6854	-0.6625	-0.7999
5.0	-0.2617	-0.3876	-0.5708	-0.6052	-0.7197
10.0	-0.1242	-0.3533	-0.4449	-0.4678	-0.5021
15.0	-0.2617	-0.3304	-0.3991	-0.4105	-0.4449
24.0	-0.2388	-0.2617	-0.3418	-0.3304	-0.3762
33.0	-0.2268	-0.2268	-0.2930	-0.2930	-0.2930
54.0	-0.1937	-0.2268	-0.2269	1.4284	-0.2599
65.0	-0.2310	-0.2484	-0.3094	-0.3181	-0.3355
78.5	-0.2745	-0.5099	-0.3791	-0.3617	-0.3966
79.5	-0.4469	-0.4587	-0.4469	24.1698	52.4734
80.5	-0.4646	-0.4528	-0.2228	-0.5471	-0.9069
81.3	-0.4823	46.5312	-0.2346	-0.2110	-0.3525
82.0	-0.4056	-0.8656	-0.2523	-0.2463	-0.2936
84.0	-0.2169	-0.2582	-0.3230	-0.2936	-0.2817
87.0	-0.2287	-0.2640	-0.3289	-0.2876	-0.3171
89.0	-0.2575	-0.2788	-0.3284	-0.3355	-0.3284
93.0	-0.1937	-0.2078	-0.2362	-0.1653	-0.3497
96.0	-0.1582	-0.1653	-0.1653	-0.1866	-0.2930
100.0	-0.1227	-0.1085	-0.0589	-0.0873	-0.1582
96.0	0.1043	0.1468	0.1681	0.1810	0.1256
84.0	0.1610	0.1894	0.2249	0.1823	0.1185
73.5	0.0711	-0.1275	0.1042	0.1042	0.1042
54.0	0.0049	-0.0282	-0.0613	-0.0613	-0.0613
33.0	0.0049	-0.1375	0.0049	0.0049	-0.0613
24.0	-0.0441	-0.0326	-0.0326	-0.0097	-0.0784
10.0	0.0361	0.0361	0.0819	0.1162	0.0711
5.0	0.0705	0.0475	0.1277	0.1735	0.2035
2.5	0.0933	0.1162	0.2079	0.2880	0.3029

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1438
43.4	-0.1176
50.4	-0.0828
56.1	-0.0741
62.5	-0.0828
69.7	-0.1176
76.9	-0.1002
83.4	-0.0828
89.4	-0.1525
95.4	-0.1874
101.1	-0.1525
---BOTTOM---	
38.2	-0.0218
43.4	-0.0218
50.4	-0.0566
56.1	-0.0043
62.5	0.0131
69.7	0.0131
76.9	-0.0130
83.4	-0.0130
89.4	-0.0043
95.4	0.0131
101.1	-0.0566

MISC.*

NO.	CP
1	-0.2599
2	0.0049
3	-0.3261
4	-0.0613
5	-0.2599
6	-0.0613
7	0.5441
8	0.0262
9	0.1752
10	0.3242
11	0.1114
12	-0.0021

TABLE A26. (Concluded)

RUN:SEQ 240 10 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 5E0

CLAERO = 0.8610 CDAERO = 0.1890 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.07	2138.92	2103.86	30.06	158.25	0.01	2.67	0.0	0.0	0.0	87.2350	0.103062E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.86	0.19	-0.01	0.00	-0.00	0.01	0.86	-0.01	0.83	0.21

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-3.8222	-2.0249
2.5	-2.0330	-1.8288
5.0	-1.3712	-1.6490
10.0	-0.0396	-1.5101
15.0	-0.8892	-1.4856
25.0	-0.6278	-1.2079
35.0	-0.4971	-0.7912
50.0	-0.3827	
56.0	-0.3418	-0.4317
65.0	-0.3684	-0.3745
76.0	-0.1948	-0.4072
79.0	-0.4807	-0.4889
80.5	-0.4941	-0.5530
81.0	-0.4823	-0.4646
82.0	-0.4587	-0.3348
84.0	-0.2051	
87.0	-0.2346	
89.0	-0.2135	-0.3007
93.0	-0.1874	
96.0	-0.1438	-0.2571
100.0	1.7825	-0.2222
96.0	0.1003	0.0567
84.0	1.7825	-0.2222
73.0	0.1003	0.0567
50.0	0.2223	0.1438
35.0	0.0667	0.0013
25.0	0.0421	-0.4072
10.0	0.1320	0.0912
5.0	0.1075	0.1402
2.5	0.5850	0.5569

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4792	-1.6817	-3.7772	-3.4911	-1.4412
2.5	-0.1700	-1.1778	-1.9336	-3.1934	-1.2923
5.0	-0.6510	-0.9030	-1.3038	-3.0673	-1.3038
10.0	-0.1586	-0.7312	-0.9831	-1.3381	-1.2236
15.0	-0.5480	-0.6396	-0.8342	-0.7541	-1.1435
24.0	-0.4792	-0.5021	-0.6052	-0.5708	-1.0518
33.0	-0.4254	-0.4916	-0.5247	-0.4916	-1.0544
54.0	-0.3261	-0.3592	-0.3923	0.8325	-0.7234
65.0	-0.3355	-0.3617	-0.4314	-0.4140	-0.6842
79.5	-0.3268	-0.5970	-0.4750	-0.4227	-0.7888
79.5	-0.5235	-0.5235	-0.5058	22.3770	52.4834
80.5	-0.5412	-0.5177	-0.2817	-0.5825	-0.7713
81.3	-0.5471	36.0757	-0.3053	-0.2817	-0.6238
82.0	-0.4882	-0.8361	-0.3230	-0.3525	-0.6120
84.0	-0.2582	-0.3289	-0.3781	-0.3584	-0.6592
87.0	-0.2935	-0.3112	-0.3820	-0.3348	-0.6769
89.0	-0.3213	-0.3355	-0.3852	-0.3852	-0.6973
93.0	-0.2504	-0.2433	-0.2859	-0.2078	-0.6618
96.0	-0.2078	-0.1866	-0.2007	-0.2220	-0.6476
100.0	-0.1724	-0.1085	-0.0873	-0.1298	-0.2007
96.0	0.0972	0.1539	0.1610	0.1397	-0.0092
84.0	0.1965	0.2249	0.2390	0.1894	0.0617
73.5	0.1042	-0.1606	0.1373	0.1373	0.0711
54.0	0.1042	0.0711	0.0711	0.0380	-0.0282
33.0	0.1373	-0.1937	0.1704	0.1704	0.0380
24.0	0.1735	0.1850	0.2079	0.2308	0.1048
10.0	0.2995	0.3224	0.3682	0.4140	0.3029
5.0	0.3567	0.4025	0.4827	0.5170	0.4353
2.5	0.4483	0.4713	0.4942	0.5285	0.4684

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2397
43.4	-0.1874
50.4	-0.1351
56.1	-0.1351
62.5	-0.1525
69.7	-0.2048
76.9	-0.1961
83.4	-0.1525
89.4	-0.1961
95.4	-0.1961
101.1	-0.1612
---BOTTOM---	
38.2	0.0567
43.4	0.0567
50.4	-0.0130
56.1	0.0916
62.5	0.1177
69.7	0.1351
76.9	0.1177
83.4	0.0828
89.4	0.0741
95.4	0.0654
101.1	-0.0566

MISC.*

NO.	CP
1	-0.4916
2	0.1704
3	-0.7896
4	0.1042
5	-0.8227
6	-0.0613
7	0.5582
8	0.0901
9	0.2958
10	0.1681
11	-0.0163
12	-0.0305

TABLE A27. RUN 241, BC1W6V, DELF=0, CMU=0.5, (BN/B)W=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
241	1	CLAERO =	0.2014	CDAERO =	0.1088	DCLC =	0.0444	DWLC =	0.1389	BASEPR =	8.1914
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.03		2138.99	2117.52	21.47	133.81	0.01	2.67	0.255	0.535	0.790	87.2365	0.868204E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.38	-0.66	-0.22	0.01	-0.01	0.01	0.23	-0.16	0.23	0.04	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.2238	0.3151
2.5	-0.0052	-0.0052
5.0	-0.1310	-0.1081
10.0	-0.1310	-0.1996
15.0	-0.2340	-0.2110
25.0	-0.2340	-0.2225
35.0	-0.1768	-0.1996
50.0	-0.2110	
56.0	-0.2454	-0.2225
65.0	-0.3026	-0.3712
76.0	-0.1424	-0.6915
79.0	-1.0803	-1.1146
80.5	-16.5437	-22.3315
81.0	2.4380	7.5818
82.0	1.6371	-0.3940
84.0	0.6298	
87.0	1.4059	
89.0	0.1960	-0.9022
93.0	0.3790	
96.0	1.0502	0.2936
100.0	3.3442	-1.3781
95.0	-0.0969	-0.0359
84.0	3.3442	-1.3781
73.0	-0.0969	-0.0359
50.0	0.0862	0.1106
35.0	-0.0260	-0.0395
25.0	-0.2225	-0.3483
10.0	-0.1538	-0.2225
5.0	-0.2568	-0.2797
2.5	-0.4855	-0.4513

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4402	0.4082	0.4883	0.5624	0.5685
2.5	0.0715	0.1837	-0.0247	-0.1049	-0.0728
5.0	0.2639	0.0875	-0.0728	-0.1530	-0.1209
10.0	-0.0247	-0.0728	-0.1850	-0.2011	-0.1690
15.0	-0.0568	-0.1209	-0.2011	-0.2011	-0.1690
24.0	-0.1209	-0.1850	-0.1850	-0.1530	-0.1369
33.0	-0.1573	-0.2036	-0.2963	-0.2963	-0.2500
54.0	-0.2036	-0.2963	-0.2963	1.9746	-0.2036
65.0	-0.2555	-0.3287	-0.3531	-0.3287	-0.3043
78.5	0.6963	-1.1951	-0.6094	-0.4385	-0.3775
79.5	21.4527	25.2014	-3.1682	-3.0031	73.1638
80.5	-18.7895	-11.2100	-0.2867	-0.3693	-1.0876
81.3	-2.3673	64.3519	-0.3693	-0.2536	-0.3032
82.0	-0.0307	-0.7325	-0.3857	-0.2784	-0.2289
84.0	-3.3829	-1.5664	-0.4683	-0.3362	-0.2619
87.0	-0.2949	-1.6985	-0.5013	-0.3362	-0.2784
89.0	-1.0841	-0.6273	-0.5280	-0.3790	-0.2797
93.0	-0.3194	-0.1009	-0.3521	-0.1705	-0.2698
95.0	-0.4088	-0.3989	-0.2598	-0.2003	-0.2400
100.0	-1.2430	-1.2132	-0.1009	-0.0811	-0.1407
96.0	-0.0016	0.0282	0.1175	0.1374	0.1275
84.0	0.0977	0.1076	0.177[illegible]	0.1473	0.1175
73.5	-0.1109	0.1671	0.0281	0.0281	0.0745
54.0	-0.1109	-0.1573	-0.1573	-0.1109	-0.1109
33.0	-0.1573	-0.0182	-0.1573	-0.1109	-0.1573
24.0	-0.1209	-0.1369	-0.0888	-0.0728	-0.0888
10.0	-0.1049	-0.1369	-0.1049	-0.1049	-0.2036
5.0	-0.1209	-0.1690	-0.1209	-0.0408	-0.1109
2.5	-0.2171	-0.2492	-0.1530	-0.1209	-0.0646

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1335
43.4	-0.1335
50.4	-0.0847
56.1	-0.0481
62.5	-0.0359
69.7	-0.0725
76.9	-0.0803
83.4	-0.0359
89.4	-0.0847
95.4	-0.1579
101.1	-0.1335
---BOTTOM---	
38.2	-0.0803
43.4	-0.0481
50.4	0.0251
56.1	-0.0115
62.5	-0.0237
69.7	-0.0359
76.9	-0.0803
83.4	-0.0481
89.4	-0.0359
95.4	0.0007
101.1	-0.0725

MISC.*

NO.	CP
1	-0.2500
2	-0.1573
3	-0.2500
4	-0.1573
5	-0.2036
6	-0.1573
7	0.6736
8	-0.2499
9	-0.1804
10	0.5545
11	0.4452
12	-0.0414

TABLE A27. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 585

241 3 CLAERO = 0.4280 CDAERO = 0.1253 DCLC = 0.0644 DWLC = 0.1798 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.08	2138.92	2117.44	21.47	133.85	0.01	2.66	0.265	0.550	0.815	87.0205	0.867703E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.67	-0.65	-0.17	0.01	-0.01	0.00	0.47	-0.11	0.46	0.04

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3608	0.0863
2.5	-0.5770	-0.8858
5.0	-0.5198	-0.6799
10.0	-0.0230	-0.5427
15.0	-0.4054	-0.5427
25.0	-0.3368	-0.4169
35.0	-0.3140	-0.3368
50.0	-0.3368	
56.0	-0.3254	-0.3597
65.0	-0.3826	-0.4855
76.0	-0.1538	-0.8058
79.0	-1.1603	-1.2060
80.5	-12.6289	-22.9404
81.0	-1.1122	0.8692
82.0	0.3573	0.1262
84.0	0.6132	
87.0	1.2820	
89.0	0.2082	-1.0852
93.0	0.3912	
96.0	1.0134	0.1960
100.0	4.3200	-1.4268
96.0	-0.0603	-0.0237
84.0	4.3200	-1.4268
73.0	-0.0603	-0.0237
50.0	0.1350	0.1228
35.0	-0.0166	-0.0624
25.0	-0.1538	-0.4283
10.0	-0.0624	-0.0967
5.0	-0.0738	-0.1081
2.5	0.1207	0.1549

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5364	0.4883	0.0074	-0.6178	-0.8263
2.5	0.0675	-0.1529	-0.6499	-1.0186	-1.0867
5.0	-0.0087	-0.2171	-0.4736	-0.6339	-0.7461
10.0	0.0394	-0.3133	-0.4576	-0.4896	-0.5697
15.0	-0.1850	-0.3613	-0.4415	-0.4415	-0.4736
24.0	-0.2171	-0.3453	-0.3613	-0.3453	-0.3934
33.0	-0.2963	-0.2963	-0.3426	-0.3890	-0.3890
54.0	-0.2500	-0.2500	-0.3426	2.7158	-0.3426
65.0	-0.2921	-0.3897	-0.3775	-0.3165	-0.3043
76.5	0.6840	-1.1462	-0.5239	-0.3897	-0.3775
79.5	18.7677	25.7933	-2.4662	-2.3423	73.2128
80.5	-15.2545	-13.2236	-0.3610	-0.4022	-1.1122
81.3	-1.7232	62.8044	-0.4105	-0.3197	-0.4270
82.0	-0.1546	-0.9306	-0.4270	-0.3444	-0.3444
84.0	-3.1844	-1.6736	-0.4848	-0.3940	-0.3610
87.0	-0.2866	-1.7892	-0.5096	-0.3940	-0.3857
89.0	-1.0542	-0.5776	-0.4584	-0.4187	-0.3988
93.0	-0.3393	-0.0811	-0.3293	-0.1903	-0.3889
96.0	-0.4088	-0.3293	-0.2400	-0.2400	-0.3393
100.0	-1.1634	-1.1435	-0.1108	-0.1606	-0.1605
96.0	0.0579	0.0977	0.1275	0.1175	0.0877
84.0	0.1672	0.1672	0.1870	0.1473	0.0877
73.5	-0.1109	0.2598	0.0745	0.0281	0.0745
54.0	-0.0183	-0.0183	-0.1109	-0.1109	-0.1109
33.0	-0.0183	0.0745	-0.0646	-0.0646	-0.1109
24.0	0.0074	-0.0568	-0.0247	0.0234	-0.0568
10.0	0.0394	-0.0247	0.0394	0.1356	0.0281
5.0	0.0074	-0.0087	0.1035	0.2158	0.1671
2.5	-0.0087	-0.0247	0.2158	0.3440	0.3062

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2189
43.4	-0.1823
50.4	-0.1335
56.1	-0.0847
62.5	-0.0847
69.7	-0.1213
76.9	-0.1091
83.4	-0.1091
89.4	-0.2067
95.4	-0.2067
101.1	-0.1701
---BOTTOM---	
38.2	-0.0481
43.4	-0.0359
50.4	0.0495
56.1	0.0007
62.5	0.0007
69.7	-0.0115
76.9	-0.0237
83.4	-0.0237
89.4	-0.0115
95.4	0.0129
101.1	-0.0603

MISC.*

NO.	CP
1	-0.2963
2	-0.0183
3	-0.3890
4	-0.1109
5	-0.3890
6	-0.1573
7	0.7927
8	-0.0612
9	0.0480
10	0.4154
11	0.2665
12	-0.1009

TABLE A27. (Concluded)

RUN:SEQ 241 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 587

CLAERO = 0.9571 CDAERO = 0.2288 DCLC = 0.1012 DWLC = 0.2529 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.06	2138.85	2117.71	21.13	132.85	0.01	2.67	0.269	0.556	0.925	87.2061	0.859718E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	1.31	-0.52	-0.09	0.01	-0.01	0.00	1.02	-0.03	1.01	0.13

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-4.4864	-2.4297
2.5	-2.1275	-1.9533
5.0	-1.5117	-2.0927
10.0	0.1501	-1.8835
15.0	-1.0617	-1.8951
25.0	-0.7795	-1.5117
35.0	-0.6169	-0.9887
50.0	-0.5239	
56.0	-0.5006	-0.5123
65.0	-0.5588	-0.5820
76.0	-0.1172	-0.8144
79.0	-1.2560	-1.1514
80.5	-13.2456	-23.3034
81.0	-1.3064	1.0823
82.0	0.1260	0.3190
84.0	0.6461	
87.0	1.1830	
89.0	0.0438	-0.9106
93.0	0.2422	
96.0	0.8745	0.1182
100.0	4.6557	-1.3943
96.0	-0.0182	-0.0038
84.0	4.6557	-1.3943
73.0	-0.0182	-0.0068
50.0	0.1802	0.1554
35.0	0.0920	0.0107
25.0	0.0107	-0.6052
10.0	0.1269	0.0920
5.0	0.1501	0.1501
2.5	0.5568	0.5568

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2817	-0.7607	-3.4810	-3.8556	-1.6077
2.5	0.1677	-0.8585	-1.7706	-3.5135	-1.4123
5.0	-0.4187	-0.7444	-1.2657	-2.7479	-1.3634
10.0	0.0211	-0.7444	-1.0213	-0.9236	-1.3145
15.0	-0.6304	-0.7119	-0.8585	-0.7933	-1.3960
24.0	-0.5978	-0.6304	-0.6630	-0.6630	-1.2820
33.0	-0.5063	-0.5554	-0.6025	-0.5554	-1.0733
54.0	-0.3200	-0.4612	-0.5063	2.8348	-0.6850
65.0	-0.4645	-0.6009	-0.5265	-0.5141	-0.7496
78.5	0.9488	-1.2579	-0.6752	-0.5389	-0.7992
79.5	17.5744	23.2196	-2.8687	-2.6590	74.3906
80.5	-10.2844	-5.3433	-0.4025	-0.4193	-0.9981
81.3	-1.2413	61.1366	-0.4276	-0.3689	-0.7128
82.0	0.2267	-0.9981	-0.4444	-0.4025	-0.6625
84.0	-2.6100	-1.5852	-0.4947	-0.4193	-0.6793
87.0	-0.2934	-1.6440	-0.4947	-0.3773	-0.7296
89.0	-1.1002	-0.6663	-0.5351	-0.4645	-0.7067
93.0	-0.3838	-0.1618	-0.3838	-0.1518	-0.7269
96.0	-0.4443	-0.3039	-0.2829	-0.2728	-0.6461
100.0	-1.2011	-1.1506	-0.1618	-0.1820	-0.1114
96.0	0.0601	0.1005	0.1307	0.1106	-0.0105
84.0	0.1711	0.2014	0.2215	0.1610	0.0399
73.5	0.0096	0.2451	0.1509	0.1509	0.1038
54.0	0.0567	0.0567	0.0567	0.0567	-0.0374
33.0	0.1509	0.1038	0.1509	0.1509	0.0096
24.0	0.1025	0.1351	0.1840	0.2166	0.1025
10.0	0.2003	0.2166	0.3632	0.4121	0.2922
5.0	0.2492	0.2817	0.3795	0.5098	0.4334
2.5	0.2980	0.3469	0.4283	0.4935	0.4805

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3281
43.4	-0.2785
50.4	-0.2041
56.1	-0.1669
62.5	-0.1669
69.7	-0.2165
76.9	-0.2165
83.4	-0.1669
89.4	-0.2165
95.4	-0.2289
101.1	-0.1793
---BOTTOM---	
38.2	0.0438
43.4	0.0682
50.4	0.1306
56.1	0.0886
62.5	0.0934
69.7	0.1058
76.9	0.1058
83.4	0.0886
89.4	0.0686
95.4	0.0682
101.1	-0.0564

MISC.*

NO.	CP
1	-0.6025
2	0.1509
3	-0.7438
4	0.1038
5	-0.9321
6	-0.0846
7	0.7462
8	-0.1114
9	0.3224
10	0.3224
11	-0.1417
12	-0.1315

TABLE A28. RUN 256, BWSV, DELF=0, CMU=0, (BN/B)W=0.5

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 598

256 3 CLAERO ********* CDAERO = ******** DCLC = 0.0 DWLC = 0.0 BASEPR = 81.8694

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
3.59	21417.40	21119.00	298.34	202.74	4512.43	-2.00	0.0	0.0	0.0	-85.4495	0.550929E+07

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
12.43	20558.70	*********	-18760.30	0.00	-0.00	0.00	-0.00	-18760.30	20558.70	*********

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	2.3422	2.3422
2.5	2.3422	2.3422
5.0	2.3422	2.3422
10.0	2.3422	2.3422
15.0	2.3422	2.3422
25.0	2.3422	2.3422
35.0	2.3422	2.3422
50.0	2.3422	
56.0	2.3422	2.3422
65.0	2.3422	2.3422
75.0	2.3422	2.3422
79.0	2.3422	2.3422
80.5	1.4164	1.4164
81.0	1.4164	1.4164
82.0	1.4164	1.4164
84.0	1.4164	
87.0	1.4164	
89.0	0.7855	0.7855
93.0	0.7855	
96.0	0.7855	0.7855
100.0	0.7855	0.7855
96.0	0.7855	0.7855
84.0	0.7855	0.7855
73.0	0.7855	0.7855
50.0	0.7855	0.7855
35.0	2.3422	2.3422
25.0	2.3422	2.3422
10.0	2.3422	2.3422
5.0	2.3422	2.3422
2.5	2.3422	2.3422

********** WING ********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.9873	0.9873	0.9873	0.9873	0.9873
2.5	0.9873	0.9873	0.9873	0.9873	0.9873
5.0	0.9873	0.9873	0.9873	0.9873	0.9873
10.0	0.9873	0.9873	0.9873	0.9873	0.9873
15.0	0.9873	0.9873	0.9873	0.9873	0.9873
24.0	0.9873	0.9873	0.9873	0.9873	0.9873
33.0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155
54.0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155
65.0	0.7855	0.7855	0.7855	0.7855	0.7855
78.5	0.7855	0.7855	0.7855	0.7855	0.7855
79.5	1.4164	1.4164	1.4164	1.4164	1.4164
80.5	1.4164	1.4164	1.4164	1.4164	1.4164
81.3	1.4164	1.4164	1.4164	1.4164	1.4164
82.0	1.4164	1.4164	1.4164	1.4164	1.4164
84.0	1.4164	1.4164	1.4164	1.4164	1.4164
87.0	1.4164	1.4164	1.4164	1.4164	1.4164
89.0	1.8029	1.8029	1.8029	1.8029	1.8029
93.0	1.8029	1.8029	1.8029	1.8029	1.8029
96.0	1.8029	1.8029	1.8029	1.8029	1.8029
100.0	1.8029	1.8029	1.8029	1.8029	1.8029
96.0	1.8029	1.8029	1.8029	1.8029	1.8029
84.0	1.8029	1.8029	1.8029	1.8029	1.8029
73.5	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155
54.0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155
33.0	-0.5155	-0.5155	-0.5155	-0.5155	-0.5155
24.0	0.9873	0.9873	0.9873	0.9873	0.9873
10.0	0.9873	0.9873	0.9873	0.9873	-0.5155
5.0	0.9873	0.9873	0.9873	0.9873	-0.5155
2.5	0.9873	0.9873	0.9873	0.9873	-0.5155

FUSELAGE

-----TOP---- XLOC	CP
38.2	0.7855
43.4	0.7855
50.4	0.7855
56.1	0.7855
62.5	0.7855
69.7	0.7855
76.9	0.7855
83.4	0.7855
89.4	0.7855
95.4	0.7855
101.1	0.7855
---BOTTOM---	
38.2	0.7855
43.4	0.7855
50.4	0.7855
56.1	0.7855
62.5	0.7855
69.7	0.7855
76.9	0.7855
83.4	0.7855
89.4	0.7855
95.4	0.7855
101.1	0.7855

MISC.*

NO.	CP
1	-0.5155
2	-0.5155
3	-0.5155
4	-0.5155
5	-0.5155
6	-0.5155
7	1.8029
8	1.8029
9	1.8029
10	1.8029
11	1.8029
12	1.8029

TABLE A26. (Continued)

RUN:SEQ 256 4 PROPLULSIVE WING / CANARD PRESSURES PAGE 599

CLAERO = 0.1397 CDAERO = 0.0376 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.01	2141.82	2111.98	29.83	160.07	0.01	2.67	0.0	0.0	0.0	87.1731	0.987642E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.14	0.04	-0.11	0.01	-0.00	0.01	0.12	-0.11	0.12	0.08

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1774	-0.1692
2.5	-0.1692	-0.1774
5.0	-0.1692	-0.1692
10.0	-0.1692	-0.1774
15.0	-0.1692	-0.1692
25.0	-0.1692	-0.1692
35.0	-0.1692	-0.1692
50.0	-0.1774	
58.0	-0.1692	-0.1774
65.0	-0.1692	-0.1774
76.0	-0.1692	-0.1692
78.0	-0.1692	-0.1692
80.5	-0.1592	-0.1533
81.0	-0.1592	-0.1414
82.0	-0.1711	-0.1651
84.0	-0.1592	
87.0	-0.1711	
89.0	0.9901	0.9550
93.0	0.9813	
96.0	0.9637	0.9550
100.0	0.9725	0.9637
96.0	0.9813	0.9637
84.0	0.9725	0.9637
73.0	0.9813	0.9637
50.0	0.9637	0.9550
35.0	-0.1774	-0.1692
25.0	-0.1692	-0.1774
10.0	-0.1692	-0.1774
5.0	-0.1610	-0.1774
2.5	-0.1692	-0.1774

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5489	0.5605	0.4682	0.5258	0.5028
2.5	-0.1433	-0.1202	-0.1087	-0.1318	-0.1433
5.0	-0.1318	-0.1548	-0.1895	-0.1664	-0.2356
10.0	-0.1779	-0.2356	-0.2587	-0.2240	-0.2471
15.0	-0.2125	-0.2125	-0.2817	-0.2240	-0.2356
24.0	-0.1664	-0.2010	-0.2125	-0.2125	-0.2356
33.0	-0.2020	-0.2020	-0.2687	-0.3020	-0.2687
54.0	-0.2020	-0.2353	-0.2687	1.3658	-0.2353
65.0	0.9901	0.9725	0.9969	0.9989	0.9969
78.5	0.9901	0.9901	1.0076	0.9901	0.9901
79.5	-0.4562	-0.4680	-0.4740	20.1518	53.8379
80.5	-0.4799	-0.4799	-0.1830	-0.5274	-0.6184
81.3	-0.4680	45.2504	-0.2127	-0.1948	-0.3077
82.0	-0.3849	-0.8244	-0.2364	-0.2306	-0.1889
84.0	-0.1592	-0.1889	-0.2067	-0.2839	-0.2067
87.0	-0.2186	-0.2661	-0.3136	-0.3017	-0.1770
89.0	-0.2662	-0.2662	-0.3662	-0.3591	-0.2805
93.0	-0.2161	-0.2161	-0.2590	-0.2018	-0.1876
96.0	-0.1804	-0.1661	-0.1804	-0.2018	-0.2590
100.0	-0.1375	-0.1232	-0.0803	-0.0875	-0.1947
96.0	0.0769	0.1126	0.1412	0.1341	0.1341
84.0	0.1412	0.0840	-0.2090	0.1627	1.7850
73.5	-0.0018	-0.1686	0.0649	0.0649	0.1316
54.0	-0.1019	-0.1353	-0.1353	-0.1353	-0.1353
33.0	-0.1019	-0.2020	-0.1353	-0.1686	-0.1686
24.0	-0.1202	-0.1318	-0.1433	-0.1433	-0.1895
10.0	-0.0656	-0.1202	-0.1087	-0.1433	-0.2353
5.0	-0.1087	-0.1087	-0.0971	-0.0625	-0.2353
2.5	-0.0971	-0.1087	-0.0510	-0.1548	-0.2353

FUSELAGE

-----TOP---- XLOC	CP
38.2	0.9637
43.4	0.9725
50.4	0.9725
56.1	0.9637
62.5	0.9637
69.7	0.9637
76.9	0.9725
83.4	0.9813
89.4	0.9725
95.4	0.9813
101.1	0.9813
---BOTTOM---	
38.2	0.9725
43.4	0.9813
50.4	0.9901
56.1	0.9725
62.5	0.9813
69.7	0.9813
76.9	0.9901
83.4	0.9901
89.4	0.9813
95.4	0.9813
101.1	0.9813

MISC.*

NO.	CP
1	-0.2353
2	-0.1353
3	-0.2687
4	-0.1686
5	-0.2353
6	-0.1686
7	-0.2161
8	-0.2018
9	-0.2018
10	-0.2018
11	-0.2018
12	-0.2018

TABLE A28. (Continued)

RUN:SEQ 256 7 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 601

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.02	2142.10	2112.04	30.06	160.76	0.01	2.66	0.0	0.0	0.0	87.0905	0.989892E+06

CLAERO = 0.3291 CDAERO = 0.0614 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1914

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	0.33	0.06	-0.13	0.01	-0.00	0.01	0.32	-0.13	0.31	0.10

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1825	-0.1825
2.5	-0.1825	-0.1743
5.0	-0.1825	-0.1743
10.0	-0.1743	-0.1743
15.0	-0.1743	-0.1743
25.0	-0.1743	-0.1825
35.0	-0.1743	-0.1661
50.0	-0.1825	
56.0	-0.1743	-0.1743
65.0	-0.1743	-0.1743
76.0	-0.1743	-0.1661
79.0	-0.1743	-0.1580
80.5	-0.1716	-0.1716
81.0	-0.1598	-0.1775
82.0	-0.1657	-0.1657
84.0	-0.1716	
87.0	-0.1657	
89.0	0.9752	0.9839
93.0	0.9926	
96.0	0.9839	1.0013
100.0	0.9839	1.0100
96.0	0.9665	1.0100
84.0	0.9839	1.0100
73.0	0.9665	1.0100
50.0	0.9839	0.9926
35.0	-0.1825	-0.1743
25.0	-0.1743	-0.1661
10.0	-0.1743	-0.1661
5.0	-0.1825	-0.1743
2.5	-0.1825	-0.1661

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2342	-0.2009	-0.5673	-0.5330	-0.4528
2.5	-0.1780	-0.7276	-0.9338	-0.7963	-1.1513
5.0	-0.4299	-0.5101	-0.6475	-0.6590	-0.7849
10.0	-0.1551	-0.4643	-0.6017	-0.5330	-0.5673
15.0	-0.3727	-0.4071	-0.4872	-0.4986	-0.4528
24.0	-0.3040	-0.3498	-0.3841	-0.3841	-0.3612
33.0	-0.2898	-0.2898	-0.3229	-0.3560	-0.3229
54.0	-0.2567	-0.2567	-0.2567	1.0344	-0.2898
65.0	0.9926	0.9752	0.9752	0.9665	0.9752
78.5	0.9839	0.9839	0.9752	0.9578	0.9752
79.5	-0.4722	-0.4722	-0.4722	21.9435	53.4298
80.5	-0.4840	-0.4722	-0.2482	-0.5724	-0.7080
81.3	-0.4663	36.5669	-0.2482	-0.2601	-0.3720
82.0	-0.3956	-0.7552	-0.2836	-0.2836	-0.3013
84.0	-0.2070	-0.2306	-0.2188	-0.3190	-0.3308
87.0	-0.2365	-0.2718	-0.3484	-0.3249	-0.1834
89.0	-0.2755	-0.2897	-0.3535	-0.3606	-0.3748
93.0	-0.2187	-0.2258	-0.2613	-0.1974	-0.1904
96.0	-0.1833	-0.1833	-0.1833	-0.2116	-0.3393
100.0	-0.1407	-0.1052	-0.0911	-0.1123	-0.1974
96.0	0.0934	0.1288	0.1501	0.1430	0.1005
84.0	0.1714	0.1005	-0.1974	0.1856	1.3418
73.5	0.0413	-0.1574	0.0744	0.1075	0.1737
54.0	-0.0249	-0.0912	-0.0580	-0.0580	-0.0912
33.0	0.0082	-0.1574	0.0082	0.0082	-0.0912
24.0	-0.0177	-0.0063	-0.0063	0.0052	-0.0979
10.0	0.0510	0.0739	0.0853	0.0968	0.0744
5.0	0.1540	0.1655	0.1884	0.2227	0.1737
2.5	0.2342	0.2342	0.3029	0.2800	0.3061

FUSELAGE

-----TOP---- XLOC	CP
38.2	0.9839
43.4	0.9926
50.4	1.0013
56.1	0.9752
62.5	0.9839
69.7	0.9839
76.9	0.9926
83.4	0.9926
89.4	1.0013
95.4	1.0013
101.1	0.9839
---BOTTOM---	
38.2	0.9752
43.4	0.9839
50.4	0.9665
56.1	0.9926
62.5	0.9926
69.7	0.9665
76.9	0.9839
83.4	0.9839
89.4	1.0013
95.4	0.9839
101.1	0.9752

MISC.

NO.	CP
1	-0.3229
2	0.0082
3	-0.3560
4	-0.0249
5	-0.2898
6	-0.0912
7	-0.2116
8	-0.2045
9	-0.1974
10	-0.1904
11	-0.1904
12	-0.1974

TABLE A28. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 603

256 9 CLAERO = 0.7415 CDAERO = 0.1727 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.93	2142.17	2112.00	30.17	161.13	0.01	2.72	0.0	0.0	0.0	88.9847	0.990759E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.74	0.17	-0.20	0.00	-0.00	0.01	0.75	-0.20	0.72	0.21

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2191	-0.2191
2.5	-0.2110	-0.2110
5.0	-0.2191	-0.2110
10.0	-0.2110	-0.2110
15.0	-0.2029	-0.2110
25.0	-0.2029	-0.2029
35.0	-0.2110	-0.2110
50.0	-0.2110	
56.0	-0.2191	-0.2110
65.0	-0.2191	-0.2110
76.0	-0.2191	-0.2029
79.0	-0.2191	-0.2110
80.5	-0.2205	-0.2029
81.0	-0.2088	-0.1735
82.0	-0.1970	-0.2146
84.0	-0.1970	
87.0	-0.1912	
89.0	0.9835	1.0096
93.0	1.0096	
96.0	1.0009	1.0009
100.0	0.9922	1.0096
96.0	1.0009	1.0096
84.0	0.9922	1.0096
73.0	1.0009	1.0096
50.0	1.0096	0.9922
35.0	-0.2110	-0.2110
25.0	-0.2029	-0.2110
10.0	-0.2029	-0.2110
5.0	-0.2110	-0.2029
2.5	-0.2191	-0.2110

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.3072	-4.2694	-5.9119	-3.1628	-1.4403
2.5	-0.1741	-1.9422	-2.4896	-2.5125	-1.3376
5.0	-1.0410	-1.5201	-1.7027	-2.3528	-1.3376
10.0	-0.1855	-1.0638	-1.2920	-2.2388	-1.2578
15.0	-0.7444	-0.8585	-1.0624	-1.6684	-1.2692
24.0	-0.5619	-0.6304	-0.7786	-1.2350	-1.1095
33.0	-0.4503	-0.5162	-0.6811	-0.6152	-1.0109
54.0	-0.3513	-0.3843	-0.4832	0.9019	-0.7141
65.0	1.0096	0.9922	1.0183	1.0009	1.0096
78.5	0.9835	1.0009	1.0096	0.9922	0.9922
79.5	-0.5317	-0.5259	-0.5259	25.2176	53.2330
80.5	-0.5376	-0.5435	-0.3262	-0.6609	-0.7607
81.3	-0.5552	36.9734	-0.3438	-0.2616	-0.6022
82.0	-0.4906	-0.7960	-0.3438	-0.3088	-0.6374
84.0	-0.2558	-0.2910	-0.2675	-0.3673	-0.6316
87.0	-0.2792	-0.3027	-0.3791	-0.3556	-0.2088
89.0	-0.3135	-0.3277	-0.3701	-0.4054	-0.6810
93.0	-0.2570	-0.2711	-0.2923	-0.2287	-0.2429
96.0	-0.2146	-0.2004	-0.2146	-0.2711	-0.6315
100.0	-0.1581	-0.1227	-0.1227	-0.1934	-0.2287
96.0	0.1106	0.1529	0.1388	0.1176	-0.0379
84.0	0.2165	0.1458	-0.2429	0.1670	1.205[illegible]
73.5	0.1104	-0.1864	0.1764	0.1434	0.1434
54.0	0.1104	0.0774	0.0774	0.0444	-0.0875
33.0	0.2094	-0.2194	0.1764	0.1764	0.0444
24.0	0.2252	0.2024	0.2138	0.2366	0.0769
10.0	0.3735	0.3963	0.4191	0.4191	0.3083
5.0	0.4876	0.5104	0.5560	0.5560	0.4732
2.5	0.6359	0.5560	0.5218	0.5560	0.5062

FUSELAGE

-----TOP---- XLOC	CP
38.2	1.0096
43.4	1.0096
50.4	1.0096
56.1	1.0096
62.5	1.0269
69.7	1.0096
76.9	1.0009
83.4	1.0096
89.4	1.0009
95.4	1.0009
101.1	1.0009
---BOTTOM---	
38.2	0.9835
43.4	0.9922
50.4	0.9835
56.1	0.9922
62.5	0.9922
69.7	1.0009
76.9	1.0096
83.4	1.0096
89.4	1.0096
95.4	1.0009
101.1	1.0183

MISC.*

NO.	CP
1	-0.5822
2	0.2094
3	-1.0439
4	0.1104
5	-0.9450
6	-0.0875
7	-0.2429
8	-0.2358
9	-0.2287
10	-0.2358
11	-0.2358
12	-0.2287

TABLE A29. RUN 257, BW6V, DELF=0, CMU=0.5, (FN/B)W=0.5

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 604

RUN:SEQ												
257	2	CLAERO =	0.1788	CDAERO =	0.1403	DCLC =	0.0	DWLC =	0.1333	BASEPR =	8.1914	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.07		2141.96	2113.03	28.03	154.53	0.01	2.67	0.0	0.518	0.518	87.2984	0.966370E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR	
-0.01		0.31	-0.36	-0.18	0.00	-0.00	0.01	0.20	-0.16	0.20	0.04	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2408	-0.2233
2.5	-0.2320	-0.2320
5.0	-0.2320	-0.2233
10.0	-0.2233	-0.2320
15.0	-0.2320	-0.2233
25.0	-0.2233	-0.2146
35.0	-0.2233	-0.2320
50.0	-0.2233	
56.0	-0.2233	-0.2320
65.0	-0.2320	-0.2320
76.0	-0.2320	-0.2233
79.0	-0.2233	-0.2496
80.5	-0.2365	-0.2238
81.0	-0.2112	-0.2365
82.0	-0.2238	-0.2112
84.0	-0.2175	
87.0	-0.2175	
89.0	0.9933	1.0027
93.0	0.9933	
96.0	0.9933	0.9933
100.0	0.9933	1.0027
96.0	1.0027	1.0027
84.0	0.9933	1.0027
73.0	1.0027	1.0027
50.0	1.0120	1.0120
35.0	-0.2320	-0.2320
25.0	-0.2233	-0.2320
10.0	-0.2233	-0.2320
5.0	-0.2233	-0.2233
2.5	-0.2233	-0.2233

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5483	0.5605	0.4991	0.5360	0.4623
2.5	-0.2132	-0.1641	-0.2255	-0.2378	-0.2255
5.0	-0.1272	0.1887	-0.2255	-0.2378	-0.3483
10.0	-0.2132	-0.2378	-0.3115	-0.2992	-0.3115
15.0	-0.2009	-0.2501	-0.2869	-0.2869	-0.3115
24.0	-0.2009	-0.2378	-0.2623	-0.2501	-0.2746
33.0	-0.2401	-0.2401	-0.2757	-0.2757	-0.2401
54.0	-0.2757	-0.2757	-0.3111	0.5055	1.5353
65.0	1.0027	0.9933	0.9933	1.0120	1.0120
78.5	0.9933	0.9933	1.0120	1.0027	1.0027
79.5	34.0489	36.2362	-3.4987	17.0612	57.2275
80.5	-21.8523	-21.8141	-0.2871	-0.6095	-1.0394
81.3	17.4405	57.2275	-0.3123	-0.2365	-0.3313
82.0	1.5211	-5.2246	-0.3376	-0.2744	-0.2618
84.0	-3.3217	-2.6917	-0.2491	-0.3503	-0.2871
87.0	4.6252	-1.2607	-0.4388	-0.3566	-0.2301
89.0	-2.3831	-0.5647	-0.4581	-0.4201	-0.3212
93.0	0.0518	-0.0168	-0.3364	-0.2527	-0.2451
96.0	-0.2527	-0.2831	-0.2451	-0.2299	-0.2831
100.0	-1.2875	-1.2570	-0.1158	-0.1082	-0.2527
96.0	-0.0016	0.0212	0.1049	0.1277	0.1125
84.0	0.1125	0.0592	-0.2527	0.1505	0.4016
73.5	-0.1336	-0.2401	0.0795	0.0795	0.1505
54.0	-0.1336	-0.1336	-0.1336	-0.1336	-0.1336
33.0	-0.1336	-0.2401	-0.1691	-0.1336	-0.1336
24.0	-0.1395	-0.1641	-0.1641	-0.1395	-0.1887
10.0	-0.1027	-0.1150	-0.1395	-0.0781	-0.1691
5.0	-0.0658	-0.1518	-0.1027	-0.0658	-0.1336
2.5	-0.0781	-0.1027	-0.0658	-0.0781	-0.0981

FUSELAGE

-----TOP----

XLOC	CP
38.2	0.9840
43.4	0.9746
50.4	0.9746
56.1	0.9933
62.5	0.9746
69.7	0.9746
76.9	0.9840
83.4	0.9933
89.4	0.9746
95.4	0.9840
101.1	0.9840

---BOTTOM---

XLOC	CP
38.2	1.0120
43.4	1.0027
50.4	0.9933
56.1	1.0027
62.5	1.0027
69.7	1.0027
76.9	0.9933
83.4	1.0027
89.4	1.0027
95.4	0.9933
101.1	0.9840

MISC.*

NO.	CP
1	-0.2757
2	-0.1336
3	-0.2401
4	-0.1336
5	-0.2046
6	-0.1691
7	-0.2451
8	-0.2527
9	-0.2527
10	-0.2527
11	-0.2603
12	-0.2603

TABLE A29. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
257 5	CLAERO =	0.3762	CDAERO =	0.1635	DCLC =	0.0	DWLC =	0.1669	BASEPR =	8.1914		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
4.05	2141.89	2113.63	28.25	155.29	0.01	2.66	0.0	0.511	0.511	87.1404	0.968135E+06	
BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	0.54	-0.32	-0.21	0.01	-0.00	0.01	0.41	-0.19	0.40	0.07		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2405	-0.2318
2.5	-0.2405	-0.2318
5.0	-0.2405	-0.2318
10.0	-0.2492	-0.2405
15.0	-0.2405	-0.2318
25.0	-0.2405	-0.2231
35.0	-0.2405	-0.2405
50.0	-0.2405	
56.0	-0.2318	-0.2318
65.0	-0.2405	-0.2231
76.0	-0.2492	-0.2405
79.0	-0.2405	-0.2318
80.5	-0.2526	-0.2401
81.0	-0.2338	-0.2338
82.0	-0.2401	-0.2401
84.0	-0.2275	
87.0	-0.2526	
89.0	1.0017	1.0110
93.0	1.0110	
96.0	1.0110	1.0017
100.0	0.9924	0.9924
96.0	1.0110	1.0017
84.0	0.9924	0.9924
73.0	1.0110	1.0017
50.0	1.0110	1.0017
35.0	-0.2405	-0.2405
25.0	-0.2405	-0.2405
10.0	-0.2492	-0.2405
5.0	-0.2405	-0.2318
2.5	-0.2403	-0.2231

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2220	-0.1313	-0.4724	-0.5334	-0.7527
2.5	-0.2410	-0.8136	-1.0938	-1.0938	-1.1790
5.0	-0.4724	-0.6552	-0.6674	-0.7039	-0.8867
10.0	-0.2288	-0.5334	-0.6187	-0.6430	-0.6430
15.0	-0.4115	-0.4237	-0.5212	-0.5455	-0.5455
24.0	-0.3384	-0.3750	-0.4359	-0.4359	-0.4481
33.0	-0.3368	-0.3720	-0.4072	-0.3720	-0.3720
54.0	-0.3016	-0.3368	-0.3720	0.3677	-0.3368
65.0	1.0202	1.0017	0.9924	1.0017	0.9924
78.5	0.9924	1.0017	1.0017	1.0017	1.0017
79.5	34.0082	36.4222	-3.5952	21.4965	56.7995
80.5	-21.8073	-21.8452	-0.3467	-0.6853	-0.9111
81.3	17.2320	44.2234	-0.3717	-0.3341	-0.4219
82.0	2.4253	-4.8368	-0.4031	-0.3467	-0.3404
84.0	-1.8017	-2.9432	-0.2777	-0.3906	-0.3780
87.0	5.5295	-1.2372	-0.4596	-0.3843	-0.2526
89.0	-2.6285	-0.6361	-0.4851	-0.4474	-0.4248
93.0	-0.0851	-0.0323	-0.3493	-0.2663	-0.2663
96.0	-0.2663	-0.3040	-0.2738	-0.2738	-0.3719
100.0	-1.2852	-1.2550	-0.1379	-0.1531	-0.2587
96.0	0.0206	0.0507	0.1111	0.1036	0.0734
84.0	0.1488	0.0809	-0.2663	0.1639	0.4054
73.5	-0.0902	-0.2311	0.0859	0.0859	0.2268
54.0	-0.0198	-0.0550	-0.0550	-0.0550	-0.0902
33.0	-0.0198	-0.2311	-0.0198	-0.0198	-0.0902
24.0	0.0027	-0.0095	-0.0338	-0.0338	-0.0826
10.0	0.0880	0.0758	0.1002	0.1002	0.0506
5.0	0.1367	0.1611	0.1733	0.2098	0.2268
2.5	0.2098	0.2342	0.3682	0.3438	0.3677

FUSELAGE

-----TOP---- XLOC	CP
38.2	1.0202
43.4	1.0295
50.4	1.0202
56.1	1.0110
62.5	1.0110
69.7	1.0017
76.9	1.0017
83.4	1.0017
89.4	1.0110
95.4	0.9924
101.1	1.0017
---BOTTOM---	
38.2	1.0110
43.4	1.0017
50.4	1.0017
56.1	1.0202
62.5	0.9831
69.7	0.9924
76.9	0.9924
83.4	1.0017
89.4	0.9924
95.4	1.0110
101.1	1.0202

MISC.*

NO.	CP
1	-0.3720
2	0.0154
3	-0.4072
4	-0.0550
5	-0.3720
6	-0.1254
7	-0.2738
8	-0.2587
9	-0.2512
10	-0.2587
11	-0.2663
12	-0.2587

TABLE A29. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

PAGE 606

RUN:SEQ											
257 7	CLAERO =	0.7438	CDAERO =	0.3048	DCLC =	0.0	DWLC =	0.2501	BASEPR =	0.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
11.65	2141.96	2114.04	27.91	154.36	0.01	2.71	0.0	0.558	0.558	88.5055	0.962044E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR	
-0.01	0.99	-0.19	-0.28	0.00	-0.00	0.02	0.82	-0.26	0.80	0.18	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2805	-0.2805
2.5	-0.2717	-0.2805
5.0	-0.2717	-0.2805
10.0	-0.2805	-0.2805
15.0	-0.2805	-0.2717
25.0	-0.2805	-0.2717
35.0	-0.2893	-0.2805
50.0	-0.2805	
56.0	-0.2805	-0.2805
65.0	-0.2805	-0.2893
76.0	-0.2805	-0.2805
79.0	-0.2893	-0.2893
80.5	-0.2855	-0.2601
81.0	-0.2918	-0.2855
82.0	-0.2791	-0.2918
84.0	-0.2791	
87.0	-0.2918	
89.0	0.9656	0.9750
93.0	0.9562	
96.0	0.9562	0.9656
100.0	0.9656	0.9843
96.0	0.9937	0.9656
84.0	0.9656	0.9843
73.0	0.9937	0.9656
50.0	0.9843	0.9656
35.0	-0.2805	-0.2893
25.0	-0.2717	-0.2893
10.0	-0.2805	-0.2893
5.0	-0.2805	-0.2805
2.5	-0.2805	-0.2805

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.0550	-3.5098	-5.8034	-5.0514	-1.6357
2.5	-0.2793	-1.8700	-2.4002	-3.6578	-1.4138
5.0	-1.0192	-1.3644	-1.6357	-2.8317	-1.4014
10.0	-0.2793	-1.0438	-1.3151	-1.4631	-1.3891
15.0	-0.7355	-0.8465	-1.0561	-1.5371	-1.3151
24.0	-0.5753	-0.6616	-0.7726	-0.8095	-1.2288
33.0	-0.4943	-0.5650	-0.6368	-0.6368	-1.0647
54.0	-0.4229	-0.4229	-0.4943	0.3614	-0.7061
65.0	1.0031	0.9843	0.9750	0.9937	0.9843
78.5	0.9937	0.9750	0.9843	0.9937	0.9843
79.5	34.8483	37.2729	-3.8337	25.2320	57.5037
80.5	-22.1911	-22.2925	-0.4187	-0.7742	-0.9773
81.3	17.2848	42.2420	-0.4886	-0.3362	-0.6599
82.0	3.1359	-4.5319	-0.4886	-0.4061	-0.6599
84.0	-0.3934	-3.0402	-0.3489	-0.4568	-0.6980
87.0	6.8936	-1.2947	-0.5394	-0.4632	-0.2981
89.0	-2.3080	-0.6927	-0.5323	-0.4941	-0.6927
93.0	-0.3872	-0.0892	-0.3872	-0.3108	-0.3032
96.0	-0.3108	-0.3261	-0.2955	-0.3337	-0.6163
100.0	-1.2504	-1.2657	-0.1962	-0.2191	-0.3032
96.0	0.0559	0.0864	0.1093	0.0941	-0.0434
84.0	0.1934	0.1246	-0.3108	0.1705	0.4837
73.5	-0.0564	-0.2803	0.1475	0.1475	0.1831
54.0	0.0762	0.0762	0.0762	0.0405	-0.0664
33.0	0.1831	-0.2803	0.1475	0.1475	0.0049
24.0	0.1646	0.1769	0.1893	0.2016	0.0538
10.0	0.3866	0.3742	0.3866	0.3989	0.2901
5.0	0.4729	0.4729	0.4975	0.5222	0.4327
2.5	0.5962	0.5592	0.5222	0.5222	0.5040

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	0.9656	38.2	0.9843
43.4	0.9468	43.4	0.9937
50.4	0.9656	50.4	0.9843
56.1	0.9562	56.1	0.9843
62.5	0.9562	62.5	0.9750
69.7	0.9562	69.7	0.9843
76.9	0.9843	76.9	0.9937
83.4	0.9843	83.4	0.9843
89.4	0.9843	89.4	0.9843
95.4	0.9843	95.4	0.9750
101.1	0.9937	101.1	0.9562

MISC.*

NO.	CP
1	-0.5656
2	0.1475
3	-0.8864
4	0.0762
5	-0.9934
6	-0.0664
7	-0.3184
8	-0.3184
9	-0.3184
10	-0.3184
11	-0.3108
12	-0.3108

TABLE A30. RUN 258, BW6V, DELF=0, CMU=1.0, (BN/B)W=0.5

RUN:SEQ																	
258	1	CLAERO =	0.1478	PROPLULSIVE		WING / CANARD		PRESSURES									
				CDAERO =	0.3242	DCLC =	0.0		DWLC =	0.2681		BASEPR =	8.1914				
ALPHA		PTOT		Q		VEL		VAW	H/C		CMUC		CMUW		CMUT	HGT	RN
-0.01		2142.38		15.93		116.37		0.01	2.66		0.0		1.037		1.037	87.1184	0.728532E+06
BETA		CL		CD		CM		CROLL	CN		CY		CNTR		CMTR	CLTR	CDTR
-0.01		0.42		-0.68		-0.23		0.00	-0.00		0.01		0.22		-0.18	0.22	0.03

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2588	-0.2434
2.5	-0.2588	-0.2434
5.0	-0.2588	-0.2434
10.0	-0.2434	-0.3204
15.0	-0.2588	-0.3204
25.0	-0.2434	-0.2896
35.0	-0.2434	-0.2588
50.0	-0.2588	
56.0	-0.2434	-0.2588
65.0	-0.2434	-0.2434
76.0	-0.2588	-0.2434
79.0	-0.2588	-0.2588
80.5	-0.2190	-0.2079
81.0	-0.2412	-0.1967
82.0	-0.2301	-0.2190
84.0	-0.1856	
87.0	-0.1967	
89.0	1.0370	1.0370
93.0	1.0370	
96.0	1.0205	1.0370
100.0	0.9876	1.0370
96.0	0.9876	1.0534
84.0	0.9876	1.0370
73.0	0.9876	1.0534
50.0	1.0041	1.0534
35.0	-0.2588	-0.2588
25.0	-0.2588	-0.2434
10.0	-0.2434	-0.2742
5.0	-0.2434	-0.2588
2.5	-0.2434	-0.2280

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5289	0.5505	0.4857	0.4641	0.5289
2.5	-0.2487	-0.1191	-0.2919	-0.2271	-0.2055
5.0	-0.1407	-0.1839	-0.2919	-0.2055	-0.3135
10.0	-0.2271	-0.3351	-0.3135	-0.2919	-0.3135
15.0	-0.2487	-0.3351	-0.2703	-0.2271	-0.3135
24.0	-0.2487	-0.2919	-0.2487	-0.2055	-0.2919
33.0	-0.2449	-0.3074	-0.3074	-0.3074	-0.2449
54.0	-0.3074	-0.3074	-0.4323	0.1298	-0.2449
65.0	1.0205	1.0205	1.0205	1.0534	1.0534
78.5	1.0205	1.0205	1.0534	1.0370	1.0534
79.5	60.6012	64.2924	-5.6787	37.6410	100.0502
80.5	-38.5679	-38.1010	-0.2635	-0.7639	-1.4421
81.3	31.8701	81.5814	-0.3302	-0.2523	-0.2630
82.0	3.0833	-8.6470	-0.3859	-0.2857	-0.2301
84.0	-5.9228	-5.4007	-0.2746	-0.3302	-0.2743
87.0	8.2761	-1.7201	-0.4410	-0.3302	-0.2079
89.0	-3.3448	-0.4545	-0.4946	-0.4410	-0.3474
93.0	0.5491	0.3018	-0.4277	-0.2672	-0.2672
96.0	-0.2136	-0.3474	-0.3206	-0.2537	-0.2939
100.0	-2.0334	-2.1137	-0.1868	-0.1333	-0.2672
96.0	-0.0664	-0.0664	0.0808	0.0942	0.0942
84.0	0.0674	0.0272	-0.2672	0.1343	0.2815
73.5	-0.2449	-0.2449	-0.0576	0.0049	0.1922
54.0	-0.1200	-0.1200	-0.1825	-0.1200	-0.1200
33.0	-0.1200	-0.2449	-0.1825	-0.1200	-0.1825
24.0	-0.1407	-0.1623	-0.1623	-0.0975	-0.1623
10.0	-0.0759	-0.1191	-0.1407	-0.0327	-0.1825
5.0	-0.0543	-0.1191	-0.0975	-0.0543	-0.1200
2.5	-0.0975	-0.1191	-0.0543	-0.0543	-0.1200

FUSELAGE

-----TOP----		
XLOC	CP	
38.2	1.0370	
43.4	1.0370	
50.4	1.0205	
53.1	1.0370	
62.5	1.0205	
69.7	1.0370	
76.9	1.0205	
83.4	1.0205	
89.4	1.0041	
95.4	1.0205	
101.1	1.0205	
---BOTTOM---		
38.2	1.0534	
43.4	1.0370	
50.4	1.0534	
56.1	1.0534	
62.5	1.0370	
69.7	1.0370	
76.9	1.0370	
83.4	1.0370	
89.4	1.0205	
95.4	1.0370	
101.1	1.0205	

MISC.*

NO.	CP
1	-0.2449
2	-0.1825
3	-0.3074
4	-0.1825
5	-0.2449
6	-0.1200
7	-0.2805
8	-0.2805
9	-0.2672
10	-0.2805
11	-0.2805
12	-0.2805

TABLE A30. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 611

258 3 CLAERO = 0.3138 CDAERO = 0.3774 DCLC = 0.0 DWLC = 0.3758 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.01	2142.38	2128.03	14.35	110.36	0.01	2.67	0.0	1.154	1.154	87.1726	0.692496E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.89	-0.71	-0.27	0.00	-0.00	0.01	0.42	-0.22	0.41	0.06

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2962	-0.2621
2.5	-0.2791	-0.2621
5.0	-0.2791	-0.2621
10.0	-0.2621	-0.2621
15.0	-0.2449	-0.2791
25.0	-0.2621	-0.2621
35.0	-0.2621	-0.2791
50.0	-0.2621	
56.0	-0.2621	-0.2621
65.0	-0.2449	-0.2621
76.0	-0.2791	-0.2621
79.0	-0.2791	-0.2621
80.5	-0.2682	-0.2435
81.0	-0.2558	-0.2558
82.0	-0.2558	-0.2558
84.0	-0.2806	
87.0	-0.2435	
89.0	1.0033	1.0033
93.0	0.9850	
96.0	0.9850	1.0033
100.0	0.9850	1.0033
96.0	1.0033	1.0033
84.0	0.9850	1.0033
73.0	1.0033	1.0033
50.0	0.9850	1.0215
35.0	-0.2791	-0.2791
25.0	-0.2621	-0.2621
10.0	-0.2791	-0.2621
5.0	-0.2621	-0.2791
2.5	-0.2621	-0.2621

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.1639	-0.1238	-0.2437	-0.3397	-0.4356
2.5	-0.2917	-0.6274	-0.9871	-1.0351	-0.9392
5.0	-0.5075	-0.5795	-0.7473	-0.7713	-0.7473
10.0	-0.2677	-0.5315	-0.6753	-0.6274	-0.6274
15.0	-0.4596	-0.4596	-0.5555	-0.5315	-0.5075
24.0	-0.3636	-0.4116	-0.4596	-0.3876	-0.4356
33.0	-0.3141	-0.3834	-0.3834	-0.4527	-0.3834
54.0	-0.3141	-0.3834	-0.3834	0.1713	-0.3834
65.0	0.9850	0.9850	1.0215	1.0215	1.0033
78.5	1.0033	1.0033	1.0033	1.0215	1.0033
79.5	68.1572	71.7132	-6.9839	35.3208	110.9315
80.5	-43.0049	-42.1898	-0.3547	-0.7867	-1.6503
81.3	35.2344	87.3406	-0.4040	-0.3300	-0.3916
82.0	4.2622	-9.1807	-0.4410	-0.3300	-0.3300
84.0	-4.7616	-5.9467	-0.2806	-0.4040	-0.3547
87.0	9.2371	-1.7496	-0.5151	-0.3916	-0.2558
89.0	-3.8643	-0.3793	-0.5417	-0.4972	-0.4378
93.0	0.5130	0.4685	-0.4080	-0.2892	-0.3041
96.0	-0.2149	-0.3189	-0.3189	-0.3189	-0.3932
100.0	-2.2650	-2.4136	-0.1852	-0.2000	-0.2892
96.0	-0.0813	-0.0515	0.0376	0.0970	0.0525
84.0	0.0970	0.0079	-0.2892	0.1416	0.2159
73.5	-0.2447	-0.2447	0.0326	-0.0367	0.1713
54.0	-0.1061	-0.1061	-0.1061	-0.1061	-0.1754
33.0	-0.0367	-0.2447	-0.1061	-0.0367	-0.1754
24.0	-0.0518	-0.0998	-0.0759	-0.0518	-0.1238
10.0	0.0440	0.0440	0.0440	0.0920	0.0326
5.0	0.0920	0.0920	0.1160	0.1879	0.1713
2.5	0.1879	0.2119	0.2359	0.2838	0.2406

FUSELAGE

-----TOP---- XLOC	CP
38.2	1.0215
43.4	1.0033
50.4	1.0033
56.1	0.9850
62.5	1.0033
69.7	1.0033
76.9	1.0215
83.4	1.0398
89.4	1.0033
95.4	1.0215
101.1	1.0215
---BOTTOM---	
38.2	1.0033
43.4	1.0033
50.4	1.0215
56.1	1.0033
62.5	0.9850
69.7	1.0033
76.9	0.9850
83.4	1.0033
89.4	1.0033
95.4	1.0033
101.1	1.0215

MISC.*

NO.	CP
1	-0.3834
2	-0.1061
3	-0.4527
4	-0.1061
5	-0.3834
6	-0.1061
7	-0.3041
8	-0.3041
9	-0.3041
10	-0.3041
11	-0.3041
12	-0.2892

TABLE A30. (Concluded)

RUN:SEQ 258 6 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 614

CLAERO = 0.7241 CDAERO = 0.4656 DCLC = 0.0 DWLC = 0.5092 BASEPR = 8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.96	2142.38	2128.03	14.35	110.31	0.01	2.73	0.0	1.123	1.123	89.1785	0.693272E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.23	-0.54	-0.35	0.00	-0.00	0.02	0.88	-0.30	0.85	0.20

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3133	-0.2962
2.5	-0.2962	-0.3133
5.0	-0.3133	-0.2962
10.0	-0.2962	-0.2962
15.0	-0.2962	-0.2962
25.0	-0.3133	-0.3133
35.0	-0.2962	-0.2962
50.0	-0.2962	
56.0	-0.3133	-0.2791
65.0	-0.3133	-0.2962
76.0	-0.3133	-0.3133
79.0	-0.3133	-0.3133
80.5	-0.3176	-0.3176
81.0	-0.3176	-0.3299
82.0	-0.3176	-0.3052
84.0	-0.3547	
87.0	-0.3052	
89.0	1.0215	1.0033
93.0	0.9667	
96.0	1.0033	0.9667
100.0	0.9850	0.9850
96.0	0.9850	0.9850
84.0	0.9850	0.9850
73.0	0.9850	0.9850
50.0	0.9850	0.9850
35.0	-0.3133	-0.2962
25.0	-0.2962	-0.3133
10.0	-0.3133	-0.2962
5.0	-0.2962	-0.3133
2.5	-0.2962	-0.2962

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.1142	-4.2485	-5.4236	-3.1694	-1.5866
2.5	-0.2917	-1.9703	-2.5459	-2.5219	-1.2749
5.0	-1.0351	-1.3468	-1.8504	-2.6898	-1.2749
10.0	-0.3157	-1.0351	-1.4187	-2.3061	-1.2509
15.0	-0.7473	-0.8672	-1.1550	-1.8984	-1.2749
24.0	-0.5795	-0.6994	-0.8193	-1.1550	-1.2030
33.0	-0.5914	-0.5914	-0.6607	-0.6607	-1.1460
54.0	-0.4527	-0.5221	-0.5914	0.0326	-0.7994
65.0	0.9850	0.9850	1.0215	1.0033	0.9850
76.5	1.0033	0.9850	0.9850	0.9850	0.9850
79.5	68.2930	71.6993	-7.6131	35.7653	110.9036
80.5	-43.0788	-43.6592	-0.5028	-0.8855	-1.6261
81.3	35.2101	75.3013	-0.5892	-0.3547	-0.7126
82.0	4.7437	-9.3416	-0.6015	-0.3670	-0.7126
84.0	-3.5395	-6.3785	-0.3216	-0.4164	-0.7250
87.0	9.8913	-1.9100	-0.6015	-0.4410	-0.3299
89.0	-4.1666	-0.5120	-0.6011	-0.5418	-0.7646
93.0	0.4239	0.4387	-0.4526	-0.3486	-0.3337
96.0	-0.2446	-0.4229	-0.3783	-0.3635	-0.7349
100.0	-2.2204	-2.3690	-0.2446	-0.2743	-0.3189
96.0	-0.0218	0.0228	0.0822	0.0674	-0.0663
84.0	0.1565	0.0822	-0.3337	0.1713	0.1416
73.5	-0.1061	-0.3141	0.1019	0.0326	0.1019
54.0	0.0326	0.0326	0.0326	0.0326	-0.1061
33.0	0.1019	-0.3141	0.1019	0.1019	0.0326
24.0	0.1879	0.2119	0.1879	0.2359	0.0440
10.0	0.3798	0.3558	0.4277	0.4517	0.3099
5.0	0.5236	0.4517	0.5476	0.5956	0.3793
2.5	0.6435	0.5476	0.5716	0.5956	0.4486

FUSELAGE

-----TOP---- XLOC	CP
38.2	1.0033
43.4	0.9667
50.4	0.9667
56.1	0.9667
62.5	0.9667
69.7	0.9485
76.9	0.9485
83.4	0.9667
89.4	0.9667
95.4	0.9850
101.1	0.9850
---BOTTOM---	
38.2	0.9667
43.4	0.9485
50.4	0.9667
56.1	0.9667
62.5	0.9667
69.7	0.9667
76.9	0.9850
83.4	0.9850
89.4	1.0033
95.4	0.9850
101.1	0.9850

MISC.*

NO.	CP
1	-0.6607
2	0.1713
3	-1.2848
4	0.0326
5	-1.0767
6	-0.1061
7	-0.3486
8	-0.3337
9	-0.3189
10	-0.3486
11	-0.3189
12	-0.3337

TABLE A31. RUN 259, BW6V, DELF=0, CMU=2.0, (BN/B)W=0.5

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES

RUN	SEQ	CLAERO =	CDAERO =	DCLC =	DWLC =	BASEPR =
259	1	0.1194	0.4941	0.0	0.4848	8.1914

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.09	2142.45	2133.98	8.48	84.66	0.01	2.65	0.0	1.884	1.884	86.8020	0.533695E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.60	-1.33	-0.31	0.00	-0.00	0.02	0.23	-0.22	0.23	0.02

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2711	-0.3001
2.5	-0.2422	-0.3001
5.0	-0.2422	-0.3001
10.0	-0.2711	-0.2711
15.0	-0.2711	-0.3001
25.0	-0.3001	-0.3001
35.0	-0.2711	-0.3001
50.0	-0.3001	
56.0	-0.2711	-0.2711
65.0	-0.3001	-0.2711
76.0	-0.3001	-0.3001
79.0	-0.3001	-0.3001
80.5	-0.2723	-0.2932
81.0	-0.2723	-0.3141
82.0	-0.2723	-0.3141
84.0	-0.3141	
87.0	-0.2514	
89.0	0.8482	0.8173
93.0	0.8482	
96.0	0.9100	0.8791
100.0	0.8791	0.8791
96.0	0.9100	0.8173
84.0	0.8791	0.8791
73.0	0.9100	0.8173
50.0	0.8791	0.8482
35.0	-0.2711	-0.3001
25.0	-0.2711	-0.3290
10.0	-0.3001	-0.2711
5.0	-0.2711	-0.2711
2.5	-0.2711	-0.3001

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5155	0.5561	0.4749	0.5155	0.4343
2.5	-0.3779	-0.1748	-0.3372	-0.1748	-0.1748
5.0	-0.2154	-0.2560	-0.2560	-0.1748	-0.3372
10.0	-0.3372	-0.2966	-0.3779	-0.2560	-0.3372
15.0	-0.2966	-0.2560	-0.3372	-0.2154	-0.3779
24.0	-0.2560	-0.2966	-0.3779	-0.1748	-0.4185
33.0	-0.2887	-0.4060	-0.4060	-0.2887	-0.4060
54.0	-0.4060	-0.4060	-0.4060	-0.6408	-0.2887
65.0	0.9718	0.9100	0.9100	0.9100	0.8791
78.5	0.9409	0.9100	0.9409	0.9100	0.9100
79.5	114.7948	121.8767	-11.9777	57.6682	187.1151
80.5	-72.7190	-73.5979	-0.3559	-1.0038	-2.5924
81.3	60.9501	136.8709	-0.4813	-0.3349	-0.3768
82.0	6.0402	-16.4712	-0.4813	-0.3559	-0.2932
84.0	-9.8037	-9.8665	-0.3559	-0.4395	-0.3141
87.0	11.8928	-2.7178	-0.5859	-0.4186	-0.3559
89.0	-4.3300	0.0217	-0.5569	-0.4813	-0.3556
93.0	1.1034	1.1537	-0.4311	-0.3053	-0.2802
96.0	-0.0789	-0.3807	-0.3053	-0.3053	-0.2802
100.0	-3.4999	-3.7264	-0.2047	-0.1543	-0.3053
96.0	-0.2047	-0.1796	0.0469	0.0972	0.0720
84.0	0.0469	-0.0286	-0.3053	0.1475	-0.5065
73.5	-0.1712	-0.4060	-0.0538	-0.0538	0.1810
54.0	-0.1712	-0.1712	-0.1712	-0.1712	-0.1712
33.0	-0.1712	-0.4060	-0.1712	-0.1712	-0.1712
24.0	-0.0936	-0.2560	-0.2560	-0.2154	-0.2966
10.0	-0.1748	-0.2560	-0.2560	-0.1748	-0.2887
5.0	-0.1342	-0.2154	-0.2154	-0.1748	-0.4060
2.5	-0.1748	-0.1748	-0.1748	-0.1342	-0.4060

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	0.8482	38.2	0.8791
43.4	0.8482	43.4	0.8482
50.4	0.8482	50.4	0.8791
56.1	0.8791	56.1	0.8482
62.5	0.8482	62.5	0.8482
69.7	0.8173	69.7	0.8791
76.9	0.9100	76.9	0.8791
83.4	0.8791	83.4	0.8482
89.4	0.8482	89.4	0.8482
95.4	0.9100	95.4	0.8791
101.1	0.8791	101.1	0.8482

MISC.*

NO.	CP
1	-0.4060
2	-0.1712
3	-0.4060
4	-0.1712
5	-0.2887
6	-0.1712
7	-0.2802
8	-0.3053
9	-0.3053
10	-0.2802
11	-0.3053
12	-0.3004

TABLE A31. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

PAGE 617

RUN:SEQ												
259 4	CLAERO =	0.3381	CDAERO =	0.4439	DCLC =	0.0	DWLC =	0.6881	BASEPR =	8.1914		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.07	2142.59	2135.59	7.01	76.89	0.01	2.69	0.0	2.106	2.106	87.9526	0.486399E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	1.03	-1.55	-0.38	0.00	-0.00	0.02	0.46	-0.28	0.46	0.06		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3319	-0.2618
2.5	-0.3319	-0.2969
5.0	-0.3319	-0.2969
10.0	-0.3319	-0.2969
15.0	-0.2969	-0.2969
25.0	-0.3319	-0.3319
35.0	-0.2969	-0.2969
50.0	-0.3319	
56.0	-0.2618	-0.3669
65.0	-0.2969	-0.3319
76.0	-0.3319	-0.3319
79.0	-0.3319	-0.3319
80.5	-0.3664	-0.3159
81.0	-0.2906	-0.3664
82.0	-0.3159	-0.2906
84.0	-0.3159	
87.0	-0.3159	
89.0	0.9989	0.9615
93.0	0.9989	
96.0	1.0363	0.9989
100.0	1.0363	0.9989
96.0	0.9989	0.9989
84.0	1.0363	0.9989
73.0	0.9989	0.9989
50.0	0.9989	1.0363
35.0	-0.3319	-0.2969
25.0	-0.3319	-0.3319
10.0	-0.2969	-0.2969
5.0	-0.2969	-0.3319
2.5	-0.3319	-0.2969

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2621	0.0657	-0.4255	-0.3764	-0.7202
2.5	-0.3272	-0.7693	-1.1621	-0.9658	-1.0640
5.0	-0.4746	-0.6219	-0.8184	-0.7202	-0.8675
10.0	-0.3272	-0.5728	-0.7693	-0.5237	-0.7693
15.0	-0.4255	-0.4746	-0.5728	-0.4746	-0.6219
24.0	-0.3764	-0.4255	-0.4255	-0.3764	-0.5237
33.0	-0.4212	-0.4212	-0.4212	-0.4212	-0.4212
54.0	-0.4212	-0.4212	-0.4212	-0.9892	-0.4212
65.0	0.9989	0.9989	1.0363	0.9989	0.9989
76.5	0.9989	0.9989	0.9989	0.9989	0.9989
79.5	139.7529	146.0943	-15.6625	62.1850	226.1765
80.5	-87.9976	-89.1364	-0.4170	-1.0491	-3.0465
81.3	73.8648	147.9922	-0.6193	-0.4170	-0.4170
82.0	4.7154	-19.8094	-0.5434	-0.3917	-0.3159
84.0	-11.6680	-10.5050	-0.3664	-0.4170	-0.4929
87.0	10.8068	-3.1476	-0.6446	-0.4675	-0.2906
89.0	-5.9386	0.0556	-0.6137	-0.5530	-0.4616
93.0	1.3944	1.4857	-0.4616	-0.3400	-0.3400
96.0	-0.0661	-0.4313	-0.3704	-0.4008	-0.4313
100.0	-4.1736	-4.5388	-0.3095	-0.2486	-0.3704
96.0	-0.2486	-0.2486	-0.0052	0.0556	0.0252
84.0	-0.0052	-0.0661	-0.3095	0.1165	-1.0397
73.5	-0.2792	-0.4212	-0.1372	-0.1372	0.1468
54.0	-0.1372	-0.1372	-0.1372	-0.1372	-0.1372
33.0	-0.1372	-0.2792	-0.1372	-0.1372	-0.2792
24.0	-0.0325	-0.0817	-0.0325	0.0165	-0.0817
10.0	0.1148	0.0657	0.1148	0.1639	0.0048
5.0	0.1639	0.0657	0.2130	0.2621	0.0048
2.5	0.1639	0.2130	0.3112	0.4095	0.0048

FUSELAGE

-----TOP---- XLOC	CP
38.2	0.9989
43.4	1.0363
50.4	0.9989
56.1	0.9615
62.5	0.9615
69.7	0.9989
76.9	0.9989
83.4	0.9989
89.4	1.0363
95.4	0.9989
101.1	0.9989
---BOTTOM---	
38.2	0.9989
43.4	1.0363
50.4	0.9989
56.1	0.9989
62.5	0.9989
69.7	0.9989
76.9	1.0363
83.4	1.0736
89.4	0.9989
95.4	0.9989
101.1	0.9615

MISC.*

NO.	CP
1	-0.4212
2	-0.1372
3	-0.4212
4	-0.1372
5	-0.4212
6	-0.2792
7	-0.3400
8	-0.3704
9	-0.3704
10	-0.3704
11	-0.3704
12	-0.4008

TABLE A31. (Concluded)

RUN:SEQ 259 7 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 619

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
12.01	2142.73	2134.82	7.91	81.68	0.01	2.73	0.0	2.130	2.130

CLAERO = 0.6345 CDAERO = 0.7661 DCLC = 0.0 DWLC = 0.9677 BASEPR = 8.1914

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.60	-1.13	-0.44	0.00	-0.00	0.02	0.93	-0.35	0.91	0.20

HQT	RN
89.2304	0.517228E+06

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3663	-0.3663
2.5	-0.3663	-0.3975
5.0	-0.3354	-0.3975
10.0	-0.3354	-0.3975
15.0	-0.3354	-0.3975
25.0	-0.3354	-0.3975
35.0	-0.3975	-0.3975
50.0	-0.3975	
56.0	-0.4284	-0.3975
65.0	-0.3975	-0.3663
76.0	-0.3975	-0.3975
79.0	-0.4594	-0.4284
80.5	-0.2556	-0.3004
81.0	-0.3004	-0.3004
82.0	-0.3004	-0.2780
84.0	-0.3229	
87.0	-0.2780	
89.0	0.8992	0.8330
93.0	0.8661	
96.0	0.8330	0.8661
100.0	0.8661	0.8330
96.0	0.8661	0.8661
84.0	0.8661	0.8330
73.0	0.8661	0.8661
50.0	0.8992	0.8330
35.0	-0.4594	-0.3975
25.0	-0.4284	-0.4284
10.0	-0.3975	-0.3975
5.0	-0.3663	-0.4284
2.5	-0.3975	-0.3975

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.9599	-4.0048	-3.7872	-3.2217	-1.6989
2.5	-0.3067	-2.0034	-2.6126	-2.5691	-1.3944
5.0	-1.0897	-1.5683	-2.0905	-2.4820	-1.3944
10.0	-0.3502	-1.1333	-1.7424	-2.5256	-1.3944
15.0	-0.7852	-0.9593	-1.2203	-2.1775	-1.3508
24.0	-0.7418	-0.8287	-0.9157	-1.0697	-1.3073
33.0	-0.5108	-0.6367	-0.7625	-0.7625	-1.2656
54.0	-0.5108	-0.5108	-0.6367	-1.2656	-0.8883
65.0	1.0317	1.0648	1.0317	0.9323	0.8992
78.5	0.9986	1.0317	0.9986	0.9323	0.9323
79.5	123.8879	130.4749	-13.5595	58.0197	200.5031
80.5	-77.7920	-78.9789	-0.5020	-1.0843	-2.5400
81.3	65.0744	119.9239	-0.6587	-0.3452	-0.8380
82.0	6.2393	-17.2543	-0.5916	-0.3900	-0.7036
84.0	-10.6251	-10.2892	-0.4572	-0.4572	-0.8156
87.0	12.5327	-3.1896	-0.6587	-0.4796	-0.3452
89.0	-5.3893	-0.2143	-0.7534	-0.5916	-0.8612
93.0	1.2681	1.1603	-0.5916	-0.4030	-0.4299
96.0	-0.1065	-0.4569	-0.4838	-0.4569	-0.8342
100.0	-3.6912	-3.9877	-0.4299	-0.3761	-0.4030
96.0	-0.1604	-0.1604	0.0283	0.0283	-0.2143
84.0	0.0822	0.0013	-0.4030	0.0552	-1.2385
73.5	-0.0077	-0.2593	0.1180	-0.0077	0.2438
54.0	0.1180	-0.0077	-0.0077	-0.0077	-0.1335
33.0	0.1180	-0.2593	0.1180	0.1180	-0.0077
24.0	0.1719	0.1719	0.1284	0.2154	-0.0456
10.0	0.4330	0.3024	0.3895	0.4330	0.2438
5.0	0.5200	0.4765	0.4765	0.5635	0.4954
2.5	0.6505	0.4765	0.5200	0.5635	0.4954

FUSELAGE

-----TOP---- XLOC	CP
38.2	0.8330
43.4	0.8661
50.4	0.7990
56.1	0.8330
62.5	0.8661
69.7	0.8992
76.9	0.8661
83.4	0.8330
89.4	0.8661
95.4	0.8330
101.1	0.8661
---BOTTOM---	
38.2	0.8330
43.4	0.7999
50.4	0.8330
56.1	0.8330
62.5	0.8330
69.7	0.7999
76.9	0.7999
83.4	0.8330
89.4	0.8330
95.4	0.8330
101.1	0.8330

MISC.*

NO.	CP
1	-0.7625
2	0.1180
3	-1.2656
4	-0.0077
5	-1.1398
6	-0.1335
7	-0.4030
8	-0.4299
9	-0.4030
10	-0.4030
11	-0.3761
12	-0.4030

TABLE A32. RUN 262, BWGV, DELF=0, CMU=0, (BN/B)W=0.25

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 620

RUN:SEQ											
262 6	CLAERO =	0.1499	CDAERO =	0.0370	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
0.05	2142.03	2112.08	29.95	158.47	0.01	2.66	0.0	0.0	0.0	86.9061	0.101979E+07
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	0.15	0.04	-0.11	0.01	-0.00	0.01	0.15	-0.11	0.15	0.07	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1817	-0.1817
2.5	-0.1817	-0.1817
5.0	-0.1817	-0.1817
10.0	-0.1817	-0.1817
15.0	-0.1734	-0.1817
25.0	-0.1734	-0.1899
35.0	-0.1899	-0.1899
50.0	-0.1817	
56.0	-0.1817	-0.1817
65.0	-0.1899	-0.1817
76.0	-0.1899	-0.1899
79.0	-0.1817	-0.1817
80.5	-0.1612	-0.1671
81.0	-0.1671	-0.1671
82.0	-0.1671	-0.1730
84.0	-0.1671	
87.0	-0.1671	
89.0	-0.1651	-0.1739
93.0	-0.1651	
96.0	-0.1651	-0.1739
100.0	-0.1651	-0.1651
96.0	-0.1739	-0.1739
84.0	-0.1651	-0.1651
73.0	-0.1739	-0.1739
50.0	-0.1826	-0.1651
35.0	-0.1899	-0.1817
25.0	-0.1817	-0.1817
10.0	-0.1899	-0.1817
5.0	-0.1734	-0.1899
2.5	-0.1817	-0.1817

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5502	0.5387	0.4927	0.5272	0.5042
2.5	-0.1509	-0.1509	-0.2199	-0.2314	-0.1854
5.0	-0.1279	-0.1624	-0.2199	-0.2314	-0.2659
10.0	-0.1509	-0.1969	-0.2889	-0.2429	-0.2773
15.0	-0.1854	-0.2084	-0.2773	-0.2314	-0.2773
24.0	-0.1854	-0.2199	-0.2314	-0.1969	-0.2199
33.0	-0.1646	-0.1979	-0.2311	-0.2311	-0.2311
54.0	-0.1646	-0.1979	-0.2311	-0.0982	-0.2311
65.0	-0.2069	-0.2526	-0.2964	-0.2701	-0.2526
78.5	0.1496	-0.4888	-0.3664	-0.3926	-0.3314
79.5	-0.4570	-0.4748	2.6374	4.0988	53.6165
80.5	-0.4629	-0.4748	-0.2145	-0.2440	-0.3446
81.3	-0.4570	53.6165	-0.2085	-0.2026	-0.3328
82.0	-0.4038	-0.4097	-0.2381	-0.2322	-0.2500
84.0	-0.1908	-0.2145	-0.2736	-0.2973	-0.2500
87.0	-0.2322	-0.2855	-0.3505	-0.3150	-0.1967
89.0	-0.2476	-0.2832	-0.3402	-0.3544	-0.2832
93.0	-0.2049	-0.2191	-0.2476	-0.2049	-0.1978
96.0	-0.1764	-0.1764	-0.1734	-0.1907	-0.2547
100.0	-0.1408	-0.1123	-0.0838	-0.0838	-0.1907
96.0	0.0728	0.1226	0.1511	0.1440	0.1298
84.0	0.1440	0.1084	-0.1978	0.1725	0.0942
73.5	0.0348	-0.1646	0.0680	0.1012	0.1677
54.0	-0.0982	-0.0982	-0.0982	-0.0982	-0.0982
33.0	-0.0982	-0.1646	-0.1314	-0.1314	-0.1314
24.0	-0.1165	-0.1394	-0.1394	-0.1165	-0.1739
10.0	-0.0934	-0.1165	-0.1165	-0.0934	-0.1646
5.0	-0.0705	-0.0934	-0.1050	-0.0360	-0.1646
2.5	-0.0590	-0.0820	-0.0820	-0.0475	-0.1646

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0951
43.4	-0.0689
50.4	-0.0302
56.1	-0.0602
62.5	-0.0602
69.7	-0.0951
76.9	-0.0684
83.4	-0.0689
89.4	-0.1126
95.4	-0.1739
101.1	-0.1478
---BOTTOM---	
38.2	-0.0689
43.4	-0.0252
50.4	-0.0951
56.1	-0.0339
62.5	-0.0339
69.7	-0.0614
76.9	-0.0602
83.4	-0.0602
89.4	-0.0426
95.4	-0.0164
101.1	-0.1039

MISC.*

NO.	CP
1	-0.2311
2	-0.0982
3	-0.2311
4	-0.1314
5	-0.1979
6	-0.1314
7	-0.1978
8	-0.2049
9	-0.2049
10	-0.2049
11	-0.1978
12	-0.1978

TABLE A32. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

PAGE 622

RUN:SEQ												
262 8	CLAERO =	0.3290	CDAERO =	0.0612	DCLC =	0.0		DWLC =	0.0	BASEPR =	8.1914	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.07	2142.10	2112.04	30.06	158.98	0.01	2.67	0.0	0.0	0.0	87.4208	0.101824E+07	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	0.33	0.06	-0.13	0.01	-0.00	0.01	0.33	-0.13	0.32	0.09		

********** CANARD ******

XX/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1743	-0.1661
2.5	-0.1580	-0.1661
5.0	-0.1743	-0.1743
10.0	-0.1743	-0.1743
15.0	-0.1661	-0.1743
25.0	-0.1743	-0.1580
35.0	-0.1743	-0.1743
50.0	-0.1661	
56.0	-0.1661	-0.1661
65.0	-0.1661	-0.1661
76.0	-0.1661	-0.1661
79.0	-0.1661	-0.1580
80.5	-0.1539	-0.1480
81.0	-0.1539	-0.1480
82.0	-0.1657	-0.1598
84.0	-0.1480	
87.0	-0.1598	
89.0	-0.1579	-0.1579
93.0	-0.1579	
96.0	-0.1666	-0.1666
100.0	-0.1666	-0.1666
96.0	-0.1666	-0.1666
84.0	-0.1666	-0.1666
73.0	-0.1666	-0.1666
50.0	-0.1579	-0.1666
35.0	-0.1743	-0.1661
25.0	-0.1580	-0.1661
10.0	-0.1661	-0.1580
5.0	-0.1661	-0.1743
2.5	-0.1661	-0.1743

********** WING **********

XX/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2342	0.0166	-0.1895	-0.2582	-0.3384
2.5	-0.1551	-0.6131	-0.8307	-0.7963	-0.8536
5.0	-0.4185	-0.5444	-0.6246	-0.6818	-0.7276
10.0	-0.1666	-0.4528	-0.5444	-0.5444	-0.5444
15.0	-0.3498	-0.3841	-0.4414	-0.4299	-0.4643
24.0	-0.2811	-0.3154	-0.3727	-0.3612	-0.3841
33.0	-0.2567	-0.2898	-0.3560	-0.3560	-0.3229
54.0	-0.2236	-0.2567	-0.2898	-0.0912	-0.2898
65.0	-0.2450	-0.2799	-0.3322	-0.3234	-0.3147
76.5	0.2169	-0.4890	-0.3845	-0.4106	-0.3832
79.5	-0.4251	-0.4663	2.5574	4.3021	53.4597
80.5	-0.4486	-0.4545	-0.2482	-0.2718	-0.8494
81.3	-0.4310	51.2608	-0.2188	-0.2188	-0.3484
82.0	-0.3720	-0.7492	-0.2365	-0.2541	-0.2954
84.0	-0.2129	-0.2129	-0.2482	-0.2954	-0.3131
87.0	-0.2246	-0.2482	-0.3306	-0.3013	-0.1598
89.0	-0.2684	-0.2967	-0.3606	-0.3677	-0.3535
93.0	-0.2045	-0.2329	-0.2471	-0.1974	-0.1904
96.0	-0.1691	-0.1762	-0.1833	-0.2045	-0.3180
100.0	-0.1407	-0.1052	-0.0768	-0.1052	-0.1904
96.0	0.0934	0.1359	0.1572	0.1572	0.1005
84.0	0.1643	0.1147	-0.1974	0.1856	0.1714
73.5	0.0413	-0.1574	0.0744	0.0744	0.1737
54.0	-0.0249	-0.0580	-0.0580	-0.0912	-0.0912
33.0	-0.0249	-0.1905	-0.0580	-0.0249	-0.0912
24.0	-0.0177	-0.0292	-0.0063	0.0166	-0.0635
10.0	0.0853	0.0853	0.1082	0.1082	0.0744
5.0	0.1540	0.1655	0.2113	0.2456	0.1737
2.5	0.2227	0.2571	0.3143	0.2914	0.2399

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0881
43.4	-0.0620
50.4	-0.0620
56.1	-0.0707
62.5	-0.0794
69.7	-0.1317
76.9	-0.1143
83.4	-0.0881
89.4	-0.1579
95.4	-0.1840
101.1	-0.1491
---BOTTOM---	
38.2	-0.0533
43.4	-0.0010
50.4	-0.0794
56.1	-0.0010
62.5	0.0252
69.7	0.0252
76.9	0.0077
83.4	-0.0010
89.4	0.0077
95.4	0.0077
101.1	-0.0794

MISC.*

NO.	CP
1	-0.3229
2	-0.0249
3	-0.3560
4	-0.0580
5	-0.3229
6	-0.1243
7	-0.2045
8	-0.1974
9	-0.1974
10	-0.1974
11	-0.1904
12	-0.1974

TABLE A32. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 262 10

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
11.93	2141.96	2111.78	30.17	159.41	0.01	2.72	0.0	0.0	0.0

CLAERO = 0.7406 CDAERO = 0.1725 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1914

HGT	RN
89.0522	0.101814E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.74	0.17	-0.19	0.00	-0.00	0.01	0.76	-0.19	0.73	0.20

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2110	-0.2110
2.5	-0.2029	-0.2110
5.0	-0.1948	-0.1866
10.0	-0.1948	-0.1948
15.0	-0.2029	-0.2029
25.0	-0.2029	-0.1948
35.0	-0.2029	-0.2029
50.0	-0.2029	
56.0	-0.1948	-0.2029
65.0	-0.2029	-0.2029
76.0	-0.2191	-0.2029
79.0	-0.2029	-0.2029
80.5	-0.2146	-0.1970
81.0	-0.2029	-0.1912
82.0	-0.1970	-0.1912
84.0	-0.1794	
87.0	-0.1970	
89.0	-0.1974	-0.1974
93.0	-0.206[illegible]	
96.0	-0.1887	-0.1974
100.0	-0.1887	-0.1887
96.0	-0.1974	-0.1887
84.0	-0.1887	-0.1887
73.0	-0.1974	-0.1887
50.0	-0.1974	-0.1974
35.0	-0.2110	-0.1948
25.0	-0.2029	-0.2029
10.0	-0.2110	-0.1948
5.0	-0.2029	-0.2029
2.5	-0.2029	-0.1948

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.9422	-3.8359	-5.5695	-3.2884	-1.4745
2.5	-0.1741	-1.8167	-2.5583	-2.5241	-1.2920
5.0	-1.0182	-1.3148	-1.8281	-2.6153	-1.3034
10.0	-0.1741	-0.9612	-1.2920	-2.1475	-1.2236
15.0	-0.6874	-0.8129	-1.0296	-1.6798	-1.2464
24.0	-0.5391	-0.6075	-0.7444	-1.2121	-1.1095
33.0	-0.4503	-0.5162	-0.5822	-0.8461	-0.9760
54.0	-0.3183	-0.3843	-0.3843	-0.0875	-0.6811
65.0	-0.3450	-0.3623	-0.4318	-0.4144	-0.6489
78.5	0.2281	-0.5360	-0.4406	-0.4231	-0.6923
79.5	-0.5082	-0.5259	2.5629	4.5769	53.2562
80.5	-0.5082	-0.5024	-0.3145	-0.3380	-0.8606
81.3	-0.5141	46.7670	-0.3027	-0.2440	-0.5905
82.0	-0.4261	-0.7079	-0.3086	-0.2969	-0.6374
84.0	-0.2381	-0.2851	-0.2969	-0.3556	-0.6551
87.0	-0.2616	-0.3203	-0.3732	-0.3497	-0.1912
89.0	-0.3065	-0.3347	-0.3630	-0.4195	-0.6527
93.0	-0.2429	-0.2499	-0.2711	-0.2358	-0.2358
96.0	-0.2004	-0.1863	-0.2075	-0.2499	-0.6315
100.0	-0.1510	-0.1227	-0.1227	-0.1793	-0.2287
96.0	0.1105	0.1458	0.1529	0.1246	-0.0521
84.0	0.2094	0.1529	-0.2287	0.1741	0.2094
73.5	0.1104	-0.1864	0.1763	0.1434	0.1763
54.0	0.0774	0.0774	0.0774	0.0774	-0.0545
33.0	0.1763	-0.1864	0.1763	0.1763	0.0444
24.0	0.2366	0.2252	0.2480	0.2708	0.0883
10.0	0.3963	0.4305	0.4077	0.4419	0.3413
5.0	0.5218	0.4990	0.5560	0.5788	0.4732
2.5	0.6473	0.5674	0.5446	0.5788	0.4732

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1019
43.4	-0.0932
50.4	-0.1192
56.1	-0.1540
62.5	-0.1974
69.7	-0.2496
76.9	-0.2234
83.4	-0.1626
89.4	-0.2321
95.4	-0.1627
101.1	-0.1453
---BOTTOM---	
38.2	0.0023
43.4	0.0631
50.4	-0.0324
56.1	0.0805
62.5	0.1413
69.7	0.1760
76.9	0.1413
83.4	0.1065
89.4	0.0805
95.4	0.0631
101.1	-0.0671

MISC.*

NO.	CP
1	-0.5492
2	0.2094
3	-1.0769
4	0.1434
5	-0.9120
6	-0.0545
7	-0.2429
8	-0.2287
9	-0.2429
10	-0.2287
11	-0.2358
12	-0.2358

TABLE A33. RUN 265, BW6V, DELF=0, CMU=0.5, (BN/B)W=0.25

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RUN:SEQ													
265	2	CLAERO =	0.1823	PROPLULSIVE		WING / CANARD		PRESSURES					
				CDAERO =	0.1163	DCLC =	0.0	DWLC =	0.1278	BASEPR =	0.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
-0.04		2139.20	2121.91	17.29	120.66	0.01	2.66	0.0	0.495	0.495	86.9884	0.769511E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR		
-0.01		0.31	-0.36	-0.19	0.00	-0.00	0.00	0.21	-0.16	0.21	0.06		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2348	-0.2206
2.5	-0.2206	-0.2064
5.0	-0.2206	-0.2064
10.0	-0.2064	-0.2206
15.0	-0.2206	-0.2348
25.0	-0.2348	-0.2348
35.0	-0.2064	-0.2206
50.0	-0.2206	
56.0	-0.2206	-0.2206
65.0	-0.2206	-0.2064
76.0	-0.2206	-0.2064
79.0	-0.2064	-0.2206
80.5	-0.2090	-0.2294
81.0	-0.2090	-0.1884
82.0	-0.2090	-0.1987
84.0	-0.1884	
87.0	-0.2192	
89.0	-0.2115	-0.2266
93.0	-0.2115	
96.0	-0.2115	-0.2266
100.0	-0.2115	-0.2266
96.0	-0.2115	-0.2266
84.0	-0.2115	-0.2266
73.0	-0.2115	-0.2266
50.0	-0.2115	-0.2266
35.0	-0.2206	-0.1922
25.0	-0.2064	-0.2206
10.0	-0.2206	-0.2064
5.0	-0.2206	-0.2206
2.5	-0.2064	-0.2206

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5228	0.5428	0.4830	0.5030	0.5030
2.5	-0.2137	-0.1340	-0.3131	-0.2733	-0.2336
5.0	-0.1738	-0.1539	-0.3131	-0.2534	-0.2534
10.0	-0.2137	-0.2534	-0.3131	-0.3131	-0.2932
15.0	-0.1937	-0.2336	-0.2932	-0.2733	-0.3131
24.0	-0.2137	-0.2137	-0.2336	-0.2137	-0.2534
33.0	-0.2079	-0.2079	-0.2079	-0.2655	-0.2079
54.0	-0.2655	-0.2655	-0.2655	1.0006	-0.2079
65.0	-0.3327	-0.6812	-0.3630	-0.3175	-0.2872
78.5	1.3036	-0.9539	-0.4539	-0.4388	-0.3781
79.5	55.7785	58.2279	8.1145	7.3146	92.3598
80.5	-36.5876	-42.7276	-0.3320	-0.3320	-1.5518
81.3	26.3187	92.3598	-0.2704	-0.2192	-0.3217
82.0	5.5107	-0.1987	-0.3217	-0.2910	-0.2807
84.0	-0.3730	-2.4230	-0.3012	-0.3320	-0.2704
87.0	10.5130	-1.9310	-0.3832	-0.3217	-0.2294
89.0	-5.0671	-0.9850	-0.4176	-0.3807	-0.3067
93.0	0.2730	-0.6396	-0.3190	-0.2573	-0.2327
96.0	-0.1094	-0.7506	-0.2327	-0.2450	-0.2696
100.0	-1.9469	-1.3179	-0.0970	-0.1094	-0.2450
96.0	-0.0477	0.0386	0.1127	0.1250	0.1127
84.0	0.1003	0.1003	-0.2573	0.1743	1.1732
73.5	-0.1504	-0.2079	0.0798	0.0798	0.1949
54.0	-0.0928	-0.0928	-0.0928	-0.1504	-0.1504
33.0	-0.0928	-0.2079	-0.0928	-0.1504	-0.1504
24.0	-0.1340	-0.1141	-0.1539	-0.1340	-0.1937
10.0	-0.0942	-0.0942	-0.1340	-0.1141	-0.1504
5.0	-0.0345	-0.0942	-0.0942	-0.0942	-0.0353
2.5	-0.0345	-0.1141	-0.0743	-0.0743	-0.0928

FUSELAGE

-----TOP----		---BOTTOM---	
X/C	CP	X/C	CP
38.2	-0.0903	38.2	-0.0751
43.4	-0.0600	43.4	-0.0297
50.4	-0.0448	50.4	-0.0903
56.1	-0.0600	56.1	-0.0448
62.5	-0.0600	62.5	-0.0448
69.7	-0.1054	69.7	-0.0448
76.9	-0.1054	76.9	-0.0600
83.4	-0.0751	83.4	-0.0600
89.4	-0.1812	89.4	-0.0448
95.4	-0.2115	95.4	-0.0297
101.1	-0.1964	101.1	-0.1206

MISC.*

NO.	CP
1	-0.2079
2	-0.0928
3	-0.2655
4	-0.1504
5	-0.2079
6	-0.1504
7	-0.2573
8	-0.2450
9	-0.2327
10	-0.2573
11	-0.2327
12	-0.2450

TABLE A33. (Continued)

PROPULSIVE WING / CANARD PRESSURES

RUN:SEQ 265 4 CLAERO = 0.3653 CDAERO = 0.1431 DCLC = 0.0 DWLC = 0.1613 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03	2139.20	2121.91	17.29	120.74	0.01	2.98	0.0	0.404	0.494	97.6337	0.768174E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	0.53	-0.32	-0.22	0.00	-0.00	-0.00	0.40	-0.19	0.40	0.09

********** CANARD ******

%X/C	BP=3.075 CP	BP=10.125 CP
0.0	-0.2206	-0.2206
2.5	-0.2206	-0.2206
5.0	-0.2206	-0.2064
10.0	-0.2348	-0.2064
15.0	-0.2206	-0.2064
25.0	-0.2206	-0.2206
35.0	-0.2206	-0.2064
50.0	-0.2206	
56.0	-0.2206	-0.2064
65.0	-0.2206	-0.2064
76.0	-0.1922	-0.2206
79.0	-0.2064	-0.2206
80.5	-0.2090	-0.2192
81.0	-0.1884	-0.1884
82.0	-0.2090	-0.1987
84.0	-0.2192	
87.0	-0.2090	
89.0	-0.2115	-0.2115
93.0	-0.2115	
96.0	-0.1964	-0.2115
100.0	-0.1964	-0.2115
96.0	-0.1964	-0.2115
84.0	-0.1964	-0.2115
73.0	-0.1964	-0.2115
50.0	-0.2115	-0.2115
35.0	-0.2064	-0.2064
25.0	-0.2206	-0.2064
10.0	-0.2064	-0.2206
5.0	-0.2348	-0.2206
2.5	-0.2064	-0.2206

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.1845	-0.0544	-0.3729	-0.2137	-0.5122
2.5	-0.1738	-0.6913	-1.0297	-0.9899	-0.9501
5.0	-0.4326	-0.5122	-0.6316	-0.6515	-0.7710
10.0	-0.1539	-0.4525	-0.5719	-0.5122	-0.5520
15.0	-0.3729	-0.3729	-0.4923	-0.3729	-0.4525
24.0	-0.2733	-0.2932	-0.3928	-0.3131	-0.3729
33.0	-0.3230	-0.3230	-0.3806	-0.3806	-0.3806
54.0	-0.3230	-0.3230	-0.3230	1.0581	-0.3230
65.0	-0.3933	-0.7266	-0.4084	-0.3781	-0.3630
78.5	1.4096	-0.9842	-0.4842	-0.4691	-0.4388
79.5	55.9219	58.5356	7.2736	7.0688	92.3714
80.5	-36.7102	-43.2707	-0.3627	-0.3422	-1.3877
81.3	25.8577	92.3714	-0.2704	-0.2294	-0.3525
82.0	6.3102	-0.3525	-0.2910	-0.2500	-0.3115
84.0	0.2318	-2.2488	-0.3012	-0.3217	-0.3012
87.0	11.6917	-1.9413	-0.4037	-0.3422	-0.1884
89.0	-5.9427	-0.9973	-0.4053	-0.4063	-0.3807
93.0	0.4580	-0.6273	-0.2943	-0.2327	-0.2080
96.0	-0.1094	-0.7260	-0.2204	-0.2327	-0.3190
100.0	-1.9716	-1.3673	-0.1217	-0.1340	-0.2204
96.0	-0.0107	0.0756	0.1373	0.1496	0.1250
84.0	0.1496	0.1250	-0.2204	0.1990	1.2595
73.5	-0.1504	-0.2079	0.1374	0.0798	0.3100
54.0	-0.0353	-0.0353	-0.0353	-0.0928	-0.0928
33.0	0.0223	-0.1504	-0.0353	-0.0353	-0.0928
24.0	0.0451	0.0252	0.0053	0.0451	-0.0146
10.0	0.1446	0.1446	0.1446	0.1845	0.0798
5.0	0.2442	0.2243	0.2243	0.3039	0.1949
2.5	0.3238	0.2840	0.3437	0.3835	0.2525

FUSELAGE

-----TOP----

XLOC	CP
38.2	-0.0903
43.4	-0.0600
50.4	-0.0600
56.1	-0.0751
62.5	-0.0903
69.7	-0.1509
76.9	-0.1509
83.4	-0.1208
89.4	-0.2115
95.4	-0.2266
101.1	-0.1964
---BOTTOM---	
38.2	-0.0600
43.4	0.0006
50.4	-0.0751
56.1	0.0006
62.5	0.0158
69.7	0.0309
76.9	0.0006
83.4	-0.0297
89.4	0.0006
95.4	0.0006
101.1	-0.1064

MISC.*

NO.	CP
1	-0.3806
2	0.0223
3	-0.3806
4	-0.0353
5	-0.3230
6	-0.0928
7	-0.2327
8	-0.2080
9	-0.2204
10	-0.2450
11	-0.2327
12	-0.2204

TABLE A33. (Concluded)

RUN:SEQ 265 6 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 629

CLAERO = 0.7406 CDAERO = 0.2860 DCLC = 0.0 DWLC = 0.2278 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.04	2139.13	2121.73	17.40	121.23	0.01	2.73	0.0	0.501	0.501	89.2687	0.769252E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.97	-0.16	-0.30	0.00	-0.00	0.01	0.81	-0.27	0.78	0.23

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2558	-0.2416
2.5	-0.2558	-0.2558
5.0	-0.2558	-0.2558
10.0	-0.2558	-0.2558
15.0	-0.2416	-0.2558
25.0	-0.2416	-0.2558
35.0	-0.2558	-0.2416
50.0	-0.2416	
56.0	-0.2558	-0.2558
65.0	-0.2558	-0.2416
76.0	-0.2558	-0.2558
79.0	-0.2698	-0.2558
80.5	-0.2629	-0.2324
81.0	-0.2426	-0.2120
82.0	-0.2426	-0.2426
84.0	-0.2528	
87.0	-0.2426	
89.0	-0.2495	-0.2646
93.0	-0.2646	
96.0	-0.2646	-0.2646
100.0	-0.2646	-0.2646
96.0	-0.2646	-0.2646
84.0	-0.2646	-0.2646
73.0	-0.2646	-0.2646
50.0	-0.2646	-0.2495
35.0	-0.2558	-0.2416
25.0	-0.2416	-0.2558
10.0	-0.2416	-0.2558
5.0	-0.2416	-0.2698
2.5	-0.2698	-0.2416

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.1445	-3.5683	-5.7835	-2.7773	-1.4722
2.5	-0.2263	-1.8083	-2.4609	-2.5202	-1.2151
5.0	-1.0371	-1.2941	-1.6304	-2.4807	-1.2348
10.0	-0.2460	-0.9778	-1.2151	-2.1643	-1.2348
15.0	-0.6811	-0.7997	-1.0173	-1.6304	-1.1557
24.0	-0.5229	-0.6416	-0.8196	-0.8789	-1.1162
33.0	-0.4867	-0.4867	-0.6010	-0.6010	-1.0584
54.0	-0.3[illegible]3	-0.4295	-0.4867	1.1142	-0.8297
65.0	-0.4603	-0.7764	-0.5054	-0.4151	-0.7011
78.5	1.4364	-1.0322	-0.5204	-0.5054	-0.7463
79.5	56.2874	58.8117	7.3344	7.3545	91.7658
80.5	-36.5782	-43.6660	-0.4157	-0.3342	-1.3832
81.3	24.6975	81.3078	-0.3546	-0.2528	-0.6398
82.0	7.8228	-1.0573	-0.3750	-0.3037	-0.6295
84.0	2.5172	-2.1572	-0.3648	-0.3444	-0.6907
87.0	13.8821	-2.0248	-0.4565	-0.3546	-0.2324
89.0	-6.8051	-1.1199	-0.4460	-0.4337	-0.7033
93.0	0.6323	-0.6910	-0.3357	-0.2621	-0.2744
96.0	-0.1274	-0.8135	-0.2621	-0.2989	-0.6297
100.0	-1.9285	-1.3894	-0.1642	-0.2009	-0.2499
96.0	0.0441	0.0932	0.1299	0.1177	-0.0294
84.0	0.2034	0.1544	-0.2744	0.1912	1.2694
73.5	-0.0865	-0.2580	0.1422	0.0850	0.0850
54.0	0.0850	0.0850	0.0850	0.0850	-0.0865
33.0	0.1994	-0.2580	0.1422	0.1422	0.0279
24.0	0.2286	0.2088	0.2286	0.2879	0.1297
10.0	0.4065	0.3868	0.4065	0.4659	0.3137
5.0	0.5450	0.5252	0.5252	0.5845	0.4281
2.5	0.6241	0.5845	0.5648	0.5845	0.4853

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.1141	38.2	-0.0237
43.4	-0.1291	43.4	0.0515
50.4	-0.1442	50.4	-0.0689
56.1	-0.1743	56.1	0.0666
62.5	-0.2194	62.5	0.0967
69.7	-0.2646	69.7	0.1569
76.9	-0.2495	76.9	0.1268
83.4	-0.2043	83.4	0.0817
89.4	-0.2798	89.4	0.0666
95.4	-0.2344	95.4	0.0365
101.1	-0.1893	101.1	-0.0990

MISC.*

NO.	CP
1	-0.6010
2	0.1422
3	-1.0584
4	0.0850
5	-0.9441
6	-0.0865
7	-0.2989
8	-0.2867
9	-0.2744
10	-0.2744
11	-0.2621
12	-0.2867

TABLE A34. RUN 266, BW6V, DELF=0. CMU=1.0, (DN/B)W=0.25

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 630

RUN:SEQ												
266	1	CLAERO =	0.1704	CDAERO =	0.1889	DCLC =	0.0	DWLC =	0.2745	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.01		2139.41	2131.39	8.02	82.22	0.01	2.66	0.0	1.060	1.060	87.0207	0.522655E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.44	-0.84	-0.25	0.00	-0.00	0.00	0.23	-0.20	0.23	0.06	

********** CANARD ******

%X/C	BP=3.875 CP	BP=10.125 CP
0.0	-0.2269	-0.2575
2.5	-0.2269	-0.2575
5.0	-0.2269	-0.2575
10.0	-0.2269	-0.2575
15.0	-0.2269	-0.2575
25.0	-0.2269	-0.2269
35.0	-0.2269	-0.2881
50.0	-0.2575	
56.0	-0.2269	-0.2881
65.0	-0.2575	-0.2575
76.0	-0.2575	-0.2575
79.0	-0.2269	-0.2575
80.5	-0.2840	-0.2620
81.0	-0.3062	-0.2840
82.0	-0.2840	-0.2399
84.0	-0.2620	
87.0	-0.3283	
89.0	-0.2762	-0.2762
93.0	-0.2109	
96.0	-0.2762	-0.2762
100.0	-0.2762	-0.2762
96.0	-0.2436	-0.2436
84.0	-0.2762	-0.2762
73.0	-0.2436	-0.2436
50.0	-0.2762	-0.2436
35.0	-0.2269	-0.2269
25.0	-0.2269	-0.2269
10.0	-0.2881	-0.2575
5.0	-0.2269	-0.2575
2.5	-0.2575	-0.2575

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5252	0.6110	0.4394	0.4394	0.4394
2.5	-0.2898	-0.1182	-0.2898	-0.3326	-0.2468
5.0	-0.2039	-0.1611	-0.2468	-0.2898	-0.3326
10.0	-0.2039	-0.2468	-0.3326	-0.3326	-0.2898
15.0	-0.3326	-0.2898	-0.3755	-0.2898	-0.3326
24.0	-0.2468	-0.2898	-0.2468	-0.2468	-0.3326
33.0	-0.2430	-0.3670	-0.2430	-0.3670	-0.3670
54.0	-0.3670	-0.3670	-0.3670	1.6169	-0.2430
65.0	-0.3741	-1.0923	-0.4088	-0.3088	-0.3088
78.5	2.3027	-1.2309	-0.5047	-0.4721	-0.3741
79.5	115.1164	120.8364	15.8828	16.1706	197.7843
80.5	-77.8725	-84.4100	-0.4607	-0.4387	-2.8902
81.3	61.4493	180.7986	-0.3503	-0.3062	-0.3944
82.0	-0.3283	5.4583	-0.4166	-0.3062	-0.3283
84.0	-16.5832	-1.7196	-0.3503	-0.3503	-0.3283
87.0	-11.5485	-3.8179	-0.4387	-0.3503	-0.3283
89.0	6.8429	-1.7400	-0.4114	-0.4114	-0.3051
93.0	0.9704	-1.1554	-0.3582	-0.2785	-0.2520
96.0	0.0669	-1.5275	-0.2785	-0.2254	-0.2785
100.0	-3.6798	-2.5106	-0.1732	-0.1457	-0.2785
96.0	-0.1722	-0.0394	0.0669	0.0935	0.0935
84.0	0.0669	0.0669	-0.2785	0.1466	1.8738
73.5	-0.2430	-0.2430	0.1290	0.0050	0.1290
54.0	-0.1191	-0.2430	-0.2430	-0.1191	-0.2430
33.0	-0.1191	-0.2430	-0.2430	-0.2430	-0.2430
24.0	-0.1182	-0.1611	-0.1611	-0.0753	-0.1611
10.0	-0.0324	-0.1611	-0.1182	-0.0324	-0.2430
5.0	-0.0753	-0.1611	-0.0753	-0.0324	-0.2430
2.5	-0.1182	-0.1182	-0.0324	-0.0324	-0.1191

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1456
43.4	-0.0803
50.4	-0.0803
56.1	-0.0803
62.5	-0.0803
69.7	-0.1130
76.9	-0.1130
83.4	-0.1456
89.4	-0.2100
95.4	-0.2436
101.1	-0.1783
---BOTTOM---	
38.2	-0.0803
43.4	-0.0477
50.4	-0.1130
56.1	-0.0803
62.5	-0.0477
69.7	-0.0477
76.9	-0.0803
83.4	-0.1130
89.4	-0.1130
95.4	-0.0803
101.1	-0.1456

MISC.*

NO.	CP
1	-0.3670
2	-0.2430
3	-0.2430
4	-0.2430
5	-0.2430
6	-0.2430
7	-0.3051
8	-0.2785
9	-0.2520
10	-0.2785
11	-0.2785
12	-0.2785

TABLE A34. (Continued)

RUN:SEQ 266 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 632

CLAERO = 0.3305 CDAERO = 0.2044 DCLC = 0.0 DWLC = 0.3465 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.04	2139.41	2131.50	7.91	81.65	0.01	2.68	0.0	1.062	1.062	87.6179	0.518690E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.68	-0.80	-0.28	0.00	-0.00	0.00	0.40	-0.23	0.40	0.09

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2428	-0.3049
2.5	-0.2428	-0.3049
5.0	-0.2428	-0.2738
10.0	-0.2428	-0.2428
15.0	-0.3049	-0.2738
25.0	-0.2738	-0.3049
35.0	-0.2428	-0.2428
50.0	-0.2118	
56.0	-0.2738	-0.2738
65.0	-0.2428	-0.2738
76.0	-0.2738	-0.2738
79.0	-0.2428	-0.2428
80.5	-0.1664	-0.2111
81.0	-0.1664	-0.1664
82.0	-0.2111	-0.2336
84.0	-0.1440	
87.0	-0.1888	
89.0	-0.2928	-0.2266
93.0	-0.2597	
96.0	-0.2597	-0.2266
100.0	-0.2597	-0.2597
96.0	-0.2597	-0.2597
84.0	-0.2597	-0.2597
73.0	-0.2597	-0.2597
50.0	-0.3259	-0.2597
35.0	-0.2738	-0.2428
25.0	-0.2428	-0.2738
10.0	-0.2738	-0.2428
5.0	-0.2738	-0.2428
2.5	-0.2738	-0.2428

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.2590	0.2155	-0.1760	-0.2195	-0.3935
2.5	-0.2630	-0.6546	-0.8721	-0.8286	-0.9156
5.0	-0.4370	-0.4805	-0.6546	-0.6111	-0.7415
10.0	-0.2630	-0.4805	-0.5676	-0.4805	-0.5676
15.0	-0.4370	-0.4805	-0.5241	-0.4370	-0.5241
24.0	-0.3935	-0.4805	-0.4370	-0.3066	-0.4370
33.0	-0.3849	-0.5107	-0.3849	-0.5107	-0.5107
54.0	-0.3849	-0.5107	-0.5107	1.6274	-0.3849
66.0	-0.4584	-1.1537	-0.4915	-0.4253	-0.4253
78.5	2.3230	-1.3555	-0.5577	-0.5577	-0.4915
79.5	116.9989	122.7111	17.0393	18.0468	200.6927
80.5	-79.0231	-85.8553	-0.4576	-0.4127	-2.6979
81.3	62.3592	161.9876	-0.2784	-0.2111	-0.3680
82.0	0.1921	5.6582	-0.3456	-0.2336	-0.2111
84.0	-16.7891	-1.6001	-0.2560	-0.2560	-0.3007
87.0	-11.5245	-3.7732	-0.4352	-0.3007	-0.2111
89.0	7.1169	-1.8584	-0.4569	-0.4299	-0.3490
93.0	0.9986	-1.2655	-0.3221	-0.2413	-0.2413
96.0	0.1091	-1.5350	-0.2683	-0.2413	-0.3221
100.0	-3.7182	-2.5592	-0.1604	-0.1335	-0.2413
96.0	-0.1335	-0.0256	0.1630	0.1361	0.1361
84.0	0.0822	0.0552	-0.2413	0.2169	1.8880
73.5	-0.2592	-0.3849	0.1182	-0.1334	0.2439
54.0	-0.2592	-0.2592	-0.2592	-0.1334	-0.2592
33.0	-0.1334	-0.3849	-0.1334	-0.1334	-0.1334
24.0	-0.0455	-0.0020	-0.0455	0.0415	-0.0890
10.0	0.0415	0.0415	0.0850	0.1720	-0.0076
5.0	0.0850	0.1285	0.1720	0.2155	0.1182
2.5	0.0850	0.1720	0.2155	0.3025	0.2439

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0810
43.4	-0.0941
50.4	-0.0810
56.1	-0.0941
62.5	-0.1273
69.7	-0.1604
76.9	-0.1935
83.4	-0.1604
89.4	-0.1935
95.4	-0.2597
101.1	-0.2266
---BOTTOM---	
38.2	-0.0941
43.4	-0.0279
50.4	-0.0941
56.1	-0.0279
62.5	0.0052
69.7	0.0052
76.9	-0.0279
83.4	-0.0279
89.4	-0.0279
95.4	-0.0279
101.1	-0.1604

MISC.*

NO.	CP
1	-0.3349
2	-0.2592
3	-0.5107
4	-0.1334
5	-0.3849
6	-0.2592
7	-0.2683
8	-0.2413
9	-0.2683
10	-0.2413
11	-0.2413
12	-0.2413

TABLE A34. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 634

266 5 CLAERO = 0.7371 CDAERO = 0.3416 DCLC = 0.0 DWLC = 0.4890 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.06	2139.48	2131.68	7.80	81.09	0.01	2.73	0.0	1.075	1.075	89.2304	0.514612E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	1.23	-0.62	-0.37	0.00	-0.00	0.01	0.85	-0.32	0.82	0.24

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2906	-0.2906
2.5	-0.2906	-0.2906
5.0	-0.2591	-0.2591
10.0	-0.2591	-0.2906
15.0	-0.2591	-0.2906
25.0	-0.2906	-0.2906
35.0	-0.3221	-0.2906
50.0	-0.2906	
56.0	-0.2906	-0.2906
65.0	-0.2906	-0.2906
76.0	-0.3221	-0.2906
79.0	-0.2906	-0.2906
80.5	-0.1816	-0.2498
81.0	-0.2724	-0.2952
82.0	-0.2724	-0.2270
84.0	-0.2498	
87.0	-0.2270	
89.0	-0.3434	-0.3098
93.0	-0.3098	
96.0	-0.2762	-0.3098
100.0	-0.3098	-0.3098
96.0	-0.3098	-0.3098
84.0	-0.3098	-0.3098
73.0	-0.3098	-0.3098
50.0	-0.3098	-0.3434
35.0	-0.2906	-0.2906
25.0	-0.2906	-0.2906
10.0	-0.2906	-0.3221
5.0	-0.2591	-0.2906
2.5	-0.2591	-0.2906

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.0007	-3.8100	-3.2804	-2.6185	-1.4711
2.5	-0.3238	-1.9124	-2.3537	-2.5744	-1.2946
5.0	-1.1181	-1.3829	-2.0889	-2.3096	-1.1622
10.0	-0.3238	-0.9857	-1.3829	-2.2214	-1.2063
15.0	-0.8091	-0.8533	-1.1622	-1.9124	-1.2063
24.0	-0.5886	-0.7209	-0.8533	-1.2504	-1.1622
33.0	-0.5309	-0.5584	-0.7860	-0.7860	-1.0412
54.0	-0.5309	-0.5309	-0.5309	1.5105	-0.9136
65.0	-0.6122	-1.2167	-0.5450	-0.5450	-0.8137
78.5	2.2093	-1.4519	-0.6458	-0.5785	-0.8473
79.5	119.3748	125.2603	17.5206	18.7260	203.5313
80.5	-80.2880	-88.5136	-0.4997	-0.4543	-2.7267
81.3	63.0174	147.8048	-0.3861	-0.2952	-0.6815
82.0	1.2501	5.6360	-0.3407	-0.2952	-0.7724
84.0	-15.8396	-1.9995	-0.3407	-0.3179	-0.7042
87.0	-9.1353	-3.8403	-0.4543	-0.3861	-0.3179
89.0	6.1951	-2.0894	-0.5035	-0.4489	-0.7223
93.0	0.8908	-1.2419	-0.3669	-0.3122	-0.2849
96.0	0.0706	-1.5425	-0.3122	-0.3122	-0.7771
100.0	-3.8119	-2.6362	-0.2029	-0.2302	-0.3122
96.0	-0.0935	-0.0115	0.1526	0.0979	-0.0661
84.0	0.1526	0.0979	-0.3395	0.1263	1.6290
73.5	-0.1481	-0.2757	0.1071	-0.0205	0.2346
54.0	-0.0205	-0.1481	-0.1481	-0.1481	-0.1481
33.0	0.1071	-0.4033	0.1071	0.1071	-0.1481
24.0	0.1175	0.2058	0.2058	0.2941	0.0293
10.0	0.2941	0.3382	0.3823	0.4706	0.2346
5.0	0.4706	0.5147	0.5589	0.6030	0.3622
2.5	0.6030	0.5147	0.5589	0.6030	0.3622

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1419
43.4	-0.1419
50.4	-0.1755
56.1	-0.2091
62.5	-0.2762
69.7	-0.3434
76.9	-0.3098
83.4	-0.2426
89.4	-0.2762
95.4	-0.2762
101.1	-0.2426
---BOTTOM---	
38.2	-0.0411
43.4	-0.0075
50.4	-0.1083
56.1	-0.0075
62.5	0.0597
69.7	0.0932
76.9	0.0597
83.4	0.0260
89.4	-0.0075
95.4	-0.0411
101.1	-0.1755

MISC.*

NO.	CP
1	-0.6584
2	0.1071
3	-1.2964
4	-0.0205
5	-1.0412
6	-0.2757
7	-0.3395
8	-0.3122
9	-0.3122
10	-0.3395
11	-0.3122
12	-0.3395

TABLE A35. RUN 267, BW6V, DELF=0, CMU=2.0, (BN/S)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
267	1	CLAERO =	0.1452	CDAERO =	0.2717	DCLC =	0.0	DWLC =	0.5354	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.01		2139.69	2135.62	4.07	58.56	0.01	2.66	0.0	2.067	2.067	86.9890	0.371625E+06
BETA		CL	CD	CN	CROLL	CN	CV	CNTR	CNTR	CLTR	CDTR	
-0.01		0.68	-1.72	-0.35	0.01	0.00	-0.00	0.26	-0.25	0.26	0.04	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3198	-0.3198
2.5	-0.2595	-0.2595
5.0	-0.2595	-0.1991
10.0	-0.2595	-0.2595
15.0	-0.1991	-0.2595
25.0	-0.3198	-0.3198
35.0	-0.3198	-0.3198
50.0	-0.3198	
56.0	-0.2595	-0.1991
65.0	-0.2595	-0.3802
76.0	-0.2595	-0.2555
79.0	-0.2595	-0.3198
80.5	-0.2120	-0.1249
81.0	-0.2555	-0.2991
82.0	-0.1249	-0.2555
84.0	-0.2120	
87.0	-0.2120	
89.0	-0.2800	-0.2800
93.0	-0.2156	
96.0	-0.3444	-0.2800
100.0	-0.3444	-0.2800
96.0	-0.2800	-0.2800
84.0	-0.3444	-0.2800
73.0	-0.2800	-0.2800
50.0	-0.2800	-0.2800
35.0	-0.3198	-0.3198
25.0	-0.3198	-0.3198
10.0	-0.2595	-0.2595
5.0	-0.2595	-0.2595
2.5	-0.2595	-0.3198

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.6692	0.6692	0.5846	0.5001	0.6692
2.5	-0.2612	-0.0920	-0.3457	-0.2612	-0.0074
5.0	-0.0920	-0.0920	-0.2612	-0.1766	-0.1766
10.0	-0.2612	-0.0920	-0.2612	-0.0920	-0.1766
15.0	-0.1766	-0.1766	-0.3457	-0.0920	-0.2612
24.0	-0.1766	-0.1766	-0.2612	-0.0074	-0.2612
33.0	-0.4596	-0.4596	-0.4596	-0.4596	-0.4596
54.0	-0.4596	-0.4596	-0.4596	2.4746	-0.4596
65.0	-0.5375	-1.8893	-0.4731	-0.4088	-0.3444
78.5	3.7112	-1.9537	-0.6019	-0.5375	-0.4088
79.5	228.0465	239.2819	34.2393	37.2004	389.1824
80.5	*********	-166.5815	-0.4733	-0.6040	-5.0899
81.3	122.6918	312.1912	-0.2120	-0.1684	-0.2120
82.0	-1.2572	12.4618	-0.2555	-0.1684	-0.1249
84.0	-35.7949	-1.9977	-0.2991	-0.2120	-0.1249
87.0	-26.5624	-6.8320	-0.4733	-0.2555	-0.1684
89.0	13.1118	-3.3416	-0.5121	-0.4597	-0.3025
93.0	2.1603	-2.1366	-0.3549	-0.3025	-0.3025
96.0	0.5359	-2.7651	-0.3025	-0.2501	-0.3025
100.0	-6.4857	-4.5468	-0.1453	-0.0929	-0.2501
96.0	-0.3025	-0.1453	0.1167	0.1167	0.0643
84.0	0.0119	0.0119	-0.3025	0.1691	2.5271
73.5	-0.2151	-0.4596	0.0295	0.0295	0.5184
54.0	-0.2151	-0.2151	-0.2151	-0.2151	-0.2151
33.0	-0.2151	-0.4596	-0.2151	-0.2151	-0.2151
24.0	0.0772	-0.0920	-0.0920	0.0772	-0.0920
10.0	-0.0074	-0.0920	0.0772	0.1617	-0.2151
5.0	0.2463	0.0772	-0.0074	0.1617	-0.2151
2.5	-0.0074	-0.0074	0.2463	0.1617	0.0295

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0869
43.4	-0.0869
50.4	-0.0869
56.1	-0.0869
62.5	-0.0225
69.7	-0.0869
76.9	-0.1512
83.4	-0.1512
89.4	-0.2156
95.4	-0.2800
101.1	-0.2156
---BOTTOM---	
38.2	-0.0869
43.4	-0.0869
50.4	-0.1512
56.1	-0.0869
62.5	-0.0869
69.7	-0.0869
76.9	-0.0869
83.4	-0.1512
89.4	-0.0869
95.4	-0.1512
101.1	-0.2156

MISC.*

NO.	CP
1	-0.4596
2	-0.2151
3	-0.4596
4	-0.2151
5	-0.2151
6	-0.2151
7	-0.3549
8	-0.3025
9	-0.3025
10	-0.3025
11	-0.3025
12	-0.3025

TABLE A35. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 637

267 3 CLAERO = 0.3148 CDAERO = 0.2856 DCLC = 0.0 DWLC = 0.6698 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03	2139.69	2135.62	4.07	58.56	0.01	2.67	0.0	2.054	2.054	87.4372	0.371525E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.98	-1.66	-0.39	0.01	-0.00	0.00	0.44	-0.29	0.44	0.07

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.3199	-0.3199	0.0	0.3309	0.0772	-0.7685	-0.3457	-0.5994	38.2	-0.1512	1	-0.4596
2.5	-0.2595	-0.3802	2.5	-0.2612	-0.5148	-1.0223	-0.9377	-1.1915	43.4	-0.0669	2	0.0295
5.0	-0.2595	-0.3802	5.0	-0.5148	-0.4303	-0.5994	-0.6840	-0.8531	50.4	-0.1512	3	-0.2151
10.0	-0.2595	-0.3802	10.0	-0.3457	-0.4303	-0.6840	-0.4303	-0.5994	56.1	-0.1512	4	0.0295
15.0	-0.3199	-0.3802	15.0	-0.4303	-0.4303	-0.4303	-0.3457	-0.5148	62.5	-0.1512	5	-0.2151
25.0	-0.3199	-0.3802	24.0	-0.2612	-0.4303	-0.4303	-0.2612	-0.4303	69.7	-0.1512	6	0.0295
35.0	-0.3199	-0.2595	33.0	-0.2151	-0.2151	-0.2151	-0.2151	-0.2151	76.9	-0.2156	7	-0.2501
50.0	-0.3199		54.0	-0.2151	-0.2151	-0.2151	2.7191	-0.2151	83.4	-0.1512	8	-0.2501
56.0	-0.3199	-0.3199	65.0	-0.5375	-2.0181	-0.4731	-0.4731	-0.4088	89.4	-0.2156	9	-0.3025
65.0	-0.2595	-0.3802	78.5	3.3249	-1.8893	-0.6663	-0.6019	-0.5375	95.4	-0.2800	10	-0.3549
76.0	-0.3802	-0.3199	79.5	228.2211	239.4146	34.9769	38.8985	389.2314	101.1	-0.2800	11	-0.3025
79.0	-0.3802	-0.3199	80.5	*********	-166.1465	-0.5169	-0.5604	-4.9156	---BOTTOM---		12	-0.3025
80.5	-0.2991	-0.2120	81.3	122.7361	280.3921	-0.1684	-0.1249	-0.2555	38.2	-0.0669		
81.0	-0.2120	-0.2555	82.0	-0.3862	13.0280	-0.2120	-0.1684	-0.2120	43.4	-0.0225		
82.0	-0.0814	-0.2555	84.0	-33.9641	-2.0412	-0.0378	-0.2555	-0.2555	50.4	-0.1512		
84.0	-0.2120		87.0	-25.3410	-7.0497	-0.5169	0.0929	-0.2991	56.1	-0.0225		
87.0	-0.1249		89.0	13.8453	-3.5513	-0.5121	-0.4073	-0.3549	62.5	-0.0669		
89.0	-0.3444	-0.2800	93.0	2.2127	-2.1366	-0.4073	-0.2501	-0.3549	69.7	-0.0225		
93.0	-0.2800		96.0	0.5883	-2.9225	-0.3549	-0.2501	-0.3549	76.9	-0.0225		
96.0	-0.2800	-0.2800	100.0	-6.5380	-4.5468	-0.1977	-0.1453	-0.3025	83.4	-0.0669		
100.0	-0.3444	-0.2800	96.0	-0.2501	-0.1453	0.0643	0.0643	0.1167	89.4	-0.0669		
96.0	-0.3444	-0.3444	84.0	0.0119	0.0119	-0.3025	0.1691	2.3698	95.4	-0.1512		
84.0	-0.3444	-0.2800	73.5	0.0295	-0.2151	0.2740	0.0295	1.0075	101.1	-0.2156		
73.0	-0.3444	-0.3444	54.0	0.0295	0.0295	0.0295	0.0295	0.0295				
50.0	-0.2800	-0.2800	33.0	0.0295	-0.2151	0.0295	0.2740	0.0295				
35.0	-0.3199	-0.3199	24.0	-0.0074	0.0772	-0.0074	0.2463	-0.0920				
25.0	-0.3199	-0.3199	10.0	0.1617	0.0772	-0.0074	0.3309	0.2740				
10.0	-0.3199	-0.4405	5.0	0.2463	0.2463	0.3309	0.4155	0.2740				
5.0	-0.3199	-0.3199	2.5	0.3309	0.3309	0.3309	0.4155	0.2740				
2.5	-0.3199	-0.3199										

TABLE A35. (Concluded)

RUN:SEQ 267 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 639

CLAERO = 0.6999 CDAERO = 0.4505 DCLC = 0.0 DWLC = 0.9673 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.00	2139.62	2135.67	3.96	57.76	0.01	2.72	0.0	2.130	2.130	89.0625	0.366065E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.67	-1.45	-0.48	0.00	-0.00	0.01	0.90	-0.38	0.87	0.24

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4416	-0.3795
2.5	-0.3795	-0.3795
5.0	-0.4416	-0.3795
10.0	-0.3174	-0.3795
15.0	-0.4416	-0.4416
25.0	-0.4416	-0.3795
35.0	-0.5036	-0.4416
50.0	-0.5036	
56.0	-0.3795	-0.4416
65.0	-0.4416	-0.5036
76.0	-0.4416	-0.4416
79.0	-0.4416	-0.5036
80.5	-0.4029	-0.3133
81.0	-0.2238	-0.3133
82.0	-0.3581	-0.4029
84.0	-0.3581	
87.0	-0.3133	
89.0	-0.4047	-0.3385
93.0	-0.4047	
96.0	-0.3385	-0.4047
100.0	-0.4047	-0.4047
96.0	-0.3385	-0.4709
84.0	-0.4047	-0.4047
73.0	-0.3385	-0.4709
50.0	-0.4047	-0.4047
35.0	-0.5036	-0.3795
25.0	-0.5036	-0.4416
10.0	-0.4416	-0.3795
5.0	-0.3795	-0.3795
2.5	-0.3174	-0.3795

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.4069	-3.4508	-3.1028	-2.0591	-1.4500
2.5	-0.4931	-2.4069	-2.6680	-1.7110	-1.1021
5.0	-1.2760	-1.4500	-2.7550	-1.7110	-1.2760
10.0	-0.4931	-1.1890	-2.4939	-1.5370	-1.1890
15.0	-0.7541	-0.8410	-2.1460	-1.5370	-1.2760
24.0	-0.7541	-0.7541	-0.9280	-1.4500	-1.2760
33.0	-0.5233	-0.5233	-0.7747	-1.0262	-1.0262
54.0	-0.5233	-0.5233	-0.2717	2.2434	-0.7747
65.0	-0.7358	-2.0601	-0.6034	-0.6696	-0.8020
78.5	2.9723	-2.0601	-0.3696	-0.5371	-0.8020
79.5	234.0284	247.0211	36.1972	40.0951	400.1919
80.5	*********	-172.3844	-0.6717	-0.7613	-5.0620
81.3	126.0636	270.3923	-0.5821	-0.4029	-0.7165
82.0	-0.9853	13.2605	-0.4477	-0.4029	-0.8061
84.0	-35.2999	-1.6574	-0.4477	-0.4477	-0.8957
87.0	-25.1308	-7.4363	-0.5821	-0.3133	-0.2238
89.0	14.1371	-3.8649	-0.5772	-0.5233	-0.9003
93.0	2.0099	-2.5176	-0.4694	-0.4155	-0.4694
96.0	0.3930	-3.1104	-0.4694	-0.4155	-0.8467
100.0	-6.7756	-4.7812	-0.3516	-0.4155	-0.4694
96.0	-0.3616	-0.1461	0.0157	-0.0382	-0.1461
84.0	0.0157	0.0695	-0.4694	-0.0382	1.7405
73.5	-0.0202	-0.2717	0.2314	0.2314	0.7343
54.0	0.2314	0.2314	0.2314	0.2314	-0.0202
33.0	0.2314	-0.2717	0.2314	0.2314	0.2314
24.0	0.2029	0.0289	0.1159	0.2899	-0.0581
10.0	0.3769	0.2899	0.3769	0.4639	0.4828
5.0	0.5508	0.3769	0.4639	0.6379	0.4828
2.5	0.7248	0.4639	0.5508	0.5508	0.4828

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.2061	38.2	-0.0736
43.4	-0.2061	43.4	-0.0074
50.4	-0.2061	50.4	-0.2061
56.1	-0.2061	56.1	-0.0074
62.5	-0.2723	62.5	-0.0074
69.7	-0.3385	69.7	0.0588
76.9	-0.2723	76.9	-0.0074
83.4	-0.2723	83.4	-0.0074
89.4	-0.3385	89.4	-0.0736
95.4	-0.3385	95.4	-0.2061
101.1	-0.2061	101.1	-0.2723

MISC.*

NO.	CP
1	-0.5233
2	0.2314
3	-1.2777
4	0.2314
5	-1.0262
6	-0.0202
7	-0.4694
8	-0.4694
9	-0.4694
10	-0.4694
11	-0.4155
12	-0.4155

TABLE A36. RUN 271, BC1W6V, DELF=0, CHU=0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
271	2	CLAERO =	0.1509	CDAERO =	0.0576	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.11		2139.20	2109.14	30.03	160.29	0.01	2.67	0.0	0.0	0.0	87.4555	0.998841E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.15	0.06	-0.11	0.00	-0.00	0.01	0.15	-0.11	0.15	0.09	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3561	0.4051
2.5	-0.0033	-0.0196
5.0	-0.0586	-0.1013
10.0	0.9441	-0.1829
15.0	-0.2075	-0.2238
25.0	-0.1829	-0.2156
35.0	-0.1666	-0.1666
50.0	-0.1585	
55.0	-0.1666	-0.1503
65.0	-0.2156	-0.1829
76.0	-0.1829	-0.2973
79.0	-0.4117	-0.4117
80.5	-0.4196	-0.4078
81.0	-0.4196	-0.4078
82.0	-0.4313	-0.3842
84.0	-0.1483	
87.0	-0.1189	
89.0	-0.1403	-0.1664
93.0	-0.1315	
96.0	-0.0890	-0.1054
100.0	0.8357	-0.1141
96.0	0.0776	0.1037
84.0	0.8357	-0.1141
73.0	0.0776	0.1037
50.0	0.1299	0.1299
35.0	-0.0115	-0.0278
25.0	-0.1829	-0.2156
10.0	-0.1666	-0.2075
5.0	-0.2238	-0.2728
2.5	-0.3708	-0.3790

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5548	0.5319	0.5090	0.5090	0.5205
2.5	-0.1779	-0.1207	-0.2123	-0.1207	-0.1207
5.0	-0.0978	-0.1550	-0.2237	-0.1894	-0.2008
10.0	-0.1779	-0.2237	-0.2352	-0.2352	-0.2008
15.0	-0.1665	-0.2123	-0.2466	-0.2581	-0.2237
24.0	-0.1665	-0.2237	-0.2123	-0.2008	-0.2123
33.0	-0.1903	-0.1903	-0.2234	-0.2565	-0.2234
54.0	-0.1903	-0.1903	-0.2565	0.9351	-0.2234
65.0	-0.2187	-0.2536	-0.2884	-0.2710	-0.2449
75.5	1.2017	-0.4801	-0.3581	-0.3756	-0.3145
73.5	-0.4313	-0.4549	1.3669	7.9112	53.5396
80.5	-0.4313	-0.4490	-0.2132	-0.2840	-0.9148
81.3	-0.4373	53.5396	-0.1896	-0.1896	-0.3016
82.0	-0.3342	-0.9148	-0.2073	-0.2132	-0.2191
84.0	-0.2014	-0.2604	-0.2427	-0.2899	-0.2604
87.0	-0.2368	-0.2780	-0.3252	-0.2899	-0.1955
89.0	-0.2613	-0.2897	-0.3535	-0.3819	-0.2755
93.0	-0.2187	-0.2258	-0.2613	-0.2046	-0.1975
96.0	-0.1762	-0.1691	-0.1833	-0.1975	-0.2471
100.0	-0.1336	-0.1194	-0.0840	-0.0840	-0.1904
96.0	0.0650	0.1217	0.1359	0.1430	0.1288
84.0	0.1359	0.0934	-0.2258	0.1643	1.1645
73.5	-0.0248	-0.1572	0.0414	0.0745	0.2069
54.0	-0.0910	-0.0910	-0.1241	-0.1241	-0.1241
33.0	-0.1241	-0.1572	-0.1572	-0.1572	-0.1572
24.0	-0.1321	-0.1665	-0.1665	-0.0978	-0.1550
10.0	-0.1092	-0.1321	-0.1207	-0.0978	-0.1572
5.0	-0.0978	-0.1665	-0.1321	-0.0520	-0.1572
2.5	-0.0863	-0.1436	-0.0863	-0.0520	-0.0910

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0967
43.4	-0.0880
50.4	-0.0619
56.1	-0.0531
62.5	-0.0531
69.7	-0.0880
76.9	-0.0793
83.4	-0.0531
89.4	-0.0880
95.4	-0.1664
101.1	-0.1403
---BOTTOM---	
38.2	-0.0619
43.4	-0.0096
50.4	-0.0793
56.1	-0.0183
62.5	-0.0183
69.7	-0.0619
76.9	-0.0531
83.4	-0.0444
89.4	-0.0357
95.4	-0.0008
101.1	-0.0705

MISC.*

NO.	CP
1	-0.2234
2	-0.1241
3	-0.2234
4	-0.1572
5	-0.1903
6	-0.1572
7	-0.2046
8	-0.1904
9	-0.2046
10	-0.1904
11	-0.1975
12	-0.1975

TABLE A36. (Continued)

RUN:SEQ 271 4 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 842

CLAERO = 0.3537 CDAERO = 0.0769 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.02	2139.13	2109.07	30.06	160.44	0.01	2.66	0.0	0.0	0.0	86.9525	0.996487E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
-0.01	0.35	0.08	-0.07	0.00	-0.00	0.01	0.36	-0.07	0.35	0.11

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3610	0.1241
2.5	-0.5375	-0.8152
5.0	-0.4477	-0.6845
10.0	0.9062	-0.5048
15.0	-0.4231	-0.4395
25.0	-0.3415	-0.3496
35.0	-0.2435	-0.2353
50.0	-0.2516	
56.0	-0.2190	-0.2190
65.0	-0.2598	-0.2680
76.0	-0.1945	-0.3496
79.0	-0.4395	-0.4395
80.5	-0.4347	-0.4347
81.0	-0.4288	-0.4288
82.0	-0.4111	-0.3463
84.0	-0.1753	
87.0	-0.1576	
89.0	-0.1523	-0.1872
93.0	-0.1349	
96.0	-0.1000	-0.1262
100.0	0.6930	-0.0913
96.0	0.1091	0.1004
84.0	0.6930	-0.0913
73.0	0.1091	0.1004
50.0	0.1788	0.1527
35.0	0.0016	-0.0556
25.0	-0.1209	-0.2925
10.0	-0.0883	-0.0964
5.0	-0.1454	-0.1046
2.5	0.0996	0.1568

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4599	0.3340	-0.0095	-0.4102	-0.5705
2.5	-0.2042	-0.4217	-0.7652	-0.7766	-0.9713
5.0	-0.2385	-0.3988	-0.5019	-0.6278	-0.6965
10.0	-0.1927	-0.3873	-0.4675	-0.4789	-0.5133
15.0	-0.2958	-0.3186	-0.4217	-0.4560	-0.4446
24.0	-0.2499	-0.3072	-0.3415	-0.3415	-0.3645
33.0	-0.2267	-0.2598	-0.2929	-0.2929	-0.2929
54.0	-0.1936	-0.2598	-0.2929	0.8987	-0.2929
65.0	-0.2569	-0.3004	-0.3353	-0.3179	-0.3179
78.5	1.1723	-0.4922	-0.3789	-0.3789	-0.3876
79.5	-0.4347	-0.4406	1.4049	8.4271	53.5613
80.5	-0.4465	-0.4524	-0.2224	-0.2932	-0.8651
81.3	-0.4583	53.5613	-0.2224	-0.2165	-0.3521
82.0	-0.4052	-0.8356	-0.2460	-0.2401	-0.2696
84.0	-0.2224	-0.2401	-0.2519	-0.2755	-0.3050
87.0	-0.2401	-0.2932	-0.3345	-0.2873	-0.1458
89.0	-0.2930	-0.3143	-0.3498	-0.3640	-0.3569
93.0	-0.2383	-0.2504	-0.2434	-0.2150	-0.2150
96.0	-0.1937	-0.1937	-0.1795	-0.2008	-0.3214
100.0	-0.1653	-0.1228	-0.0802	-0.1086	-0.2079
96.0	0.0830	0.1326	0.1610	0.1397	0.1113
84.0	0.1468	0.1042	-0.2008	0.1894	1.1186
73.5	0.0381	-0.1605	0.1043	0.1043	0.1705
54.0	-0.0612	-0.0612	-0.0612	-0.0612	-0.0943
33.0	-0.0281	-0.1605	-0.0281	-0.0281	-0.0612
24.0	-0.0324	-0.0568	-0.0439	-0.0095	-0.1011
10.0	0.0134	0.0019	0.0363	0.1279	0.1043
5.0	0.0821	0.0592	0.1393	0.1851	0.1705
2.5	0.0706	0.0935	0.2195	0.2882	0.3360

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1436
43.4	-0.1262
50.4	-0.0913
56.1	-0.0652
62.5	-0.0739
69.7	-0.1174
76.9	-0.1087
83.4	-0.0739
89.4	-0.1610
95.4	-0.1784
101.1	-0.1523
---BOTTOM---	
38.2	-0.0390
43.4	0.0220
50.4	-0.0652
56.1	0.0046
62.5	0.0046
69.7	0.0133
76.9	-0.0129
83.4	-0.0129
89.4	0.0046
95.4	0.0133
101.1	-0.0739

MISC.*

NO.	CP
1	-0.2929
2	0.0050
3	-0.3260
4	-0.0281
5	-0.2929
6	-0.0943
7	-0.2221
8	-0.2076
9	-0.2150
10	-0.2150
11	-0.2221
12	-0.2150

TABLE A36. (Concluded)

RUN:SEQ 271 6 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 644

CLAERO = 0.8522 CDAERO = 0.2007 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	XQT	RN
12.07	2138.99	2108.93	30.06	160.57	0.01	2.71	0.0	0.0	0.0	88.7508	0.894565E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.85	0.20	0.00	0.00	-0.00	0.01	0.87	-0.00	0.84	0.23

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-4.7896	-1.7838
2.5	-2.1759	-1.5551
5.0	-1.4653	-1.4489
10.0	0.8707	-1.3999
15.0	-0.9344	-1.3428
25.0	-0.6812	-1.3673
35.0	-0.5342	-1.1467
50.0	-0.3953	
56.0	-0.3871	-0.5178
65.0	-0.3708	-0.4607
76.0	-0.2565	-0.4280
79.0	-0.5015	-0.5097
80.5	-0.4785	-0.5552
81.0	-0.4962	-0.4078
82.0	-0.4196	-0.3193
84.0	-0.2132	
87.0	-0.2191	
89.0	-0.3187	-0.3146
93.0	-0.1838	
96.0	-0.1490	-0.2623
100.0	0.5830	-0.2710
96.0	0.1125	0.0340
84.0	0.5830	-0.2710
73.0	0.1125	0.0340
50.0	0.2258	0.1125
35.0	0.0702	-0.0033
25.0	0.0376	-0.4280
10.0	0.1356	0.0947
5.0	0.1438	0.1438
2.5	0.5685	0.5766

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.3268	-1.5518	-3.9333	-3.7044	-1.5747
2.5	-0.2352	-1.1168	-1.8152	-3.3494	-1.3916
5.0	-0.6130	-0.8878	-1.3343	-2.9030	-1.3916
10.0	-0.2123	-0.7275	-0.9908	-0.9107	-1.3114
15.0	-0.5557	-0.6702	-0.8535	-0.6931	-1.2427
24.0	-0.4870	-0.5443	-0.6702	-0.6244	-1.0710
33.0	-0.4551	-0.5213	-0.5544	-0.5544	-1.1171
54.0	-0.3558	-0.4220	-0.4220	0.7697	-0.8854
65.0	-0.3663	-0.4104	-0.4366	-0.4104	-0.8026
78.5	1.1147	-0.5760	-0.4801	-0.4714	-0.7416
79.5	-0.5198	-0.5257	1.3905	8.9963	53.5646
80.5	-0.5198	-0.4962	-0.3193	-0.3665	-0.8912
81.3	-0.5080	52.7030	-0.2958	-0.3135	-0.6672
82.0	-0.4667	-0.8382	-0.3135	-0.3135	-0.7085
84.0	-0.2722	-0.2899	-0.3193	-0.3429	-0.7026
87.0	-0.2899	-0.3193	-0.3665	-0.3429	-0.2309
89.0	-0.3464	-0.3677	-0.4103	-0.3890	-0.7224
93.0	-0.2226	-0.2826	-0.3039	-0.2755	-0.2755
96.0	-0.2400	-0.2329	-0.2289	-0.2329	-0.6018
100.0	-0.1904	-0.1336	-0.1123	-0.1691	-0.2613
96.0	0.0934	0.1430	0.1501	0.1218	-0.0130
84.0	0.1927	0.1430	-0.2684	0.1714	1.0368
73.5	0.1076	-0.2896	0.1076	0.1076	0.1738
54.0	0.0745	0.0414	0.0414	0.0083	-0.0579
33.0	0.1076	-0.2896	0.1076	0.1076	-0.0248
24.0	0.1541	0.1541	0.1885	0.2114	0.0854
10.0	0.2800	0.2800	0.3716	0.4060	0.2731
5.0	0.3716	0.3831	0.4280	0.5205	0.4055
2.5	0.4861	0.4403	0.4976	0.5205	0.4386

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2536
43.4	-0.2013
50.4	-0.1490
56.1	-0.1490
62.5	-0.1838
69.7	-0.2361
76.9	-0.2187
83.4	-0.1751
89.4	-0.2100
95.4	-0.2274
101.1	-0.1926
---BOTTOM---	
38.2	0.0427
43.4	0.0863
50.4	-0.0270
56.1	0.0689
62.5	0.1038
69.7	0.1212
76.9	0.1212
83.4	0.0776
89.4	0.0602
95.4	0.0515
101.1	-0.0792

MISC.*

NO.	CP
1	-0.5213
2	0.1076
3	-0.8854
4	0.0745
5	-0.8854
6	-0.0910
7	-0.2826
8	-0.2755
9	-0.2755
10	-0.2755
11	-0.2826
12	-0.2826

TABLE A37. RUN 272, BC1W6V, DELF=0, CMU=0.5, (BN/B)W=0.25

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 645

RUN:SEQ												
272	1	CLAERO =	0.2138	CDAERO =	0.1794	DCLC =	0.0491	DWLC =	0.1273	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.03		2138.77	2120.02	18.76	126.58	0.01	2.66	0.282	0.491	0.773	87.0914	0.787667E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.39	-0.57	-0.19	0.00	-0.00	0.01	0.22	-0.15	0.22	0.05	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	0.4001	0.5179
2.5	-0.0448	-0.1365
5.0	-0.1234	-0.1888
10.0	0.8844	-0.2150
15.0	-0.2673	-0.2542
25.0	-0.2412	-0.2281
35.0	-0.2281	-0.2542
50.0	-0.2542	
56.0	-0.2804	-0.2673
65.0	-0.3851	-0.4374
76.0	-0.3066	-0.7778
79.0	-1.1835	-1.2489
80.5	-17.6854	-25.5552
81.0	0.3220	6.4630
82.0	0.3031	-0.6890
84.0	0.2086	
87.0	1.5123	
89.0	0.2341	-1.0785
93.0	0.4715	
96.0	1.1977	0.3598
100.0	0.3878	-1.5812
96.0	-0.0871	-0.0452
84.0	0.3078	-1.5812
73.0	-0.0871	-0.0452
50.0	0.0945	0.0945
35.0	-0.0448	-0.0710
25.0	-0.2019	-0.3851
10.0	-0.1888	-0.2019
5.0	-0.2673	-0.2542
2.5	-0.3066	-0.3066

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.2683	0.3234	0.4334	0.4885	0.4885
2.5	-0.3187	0.3050	-0.0619	-0.1903	-0.2821
5.0	0.4518	0.1766	-0.1353	-0.2086	-0.2454
10.0	-0.3004	-0.0252	-0.2637	-0.2454	-0.2821
15.0	-0.0436	-0.1536	-0.2454	-0.2454	-0.2821
24.0	-0.1720	-0.2086	-0.2454	-0.1903	-0.2454
33.0	-0.1647	-0.2178	-0.2178	-0.2178	-0.2178
54.0	-0.1647	-0.2178	-0.2708	0.8961	-0.2178
65.0	-0.3105	-0.5478	-0.3943	-0.3244	-0.2825
78.5	1.2675	-1.3717	-0.6177	-0.4640	-0.3663
79.5	61.5331	64.5663	2.1076	9.2500	65.2284
80.5	-32.5369	-45.2721	-0.2260	-0.3110	-1.4447
81.3	-3.3437	65.2284	-0.2543	-0.1977	-0.2733
82.0	30.3652	-8.3698	-0.3016	-0.2449	-0.2071
84.0	3.0523	0.1614	-0.4055	-0.3110	-0.2638
87.0	-9.3427	-2.0589	-0.5378	-0.3205	-0.2638
89.0	-11.4670	-1.1461	-0.5892	-0.4300	-0.3277
93.0	3.5938	-0.8051	-0.4414	-0.3618	-0.3504
96.0	-0.3504	-0.8733	-0.3163	-0.2481	-0.2709
100.0	-1.9304	-1.2484	-0.1572	-0.1345	-0.3618
96.0	-0.0663	0.0247	0.0615	0.1042	0.1042
84.0	0.0701	0.0474	-0.3504	0.1270	0.9340
73.5	-0.2178	-0.2708	0.1535	0.1005	0.2596
54.0	-0.0586	-0.1117	-0.1117	-0.1117	-0.1117
33.0	-0.1117	-0.2708	-0.1647	-0.1117	-0.1117
24.0	-0.1353	-0.1903	-0.1536	-0.0802	-0.1536
10.0	-0.1170	-0.1720	-0.1353	-0.0252	-0.1117
5.0	-0.1536	-0.2086	-0.1170	-0.0252	-0.1117
2.5	-0.2637	-0.2821	-0.1536	-0.0986	-0.0056

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1568
43.4	-0.1429
50.4	-0.0731
56.1	-0.0591
62.5	-0.0452
69.7	-0.0731
76.9	-0.0591
83.4	-0.0731
89.4	-0.1568
95.4	-0.2267
101.1	-0.2127
---BOTTOM---	
38.2	-0.0691
43.4	-0.0172
50.4	-0.1010
56.1	-0.0172
62.5	-0.0172
69.7	-0.0452
76.9	-0.0591
83.4	-0.0591
89.4	-0.0452
95.4	-0.0312
101.1	-0.1289

MISC.*

NO.	CP
1	-0.2708
2	-0.1117
3	-0.2178
4	-0.1117
5	-0.2178
6	-0.1647
7	-0.3618
8	-0.3618
9	-0.3391
10	-0.3618
11	-0.3618
12	-0.3504

TABLE A37. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 647

RUN	SEQ	CLAERO =	CDAERO =	DCLC =	DWLC =	BASEPR =
272	3	0.3997	0.2141	0.0707	0.1654	8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.03	2138.63	2120.21	18.42	125.45	0.01	2.66	0.291	0.507	0.799	87.0853	0.780197E+06

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.64	-0.55	-0.16	0.00	-0.00	0.01	0.43	-0.11	0.42	0.08

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.2659	0.1059
2.5	-0.6138	-0.8404
5.0	-0.4806	-0.6405
10.0	1.1455	-0.6005
15.0	-0.4539	-0.5605
25.0	-0.3872	-0.4139
35.0	-0.3206	-0.3339
50.0	-0.3339	
56.0	-0.3473	-0.3339
65.0	-0.4139	-0.4939
76.0	-0.3339	-0.8537
79.0	-1.2536	-1.3202
80.5	-18.4560	-25.9891
81.0	0.9412	6.7910
82.0	0.6044	-0.9446
84.0	0.1522	
87.0	1.4992	
89.0	0.2452	-1.2338
93.0	0.4727	
96.0	1.1980	0.2736
100.0	0.6292	-1.6178
96.0	-0.0392	-0.0108
84.0	0.6292	-1.6178
73.0	-0.0392	-0.0108
50.0	0.1457	0.1314
35.0	-0.0007	-0.0674
25.0	-0.1207	-0.4406
10.0	-0.0940	-0.1074
5.0	-0.1607	-0.1207
2.5	0.0793	0.1859

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4944	0.4197	0.1021	-0.1408	-0.2716
2.5	-0.3276	0.0087	-0.5705	-0.7574	-0.8255
5.0	0.2142	-0.1034	-0.4584	-0.5705	-0.7387
10.0	-0.3463	-0.3090	-0.4584	-0.4771	-0.5332
15.0	-0.1595	-0.3837	-0.4211	-0.4211	-0.4958
24.0	-0.2155	-0.3276	-0.3650	-0.3463	-0.3837
33.0	-0.2435	-0.2975	-0.3515	-0.2975	-0.2975
54.0	-0.2435	-0.2975	-0.3515	1.0629	-0.2975
65.0	-0.3663	-0.5939	-0.3948	-0.3521	-0.3521
78.5	1.4642	-1.4044	-0.5512	-0.4659	-0.4090
79.5	62.6819	65.6352	2.1439	9.7834	86.7753
80.5	-33.0613	-45.9539	-0.2807	-0.3866	-1.4257
81.3	-3.7541	82.7416	-0.3385	-0.2615	-0.3577
82.0	31.0467	-6.4300	-0.3577	-0.2711	-0.3000
84.0	3.0290	-0.0694	-0.3866	-0.3385	-0.3289
87.0	-9.4788	-2.2243	-0.4828	-0.3192	-0.3096
89.0	-11.8040	-1.2121	-0.5175	-0.4018	-0.3756
93.0	3.7540	-0.7838	-0.3670	-0.3439	-0.3439
96.0	-0.0198	-0.8879	-0.2629	-0.2397	-0.3323
100.0	-1.9298	-1.2468	-0.1239	-0.1355	-0.3439
96.0	-0.0429	0.0381	0.1423	0.1307	0.0960
84.0	0.1075	0.0844	-0.3439	0.1770	1.1494
73.5	-0.1895	-0.2975	0.1887	0.0806	0.1346
54.0	-0.0274	-0.0814	-0.0814	-0.0814	-0.1354
33.0	-0.0814	-0.2975	-0.0814	-0.0274	-0.0814
24.0	-0.1034	-0.1034	-0.0847	-0.0287	-0.0847
10.0	-0.0847	-0.0847	-0.0100	0.0847	0.0266
5.0	-0.0661	-0.0661	0.0847	0.1768	0.1887
2.5	-0.1221	-0.1221	0.1208	0.2142	0.2427

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1814
43.4	-0.1872
50.4	-0.1103
56.1	-0.0819
62.5	-0.0677
69.7	-0.0961
76.9	-0.0961
83.4	-0.0819
89.4	-0.1814
95.4	-0.2383
101.1	-0.2098
---BOTTOM---	
38.2	-0.0250
43.4	0.0176
50.4	-0.0819
56.1	0.0319
62.5	0.0319
69.7	-0.0108
76.9	-0.0108
83.4	-0.0250
89.4	-0.0108
95.4	-0.0108
101.1	-0.1245

MISC.*

NO.	CP
1	-0.2975
2	-0.0274
3	-0.3515
4	-0.0274
5	-0.2975
6	-0.1354
7	-0.3555
8	-0.3555
9	-0.3555
10	-0.3670
11	-0.3323
12	-0.3439

TABLE A37. (Concluded)

RUN:SEQ 272 B

PROPLULSIVE WING / CANARD PRESSURES

CLAERO = 0.9372 CDAERO = 0.3358 DCLC = 0.1093 DWLC = 0.2310 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.14	2138.35	2120.38	17.97	123.92	0.01	2.73	0.290	0.506	0.797	89.2004	0.770235E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.28	-0.38	-0.08	-0.00	-0.01	0.02	0.99	-0.03	0.97	0.22

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-5.3245	-2.3869
2.5	-2.2776	-1.9770
5.0	-1.6900	-1.9633
10.0	1.4800	-1.8813
15.0	-1.0615	-1.7447
25.0	-0.8019	-1.7993
35.0	-0.6515	-1.4031
50.0	-0.5695	
56.0	-0.5550	-0.6106
65.0	-0.6106	-0.6242
76.0	-0.3920	-0.8292
79.0	-1.3894	-1.2391
80.5	-19.3366	-26.5767
81.0	1.1596	7.2061
82.0	0.5875	-0.9710
84.0	0.0746	
87.0	1.4555	
89.0	0.1416	-1.2726
93.0	0.3749	
96.0	1.1184	0.2728
100.0	0.8560	-1.6808
96.0	-0.0042	-0.0333
84.0	0.8560	-1.6808
73.0	-0.0042	-0.0333
50.0	0.1999	0.1271
35.0	0.0863	-0.0094
25.0	0.0316	-0.6379
10.0	0.1546	0.0999
5.0	0.1409	0.1409
2.5	0.5782	0.5645

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4271	-0.2816	-3.1546	-4.3230	-1.6606
2.5	-0.3773	-0.7412	-1.6606	-3.5760	-1.4499
5.0	-0.3007	-0.6646	-1.1626	-2.5034	-1.3925
10.0	-0.3773	-0.6838	-0.9902	-0.8753	-1.5268
15.0	-0.5306	-0.6838	-0.8178	-0.8178	-1.3733
24.0	-0.5497	-0.6071	-0.6838	-0.6455	-1.2392
33.0	-0.5488	-0.6041	-0.6041	-0.6595	-1.1025
54.0	-0.4380	-0.4934	-0.5488	1.2233	-0.8810
65.0	-0.5144	-0.7623	-0.5290	-0.5144	-0.8206
78.5	1.4683	-1.4766	-0.6019	-0.5873	-0.9081
79.5	64.5035	67.4644	2.3136	10.9539	88.9470
80.5	-33.8858	-47.2312	-0.4482	-0.5074	-1.5036
81.3	-4.3738	76.9089	-0.4186	-0.3988	-0.7244
82.0	32.0125	-8.8518	-0.4580	-0.4186	-0.6947
84.0	3.3197	0.2324	-0.4679	-0.4778	-0.7835
87.0	-10.6868	-2.3025	-0.5074	-0.4383	-0.3988
89.0	-12.0876	-1.2767	-0.5410	-0.5172	-0.7190
93.0	3.5057	-0.8139	-0.4223	-0.4342	-0.4223
96.0	-0.3867	-0.8970	-0.3036	-0.3155	-0.6715
100.0	-1.9887	-1.3123	-0.1612	-0.2205	-0.4223
96.0	-0.0069	0.0880	0.1592	0.0999	-0.0188
84.0	0.1711	0.1236	-0.4342	0.1711	1.2154
73.5	-0.1058	-0.4380	0.1711	0.1158	0.2819
54.0	0.0050	0.0050	0.0050	0.0050	-0.1058
33.0	0.1158	-0.3826	0.1158	0.1158	0.0050
24.0	0.1207	0.1207	0.1590	0.2548	0.1207
10.0	0.1973	0.1973	0.3122	0.4271	0.2819
5.0	0.2165	0.2548	0.4271	0.5229	0.4480
2.5	0.2548	0.3122	0.4655	0.5038	0.4480

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3541
43.4	-0.2958
50.4	-0.2063
56.1	-0.1645
62.5	-0.1937
69.7	-0.2228
76.9	-0.2228
83.4	-0.2063
89.4	-0.2686
95.4	-0.2958
101.1	-0.2520
---BOTTOM---	
38.2	0.0250
43.4	0.0687
50.4	-0.0771
56.1	0.0687
62.5	0.0833
69.7	0.0833
76.9	0.0687
83.4	0.0541
89.4	0.0396
95.4	0.0250
101.1	-0.1354

MISC.*

NO.	CP
1	-0.6041
2	0.1158
3	-0.6041
4	0.1158
5	-0.9364
6	-0.1058
7	-0.4579
8	-0.4342
9	-0.4342
10	-0.4400
11	-0.4342
12	-0.4342

TABLE A38. RUN 273, BC1W6V, DELF=0, CMU=1.0, (BN/B)W=0.25

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 650

273	1	CLAERO =	0.2216	CDAERO =	0.3187	DCLC =	0.1037	DWLC =	0.2683	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
0.06		2138.35	2129.31	9.04	87.76	0.01	2.66	0.594	1.033	1.626	87.0442	0.647491E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.59	-1.26	-0.30	0.00	-0.00	0.02	0.24	-0.20	0.24	0.05	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3819	0.4090
2.5	0.0017	-0.2155
5.0	-0.1341	-0.2427
10.0	2.5541	-0.2698
15.0	-0.3241	-0.2970
25.0	-0.3241	-0.2970
35.0	-0.3241	-0.3241
50.0	-0.3784	
56.0	-0.4327	-0.3784
65.0	-0.5413	-0.5956
76.0	-0.5142	-1.1930
79.0	-1.8719	-1.9803
80.5	-37.7052	-52.7184
81.0	3.3193	15.0597
82.0	2.1040	-1.3261
84.0	0.4968	
87.0	3.5936	
89.0	0.7746	-1.8328
93.0	1.2092	
96.0	2.5998	0.9485
100.0	0.8905	-3.0785
96.0	-0.2683	-0.2394
84.0	0.8905	-3.0785
73.0	-0.2683	-0.2394
50.0	0.0503	0.0503
35.0	-0.2427	-0.1882
25.0	-0.3784	-0.5413
10.0	-0.2427	-0.3241
5.0	-0.2970	-0.2970
2.5	-0.4327	-0.4327

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4780	-0.1736	0.3593	0.4735	0.4735
2.5	-0.4780	1.1586	0.0929	-0.1355	-0.2116
5.0	1.4631	0.7400	-0.1736	-0.2116	-0.3258
10.0	-0.5161	0.0168	-0.3258	-0.2496	-0.3639
15.0	-0.0974	-0.2116	-0.3639	-0.2116	-0.3639
24.0	-0.2877	-0.3258	-0.4780	-0.2116	-0.3639
33.0	-0.2041	-0.2041	-0.3141	-0.3141	-0.3141
54.0	-0.3141	-0.3141	-0.3141	1.6666	-0.3141
65.0	-0.4132	-0.9637	-0.5291	-0.4132	-0.2973
78.5	2.0493	-2.2963	-0.9347	-0.6740	-0.3842
79.5	130.0376	135.0918	4.8677	24.4680	175.6912
80.5	-66.9686	-91.2729	-0.2677	-0.5813	-2.6785
81.3	-8.8723	165.7266	-0.2872	-0.2284	-0.3068
82.0	65.2964	-13.2427	-0.3852	-0.2677	-0.2088
84.0	6.7494	1.6532	-0.5421	-0.3852	-0.1892
87.0	-17.6731	-4.0097	-1.0125	-0.3656	-0.4048
89.0	-23.6839	-2.1300	-0.9037	-0.5500	-0.3614
93.0	8.2696	-1.1336	-0.6443	-0.5029	-0.5736
96.0	0.4640	-1.5168	-0.4321	-0.2906	-0.3614
100.0	-3.7336	-2.5781	-0.2199	-0.1256	-0.5500
96.0	-0.1727	-0.0070	0.1574	0.0867	0.1103
84.0	0.0867	0.0395	-0.4792	0.1339	1.3130
73.5	-0.3141	-0.5342	0.1261	0.0160	-0.0940
54.0	-0.0940	-0.0940	-0.0940	-0.0940	-0.2041
33.0	-0.0940	-0.4242	-0.0940	-0.0940	-0.0940
24.0	-0.2496	-0.2877	-0.3258	-0.1355	-0.2877
10.0	-0.1736	-0.2877	-0.3639	-0.1355	-0.2041
5.0	-0.2877	-0.3639	-0.2496	-0.0974	-0.0940
2.5	-0.5161	-0.6683	-0.3258	-0.1355	-0.2041

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1814
43.4	-0.2104
50.4	-0.1234
56.1	-0.0945
62.5	-0.0366
69.7	-0.0945
76.9	-0.0945
83.4	-0.0945
89.4	-0.1814
95.4	-0.2973
101.1	-0.2683
---BOTTOM---	
38.2	-0.1234
43.4	-0.0656
50.4	-0.1814
56.1	-0.0366
62.5	-0.1524
69.7	-0.1524
76.9	-0.1814
83.4	-0.0945
89.4	-0.0945
95.4	-0.0656
101.1	-0.2394

MISC.*

NO.	CP
1	-0.3141
2	-0.0940
3	-0.3141
4	-0.0940
5	-0.2041
6	-0.0940
7	-0.5029
8	-0.5029
9	-0.4792
10	-0.5029
11	-0.5029
12	-0.5029

TABLE A38. (Continued)

RUN:SEQ 273 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 652

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
4.02	2138.35	2129.31	9.04	87.73	0.01	2.66	0.595	1.028	1.623

CLAERO = 0.4296 CDAERO = 0.3226 DCLC = 0.1441 DWLC = 0.3350 BASEPR = 8.1904

HQT	RN
86.9369	0.548005E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.91	-1.23	-0.25	0.00	-0.01	0.02	0.47	-0.15	0.46	0.07

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.4633	-0.0797
2.5	-0.6771	-1.0300
5.0	-0.5685	-0.7314
10.0	2.7712	-0.6771
15.0	-0.5685	-0.6499
25.0	-0.4870	-0.5142
35.0	-0.3784	-0.4599
50.0	-0.4327	
56.0	-0.4599	-0.4870
65.0	-0.5956	-0.6771
76.0	-0.4599	-1.2744
79.0	-1.8719	-2.0076
80.5	-39.2922	-52.4054
81.0	5.2989	18.1176
82.0	2.9665	-1.2281
84.0	0.3204	
87.0	3.3977	
89.0	0.7167	-2.2384
93.0	1.1802	
96.0	2.5998	0.7746
100.0	0.9485	-3.1075
96.0	-0.2683	-0.1814
84.0	0.9485	-3.1075
73.0	-0.2683	-0.1814
50.0	0.0214	0.0214
35.0	-0.0797	-0.1340
25.0	-0.2155	-0.5685
10.0	-0.1069	-0.1884
5.0	-0.1884	-0.2155
2.5	0.0561	0.2189

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0974	0.2451	0.1690	-0.3258	-0.4019
2.5	-0.4400	0.5877	-0.5923	-0.7825	-1.0109
5.0	1.1967	0.2832	-0.5161	-0.6303	-0.8206
10.0	-0.4780	-0.1736	-0.5541	-0.5161	-0.6683
15.0	-0.1355	-0.3258	-0.5161	-0.4400	-0.5541
24.0	-0.2877	-0.4019	-0.4780	-0.3639	-0.4400
33.0	-0.3141	-0.3141	-0.5342	-0.4242	-0.3141
54.0	-0.3141	-0.4242	-0.5342	1.5566	-0.3141
65.0	-0.4712	-0.9926	-0.5581	-0.4422	-0.4422
78.5	2.1073	-2.3832	-0.8767	-0.5871	-0.5001
79.5	131.9177	136.5406	5.0245	26.1734	175.6027
80.5	-66.2232	-91.7430	-0.3656	-0.6401	-2.7374
81.3	-16.9874	150.0442	-0.4245	-0.4048	-0.4636
82.0	66.9818	-13.8111	-0.4441	-0.3852	-0.4245
84.0	4.9657	0.8104	-0.6989	-0.4832	-0.5029
87.0	-9.9307	-4.3838	-0.8753	-0.4636	-0.5225
89.0	-22.3634	-2.1300	-0.8801	-0.5029	-0.4557
93.0	7.0434	-1.2339	-0.6208	-0.5265	-0.5029
96.0	1.4544	-1.5641	-0.4321	-0.2906	-0.3849
100.0	-3.6626	-2.5309	-0.2199	-0.1727	-0.5029
96.0	-0.1963	-0.0312	0.1103	0.1103	0.0631
84.0	0.0631	0.0159	-0.5029	0.1574	1.2422
73.5	-0.0940	-0.5342	0.1261	-0.0940	0.1261
54.0	-0.0940	-0.0940	-0.0940	-0.0940	-0.0940
33.0	-0.0940	-0.5342	-0.0940	-0.0940	-0.0940
24.0	-0.0974	-0.1355	-0.0974	-0.0593	-0.1736
10.0	-0.0974	-0.0974	-0.0593	0.0548	-0.0940
5.0	-0.1736	-0.1355	0.0168	0.1310	0.1261
2.5	-0.2877	-0.2496	0.0548	0.2832	0.2361

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2973
43.4	-0.2394
50.4	-0.2104
56.1	-0.1524
62.5	-0.1234
69.7	-0.1814
76.9	-0.1814
83.4	-0.1524
89.4	-0.2104
95.4	-0.3263
101.1	-0.3263
---BOTTOM---	
38.2	-0.0656
43.4	-0.0366
50.4	-0.1814
56.1	-0.0076
62.5	-0.0076
69.7	-0.0656
76.9	-0.0656
83.4	-0.0656
89.4	-0.0945
95.4	-0.0656
101.1	-0.2683

MISC.*

NO.	CP
1	-0.4242
2	-0.0940
3	-0.4242
4	-0.0940
5	-0.4242
6	-0.0940
7	-0.5265
8	-0.5265
9	-0.5265
10	-0.5265
11	-0.5029
12	-0.5265

TABLE A38. (Concluded)

RUN:SEQ 273 6 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 654

CLAERO = 0.9809 CDAERO = 0.4274 DCLC = 0.2187 DWLC = 0.4573 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.06	2138.28	2129.24	9.04	87.69	0.01	2.73	0.582	1.006	1.587	89.2749	0.548478E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.66	-1.01	-0.16	-0.00	-0.01	0.02	1.06	-0.06	1.03	0.22

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-5.2392	-2.5508
2.5	-2.4422	-2.1164
5.0	-1.7362	-2.1707
10.0	2.6630	-2.0893
15.0	-1.1931	-1.9262
25.0	-0.8944	-2.0348
35.0	-0.7585	-1.5733
50.0	-0.7042	
56.0	-0.6771	-0.8129
65.0	-0.8129	-0.8129
76.0	-0.5142	-1.1931
79.0	-1.9805	-1.8176
80.5	-39.5321	-52.2542
81.0	6.1032	18.4338
82.0	3.3001	-0.7185
84.0	0.3400	
87.0	3.3785	
89.0	0.6878	-2.0357
93.0	1.0645	
96.0	2.5132	0.8906
100.0	1.1514	-3.0499
96.0	-0.1814	-0.1234
84.0	1.1514	-3.0499
73.0	-0.1814	-0.1234
50.0	0.1663	0.1373
35.0	0.0018	-0.0797
25.0	-0.0525	-0.8129
10.0	0.1104	0.0561
5.0	0.0833	0.1104
2.5	0.5449	0.5177

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4736	0.1310	-2.7239	-4.2467	-1.6962
2.5	-0.5161	-0.5161	-1.6201	-3.7518	-1.5440
5.0	0.0929	-0.6684	-1.2774	-3.6614	-1.5058
10.0	-0.5541	-0.7825	-1.0110	-0.7825	-1.4297
15.0	-0.5161	-0.7825	-0.9348	-0.7825	-1.5820
24.0	-0.5161	-0.7064	-0.7445	-0.7064	-1.3536
33.0	-0.4242	-0.5342	-0.6443	-0.5342	-1.5247
54.0	-0.4242	-0.4242	-0.4242	1.5566	-0.8644
65.0	-0.6451	-1.1956	-0.6739	-0.5871	-0.8768
78.5	2.1075	-2.5284	-0.8183	-0.6160	-0.9927
79.5	133.4255	142.4843	5.8876	27.9991	175.8876
80.5	-65.6432	-85.2450	-0.4441	-0.6401	-2.4240
81.3	-20.6935	125.4397	-0.4637	-0.3657	-0.5814
82.0	67.7735	-10.1084	-0.4637	-0.4048	-0.7185
84.0	4.5154	-0.3852	-0.5225	-0.4637	-0.6009
87.0	-6.2464	-3.2276	-0.6205	-0.4245	-0.4832
89.0	-21.0452	-1.5878	-0.7152	-0.5265	-0.7152
93.0	5.1574	-1.1397	-0.5029	-0.5029	-0.5500
96.0	1.4311	-1.3755	-0.3849	-0.3142	-0.6680
100.0	-3.5453	-2.2245	-0.1963	-0.2199	-0.5029
96.0	-0.1020	-0.0076	0.1339	0.1103	-0.0076
84.0	0.1339	0.0867	-0.5265	0.1575	1.4782
73.5	0.0160	-0.4242	0.2361	0.1261	0.3462
54.0	0.0160	0.0160	0.0160	0.0160	0.0160
33.0	0.1261	-0.4242	0.1261	0.1261	0.0160
24.0	0.0929	0.0929	0.0929	0.2833	0.0929
10.0	0.1310	0.1691	0.3213	0.4355	0.3462
5.0	0.2071	0.2071	0.3594	0.5497	0.4563
2.5	0.1691	0.2833	0.4736	0.5117	0.5663

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4132
43.4	-0.3553
50.4	-0.2683
56.1	-0.2104
62.5	-0.2394
69.7	-0.2394
76.9	-0.2394
83.4	-0.2104
89.4	-0.2973
95.4	-0.3853
101.1	-0.2973
---BOTTOM---	
38.2	0.0214
43.4	0.0504
50.4	-0.1524
56.1	0.0793
62.5	0.0793
69.7	0.0793
76.9	0.0504
83.4	-0.0076
89.4	0.0214
95.4	-0.0076
101.1	-0.2104

MISC.

NO.	CP
1	-0.6443
2	0.1261
3	-0.7543
4	0.1261
5	-0.8644
6	-0.0940
7	-0.5500
8	-0.5265
9	-0.5265
10	-0.5265
11	-0.5500
12	-0.5500

TABLE A39. RUN 274, BC1W6V, DELF=0, CMU=2.0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
274	1	CLAERO =	-1.0788	CDAERO =	-0.3775	DCLC =	0.2193	DWLC =	1.8754	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.01		2138.21	2133.91	4.29	60.38	0.01	2.66	1.262	2.165	3.427	87.0442	0.378549E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		1.02	-2.70	-0.48	0.00	-0.01	0.02	0.29	-0.28	0.29	0.04	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.3153	0.3724
2.5	-0.1992	-0.2564
5.0	-0.3136	-0.4279
10.0	3.5735	-0.4279
15.0	-0.4851	-0.5422
25.0	-0.4279	-0.4851
35.0	-0.4279	-0.5994
50.0	-0.5422	
56.0	-0.5994	-0.6566
65.0	-0.8280	-0.8852
76.0	-0.7709	-1.9141
79.0	-2.7717	-3.1146
80.5	-72.9447	-112.1469
81.0	-0.9383	21.1795
82.0	1.3726	-3.3316
84.0	0.9599	
87.0	7.7273	
89.0	2.0380	-2.9633
93.0	2.7090	
96.0	5.5757	2.2210
100.0	-0.2187	-5.8300
96.0	-0.6456	-0.5236
84.0	-0.2187	-5.8300
73.0	-0.6456	-0.5236
50.0	-0.2187	-0.0967
35.0	-0.3136	-0.3136
25.0	-0.4851	-0.7709
10.0	-0.370[illegible]	-0.4279
5.0	-0.4279	-0.4279
2.5	-0.4279	-0.2564

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.9637	-0.8419	0.2800	0.4403	0.3601
2.5	-0.8419	2.6840	0.4403	-0.2809	-0.5213
5.0	3.1648	1.5621	-0.2008	-0.3611	-0.6015
10.0	-0.8419	-0.0405	-0.2809	-0.2809	-0.5213
15.0	-0.2008	-0.3611	-0.5213	-0.3611	-0.6015
24.0	-0.5213	-0.5213	-0.6015	-0.3611	-0.5213
33.0	-0.3889	-0.3889	-0.3889	-0.3889	-0.3889
54.0	-0.3889	-0.6205	-0.6205	0.5379	-0.3889
65.0	-0.6456	-1.8655	-0.7676	-0.5236	-0.4627
78.5	1.6721	-4.1221	-1.4994	-1.0115	-0.5846
79.5	268.9001	287.5503	11.2761	56.4204	368.5532
80.5	*********	-168.3904	-0.3193	-0.8145	-5.0649
81.3	7.0672	284.6631	-0.5256	-0.3193	-0.3606
82.0	128.3452	-24.0867	-0.5256	-0.3606	-0.3606
84.0	19.9418	-0.4844	-1.0208	-0.6494	-0.4844
87.0	-62.2575	-6.1790	-1.4336	-0.5669	-0.8970
89.0	-39.5609	-2.9209	-1.3813	-0.7861	-0.3889
93.0	7.1077	-1.8287	-0.9350	-0.7861	-0.7364
96.0	-0.9350	-2.3745	-0.6868	-0.3889	-0.2897
100.0	-7.2402	-4.8572	-0.2393	-0.1407	-0.7364
96.0	-0.2400	-0.0414	0.2068	0.0579	0.1075
84.0	0.2068	0.1075	-0.7364	0.1572	0.0062
73.5	-0.6205	-0.6205	0.3062	0.0745	-0.8522
54.0	-0.1572	-0.1572	-0.1572	-0.1572	-0.1572
33.0	-0.1572	-0.6205	-0.3889	-0.1572	-0.1572
24.0	-0.2809	-0.4412	-0.3611	-0.2008	-0.3611
10.0	-0.2809	-0.4412	-0.3611	-0.0405	-0.1572
5.0	-0.4412	-0.7617	-0.3611	-0.0405	-0.1572
2.5	-1.0021	-1.2425	-0.5213	-0.2809	0.0745

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.3407	38.2	-0.2187
43.4	-0.3407	43.4	-0.0967
50.4	-0.2797	50.4	-0.3407
56.1	-0.1577	56.1	-0.0967
62.5	-0.0967	62.5	-0.0967
69.7	-0.1577	69.7	-0.1577
76.9	-0.1577	76.9	-0.1577
83.4	-0.2187	83.4	-0.2187
89.4	-0.1577	89.4	-0.2187
95.4	-0.4627	95.4	-0.2187
101.1	-0.4627	101.1	-0.5236

MISC.*

NO.	CP
1	-0.3889
2	-0.1572
3	-0.3889
4	-0.1572
5	-0.3889
6	-0.1572
7	-0.7861
8	-0.7364
9	-0.7364
10	-0.6868
11	-0.7861
12	-0.7364

TABLE A39. (Continued)

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 656

274 2 CLAERO = -1.1807 CDAERO = -0.3549 DCLC = 0.1738 DWLC = 1.8188 BASEPR = 8.1903

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-1.99	2138.21	2133.91	4.29	60.38	0.01	2.67	1.248	2.145	3.393	87.2558	0.378487E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.81	-2.73	-0.51	0.00	-0.01	0.03	0.18	-0.31	0.18	0.03

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.1421	0.0866
2.5	-0.0849	-0.1421
5.0	-0.1992	-0.1992
10.0	3.0019	-0.3136
15.0	-0.3707	-0.4851
25.0	-0.3707	-0.4851
35.0	-0.3707	-0.4851
50.0	-0.4851	
56.0	-0.5994	-0.5994
65.0	-0.8260	-0.8852
76.0	-0.8280	-1.9141
79.0	-2.7717	-3.1717
80.5	-72.6140	-112.2703
81.0	-1.0208	19.7767
82.0	1.6201	-3.0015
84.0	1.0837	
87.0	7.7273	
89.0	2.0380	-2.9023
93.0	2.8309	
96.0	5.5757	2.2820
100.0	-0.4017	-5.7060
96.0	-0.5846	-0.5236
84.0	-0.4017	-5.7060
73.0	-0.5846	-0.5236
50.0	-0.1577	-0.0357
35.0	-0.3707	-0.4851
25.0	-0.4851	-0.8280
10.0	-0.4851	-0.4851
5.0	-0.4851	-0.5994
2.5	-0.8852	-0.8280

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.9637	-0.8419	0.3601	0.2800	0.2800
2.5	-0.7617	2.9244	0.9210	-0.0405	-0.0405
5.0	3.2448	1.6422	0.4403	-0.1207	-0.3611
10.0	-0.8419	0.0396	-0.2809	-0.2008	-0.3611
15.0	-0.2008	-0.3611	-0.3611	-0.2008	-0.4412
24.0	-0.4412	-0.5213	-0.5213	-0.2809	-0.4412
33.0	-0.3889	-0.1572	-0.3889	-0.1572	-0.1572
54.0	-0.1572	-0.3889	-0.6205	0.3062	-0.3889
65.0	-0.5846	-1.8045	-0.7066	-0.5236	-0.3407
78.5	1.0621	-4.0611	-1.4384	-0.9506	-0.4627
79.5	268.5276	285.3149	12.5965	58.1107	368.7163
80.5	*********	-170.1250	-0.2368	-0.6907	-4.9410
81.3	9.2542	275.5361	-0.2780	-0.1542	-0.3193
82.0	126.5291	-24.5011	-0.4844	-0.2368	-0.1955
84.0	20.3131	0.2584	-0.8970	-0.6062	-0.2780
87.0	-62.1341	-6.1375	-1.2684	-0.6494	-0.6907
89.0	-39.2117	-2.7719	-1.3819	-0.8854	-0.3889
93.0	5.8665	-1.8287	-0.9350	-0.7364	-0.6868
96.0	-1.0840	-2.4741	-0.6868	-0.4882	-0.2897
100.0	-7.3892	-4.9067	-0.3393	-0.1904	-0.7851
96.0	-0.2400	-0.0911	0.2068	0.1572	0.0579
84.0	0.1572	0.0579	-0.7364	0.1572	-0.4386
73.5	-0.6205	-0.8522	0.3062	0.0745	-0.3889
54.0	-0.1572	-0.1572	-0.1572	-0.1572	-0.1572
33.0	-0.1572	-0.6205	-0.3889	-0.1572	-0.1572
24.0	-0.2809	-0.4412	-0.3611	-0.2809	-0.4412
10.0	-0.3611	-0.6015	-0.4412	-0.2008	-0.3889
5.0	-0.4412	-0.7617	-0.5213	-0.4412	-0.3889
2.5	-1.1624	-1.5630	-1.2425	-0.6015	-0.6205

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2187
43.4	-0.2797
50.4	-0.1577
56.1	-0.0967
62.5	-0.0357
69.7	-0.0967
76.9	-0.0967
83.4	-0.0967
89.4	-0.2797
95.4	-0.4627
101.1	-0.4017
---BOTTOM---	
38.2	-0.1577
43.4	-0.0357
50.4	-0.2797
56.1	-0.0967
62.5	-0.0967
69.7	-0.0967
76.9	-0.1577
83.4	-0.1577
89.4	-0.2187
95.4	-0.1577
101.1	-0.4017

MISC.*

NO.	CP
1	-0.3889
2	-0.1572
3	-0.1572
4	-0.1572
5	-0.3889
6	-0.1572
7	-0.7364
8	-0.6868
9	-0.7364
10	-0.7364
11	-0.7364
12	-0.7364

TABLE A39. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 660

RUN:SEQ												
274	6	CLAERO =	-0.0779	CDAERO =	-0.4764	DCLC =	0.4580	DWLC =	1.9931	BASEPR =	8.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.09		2138.00	2133.59	4.41	61.15	0.01	2.74	1.218	2.095	3.312	89.5629	0.384019E+06
BETA		CL	CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01		2.37	-2.25	-0.33	-0.01		-0.01	0.03	1.14	-0.13	1.12	0.21

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-5.7396	-2.7328
2.5	-2.7328	-2.2317
5.0	-2.0646	-2.3431
10.0	1.4990	-2.2873
15.0	-1.3409	-2.2873
25.0	-1.0067	-2.0646
35.0	-0.8953	-1.9533
50.0	-0.7839	
56.0	-0.8953	-1.2294
65.0	-1.0067	-1.1180
76.0	-0.6726	-1.6750
79.0	-2.9555	-2.6214
80.5	-71.3939	-108.8543
81.0	-1.6951	18.5228
82.0	1.0380	-2.5794
84.0	0.7989	
87.0	7.3888	
89.0	1.7702	-2.8044
93.0	2.6020	
96.0	5.3349	2.1287
100.0	-0.9033	-5.5968
96.0	-0.4280	-0.3685
84.0	-0.9033	-5.5968
73.0	-0.4280	-0.3685
50.0	0.1067	0.0473
35.0	-0.0601	-0.1714
25.0	-0.0601	-1.0624
10.0	0.0513	-0.0044
5.0	0.0513	0.0513
2.5	0.4967	0.4967

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0949	0.3734	-2.2023	-4.1536	-1.8122
2.5	-0.6412	0.2954	-1.3438	-3.7635	-1.6559
5.0	1.6223	-0.0949	-1.2657	-3.6853	-1.5779
10.0	-0.7974	-0.7974	-1.1878	-0.7974	-1.5779
15.0	-0.2510	-0.7193	-1.1096	-0.8755	-1.4999
24.0	-0.6412	-0.8755	-0.9535	-0.6412	-1.8122
33.0	-0.8074	-0.8074	-1.0331	-0.8074	-1.4844
54.0	-0.8074	-0.8074	-0.8074	-1.0331	-1.2588
65.0	-0.7845	-1.9133	-0.7845	-0.6657	-0.9033
78.5	0.0473	-4.1114	-1.2597	-0.7845	-0.9627
79.5	261.6572	274.0659	12.2123	57.0693	359.1807
80.5	*********	-170.4345	-0.4894	-1.0923	-4.6896
81.3	9.8408	235.5303	-0.4894	-0.5698	-0.7707
82.0	122.6277	-26.2951	-0.6903	-0.5295	-0.9315
84.0	19.6083	1.9224	-0.9315	-0.6501	-0.9315
87.0	-59.8567	-7.2823	-1.1726	-0.6501	-0.6903
89.0	-36.8185	-3.0643	-1.1783	-0.6947	-0.9365
93.0	4.7217	-1.9521	-0.8881	-0.7431	-0.7914
96.0	-0.9849	-2.7742	-0.6463	-0.5013	-0.8881
100.0	-5.5461	-4.5152	-0.4045	-0.3562	-0.7914
96.0	-0.3562	-0.1628	0.0307	-0.0177	-0.1628
84.0	0.0307	-0.0177	-0.8398	0.0307	-0.8881
73.5	-0.3561	-0.8074	0.0952	-0.1305	-0.5818
54.0	-0.1305	-0.1305	-0.1305	-0.1305	-0.3561
33.0	-0.1305	-1.0331	-0.1305	-0.1305	-0.1305
24.0	0.2173	0.2173	0.0612	0.3734	0.0612
10.0	0.2954	0.1393	0.2173	0.4515	0.0952
5.0	0.2173	0.1393	0.3734	0.6076	0.3209
2.5	0.0612	0.0612	0.3734	0.4515	0.3209

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.4874	38.2	-0.0121
43.4	-0.4280	43.4	0.1067
50.4	-0.3092	50.4	-0.1904
56.1	-0.2498	56.1	0.1067
62.5	-0.2498	62.5	0.1067
69.7	-0.2498	69.7	0.1067
76.9	-0.2498	76.9	0.0473
83.4	-0.1904	83.4	-0.0121
89.4	-0.2498	89.4	-0.0715
95.4	-0.3685	95.4	-0.0715
101.1	-0.3685	101.1	-0.3092

MISC.*

NO.	CP
1	-1.0331
2	-0.1305
3	-1.0331
4	-0.1305
5	-1.2588
6	-0.3561
7	-0.7431
8	-0.7914
9	-0.7914
10	-0.7914
11	-0.7914
12	-0.7914

TABLE A40. RUN 277, BW6V, DELF=45, CMU=0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
277	1	CLAERO =	0.5209	CDAERO =	0.2419	DCLC =	0.0	DWLC =	0.0	BASEPR =	5.1904	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.10		2133.40	2103.34	30.06	160.15	0.01	2.67	0.0	0.0	0.0	87.4810	0.100348E+07
BETA		CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01		0.52	0.24	-0.32	0.00	-0.00	0.00	0.52	-0.33	0.52	0.30	

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.2646	-0.2646
2.5	-0.2565	-0.2646
5.0	-0.2565	-0.2728
10.0	-0.2728	-0.2728
15.0	-0.2565	-0.2728
25.0	-0.2646	-0.2728
35.0	-0.2728	-0.2728
50.0	-0.2646	
56.0	-0.2646	-0.2810
65.0	-0.2646	-0.2810
76.0	-0.2891	-0.2728
79.0	-0.2810	-0.2646
80.5	-0.2191	-0.2486
81.0	-0.2309	-0.2427
82.0	-0.2309	-0.2545
84.0	-0.2368	
87.0	-0.2309	
89.0	-0.2623	-0.2710
93.0	-0.2623	
96.0	-0.2623	-0.2710
100.0	-0.2710	-0.2623
96.0	-0.2710	-0.2623
84.0	-0.2710	-0.2623
73.0	-0.2710	-0.2623
50.0	-0.2797	-0.2710
35.0	-0.2728	-0.2810
25.0	-0.2728	-0.2891
10.0	-0.2810	-0.2891
5.0	-0.2646	-0.2810
2.5	-0.2810	-0.2646

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	0.5205	0.4976	0.1198	0.2571	-0.0291
2.5	-0.2581	-0.3611	-0.5443	-0.5443	-0.7618
5.0	-0.2237	-0.2924	-0.4413	-0.4413	-0.6244
10.0	-0.2466	-0.3382	-0.4069	-0.4641	-0.5214
15.0	-0.2581	-0.3268	-0.3840	-0.4184	-0.4069
24.0	-0.2466	-0.3039	-0.3497	-0.3497	-0.3840
33.0	-0.2565	-0.3227	-0.3558	-0.3558	-0.3558
54.0	-0.2896	-0.3227	-0.4220	0.6703	-0.3558
65.0	-0.3320	-0.4191	-0.5237	-0.5150	-0.4888
78.5	0.4523	-0.6544	-0.8026	-0.8287	-0.8461
79.5	-0.4962	-0.5139	-0.7439	-0.7085	-0.8146
80.5	-0.4844	-0.5198	-0.7262	-0.6908	-0.2663
81.3	-0.5080	53.5163	-0.7085	-0.7085	-0.8205
82.0	-0.4962	-1.1212	-0.7320	-0.6908	-0.7969
84.0	-0.5021	-0.5552	-0.6554	-0.7085	-0.8146
87.0	-0.5316	-0.5846	-0.7262	-0.6908	-0.2663
89.0	-0.5805	-0.6373	-0.7791	-0.7578	-0.8430
93.0	-0.5663	-0.6443	-0.7437	-0.3252	-0.2968
96.0	-0.5592	-0.6656	-0.7437	-0.7437	-0.8713
100.0	-0.5167	-0.7011	-0.6514	-0.7295	-0.2897
96.0	0.3346	0.3558	0.3346	0.3558	0.2565
84.0	0.5332	0.6538	-0.3039	0.6183	0.4977
73.5	0.4717	-0.2896	0.6041	0.5710	0.3724
54.0	0.1738	0.2069	0.2400	0.2731	0.1738
33.0	-0.0248	-0.2896	0.1076	0.1076	0.0414
24.0	0.0511	0.0167	0.0396	0.0739	0.0282
10.0	0.0511	0.0396	0.0511	0.0969	0.0414
5.0	0.0739	0.0511	0.1083	0.1541	0.1407
2.5	0.1083	0.0969	0.1884	0.2228	0.2400

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.0880
43.4	-0.0706
50.4	-0.0706
56.1	-0.0793
62.5	-0.0367
69.7	-0.1403
76.9	-0.1490
83.4	-0.1403
89.4	-0.2187
95.4	-0.2797
101.1	-0.2361
---BOTTOM---	
38.2	-0.0619
43.4	-0.0270
50.4	-0.0120
56.1	-0.0008
62.5	0.0253
69.7	0.0802
76.9	0.0889
83.4	0.1473
89.4	0.1996
95.4	0.1212
101.1	-0.3145

MISC.*

NO.	CP
1	-0.3227
2	0.0745
3	-0.3889
4	0.1076
5	-0.3227
6	0.0083
7	-0.3252
8	-0.3110
9	-0.2968
10	-0.3110
11	-0.3039
12	-0.3180

TABLE A40. (Continued)

RUN:SEQ 277 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 663

CLAERO = 0.6921 CDAERO = 0.2946 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.05	2133.26	2102.86	30.40	161.26	0.01	2.66	0.0	0.0	0.0	87.0177	0.100600E+07

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	0.69	0.29	-0.35	0.00	-0.00	0.00	0.71	-0.36	0.69	0.35

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.2760	-0.2599
2.5	-0.2599	-0.2518
5.0	-0.2680	-0.2680
10.0	-0.2599	-0.2680
15.0	-0.2599	-0.2599
25.0	-0.2680	-0.2599
35.0	-0.2760	-0.2599
50.0	-0.2599	
56.0	-0.2599	-0.2599
65.0	-0.2680	-0.2518
76.0	-0.2760	-0.2680
79.0	-0.2599	-0.2680
80.5	-0.2651	-0.2592
81.0	-0.2476	-0.2476
82.0	-0.2476	-0.2534
84.0	-0.2359	
87.0	-0.2534	
89.0	-0.2322	-0.2581
93.0	-0.2581	
96.0	-0.2495	-0.2409
100.0	-0.2495	-0.2581
96.0	-0.2495	-0.2495
84.0	-0.2495	-0.2581
73.0	-0.2495	-0.2495
50.0	-0.2581	-0.2495
35.0	-0.2680	-0.2680
25.0	-0.2599	-0.2599
10.0	-0.2680	-0.2599
5.0	-0.2680	-0.2599
2.5	-0.2680	-0.2599

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0868	-0.4151	-1.3435	-1.6152	-2.8833
2.5	-0.2453	-0.8793	-1.0605	-1.2869	-1.6152
5.0	-0.5057	-0.5963	-0.9019	-0.9925	-1.1397
10.0	-0.2566	-0.5623	-0.7661	-0.7661	-0.8680
15.0	-0.4265	-0.5283	-0.6642	-0.6869	-0.6869
24.0	-0.3812	-0.4378	-0.5170	-0.5510	-0.5283
33.0	-0.3747	-0.4074	-0.4729	-0.5383	-0.5056
54.0	-0.3420	-0.4074	-0.4729	0.4763	-0.4729
65.0	-0.3701	-0.4563	-0.5855	-0.5855	-0.5942
78.5	0.4658	-0.6631	-0.9044	-0.8871	-0.9388
79.5	-0.5274	-0.5274	-0.9239	-0.7607	-0.9239
80.5	-0.5099	-0.5158	-0.8772	-0.7198	-0.2359
81.3	-0.5099	52.9623	-0.8772	-0.7315	-0.8831
82.0	-0.5216	-1.0521	-0.8656	-0.7432	-0.8772
84.0	-0.5216	-0.5508	-0.7607	-0.7315	-0.8831
87.0	-0.5333	-0.5624	-0.8714	-0.7607	-0.2534
89.0	-0.6132	-0.6624	-0.9850	-0.8167	-0.9289
93.0	-0.6132	-0.6694	-0.8586	-0.3257	-0.2836
96.0	-0.5852	-0.6834	-0.8447	-0.8097	-0.9289
100.0	-0.5361	-0.7395	-0.7395	-0.8027	-0.2906
96.0	0.3687	0.3617	0.3407	0.3477	0.2144
84.0	0.5722	0.6563	-0.3046	0.5862	0.4950
73.5	0.5091	-0.2765	0.5745	0.5745	0.4109
54.0	0.2145	0.2472	0.3127	0.3127	0.1818
33.0	0.0508	-0.3092	0.1818	0.1818	0.0836
24.0	0.1622	0.1170	0.1509	0.1962	0.1057
10.0	0.1962	0.2075	0.2189	0.3094	0.2145
5.0	0.2642	0.2868	0.3547	0.3773	0.3781
2.5	0.3773	0.3773	0.4453	0.5019	0.4436

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.0857
43.4	-0.0599
50.4	-0.0771
56.1	-0.0857
62.5	-0.1202
69.7	-0.1719
76.9	-0.1719
83.4	-0.1633
89.4	-0.2150
95.4	-0.2667
101.1	-0.2238
---BOTTOM---	
38.2	-0.0254
43.4	0.0004
50.4	-0.0599
56.1	0.0521
62.5	0.0952
69.7	0.1211
76.9	0.1469
83.4	0.1986
89.4	0.2417
95.4	0.1728
101.1	-0.3012

MISC.*

NO.	CP
1	-0.4401
2	0.1818
3	-0.5383
4	0.1818
5	-0.4401
6	0.0181
7	-0.2976
8	-0.2976
9	-0.3046
10	-0.3046
11	-0.2976
12	-0.3046

TABLE A40. (Concluded)

RUN:SEQ 277 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 665

CLAERO = 1.0858 CDAERO = 0.4915 DCLC = 0.0 DWLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.03	2133.12	2103.06	30.06	160.49	0.01	2.75	0.0	0.0	0.0	89.9482	0.998211E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.09	0.49	-0.44	0.01	-0.00	0.00	1.17	-0.45	1.07	0.55

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3088	-0.3006
2.5	-0.3006	-0.3006
5.0	-0.3006	-0.3088
10.0	-0.2925	-0.3088
15.0	-0.2925	-0.3006
25.0	-0.3006	-0.2925
35.0	-0.3088	-0.3006
50.0	-0.2925	
56.0	-0.3006	-0.3006
65.0	-0.3088	-0.3170
76.0	-0.3251	-0.3006
79.0	-0.3170	-0.3088
80.5	-0.3050	-0.2932
81.0	-0.2814	-0.2873
82.0	-0.2755	-0.2873
84.0	-0.2814	
87.0	-0.2814	
89.0	-0.2917	-0.2917
93.0	-0.3004	
96.0	-0.3004	-0.3004
100.0	-0.3004	-0.3004
96.0	-0.2917	-0.3004
84.0	-0.3004	-0.3004
73.0	-0.2917	-0.3004
50.0	-0.3004	-0.3004
35.0	-0.3006	-0.3088
25.0	-0.3088	-0.3006
10.0	-0.3088	-0.3088
5.0	-0.2925	-0.3088
2.5	-0.3006	-0.3088

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-2.8145	-5.5511	-7.0052	-2.9063	-1.5895
2.5	-0.2728	-2.2765	-2.6887	-2.7689	-1.5437
5.0	-1.2461	-1.6239	-2.0933	-2.7346	-1.5323
10.0	-0.2843	-1.2575	-1.6124	-2.7117	-1.5208
15.0	-0.8453	-1.0056	-1.3605	-2.5171	-1.4521
24.0	-0.6736	-0.7652	-1.0400	-1.3720	-1.4407
33.0	-0.6239	-0.6901	-0.8557	-0.9219	-1.4184
54.0	-0.4915	-0.5577	-0.6901	0.3029	-1.2860
65.0	-0.4835	-0.5619	-0.7536	-0.6752	-1.1545
78.5	0.4142	-0.6403	-1.2852	-1.1981	-1.2068
79.5	-0.5113	-0.5054	-1.4960	-1.0950	-1.3014
80.5	-0.5173	-0.5113	-1.3781	-0.9889	-0.2873
81.3	-0.4877	51.7582	-1.4429	-1.0184	-1.2012
82.0	-0.4937	-0.9948	-1.3427	-1.0381	-1.1540
84.0	-0.4996	-0.5467	-1.1776	-1.0184	-1.0892
87.0	-0.5173	-0.5821	-1.1835	-1.0007	-0.2873
89.0	-0.6051	-0.6406	-1.1442	-1.0591	-1.1159
93.0	-0.5696	-0.6546	-0.8889	-0.3781	-0.3498
96.0	-0.5596	-0.6619	-0.8038	-0.9740	-1.0875
100.0	-0.5484	-0.6974	-0.7328	-0.9101	-0.3356
96.0	0.3241	0.4235	0.3809	0.3454	0.1823
84.0	0.6646	0.7427	-0.3498	0.6150	0.4093
73.5	0.6008	-0.3260	0.6339	0.5677	0.3360
54.0	0.3360	0.3360	0.3691	0.3691	0.2036
33.0	0.2367	-0.3260	0.3360	0.3360	0.1705
24.0	0.3340	0.3340	0.3569	0.4027	0.2080
10.0	0.4828	0.4828	0.4943	0.5286	0.4353
5.0	0.5973	0.5630	0.5401	0.5744	0.5015
2.5	0.6775	0.5744	0.5057	0.5171	0.4684

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1087
43.4	-0.1174
50.4	-0.1436
56.1	-0.1959
62.5	-0.2482
69.7	-0.3266
76.9	-0.3092
83.4	-0.2569
89.4	-0.3353
95.4	-0.2743
101.1	-0.2133
---BOTTOM---	
38.2	0.0046
43.4	0.0307
50.4	-0.0390
56.1	0.1092
62.5	0.1876
69.7	0.2486
76.9	0.2486
83.4	0.2747
89.4	0.2834
95.4	0.1701
101.1	-0.3266

MISC.*

NO.	CP
1	-0.6901
2	0.3360
3	-1.6832
4	0.2698
5	-1.2860
6	0.0381
7	-0.3569
8	-0.3498
9	-0.3569
10	-0.3498
11	-0.3498
12	-0.3427

TABLE A41. RUN 280, BW6V, DELF=45, CMU=0.5, (BN/B)W=0.25

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 666

280 2	CLAERO =	0.7105	CDAERO =	0.4412	DCLC =	0.0	DWLC =	0.4631	BASCPR =	8.1904			
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN		
-0.01	2133.12	2122.94	10.17	93.10	0.01	2.69	0.0	0.535	0.535	87.9400	0.581730E+06		
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR			
-0.01	1.17	0.17	-0.78	0.01	-0.01	0.01	0.81	-0.51	0.81	0.40			

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3489	-0.3489
2.5	-0.3489	-0.3247
5.0	-0.3730	-0.3247
10.0	-0.3489	-0.3489
15.0	-0.3489	-0.3489
25.0	-0.3247	-0.3489
35.0	-0.3489	-0.3489
50.0	-0.3489	
56.0	-0.3247	-0.3489
65.0	-0.3247	-0.3489
76.0	-0.3247	-0.3247
79.0	-0.3247	-0.3489
80.5	-0.3389	-0.3389
81.0	-0.3389	-0.3389
82.0	-0.2866	-0.3737
84.0	-0.3389	
87.0	-0.3563	
89.0	-0.3106	-0.3364
93.0	-0.3364	
96.0	-0.3364	-0.3364
100.0	-0.3364	-0.3364
96.0	-0.3364	-0.3364
84.0	-0.3364	-0.3364
73.0	-0.3364	-0.3364
50.0	-0.3364	-0.3106
35.0	-0.3247	-0.3489
25.0	-0.3247	-0.3247
10.0	-0.3489	-0.3489
5.0	-0.3489	-0.3730
2.5	-0.3247	-0.3247

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.5291	0.3938	0.0892	-0.2829	-0.6889
2.5	-0.3844	-0.3844	-0.7227	-0.8243	-1.0611
5.0	-0.3187	-0.3844	-0.6213	-0.7227	-0.8581
10.0	-0.4183	-0.3844	-0.6213	-0.5874	-0.7227
15.0	-0.3506	-0.4521	-0.6213	-0.4859	-0.6889
24.0	-0.3844	-0.4183	-0.5536	-0.4521	-0.5874
33.0	-0.3667	-0.3667	-0.3667	-0.4645	-0.4645
54.0	-0.5623	-0.5623	-0.5623	-0.4645	-0.5623
65.0	-0.7999	-1.2119	-0.7999	-0.7742	-0.7999
78.5	-0.0531	-2.7828	-1.2634	-1.3922	-1.4953
79.5	-55.6400	-20.7235	-1.2100	-1.3320	-1.5933
80.5	-48.7406	-1.9940	-1.2823	-1.3320	-0.3214
81.3	-33.0072	156.2230	-1.2448	-1.3145	-1.5585
82.0	-22.2572	-3.6492	-1.2971	-1.3320	-1.5759
84.0	1.1421	-1.8546	-1.1229	-1.3668	-1.5759
87.0	-0.1985	-1.6630	-1.3494	-1.3842	-0.3911
89.0	-1.4498	-1.6594	-1.3869	-1.5127	-1.6804
93.0	-0.6533	-1.4708	-1.4079	-0.4227	-0.3807
96.0	-0.2340	-1.2612	-1.3451	-1.4708	-1.7433
100.0	-0.8419	-1.3451	-1.1773	-1.3451	-0.3807
96.0	0.5206	0.3739	0.1433	0.2272	0.1014
84.0	0.5625	0.6674	-0.3598	0.6464	0.4368
73.5	0.5137	-0.3667	0.6115	0.6115	0.4158
54.0	0.2202	0.2202	0.3180	0.2202	0.1224
33.0	0.0246	-0.2689	0.1224	0.1224	0.0246
24.0	0.0892	0.0554	-0.0123	0.1569	-0.0123
10.0	0.0892	0.0554	0.0554	0.1569	0.2202
5.0	0.0554	0.0554	0.1569	0.2246	0.2202
2.5	0.1569	0.0892	0.2923	0.3261	0.3180

FUSELAGE

-----TOP----		
XLOC	CP	
38.2	-0.0788	
43.4	-0.0788	
50.4	-0.0788	
56.1	-0.1046	
62.5	-0.0788	
69.7	-0.1819	
76.9	-0.2076	
83.4	-0.2076	
89.4	-0.3106	
95.4	-0.3106	
101.1	-0.2849	
---BOTTOM---		
38.2	-0.0531	
43.4	-0.0273	
50.4	-0.1561	
56.1	-0.0016	
62.5	0.0499	
69.7	0.0757	
76.9	0.1014	
83.4	0.1787	
89.4	0.2302	
95.4	0.0499	
101.1	-0.4908	

MISC.*

NO.	CP
1	-0.3667
2	0.1224
3	-0.5623
4	0.1224
5	-0.3667
6	0.0246
7	-0.4017
8	-0.4017
9	-0.3807
10	-0.3807
11	-0.3807
12	-0.3807

TABLE A41. (Continued)

RUN:SEQ 280 4 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 668

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT
4.08	2133.19	2123.13	10.06	92.61	0.01	2.68	0.0	0.550	0.550

CLAERO = 0.8726 CDAERO = 0.5121 DCLC = 0.0 DWLC = 0.4945 BASEPR = 8.1904

BETA	CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	HGT	RN	CLTR	CDTR
-0.01	1.37	0.27	-0.83	0.01	-0.01	0.01	1.01	-0.55	87.6534	0.578059E+06	0.98	0.47

********** CANARD ******

%X/C	BP=3.875 CP	BP=10.125 CP
0.0	-0.3628	-0.3628
2.5	-0.3383	-0.3383
5.0	-0.3383	-0.3383
10.0	-0.3383	-0.3628
15.0	-0.3383	-0.3628
25.0	-0.3383	-0.3383
35.0	-0.3628	-0.3628
50.0	-0.3628	
56.0	-0.3628	-0.3871
65.0	-0.3628	-0.3628
76.0	-0.3628	-0.3383
79.0	-0.3628	-0.3628
80.5	-0.2821	-0.3173
81.0	-0.3173	-0.3173
82.0	-0.3173	-0.3349
84.0	-0.3349	
87.0	-0.2997	
89.0	-0.3502	-0.3502
93.0	-0.3502	
96.0	-0.3502	-0.3762
100.0	-0.3502	-0.3502
96.0	-0.3502	-0.3502
84.0	-0.3502	-0.3502
73.0	-0.3502	-0.3502
60.0	-0.3241	-0.3502
35.0	-0.3628	-0.3383
25.0	-0.3628	-0.3628
10.0	-0.3628	-0.3383
5.0	-0.3628	-0.3871
2.5	-0.3628	-0.3383

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.1250	-0.6040	-1.8015	-2.3489	-2.9647
2.5	-0.3303	-0.8778	-1.2540	-1.5620	-2.4857
5.0	-0.5698	-0.7066	-1.0145	-1.1172	-1.5278
10.0	-0.3303	-0.6040	-0.8778	-0.8778	-1.0830
15.0	-0.5356	-0.5698	-0.8093	-0.7408	-0.8436
24.0	-0.4672	-0.5356	-0.7066	-0.6383	-0.7408
33.0	-0.4797	-0.4797	-0.5786	-0.6775	-0.6775
54.0	-0.5786	-0.6775	-0.6775	-0.3808	-0.6775
65.0	-0.8710	-1.3137	-0.8970	-0.9231	-0.9231
78.5	-0.0897	-2.8501	-1.3137	-1.4439	-1.6002
79.5	-55.3584	-20.7556	-1.3040	-1.3392	-1.6388
80.5	-49.1918	-2.0440	-1.2864	-1.3392	-0.3527
81.3	-34.0402	158.0376	-1.2511	-1.3216	-1.5507
82.0	-24.0501	-3.4359	-1.2159	-1.3392	-1.6036
84.0	1.0217	-1.8326	-1.1102	-1.3216	-1.6564
87.0	-0.0178	-1.6388	-1.3392	-1.4097	-0.3349
89.0	-1.4337	-1.7729	-1.4548	-1.5185	-1.7517
93.0	-0.6494	-1.5396	-1.4548	-0.4374	-0.3527
96.0	-0.2890	-1.3913	-1.4125	-1.5609	-1.8788
100.0	-0.8825	-1.4337	-1.2853	-1.3913	-0.3738
96.0	0.5377	0.3469	0.1349	0.1985	0.0077
84.0	0.6013	0.7709	-0.3950	0.6649	0.4741
73.5	0.6084	-0.2819	0.7073	0.7073	0.4105
54.0	0.3116	0.3116	0.3116	0.3116	0.1138
33.0	0.1138	-0.2819	0.2127	0.2127	0.0149
24.0	0.2171	0.1829	0.1829	0.2858	0.0803
10.0	0.2858	0.2513	0.2513	0.4224	0.3116
5.0	0.3882	0.3540	0.4224	0.4908	0.3116
2.5	0.4224	0.4566	0.5251	0.5251	0.5095

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1157
43.4	-0.1157
50.4	-0.1157
56.1	-0.1157
62.5	-0.1678
69.7	-0.2460
76.9	-0.2720
83.4	-0.2981
89.4	-0.3241
95.4	-0.3762
101.1	-0.3241
---BOTTOM---	
38.2	-0.0636
43.4	-0.0376
50.4	-0.1418
56.1	0.0145
62.5	0.0666
69.7	0.1186
76.9	0.1447
83.4	0.1967
89.4	0.2488
95.4	0.0666
101.1	-0.4803

MISC.*

NO.	CP
1	-0.4797
2	0.2127
3	-0.6775
4	0.2127
5	-0.6775
6	0.0149
7	-0.4162
8	-0.3950
9	-0.3950
10	-0.4162
11	-0.3950
12	-0.3738

TABLE A41. (Concluded)

RUN:SEQ 280 7 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 671

CLAERO =	1.3745	CDAERO =	0.7977	DCLC =	0.0	DWLC =	0.5143	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.07	2133.19	2123.24	9.95	92.09	0.01	2.75	0.0	0.541	0.541	90.0125	0.574659E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
-0.01	1.89	0.63	-0.99	0.01	-0.01	0.00	1.60	-0.71	1.47	0.78	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.4263	-0.4016
2.5	-0.4016	-0.4263
5.0	-0.4263	-0.4263
10.0	-0.4263	-0.4016
15.0	-0.4016	-0.4016
25.0	-0.4016	-0.4263
35.0	-0.4016	-0.4263
50.0	-0.3770	
56.0	-0.3770	-0.4016
65.0	-0.3770	-0.4016
76.0	-0.4016	-0.4263
79.0	-0.4016	-0.3770
80.5	-0.4914	-0.3845
81.0	-0.4736	-0.4380
82.0	-0.4558	-0.3845
84.0	-0.4914	
87.0	-0.4023	
89.0	-0.4432	-0.4169
93.0	-0.4169	
96.0	-0.4169	-0.4432
100.0	-0.4432	-0.4169
96.0	-0.4169	-0.4169
84.0	-0.4432	-0.4169
73.0	-0.4169	-0.4169
50.0	-0.4169	-0.4169
35.0	-0.4016	-0.3770
25.0	-0.4016	-0.4263
10.0	-0.3522	-0.4016
5.0	-0.4016	-0.4016
2.5	-0.4016	-0.4016

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-3.6316	-5.5003	-2.8703	-2.2128	-1.7283
2.5	-0.4133	-4.9812	-2.9741	-2.0744	-1.8667
5.0	-1.5553	-1.6591	-3.4585	-2.5588	-1.7976
10.0	-0.4480	-1.2784	-2.9395	-2.6973	-1.5899
15.0	-1.0706	-1.1055	-3.1125	-2.5934	-1.4860
24.0	-0.8632	-0.9670	-1.2438	-2.4896	-1.5553
33.0	-0.7954	-0.7954	-0.7954	-2.1960	-1.5957
54.0	-0.7954	-0.8954	-0.7954	-0.3952	-1.7958
65.0	-1.2070	-1.7075	-1.0753	-1.2334	-2.0236
78.5	-0.1008	-3.9464	-2.6031	-1.7338	-1.7075
79.5	-55.4493	-27.7739	-4.4119	-3.5565	-1.7923
80.5	-49.6576	-3.8534	-3.6634	-2.6120	-0.4023
81.3	-38.7561	153.9220	-3.6456	-2.0062	-1.7566
82.0	-27.5246	-4.7326	-2.3625	-2.3090	-1.8101
84.0	-0.3310	-3.5387	-2.2200	-1.5428	-1.4715
87.0	0.3996	-2.1666	-2.2556	-1.2577	-0.4201
89.0	-1.1814	-2.1461	-2.3391	-1.4173	-1.4815
93.0	-0.6025	-1.0528	-1.5245	-0.4524	-0.4310
96.0	-0.3881	-1.4387	-0.9670	-1.1814	-1.5459
100.0	-0.5811	-1.2243	-0.7740	-1.1385	-0.4310
96.0	0.5338	0.4695	0.3403	0.2980	0.0192
84.0	0.7268	0.8340	-0.4095	0.7268	0.4266
73.5	0.8054	-0.3952	0.7053	0.7053	0.5053
54.0	0.4052	0.3052	0.4052	0.4052	0.2051
33.0	0.3052	-0.3952	0.4052	0.3052	0.2051
24.0	0.4518	0.4518	0.4518	0.5210	0.2788
10.0	0.5902	0.5902	0.5902	0.6248	0.4052
5.0	0.7286	0.6594	0.6248	0.6594	0.5053
2.5	0.7633	0.5556	0.5556	0.5210	0.4052

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1535
43.4	-0.1271
50.4	-0.2061
56.1	-0.2325
62.5	-0.3379
69.7	-0.4169
76.9	-0.3905
83.4	-0.3642
89.4	-0.4169
95.4	-0.2642
101.1	-0.3115
---BOTTOM---	
38.2	0.0046
43.4	0.0309
50.4	-0.1006
56.1	0.1363
62.5	0.1889
69.7	0.2943
76.9	0.2943
83.4	0.3207
89.4	0.3207
95.4	0.1363
101.1	-0.5222

MISC.*

NO.	CP
1	-0.7954
2	0.4052
3	-1.9959
4	0.3052
5	-1.5957
6	0.0050
7	-0.4310
8	-0.4524
9	-0.4524
10	-0.4310
11	-0.4310
12	-0.4310

TABLE A42. RUN 281, BWGV, DELF=45, CMU=1.0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
281	1	CLAERO =	0.6447	CDAERO =	0.5116	DCLC =	0.0	DWLC =	1.0008	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.07		2133.47	2128.72	4.75	63.57	0.01	2.67	0.0	1.157	1.157	87.4063	0.397499E+06
BETA		CL	CD	CM	CROLL		CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01		1.65	-0.07	-1.13	0.01		-0.02	0.03	0.86	-0.54	0.86	0.42

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3852	-0.4369
2.5	-0.3852	-0.4369
5.0	-0.3852	-0.4369
10.0	-0.3852	-0.3852
15.0	-0.4369	-0.3335
25.0	-0.3852	-0.3852
35.0	-0.4369	-0.3852
50.0	-0.4369	
56.0	-0.4369	-0.4369
65.0	-0.3852	-0.4369
76.0	-0.4369	-0.4369
79.0	-0.3852	-0.4369
80.5	-0.4074	-0.3702
81.0	-0.2955	-0.3702
82.0	-0.4448	-0.4074
84.0	-0.2208	
87.0	-0.2955	
89.0	-0.3684	-0.3684
93.0	-0.4235	
96.0	-0.4235	-0.4235
100.0	-0.4235	-0.4235
96.0	-0.4235	-0.4235
84.0	-0.4235	-0.4235
73.0	-0.4235	-0.4235
50.0	-0.3684	-0.4235
35.0	-0.3852	-0.3852
25.0	-0.4369	-0.4369
10.0	-0.3852	-0.3852
5.0	-0.3852	-0.4886
2.5	-0.4369	-0.3852

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.4311	0.3586	-0.5112	-0.5112	-0.8012
2.5	-0.3663	-0.5112	-1.0186	-1.0911	-1.0186
5.0	-0.4388	-0.4388	-0.6562	-0.7287	-1.0186
10.0	-0.3663	-0.5112	-0.7287	-0.6562	-0.8012
15.0	-0.5112	-0.4388	-0.7287	-0.5112	-0.6562
24.0	-0.4388	-0.5112	-0.5837	-0.5112	-0.5837
33.0	-0.6655	-0.6655	-0.6655	-0.6655	-0.6655
54.0	-0.8751	-0.8751	-0.8751	-1.9229	-0.8751
65.0	-0.9752	-1.9684	-0.9201	-0.8649	-0.9752
78.5	-0.9201	-3.5683	-1.5270	-1.5821	-1.7477
79.5	********	-43.7068	-1.4526	-1.5272	-2.0125
80.5	-95.6655	-2.6098	-1.4153	-1.4526	-0.3702
81.3	-71.2532	333.2561	-1.4526	-1.4899	-1.9006
82.0	-52.7022	-5.6706	-1.3406	-1.3780	-1.7885
84.0	2.1308	-2.7591	-1.3033	-1.4899	-1.8258
87.0	2.4667	-2.6098	-1.3780	-1.5272	-0.3328
89.0	-1.2344	-2.5817	-1.5938	-1.6386	-1.9082
93.0	-0.3363	-1.9979	-1.6386	-0.4261	-0.3363
96.0	0.1577	-1.5489	-1.5040	-1.6386	-2.0427
100.0	-1.8633	-1.8633	-1.4141	-1.6837	-0.3812
96.0	0.4721	0.3823	0.0679	0.2475	0.0679
84.0	0.5619	0.7415	-0.3812	0.6517	0.5170
73.5	0.3823	-0.6655	0.3823	0.5918	0.3823
54.0	0.1727	0.1727	0.1727	0.1727	-0.0369
33.0	-0.0369	-0.6655	-0.0369	-0.0369	-0.0369
24.0	0.1411	0.0686	-0.0039	0.2136	-0.0039
10.0	0.1411	0.1411	-0.0039	0.2861	-0.0369
5.0	0.2136	0.1411	0.2136	0.3586	0.1727
2.5	0.0686	0.1411	0.3586	0.5036	0.1727

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1477
43.4	-0.1477
50.4	-0.1477
56.1	-0.2029
62.5	-0.2029
69.7	-0.3132
76.9	-0.3132
83.4	-0.3132
89.4	-0.3684
95.4	-0.3684
101.1	-0.3684
---BOTTOM---	
38.2	-0.0926
43.4	-0.0373
50.4	-0.2029
56.1	0.0178
62.5	0.0178
69.7	0.0178
76.9	0.0178
83.4	0.0730
89.4	0.1282
95.4	-0.1477
101.1	-0.5891

MISC.*

NO.	CP
1	-0.6655
2	-0.0369
3	-0.6655
4	-0.0369
5	-0.6655
6	-0.2465
7	-0.4261
8	-0.3812
9	-0.3812
10	-0.4261
11	-0.3812
12	-0.3812

TABLE A42. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 674

281 3 CLAERO = 0.8725 CDAERO = 0.5326 DCLC = 0.0 DWLC = 0.9656 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.02	2133.40	2128.65	4.75	63.55	0.01	2.67	0.0	1.074	1.074	87.2774	0.397797E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	1.84	0.06	-1.17	0.01	-0.02	0.03	1.05	-0.58	1.01	0.49

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3029	-0.3546
2.5	-0.3546	-0.3029
5.0	-0.3029	-0.3029
10.0	-0.3029	-0.3546
15.0	-0.3029	-0.3029
25.0	-0.3546	-0.3029
35.0	-0.3029	-0.3029
50.0	-0.3546	
56.0	-0.3546	-0.3546
65.0	-0.3546	-0.4580
76.0	-0.3546	-0.4580
79.0	-0.3546	-0.4063
80.5	-0.3539	-0.2419
81.0	-0.4286	-0.4286
82.0	-0.3539	-0.3166
84.0	-0.3913	
87.0	-0.3913	
89.0	-0.3895	-0.4446
93.0	-0.4446	
96.0	-0.4446	-0.3895
100.0	-0.4446	-0.4446
96.0	-0.4446	-0.4446
84.0	-0.4446	-0.4446
73.0	-0.4446	-0.4446
50.0	-0.4446	-0.4446
35.0	-0.3029	-0.4063
25.0	-0.3546	-0.5097
10.0	-0.3029	-0.4063
5.0	-0.3029	-0.3029
2.5	-0.3029	-0.3029

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.0249	-0.6048	-2.1995	-3.0693	-3.0693
2.5	-0.3873	-0.9673	-1.4746	-2.6344	-2.8518
5.0	-0.6048	-0.7498	-1.2572	-1.6197	-1.7646
10.0	-0.3873	-0.7498	-1.0397	-1.1122	-1.2572
15.0	-0.6048	-0.6773	-0.9673	-0.8948	-1.2572
24.0	-0.5323	-0.6048	-0.7498	-0.8223	-0.8223
33.0	-0.6866	-0.6866	-0.8961	-0.8961	-0.8961
54.0	-0.8961	-0.8961	-0.8961	-1.7344	-0.8961
65.0	-1.0515	-1.9895	-0.9963	-0.9963	-1.1067
78.5	-0.9412	-3.5893	-1.5480	-1.6584	-1.8791
79.5	********	-43.3172	-1.5110	-1.5485	-2.2204
80.5	-95.1264	-2.6308	-1.5110	-1.5857	-0.6152
81.3	-72.7675	312.5525	-1.4737	-1.5485	-2.0338
82.0	-55.3370	-5.3930	-1.4364	-1.5857	-1.9590
84.0	1.1765	-2.8174	-1.3616	-1.6603	-1.9217
87.0	3.1175	-2.7056	-1.6976	-1.7723	-0.4286
89.0	-1.1658	-2.6476	-1.7496	-1.8844	-2.1089
93.0	-0.4023	-2.1987	-1.8393	-0.4921	-0.4472
96.0	0.0917	-1.7945	-1.5251	-1.8393	-2.1987
100.0	-1.9741	-1.9293	-1.6148	-1.7048	-0.4472
96.0	0.4061	0.3162	-0.0430	0.1366	-0.0430
84.0	0.5408	0.7653	-0.4472	0.5857	0.4510
73.5	0.3612	-0.6866	0.3612	0.5708	0.3612
54.0	0.1516	0.1516	-0.0579	0.1516	-0.0579
33.0	-0.0579	-0.6866	-0.0579	-0.0579	-0.0579
24.0	0.2650	0.1200	0.1925	0.2650	0.1200
10.0	0.3375	0.2650	0.3375	0.5550	0.1516
5.0	0.4825	0.2650	0.4100	0.5550	0.3612
2.5	0.4825	0.4100	0.5550	0.5550	0.3612

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2791
43.4	-0.2791
50.4	-0.2791
56.1	-0.2791
62.5	-0.3343
69.7	-0.3343
76.9	-0.3343
83.4	-0.2791
89.4	-0.3343
95.4	-0.3895
101.1	-0.3895
---BOTTOM---	
38.2	-0.1136
43.4	-0.0584
50.4	-0.2239
56.1	-0.0584
62.5	-0.0032
69.7	0.1071
76.9	0.0519
83.4	0.1071
89.4	0.1071
95.4	-0.1688
101.1	-0.7205

MISC.

NO.	CP
1	-0.8961
2	0.1516
3	-0.8961
4	-0.0579
5	-0.8961
6	-0.2675
7	-0.4921
8	-0.5370
9	-0.5370
10	-0.5819
11	-0.5370
12	-0.5819

TABLE A42. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 676

281 5 CLAERO = 1.3798 CDAERO = 0.7991 DCLC = 0.0 DWLC = 0.9847 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.20	2133.40	2128.65	4.75	63.55	0.01	2.75	0.0	1.034	1.034	90.0912	0.397730E+06

BETA	CL	CD	CM	CROLL	CN	CY	CMTR	CMTR	CLTR	CDTR
-0.01	2.36	0.48	-1.31	0.03	-0.02	0.01	1.62	-0.72	1.49	0.79

CANARD BP=3.375 XX/C	CP	CANARD BP=10.125 CP	WING BP = 2 XX/C	CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4580	-0.4580	0.0	-3.7941	-5.3164	-3.7216	-2.4168	-1.6921
2.5	-0.4063	-0.4063	2.5	-0.6048	-5.4613	-3.5767	-2.4168	-1.5472
5.0	-0.4580	-0.5097	5.0	-1.6197	-2.1995	-3.2867	-2.3444	-1.7646
10.0	-0.4580	-0.4580	10.0	-0.6048	-1.1848	-3.3591	-2.1995	-1.6197
15.0	-0.4580	-0.5097	15.0	-1.1848	-1.1848	-3.4318	-2.8518	-1.5472
25.0	-0.4580	-0.4580	24.0	-0.9673	-1.1122	-1.6921	-2.7068	-1.9095
35.0	-0.4063	-0.4580	33.0	-0.8962	-0.8962	-1.1058	-1.9440	-1.5249
50.0	-0.4580		54.0	-0.8962	-1.1058	-0.8962	-1.3153	-1.7344
56.0	-0.5097	-0.4580	66.0	-1.3273	-2.3204	-1.3273	-1.0515	-2.0447
65.0	-0.4580	-0.4580	78.5	-0.7204	-4.8031	-3.5342	-3.1481	-1.5480
76.0	-0.4580	-0.4580	79.5	********	-49.6251	-3.0788	-3.0788	-1.6230
79.0	-0.4580	-0.4580	80.5	-93.6331	-4.1613	-2.5188	-1.6603	-0.4659
80.5	-0.4285	-0.5032	81.3	-73.6647	299.5578	-3.6013	-1.7351	-2.1456
81.0	-0.5032	-0.5405	82.0	-57.2030	-7.6327	-2.6681	-1.8469	-2.0338
82.0	-0.5405	-0.4285	84.0	-0.7272	-4.0120	-2.6681	-1.8469	-1.6603
84.0	-0.4659		87.0	3.6400	-3.3027	-2.5936	-1.4737	-0.4285
87.0	-0.5779		89.0	-0.8514	-3.4560	-1.7048	-1.2555	-1.4800
89.0	-0.4446	-0.4446	93.0	-0.3574	-2.3783	-1.2555	-0.5370	-0.4921
93.0	-0.4446		96.0	0.0019	-0.8065	-0.9862	-1.4352	-1.5699
96.0	-0.4446	-0.4446	100.0	-1.4800	-0.4921	-0.6269	-1.0311	-0.4921
100.0	-0.4997	-0.4446	96.0	0.5857	0.4959	0.3162	0.2264	0.0468
96.0	-0.4446	-0.3895	84.0	0.7204	0.8102	-0.4921	0.7204	0.4061
84.0	-0.4997	-0.4446	73.5	0.5708	-0.4771	0.5708	0.5708	0.5708
73.0	-0.4446	-0.3895	54.0	0.3612	0.5708	0.5708	0.3612	0.1516
50.0	-0.3895	-0.4446	33.0	0.3612	-0.4771	0.3612	0.3612	0.1516
35.0	-0.5097	-0.4063	24.0	0.3375	0.4100	0.3375	0.5550	0.1925
25.0	-0.4580	-0.4580	10.0	0.5550	0.4825	0.4825	0.6274	0.5708
10.0	-0.4580	-0.4063	5.0	0.6274	0.5550	0.5550	0.5550	0.5708
5.0	-0.4063	-0.5097	2.5	0.6274	0.4825	0.4825	0.5550	0.3612
2.5	-0.4063	-0.4063						

FUSELAGE TOP XLOC	CP
38.2	-0.1136
43.4	-0.1136
50.4	-0.2239
56.1	-0.2239
62.5	-0.3343
69.7	-0.3895
76.9	-0.3895
83.4	-0.3895
89.4	-0.3895
95.4	-0.3895
101.1	-0.2791
BOTTOM	
38.2	-0.0032
43.4	-0.0032
50.4	-0.1136
56.1	0.1071
62.5	0.2175
69.7	0.2726
76.9	0.3278
83.4	0.3278
89.4	0.3278
95.4	-0.0032
101.1	-0.6653

MISC. NO.	CP
1	-0.6868
2	0.3612
3	-1.9440
4	0.5708
5	-1.5249
6	-0.0579
7	-0.4921
8	-0.4921
9	-0.4921
10	-0.4921
11	-0.4921
12	-0.4921

TABLE A43. RUN 282, BW6V, DELF=45, CMU=2.0, (BN/B)W=0.25

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 679

282 1 CLAERO = 0.6069 CDAERO = 0.5751 DCLC = 0.0 DWLC = 1.7321 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.02	2133.47	2130.76	2.71	46.00	0.01	2.66	0.0	2.000	2.000	87.0273	0.301292E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	2.34	-0.42	-1.63	0.03	-0.03	0.06	0.95	-0.60	0.96	0.43

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.3621	-0.4526
2.5	-0.4526	-0.4526
5.0	-0.4526	-0.4526
10.0	-0.3621	-0.4526
15.0	-0.4526	-0.5430
25.0	-0.4526	-0.4526
35.0	-0.4526	-0.3621
50.0	-0.4526	
56.0	-0.4526	-0.4526
65.0	-0.4526	-0.4526
76.0	-0.4526	-0.4526
79.0	-0.3621	-0.4526
80.5	-0.3766	-0.3113
81.0	-0.1153	-0.3766
82.0	-0.3113	-0.4419
84.0	-0.3766	
87.0	-0.2459	
89.0	-0.2600	-0.3565
93.0	-0.2600	
96.0	-0.2600	-0.2600
100.0	-0.3565	-0.2600
96.0	-0.2600	-0.2600
84.0	-0.3565	-0.2600
73.0	-0.2600	-0.2600
50.0	-0.3565	-0.2600
35.0	-0.4526	-0.4526
25.0	-0.4526	-0.4526
10.0	-0.3621	-0.4526
5.0	-0.5430	-0.4526
2.5	-0.4526	-0.4526

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.3607	-0.0197	-0.6537	-0.6537	-1.2878
2.5	-0.4001	-0.5270	-1.2878	-1.0342	-1.7950
5.0	-0.4001	-0.4001	-0.9074	-0.6537	-1.1610
10.0	-0.4001	-0.5270	-0.7806	-0.5270	-0.9074
15.0	-0.4001	-0.5270	-0.9074	-0.5270	-0.7803
24.0	-0.5270	-0.6537	-0.6537	-0.4001	-0.9074
33.0	-0.1051	-0.4717	-0.4717	-0.4717	-0.4717
54.0	-0.4717	-0.4717	-0.8383	-2.6715	-0.8383
65.0	-0.9357	-2.4800	-0.9357	-0.9357	-1.1287
78.5	-1.5975	-4.2176	-1.3218	-1.6114	-3.8315
79.5	********	-69.9916	-1.3561	-1.8132	-3.6416
80.5	********	-4.1642	-1.5521	-1.5521	-0.1806
81.3	********	568.0336	-1.2908	-1.8785	-2.9235
82.0	-89.9739	-5.4089	-1.2908	-2.4010	-3.5112
84.0	7.9822	-3.5764	-1.4867	-1.9438	-2.7927
87.0	5.1088	-3.6416	-1.4867	-1.9438	-0.2459
89.0	-1.9908	-3.3262	-1.9122	-2.5408	-2.3836
93.0	-0.1837	-2.3836	-2.3836	-0.4980	-0.4980
96.0	0.4448	-1.6765	-1.9908	-1.9908	-1.7550
100.0	-3.3262	-2.0693	-1.2050	-0.8122	-0.3409
96.0	0.4448	0.2878	0.0521	0.2878	0.1306
84.0	0.6020	0.7592	-0.4194	0.7592	0.4448
73.5	0.6282	-0.4717	0.6282	0.6282	0.6282
54.0	0.2616	0.2616	0.2616	0.2616	0.2616
33.0	0.2616	-0.4717	0.2616	0.2616	0.2616
24.0	0.2340	-0.1465	-0.0197	0.3607	-0.0197
10.0	0.1071	0.1071	0.1071	0.3607	0.2616
5.0	0.1071	0.1071	0.1071	0.4876	0.2616
2.5	0.1071	0.1071	0.3607	0.3607	0.6282

FUSELAGE

-----TOP---- XLOC	CP	---BOTTOM--- XLOC	CP
38.2	-0.0669	38.2	0.0296
43.4	0.0296	43.4	0.0296
50.4	-0.0669	50.4	-0.0669
56.1	-0.0669	56.1	0.1260
62.5	-0.0669	62.5	0.1260
69.7	-0.0669	69.7	0.1260
76.9	-0.1635	76.9	0.1260
83.4	-0.1635	83.4	0.2226
89.4	-0.2600	89.4	0.1260
95.4	-0.1635	95.4	-0.1635
101.1	-0.2600	101.1	-0.6461

MISC.*

NO.	CP
1	-0.4717
2	0.2616
3	-0.4717
4	0.2616
5	-0.4717
6	-0.1051
7	-0.4194
8	-0.4194
9	-0.4194
10	-0.4194
11	-0.3409
12	-0.4194

TABLE A43. (Continued)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 681

282 3 CLAERO = 0.6805 CDAERO = 0.6649 DCLC = 0.0 DWLC = 1.9312 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.02	2133.54	2130.94	2.60	46.38	0.01	2.68	0.0	2.148	2.148	87.7956	0.295076E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
-0.01	2.61	-0.28	-1.72	0.02	-0.03	0.04	1.13	-0.64	1.10	0.51

CANARD BP=3.375 %X/C	CP	CANARD BP=10.125 CP
0.0	-0.4164	-0.3219
2.5	-0.4164	-0.4164
5.0	-0.4164	-0.4164
10.0	-0.3219	-0.4164
15.0	-0.3219	-0.4164
25.0	-0.4164	-0.3219
35.0	-0.4164	-0.3219
50.0	-0.4164	
56.0	-0.4164	-0.4164
65.0	-0.3219	-0.3219
76.0	-0.3219	-0.4164
79.0	-0.3219	-0.3219
80.5	-0.5677	-0.6358
81.0	-0.5677	-0.4995
82.0	-0.4995	-0.4314
84.0	-0.4995	
87.0	-0.3632	
89.0	-0.5112	-0.5112
93.0	-0.5112	
96.0	-0.5112	-0.5112
100.0	-0.5112	-0.5112
96.0	-0.5112	-0.5112
84.0	-0.5112	-0.5112
73.0	-0.5112	-0.5112
50.0	-0.5112	-0.6119
35.0	-0.4164	-0.3219
25.0	-0.4164	-0.3219
10.0	-0.4164	-0.3219
5.0	-0.3219	-0.4164
2.5	-0.3219	-0.4164

WING %X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.4560	-1.3823	-3.2351	-2.8380	-2.5734
2.5	-0.4560	-1.3823	-2.0440	-2.0440	-2.9705
5.0	-0.8530	-0.9853	-1.3823	-1.5146	-1.6470
10.0	-0.4560	-0.8530	-1.1176	-1.1176	-1.5146
15.0	-0.7206	-0.7206	-0.9853	-0.7206	-0.9853
24.0	-0.5883	-0.7206	-0.7206	-0.5883	-0.8530
33.0	-0.5307	-0.5307	-0.5307	-0.9133	-0.5307
54.0	-0.9133	-0.9133	-0.9133	-2.4438	-0.9133
65.0	-1.3170	-2.9285	-1.2163	-1.3170	-1.3170
78.5	-2.2235	-4.8424	-1.8206	-1.8206	-3.7345
79.5	********	-76.6900	-1.7261	-2.0669	-2.9528
80.5	********	-3.4299	-1.7261	-1.7261	-0.6358
81.3	********	559.0640	-1.7261	-1.8625	-3.0893
82.0	-95.1588	-9.0184	-1.7943	-2.3395	-2.3395
84.0	7.3375	-4.1113	-1.5218	-1.6581	-2.6802
87.0	5.3614	-3.7706	-1.8625	-1.7943	-0.4995
89.0	-2.0339	-3.9199	-2.1979	-2.1979	-2.4438
93.0	-0.3121	-2.9359	-2.1979	-0.5581	-0.4761
96.0	0.1799	-2.0339	-1.8699	-2.1979	-2.5258
100.0	-3.5096	-2.4438	-1.7060	-1.9519	-0.4761
96.0	0.3439	0.2619	-0.1480	0.1799	-0.1480
84.0	0.5079	0.7538	-0.4761	0.6718	0.4258
73.5	0.6172	-0.1480	0.2346	0.6172	0.2346
54.0	0.2346	0.2346	0.2346	0.2346	0.2346
33.0	0.2346	-0.5307	0.2346	0.2346	0.2346
24.0	0.0734	0.2058	-0.0590	0.4704	0.0734
10.0	0.3381	0.0734	0.2058	0.4704	0.2346
5.0	0.4704	0.4704	0.4704	0.6027	0.2346
2.5	0.6027	0.6027	0.4704	0.6027	0.6172

FUSELAGE TOP XLOC	CP
38.2	-0.3097
43.4	-0.2090
50.4	-0.2090
56.1	-0.3097
62.5	-0.4104
69.7	-0.4104
76.9	-0.5112
83.4	-0.5112
89.4	-0.5112
95.4	-0.5112
101.1	-0.4104
BOTTOM	
38.2	-0.1082
43.4	-0.2090
50.4	-0.3097
56.1	-0.1082
62.5	-0.1082
69.7	0.0632
76.9	-0.0075
83.4	-0.0075
89.4	-0.0075
95.4	-0.3097
101.1	-0.9141

MISC. NO.	CP
1	-0.5307
2	0.2346
3	-0.5307
4	0.2346
5	-0.9133
6	-0.1480
7	-0.5581
8	-0.3941
9	-0.4761
10	-0.4761
11	-0.4761
12	-0.4761

TABLE A43. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 683

282 5	CLAERO =	1.2496	CDAERO =	0.8061	DCLC =	0.0	DWLC =	1.8461	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
12.13	2133.54	2130.94	2.60	46.98	0.01	2.74	0.0	1.940	1.940	89.7773	0.295104E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
-0.01	3.10	0.21	-1.81	0.03	-0.03	0.02	1.61	-0.73	1.48	0.78		

********** CANARD ******			********** WING **********						**FUSELAGE**		**MISC.***	
	BP=3.375	BP=10.125		BP = 2	BP = 6	BP =12	BP =16	BP = 22	-----TOP----			
%X/C	CP	CP	%X/C	CP	CP	CP	CP	CP	XLOC	CP	NO.	CP
0.0	-0.5107	-0.5107	0.0	-3.7645	-5.0877	-3.2351	-2.3087	-1.9116	38.2	-0.1082	1	-0.5307
2.5	-0.5107	-0.4163	2.5	-0.5883	-4.9554	-3.3673	-2.1764	-1.7793	43.4	-0.2090	2	0.6172
5.0	-0.5107	-0.6050	5.0	-1.7793	-4.6908	-2.7057	-2.5734	-2.0440	50.4	-0.3097	3	-1.6785
10.0	-0.5107	-0.6050	10.0	-0.5883	-1.1176	-3.4996	-2.4410	-2.0440	56.1	-0.3097	4	0.6172
15.0	-0.6050	-0.6050	15.0	-1.1176	-1.1176	-3.6323	-2.1764	-2.0440	62.5	-0.3097	5	-1.6785
25.0	-0.5107	-0.5107	24.0	-0.9853	-0.9853	-1.9116	-2.3087	-2.3087	69.7	-0.5112	6	0.2346
35.0	-0.5107	-0.6050	33.0	-0.5307	-0.9133	-1.6785	-2.0610	-1.6785	76.9	-0.4104	7	-0.4761
50.0	-0.5107		54.0	-0.9133	-0.9133	-0.9133	-2.8263	-2.0610	83.4	-0.4104	8	-0.5581
56.0	-0.6050	-0.6050	65.0	-1.4178	-3.2308	-1.4178	-1.3169	-1.5185	89.4	-0.5112	9	-0.5581
65.0	-0.5107	-0.5107	78.5	-2.1228	-5.8499	-3.2308	-1.7199	-1.3169	95.4	-0.4104	10	-0.5581
76.0	-0.4163	-0.6050	79.5	********	-95.0204	-5.0654	-6.1559	-1.5218	101.1	-0.4104	11	-0.5581
79.0	-0.5107	-0.6050	80.5	********	-6.4285	-3.8387	-4.1113	-0.4995	---BOTTOM---		12	-0.5581
80.5	-0.5677	-0.4314	81.3	********	535.8376	-1.9988	-1.6580	-1.2491	38.2	0.0932		
81.0	-0.4995	-0.3632	82.0	********	-11.6077	-2.4759	-1.5218	-1.5218	43.4	0.0932		
82.0	-0.3632	-0.4995	84.0	6.9969	-5.4061	-2.6122	-1.3855	-1.4536	50.4	-0.2090		
84.0	-0.4995		87.0	6.2474	-4.5886	-3.1573	-1.2491	-0.6358	56.1	0.0932		
87.0	-0.5677		89.0	-1.7879	-2.7719	-1.6239	-1.5420	-1.5420	62.5	0.2947		
89.0	-0.4104	-0.5112	93.0	-0.4761	-1.7879	-1.4600	-0.5581	-0.5581	69.7	0.1940		
93.0	-0.4104		96.0	-0.0661	-1.2960	-1.2140	-1.4600	-2.2799	76.9	0.2947		
96.0	-0.4104	-0.4104	100.0	-2.9359	-1.2960	-1.0500	-1.4600	-0.6401	83.4	0.1940		
100.0	-0.5112	-0.3097	96.0	0.4258	0.2618	0.0979	0.1799	-0.0661	89.4	0.1940		
96.0	-0.5112	-0.3097	84.0	0.5079	0.7538	-0.5581	0.5898	0.4258	95.4	-0.1082		
84.0	-0.5112	-0.3097	73.5	0.9998	-0.1480	0.6172	0.9998	0.2346	101.1	-0.9141		
73.0	-0.5112	-0.3097	54.0	0.6172	0.2346	0.6172	0.6172	0.2346				
50.0	-0.5112	-0.4104	33.0	0.2346	-0.1480	0.2346	0.2346	0.2346				
35.0	-0.5107	-0.5107	24.0	0.2058	0.2058	0.2058	0.4704	-0.0590				
25.0	-0.5107	-0.5107	10.0	0.4704	0.4704	0.4704	0.6027	0.2346				
10.0	-0.5107	-0.5107	5.0	0.6027	0.6027	0.6027	0.6027	0.6172				
5.0	-0.5107	-0.6050	2.5	0.7351	0.6027	0.6027	0.4704	0.2346				
2.5	-0.5107	-0.5107										

TABLE A44. RUN 289, BC1W6V, DELF=45, CMU=0,(BN/B)W=0.25

RUN:SEQ — PROPLULSIVE — WING / CANARD — PRESSURES — PAGE 711

RUN:SEQ											
289	2	CLAERO =	0.5316	CDAERO =	0.2390	DCLC =	0.0	DWLC =	0.0	BASEPR =	8.1904
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.06		2133.19	2103.01	30.17	160.35	-0.05	2.66	0.0	0.0	0.0	87.0440	0.100699E+07
BETA		CL	CD	CN	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.06		0.53	0.24	-0.18	0.00	-0.00	0.00	0.52	-0.17	0.52	0.27	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.5127	0.5452
2.5	-0.0895	-0.2522
5.0	-0.1383	-0.2603
10.0	0.7242	-0.2847
15.0	-0.2441	-0.2929
25.0	-0.2441	-0.2685
35.0	-0.2359	-0.2441
50.0	-0.2359	
56.0	-0.2522	-0.2685
65.0	-0.3173	-0.3417
76.0	-0.2603	-0.5207
79.0	-0.4882	-0.6590
80.5	-0.4852	-0.6496
81.0	-0.4817	-0.6261
82.0	-0.4734	-0.6498
84.0	-0.2798	
87.0	-0.4910	
89.0	-0.4837	-0.6834
93.0	-0.4924	
96.0	-0.5010	0.6189
100.0	0.7491	-0.6486
96.0	0.3411	0.3411
84.0	0.7491	-0.6486
73.0	0.3411	0.3411
50.0	0.6536	0.6102
35.0	0.5289	0.5452
25.0	0.1709	-0.3336
10.0	0.0407	0.0407
5.0	-0.0895	-0.0081
2.5	-0.1790	-0.0488

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0542	0.1112	0.2937	0.0998	0.0770
2.5	-0.2310	0.2025	-0.2082	-0.4363	-0.6872
5.0	0.1454	0.0998	-0.2310	-0.3793	-0.5732
10.0	-0.2424	-0.0371	-0.2766	-0.3907	-0.5047
15.0	-0.0143	-0.1055	-0.2994	-0.3222	-0.4249
24.0	-0.0941	-0.1733	-0.2994	-0.2994	-0.3564
33.0	-0.1533	-0.2192	-0.3511	-0.3511	-0.3511
54.0	-0.2522	-0.3192	-0.4171	0.1765	-0.3511
65.0	-0.3101	-0.3622	-0.5097	-0.4576	-0.4490
78.5	0.1587	-0.6313	-0.8830	-0.7702	-0.7789
79.5	-0.5028	-0.5233	-0.8904	-0.6906	-0.7671
80.5	-0.5439	-0.5380	-0.8963	-0.7025	-0.2796
81.3	-0.4969	53.2363	-0.7965	-0.6614	-0.7495
82.0	-0.5263	-1.1606	-0.8435	-0.6436	-0.7436
84.0	-0.5204	-0.5615	-0.6906	-0.6906	-0.7612
87.0	-0.5615	-0.5909	-0.7965	-0.6849	-0.2913
89.0	-0.5891	-0.6245	-0.8715	-0.7163	-0.7588
93.0	-0.5750	-0.6457	-0.7870	-0.2994	-0.2711
96.0	-0.5609	-0.6527	-0.7588	-0.7022	-0.7941
100.0	-0.5185	-0.6598	-0.6810	-0.6952	-0.2641
96.0	0.2165	0.2871	0.3295	0.3790	0.2801
84.0	0.4073	0.5274	-0.2782	0.5840	0.5204
73.5	0.3414	-0.2522	0.4733	0.5062	0.4733
54.0	0.1435	0.1435	0.2095	0.2424	0.1765
33.0	0.0116	-0.2522	0.0775	0.0775	0.0446
24.0	-0.0371	-0.0143	0.0428	0.0884	0.0199
10.0	-0.1055	-0.1055	0.0085	0.0998	0.0775
5.0	-0.1169	-0.1625	0.0085	0.1568	0.1435
2.5	-0.1511	-0.2980	-0.0257	0.1682	0.2095

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.1625
43.4	-0.1538
50.4	-0.1191
56.1	-0.0843
62.5	-0.0583
69.7	-0.0843
76.9	-0.0930
83.4	-0.0930
89.4	-0.1972
95.4	-0.2406
101.1	-0.2059
---BOTTOM---	
38.2	0.0025
43.4	0.0285
50.4	-0.0583
56.1	0.0198
62.5	0.0112
69.7	0.0025
76.9	0.0459
83.4	0.1067
89.4	0.1587
95.4	0.0719
101.1	-0.3101

MISC.*

NO.	CP
1	-0.2522
2	0.0446
3	-0.3841
4	0.0775
5	-0.2852
6	0.0116
7	-0.2923
8	-0.2782
9	-0.2782
10	-0.2853
11	-0.2853
12	-0.2853

TABLE A44. (Continued)

RUN: SEQ
289 4

PROPLULSIVE WING / CANARD PRESSURES

CLAERO = 0.7639 CDAERO = 0.3084 DCLC = 0.0 DWLC = 0.0

BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.15	2133.33	2103.04	30.29	160.84	-0.05	2.67	0.0	0.0	0.0	87.2514	0.100584E+07

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR
0.05	0.76	0.31	-0.16	0.00	-0.00	0.00	0.77	-0.14	0.75	0.33

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.0844	-0.6127
2.5	-0.7425	-1.0181
5.0	-0.5884	-0.6857
10.0	0.6195	-0.6209
15.0	-0.4912	-0.5722
25.0	-0.3858	-0.4344
35.0	-0.3209	-0.3533
50.0	-0.3209	
56.0	-0.3209	-0.3452
65.0	-0.3776	-0.4425
76.0	-0.2804	-0.6127
79.0	-0.5398	-0.7263
80.5	-0.5386	-0.7200
81.0	-0.5152	-0.7024
82.0	-0.5620	-0.7200
84.0	-0.3045	
87.0	-0.5327	
89.0	-0.5478	-0.7208
93.0	-0.5391	
96.0	-0.5391	0.7064
100.0	0.6891	-0.7035
96.0	0.3431	0.3345
84.0	0.6891	-0.7035
73.0	0.3431	0.3345
50.0	0.6286	0.6632
35.0	0.5708	0.5627
25.0	0.2141	-0.4101
10.0	0.1249	0.1249
5.0	0.0115	0.1249
2.5	0.2385	0.3763

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.1255	0.2391	-0.3859	-1.1700	-2.6587
2.5	-0.2609	-0.0336	-0.7041	-1.0109	-1.5905
5.0	-0.0336	-0.1132	-0.5336	-0.8632	-1.1814
10.0	-0.2609	-0.2268	-0.4996	-0.6473	-0.8064
15.0	-0.1586	-0.2609	-0.4427	-0.5564	-0.6586
24.0	-0.2268	-0.3177	-0.4314	-0.4654	-0.5223
33.0	-0.2480	-0.3137	-0.4123	-0.4451	-0.4123
54.0	-0.3137	-0.3465	-0.4780	0.1134	-0.4123
65.0	-0.3662	-0.4526	-0.6170	-0.5651	-0.5564
78.5	0.1874	-0.6429	-1.0841	-0.9543	-0.8332
79.5	-0.5678	-0.5444	-1.2116	-0.8780	-0.8253
80.5	-0.5444	-0.5678	-1.1940	-0.8604	-0.2518
81.3	-0.5327	53.1121	-1.1062	-0.8604	-0.7961
82.0	-0.5386	-1.1121	-1.1413	-0.8838	-0.8312
84.0	-0.5503	-0.5912	-1.0360	-0.9014	-0.7961
87.0	-0.6029	-0.6088	-1.1706	-0.8780	-0.2694
89.0	-0.6470	-0.6682	-1.1610	-0.9286	-0.8230
93.0	-0.6330	-0.6682	-0.9639	-0.3161	-0.2950
96.0	-0.6259	-0.6611	-0.8653	-0.9146	-0.8864
100.0	-0.5837	-0.6470	-0.7245	-0.9146	-0.2880
96.0	0.2612	0.3316	0.3598	0.3527	0.2471
84.0	0.4443	0.5851	-0.3020	0.5851	0.4583
73.5	0.4420	-0.2480	0.5734	0.5405	0.4420
54.0	0.2448	0.2448	0.3106	0.3106	0.2120
33.0	0.1134	-0.2480	0.1791	0.1791	0.1134
24.0	0.0459	0.0800	0.1141	0.1709	0.1028
10.0	0.0118	0.0459	0.1596	0.2618	0.2448
5.0	0.0005	-0.0109	0.2164	0.3187	0.3763
2.5	-0.0677	-0.0336	0.2732	0.3982	0.4420

FUSELAGE

-----TOP----

XLOC	CP
38.2	-0.2018
43.4	-0.1932
50.4	-0.1499
56.1	-0.1067
62.5	-0.0894
69.7	-0.1326
76.9	-0.1413
83.4	-0.1326
89.4	-0.2106
95.4	-0.2451
101.1	-0.1932

---BOTTOM---

XLOC	CP
38.2	0.0404
43.4	0.0577
50.4	-0.0288
56.1	0.0490
62.5	0.0663
69.7	0.0750
76.9	0.1009
83.4	0.1615
89.4	0.1961
95.4	0.0923
101.1	-0.3143

MISC.*

NO.	CP
1	-0.3794
2	0.1463
3	-0.4451
4	0.1791
5	-0.4123
6	0.0149
7	-0.3161
8	-0.3020
9	-0.3020
10	-0.3020
11	-0.2950
12	-0.2950

TABLE A44. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 715

289 6 CLAERO = 1.2544 CDAERO = 0.5250 DCLC = 0.0 DMLC = 0.0 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.15	2133.47	2103.64	29.83	159.73	-0.04	2.72	0.0	0.0	0.0	89.0004	0.996620E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.04	1.25	0.52	-0.10	-0.00	-0.00	0.00	1.32	-0.08	1.23	0.55

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-5.8034	-1.6394
2.5	-2.4459	-1.5407
5.0	-1.7711	-1.4748
10.0	0.5331	-1.4584
15.0	-1.1292	-1.4419
25.0	-0.8082	-1.4666
35.0	-0.6613	-1.3596
50.0	0.5256	
56.0	-0.5120	-1.0222
65.0	-0.5531	-0.8247
76.0	-0.3392	-0.8412
79.0	-0.6766	-0.8905
80.5	-0.6906	-0.9463
81.0	-0.6611	-0.9284
82.0	-0.6611	-0.9284
84.0	-0.3819	
87.0	-0.6849	
89.0	-0.6748	-0.9207
93.0	-0.6661	
96.0	-0.6748	0.7037
100.0	0.6422	-0.8504
96.0	0.3349	0.3086
84.0	0.6422	-0.8504
73.0	0.3349	0.3086
50.0	0.8090	0.6422
35.0	0.6813	0.5825
25.0	0.3356	-0.5778
10.0	0.3109	0.2698
5.0	0.2780	0.3109
2.5	0.6236	0.5825

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	0.0217	-1.0627	-4.9849	-3.7390	-1.5933
2.5	-0.3244	-0.7974	-1.9971	-3.7390	-1.6279
5.0	-0.4398	-0.7628	-1.4318	-3.8889	-1.6164
10.0	-0.3129	-0.6705	-1.1088	-1.4895	-1.5703
15.0	-0.5090	-0.6474	-0.9589	-0.7858	-1.5587
24.0	-0.4974	-0.5782	-0.7397	-0.7166	-1.5703
33.0	-0.4986	-0.5653	-0.6654	-0.6654	-1.4991
54.0	-0.4652	-0.5319	-0.6320	0.0350	-1.1656
65.0	-0.4992	-0.5783	-0.7363	-0.7363	-0.9470
76.5	0.1856	-0.6661	-1.4475	-1.3070	-1.2631
79.5	-0.5839	-0.5958	-1.8433	-1.1601	-1.3562
80.5	-0.6017	-0.5780	-1.7542	-1.1364	-0.3106
81.2	-0.6136	53.9150	-1.6473	-1.1483	-1.1898
82.0	-0.5601	-1.1007	-1.7007	-1.1780	-1.1898
84.0	-0.6017	-0.6314	-1.3977	-1.0888	-1.1245
87.0	-0.6077	-0.6671	-1.2077	-1.1601	-0.3225
89.0	-0.6774	-0.7203	-1.1133	-1.1920	-1.1133
93.0	-0.7131	-0.7131	-0.8918	-0.3915	-0.3486
96.0	-0.6559	-0.6845	-0.8560	-1.1062	-1.1062
100.0	-0.6059	-0.6130	-0.7417	-1.0204	-0.3486
96.0	0.3232	0.4090	0.3875	0.3589	0.2017
84.0	0.6020	0.7092	-0.3557	0.5877	0.3804
73.5	0.5086	-0.3318	0.6353	0.5686	0.4019
54.0	0.3352	0.3352	0.3685	0.3685	0.2351
33.0	0.2351	-0.3318	0.3018	0.3352	0.2018
24.0	0.1947	0.2524	0.3101	0.3677	0.2178
10.0	0.2178	0.2870	0.4254	0.4716	0.4352
5.0	0.2524	0.3447	0.4600	0.5293	0.5019
2.5	0.2985	0.4024	0.4369	0.4485	0.4352

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3236
43.4	-0.2797
50.4	-0.2095
56.1	-0.1568
62.5	-0.1744
69.7	-0.2271
76.9	-0.2358
83.4	-0.2007
89.4	-0.2709
95.4	-0.2709
101.1	-0.2095
---BOTTOM---	
38.2	0.1154
43.4	0.1242
50.4	0.0100
56.1	0.1154
62.5	0.1593
69.7	0.1944
76.9	0.2295
83.4	0.2559
89.4	0.2822
95.4	0.1593
101.1	-0.3500

MISC.*

NO.	CP
1	-0.6320
2	0.3018
3	-1.0656
4	0.3018
5	-1.2990
6	0.1017
7	-0.3629
8	-0.3557
9	-0.3629
10	-0.3557
11	-0.3701
12	-0.3557

TABLE A45. RUN 290, BC1W6V, DELF=45, CMU=0.5, (BN/B)W=0.25

RUN:SEQ — PROPLULSIVE WING / CANARD PRESSURES — PAGE 716

290 1 CLAERO = 0.4167 CDAERO = 0.4210 DCLC = 0.2138 DWLC = 0.4139 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
0.05	2133.75	2119.97	13.79	108.27	-0.05	2.65	0.261	0.478	0.739	86.5494	0.679618E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	1.04	0.03	-0.06	-0.01	-0.00	0.02	0.62	0.05	0.62	0.33

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.4845	-0.3524
2.5	-0.4771	-1.0825
5.0	-0.4593	-0.9045
10.0	0.0571	-0.7620
15.0	-0.5127	-0.7264
25.0	-0.5127	-0.6730
35.0	-0.5483	-0.6373
50.0	-0.7086	
56.0	-0.6393	-0.9757
66.0	-1.1894	-1.4744
76.0	-0.3347	-3.4154
79.0	-5.1605	-5.8728
80.5	-8.5370	-19.2195
81.0	-5.5035	-11.2882
82.0	-4.1281	-7.8175
84.0	-0.7473	
87.0	-1.5571	
89.0	-1.5641	-2.6090
93.0	-0.9561	
96.0	-0.3481	0.6779
100.0	0.3169	-0.2911
96.0	0.6209	0.5829
84.0	0.3169	-0.2911
73.0	0.6209	0.5829
50.0	0.7919	0.6019
35.0	0.6626	0.5914
25.0	0.3064	-1.1182
10.0	0.2174	0.1818
5.0	0.0571	0.1640
2.5	0.1105	0.3598

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.7215	-0.6216	-0.7464	0.3519	0.1272
2.5	-0.2971	0.7763	-0.3970	-0.5967	-0.9212
5.0	0.6265	0.5766	-0.3471	-0.4968	-0.7714
10.0	-0.3221	0.3269	-0.2721	-0.4469	-0.5967
15.0	0.2770	0.1772	-0.3221	-0.4469	-0.4968
24.0	0.0274	-0.0475	-0.4219	-0.4469	-0.4718
33.0	-0.1538	-0.1538	-0.4425	-0.5868	-0.4425
54.0	-0.4425	-0.4425	-0.5868	-0.8755	-0.5146
65.0	-0.5951	-0.9561	-0.5951	-0.6711	-0.6141
78.5	-0.6141	-2.2481	-0.9561	-1.0511	-1.1271
79.5	-22.8186	-27.8581	-0.8501	-0.9915	-1.2101
80.5	-22.1375	-1.4929	-0.8501	-0.9787	-0.3360
81.3	-22.5234	115.4856	-0.8630	-1.0044	-1.0944
82.0	-22.1759	-2.6627	-0.8758	-0.9530	-1.1458
84.0	-21.8031	-1.5443	-0.8758	-0.9915	-1.1330
87.0	-11.3136	-1.4929	-0.9273	-1.0173	-0.3745
89.0	-1.8912	-1.4581	-1.0251	-1.1334	-1.2725
93.0	2.0217	-1.1179	-0.9787	-0.3910	-0.3601
96.0	0.3514	-0.9478	-0.9787	-1.1024	-1.2725
100.0	-0.9787	-1.0714	-0.8859	-0.9478	-0.3601
96.0	0.1348	0.0111	0.0885	0.3514	0.2586
84.0	0.1194	0.0111	-0.3801	0.6762	0.6298
73.5	-0.1538	-0.3703	0.0627	0.3514	0.4958
54.0	-0.5146	-0.4425	-0.4425	-0.1538	0.0627
33.0	-0.5868	-0.3703	-0.6590	-0.4425	-0.1538
24.0	-0.5717	-0.6216	-0.5967	-0.3720	-0.1723
10.0	-0.6466	-0.5967	-0.6466	-0.3471	-0.0816
5.0	-0.6466	-0.5717	-0.6466	-0.2472	-0.0816
2.5	-0.5717	-0.5468	-0.6466	-0.2222	0.1349

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3101
43.4	-0.3101
50.4	-0.1580
56.1	-0.0441
62.5	0.0129
69.7	-0.0061
76.9	-0.0441
83.4	-0.0820
89.4	-0.1961
95.4	-0.2531
101.1	-0.2151
---BOTTOM---	
38.2	0.1459
43.4	0.2029
50.4	0.0509
56.1	0.0319
62.5	-0.3101
69.7	-0.4241
76.9	-0.3861
83.4	-0.2721
89.4	-0.0630
95.4	-0.0441
101.1	-0.3481

MISC.*

NO.	CP
1	-0.2981
2	-0.6590
3	-0.5146
4	-0.2259
5	-0.4425
6	-0.1538
7	-0.3910
8	-0.3601
9	-0.3446
10	-0.3601
11	-0.3755
12	-0.3601

TABLE A45. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
290 3	CLAERO =	0.6265	CDAERO =	0.4831	DCLC =	0.2276	DWLC =	0.4383	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.08	2133.75	2120.08	13.67	107.82	-0.05	2.66	0.265	0.487	0.753	87.0614	0.676836E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.05	1.29	0.13	-0.03	0.00	-0.00	0.02	0.87	0.09	0.84	0.40		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-1.0809	-3.2713
2.5	-1.2065	-2.1222
5.0	-0.8116	-1.5118
10.0	0.2657	-1.2065
15.0	-0.7756	-1.1168
25.0	-0.6679	-0.8475
35.0	-0.6679	-0.8295
50.0	-0.8116	
56.0	-0.9552	-1.1168
65.0	-1.2963	-1.6375
76.0	-0.3268	-3.7023
79.0	-5.3720	-6.2518
80.5	-8.6800	-20.0468
81.0	-5.6341	-11.8554
82.0	-4.2343	-8.3041
84.0	-0.7349	
87.0	-1.6162	
89.0	-1.6418	-2.9253
93.0	-1.0[illegible]79	
96.0	-0.39[illegible]	0.7337
100.0	0.4655	-0.3391
96.0	0.6379	0.5421
84.0	0.4655	-0.3391
73.0	0.6379	0.5421
50.0	0.7720	0.6187
35.0	0.7505	0.6069
25.0	0.3735	-1.2245
10.0	0.3376	0.2837
5.0	0.1939	0.2657
2.5	0.4094	0.5530

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.7348	-0.6341	-0.9109	-0.5334	-2.1191
2.5	-0.3069	0.5992	-0.9864	-0.6844	-1.6156
5.0	0.5489	0.5237	-1.1627	-0.6341	-1.1627
10.0	-0.3573	0.2972	-1.1123	-0.6089	-0.8606
15.0	0.1965	0.0958	-1.0368	-0.6089	-0.7096
24.0	0.0203	-0.1055	-0.9361	-0.6089	-0.5586
33.0	-0.0896	-0.1624	-0.6717	-0.6717	-0.4534
54.0	-0.3807	-0.3807	-0.5262	-0.6717	-0.5262
65.0	-0.6456	-1.0479	-0.6073	-0.9139	-0.7223
78.5	-0.4924	-2.6380	-0.9904	-1.1437	-1.4694
79.5	-22.9757	-31.4524	-0.9034	-0.9682	-1.3181
80.5	-22.1596	-1.4218	-1.0201	-0.9812	-0.3201
81.3	-22.4185	112.0886	-0.9652	-1.0330	-1.5255
82.0	-22.5355	-2.8216	-0.9812	-1.0070	-1.6292
84.0	-21.9782	-1.9143	-0.8256	-1.0459	-1.5903
87.0	-12.5033	-1.7458	-1.0978	-1.0070	-0.3201
89.0	-3.0213	-1.5654	-1.0720	-1.2436	-1.6178
93.0	2.0779	-1.1657	-1.0409	-0.4171	-0.3548
96.0	0.2846	-1.0565	-1.0565	-1.0876	-1.4433
100.0	-0.8382	-1.0720	-0.9785	-0.9629	-0.3548
96.0	0.2378	0.0975	0.1286	0.4093	0.3158
84.0	0.2222	0.1754	-0.3704	0.7992	0.6120
73.5	-0.0168	-0.3807	0.2742	0.5653	0.4925
54.0	-0.3807	-0.2351	-0.1624	-0.0896	0.1287
33.0	-0.5990	-0.3807	-0.4534	-0.2351	-0.0168
24.0	-0.5334	-0.6089	-0.6341	-0.2817	-0.0552
10.0	-0.6844	-0.5082	-0.6592	-0.3824	0.2015
5.0	-0.5837	-0.5082	-0.6341	-0.3320	0.2015
2.5	-0.6341	-0.5082	-0.6844	-0.3069	0.3470

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3966
43.4	-0.3966
50.4	-0.2242
56.1	-0.0901
62.5	-0.0326
69.7	-0.0710
76.9	-0.1093
83.4	-0.1475
89.4	-0.2242
95.4	-0.2816
101.1	-0.2433
---BOTTOM---	
38.2	0.1590
43.4	0.2356
50.4	0.0632
56.1	0.0440
62.5	-0.2242
69.7	-0.3774
76.9	-0.3583
83.4	-0.1858
89.4	0.0057
95.4	-0.0135
101.1	-0.3966

MISC.*

NO.	CP
1	-0.3079
2	-0.5990
3	-0.5262
4	-0.0896
5	-0.4534
6	-0.0896
7	-0.3859
8	-0.3859
9	-0.3859
10	-0.3548
11	-0.3704
12	-0.3704

TABLE A45. (Concluded)

RUN:SEQ 290 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 720

CLAERO = 1.3032 CDAERO = 0.8575 DCLC = 0.2567 DWLC = 0.4893 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.15	2133.68	2120.46	13.22	106.02	-0.04	2.72	0.279	0.514	0.793	88.9776	0.665757E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.04	2.05	0.59	-0.03	-0.00	-0.00	0.01	1.68	0.09	1.54	0.80

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-8.0059	-3.0120
2.5	-5.2213	-3.0863
5.0	-1.6196	-3.0863
10.0	0.4226	-3.2534
15.0	-1.4710	-3.4391
25.0	-1.1555	-4.1631
35.0	-1.0626	-3.2348
50.0	-1.1184	
56.0	-1.2483	-1.0626
65.0	-1.6010	-1.3225
76.0	-0.5428	-3.6062
79.0	-5.9640	-6.3539
80.5	-9.3618	-21.8255
81.0	-6.1183	-13.0203
82.0	-4.7244	-9.4419
84.0	-1.0390	
87.0	-1.9235	
89.0	-1.9335	-3.6965
93.0	-1.2601	
96.0	-0.5866	0.8199
100.0	0.5822	-0.4677
96.0	0.6812	0.4831
84.0	0.5822	-0.4677
73.0	0.6812	0.4831
50.0	0.8397	0.6614
35.0	0.8125	0.5897
25.0	0.5155	-1.5825
10.0	0.4783	0.3855
5.0	0.4412	0.4226
2.5	0.6083	0.4412

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.2132	-0.7976	-0.7716	-1.9427	-4.1288
2.5	-0.5633	0.1654	-0.9277	-1.4742	-3.3603
5.0	0.4256	0.1394	-1.0318	-1.1879	-3.7124
10.0	-0.6414	-0.0949	-1.1619	-0.9277	-3.0878
15.0	-0.0168	-0.2771	-1.2921	-0.8756	-2.0207
24.0	-0.1730	-0.4853	-1.2921	-0.8496	-0.9277
33.0	-0.2809	-0.4314	-1.2591	-0.8829	-0.7324
54.0	-0.5067	-0.6572	-1.1839	-0.5819	-0.8076
65.0	-0.8441	-1.2006	-1.2795	-1.1214	-0.9828
78.5	-0.3092	-3.8154	-1.5988	-1.4384	-1.5374
79.5	-23.8491	-46.0428	-1.4276	-1.2936	-1.3740
80.5	-22.9106	-5.8904	-1.7090	-1.2266	-0.5029
81.3	-23.3530	107.8493	-1.5616	-1.2668	-1.4008
82.0	-23.6078	-5.2606	-1.5482	-1.2936	-1.4544
84.0	-23.0316	-3.7461	-1.4678	-1.2936	-1.4142
87.0	-16.0224	-2.9822	-1.8029	-1.3338	-0.5430
89.0	-6.2900	-2.5170	-1.9526	-1.5979	-1.5172
93.0	1.7399	-1.6462	-1.7269	-0.6626	-0.6143
96.0	0.1597	-1.0657	-1.5818	-1.4043	-1.5979
100.0	-0.0821	-0.5820	-1.7430	-1.5011	-0.6143
96.0	0.6596	0.4177	0.0630	0.1597	0.1920
84.0	0.3210	0.4177	-0.6465	0.4661	0.6596
73.5	0.2457	-0.5067	0.2457	0.3210	0.4715
54.0	0.1705	0.1705	0.1705	0.1705	0.1705
33.0	-0.1305	-0.5067	0.0200	0.1705	0.1705
24.0	-0.2771	-0.3291	-0.0428	0.1654	0.1654
10.0	-0.8236	-0.6414	-0.3291	0.1654	0.3962
5.0	-1.0578	-0.7195	-0.4593	0.1914	0.3962
2.5	-1.0578	-0.7455	-0.4593	0.2174	0.3210

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.5866
43.4	-0.5074
50.4	-0.3290
56.1	-0.1903
62.5	-0.1112
69.7	-0.1503
76.9	-0.1705
83.4	-0.1705
89.4	-0.3092
95.4	-0.3489
101.1	-0.2894
---BOTTOM---	
38.2	0.2454
43.4	0.2851
50.4	0.0870
56.1	0.1068
62.5	-0.0715
69.7	-0.1310
76.9	-0.0319
83.4	0.1860
89.4	0.2454
95.4	0.0870
101.1	-0.6460

MISC.*

NO.	CP
1	-0.9581
2	0.0200
3	-0.8076
4	0.1705
5	-1.3343
6	0.0200
7	-0.6304
8	-0.6143
9	-0.6143
10	-0.5981
11	-0.5820
12	-0.5981

TABLE A46. RUN 291, BC1W6V, DELF=45, CMU=1.0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ													
291	1	CLAERO =	0.2638	CDAERO =	0.5784	DCLC =	0.4660	DWLC =	0.9072	BASEPR =	8.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
-0.08		2133.96	2127.52	6.44	73.91	-0.05	2.66	0.569	1.048	1.618	86.9237	0.465251E+06	
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.05		1.64	-0.27	-0.24	-0.01	0.00	0.01	0.71	0.01	0.71	0.37		

********** CANARD ******

%X/C	BP=3.975 CP	BP=10.125 CP
0.0	0.4754	-0.3630
2.5	-0.4773	-1.2014
5.0	-0.4773	-0.9346
10.0	0.4754	-0.8584
15.0	-0.5535	-0.8204
25.0	-0.5917	-0.7822
35.0	-0.6298	-0.7822
50.0	-0.7822	
56.0	-0.9728	-1.1633
65.0	-1.4301	-1.7731
76.0	-0.3630	-4.2501
79.0	-6.4223	-7.4132
80.5	-13.1310	-32.5251
81.0	-8.0692	-18.4683
82.0	-5.8135	-12.3057
84.0	-0.8067	
87.0	-1.5769	
89.0	-2.0407	-3.5451
93.0	-1.1868	
96.0	-0.1702	0.6836
100.0	0.9683	-0.9022
96.0	0.5617	0.5617
84.0	0.9683	-0.9022
73.0	0.5617	0.5617
50.0	0.7650	0.6023
35.0	0.6659	0.5516
25.0	0.2848	-1.3158
10.0	0.2087	0.1705
5.0	0.0943	0.0943
2.5	0.0943	0.3992

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.9677	-0.8609	-0.3266	0.8486	0.0474
2.5	-0.3266	1.0089	0.5282	-0.6471	-0.6471
5.0	0.7418	0.7952	0.1008	-0.4868	-0.7005
10.0	-0.3266	0.3679	-0.1129	-0.3800	-0.6471
15.0	0.3679	0.2076	-0.2732	-0.3266	-0.5403
24.0	0.2076	-0.0061	-0.3266	-0.3266	-0.4868
33.0	-0.0722	-0.2266	-0.3810	-0.5355	-0.5355
54.0	-0.3810	-0.3810	-0.5355	-1.4622	-0.5355
65.0	-0.7395	-1.4713	-0.7395	-0.6988	-0.6988
78.5	-1.5934	-3.3011	-1.1462	-1.2275	-1.3494
79.5	-50.8195	-60.7227	-1.0268	-1.0543	-1.3845
80.5	-48.9755	-1.9896	-1.1369	-0.9717	-0.3115
81.3	-48.3702	230.0229	-1.0543	-1.0268	-1.2744
82.0	-47.5723	-4.4105	-1.0268	-0.9993	-1.3294
84.0	-45.8121	-2.4022	-0.8893	-1.0268	-1.3294
87.0	-13.9014	-1.8796	-1.1004	-0.9442	-0.3115
89.0	2.2998	-1.5727	-1.1755	-1.2418	-1.5064
93.0	3.1603	-1.2749	-1.2086	-0.4143	-0.3812
96.0	0.1484	-1.0762	-1.1424	-1.2086	-1.6058
100.0	-1.9367	-1.1094	-0.9770	-1.1755	-0.3481
96.0	0.0160	0.0160	0.2146	0.3470	0.2477
84.0	0.1153	0.4132	-0.4143	0.7442	0.6780
73.5	-0.2266	-0.3810	0.2367	0.5456	0.5456
54.0	-0.3810	-0.3810	-0.3810	-0.2266	0.0822
33.0	-0.6900	-0.3810	-0.6900	-0.2266	-0.2266
24.0	-0.8609	-0.9677	-0.5938	-0.2732	-0.1129
10.0	-0.7539	-0.8609	-0.7539	-0.2198	-0.0722
5.0	-0.7005	-0.8073	-0.9143	-0.1663	0.0822
2.5	-0.6471	-0.7539	-1.0211	-0.2198	0.2367

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3329
43.4	-0.3329
50.4	-0.1702
56.1	-0.0483
62.5	0.0737
69.7	-0.0076
76.9	-0.0483
83.4	-0.1296
89.4	-0.1702
95.4	-0.2516
101.1	-0.2109
---BOTTOM---	
38.2	0.1550
43.4	0.1957
50.4	-0.0483
56.1	-0.1702
62.5	-0.6175
69.7	-0.6988
76.9	-0.5768
83.4	-0.3735
89.4	-0.1702
95.4	-0.1296
101.1	-0.4955

MISC.*

NO.	CP
1	-0.2266
2	-0.6900
3	-0.5355
4	-0.2266
5	-0.3810
6	-0.2266
7	-0.3481
8	-0.3481
9	-0.3150
10	-0.3481
11	-0.3481
12	-0.3481

TABLE A46. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
291 3	CLAERO =	0.3103	CDAERO =	0.5364	DCLC =	0.4821	DWLC =	0.9278	BASEPR =	8.1904		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
4.05	2134.11	2127.66	6.44	73.89	-0.05	2.68	0.562	1.032	1.594	87.6989	0.465629E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.05	1.72	-0.20	-0.09	-0.01	0.00	-0.00	0.77	0.16	0.74	0.38		

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.9501	-3.7321
2.5	-1.4075	-3.2367
5.0	-0.9501	-2.0553
10.0	0.8029	-1.2932
15.0	-0.7977	-1.1407
25.0	-0.7215	-0.9501
35.0	-0.7215	-0.9121
50.0	-0.9121	
56.0	-0.9883	-1.3313
65.0	-1.4456	-1.9029
76.0	-0.3023	-4.5705
79.0	-6.6284	-7.7716
80.5	-13.2566	-33.0917
81.0	-8.1398	-18.9230
82.0	-5.9666	-12.7064
84.0	-0.8773	
87.0	-1.6475	
89.0	-2.0969	-3.8046
93.0	-1.2023	
96.0	-0.2671	0.7495
100.0	1.2374	-0.9583
96.0	0.5868	0.5462
84.0	1.2374	-0.9583
73.0	0.5868	0.5462
50.0	0.8308	0.6681
35.0	0.6886	0.5742
25.0	0.3455	-1.4075
10.0	0.3074	0.2312
5.0	0.2312	0.2693
2.5	0.4217	0.5361

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.9298	-0.8228	-0.9298	-0.2887	-1.7844
2.5	-0.3955	0.8331	-0.7694	-0.6626	-1.8914
5.0	0.6195	0.7263	-0.5023	-0.5558	-1.0366
10.0	-0.3421	0.4058	-0.6093	-0.6093	-0.7694
15.0	0.2455	0.1921	-0.6626	-0.5023	-0.7694
24.0	0.0319	-0.0750	-0.6093	-0.5023	-0.5558
33.0	-0.2421	-0.2421	-0.7055	-0.7055	-0.5510
54.0	-0.3965	-0.3965	-0.7055	-1.1688	-0.5510
65.0	-0.7143	-1.4462	-0.6330	-0.8363	-0.7550
78.5	-1.5275	-3.3573	-1.0803	-1.1617	-1.2430
79.5	-49.2114	-61.8667	-0.9322	-1.0148	-1.3174
80.5	-48.4695	-2.4728	-0.9872	-0.9872	-0.4096
81.3	-48.8265	219.3437	-0.9597	-1.0698	-1.2624
82.0	-48.6340	-4.5360	-0.9597	-1.0148	-1.2899
84.0	-46.7636	-2.4452	-0.9322	-1.0148	-1.2073
87.0	-15.5951	-1.9500	-1.0423	-1.0698	-0.3546
89.0	2.9463	-1.4228	-1.1579	-1.2241	-1.4228
93.0	3.7406	-1.2571	-1.0917	-0.4297	-0.4297
96.0	0.3646	-1.0917	-1.1249	-1.1579	-1.4228
100.0	-1.6874	-1.2571	-1.0586	-0.9925	-0.3967
96.0	-0.0988	-0.1319	0.0667	0.3315	0.2653
84.0	0.0005	0.0005	-0.3967	0.6625	0.6294
73.5	-0.2421	-0.3965	-0.2421	0.5301	0.5301
54.0	-0.7055	-0.7055	-0.7055	-0.2421	0.0667
33.0	-0.7055	-0.3965	-0.8599	-0.5510	-0.2421
24.0	-0.7694	-0.8228	-0.8228	-0.5023	-0.0750
10.0	-0.6093	-0.7160	-0.8228	-0.5023	0.0667
5.0	-0.7160	-0.7694	-0.7694	-0.6093	0.2212
2.5	-0.7160	-0.7694	-0.8228	-0.5558	0.3757

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4297
43.4	-0.4297
50.4	-0.2264
56.1	-0.0638
62.5	0.0175
69.7	-0.0231
76.9	-0.0638
83.4	-0.1044
89.4	-0.1857
95.4	-0.2264
101.1	-0.2264
---BOTTOM---	
38.2	0.1802
43.4	0.2209
50.4	0.0989
56.1	-0.0638
62.5	-0.5110
69.7	-0.6736
76.9	-0.6330
83.4	-0.4704
89.4	-0.2254
95.4	-0.1857
101.1	-0.5517

MISC.*

NO.	CP
1	-0.2421
2	-0.8599
3	-0.7055
4	-0.2421
5	-0.5510
6	-0.2421
7	-0.3967
8	-0.3967
9	-0.3967
10	-0.3636
11	-0.3967
12	-0.3967

TABLE A46. (Concluded)

RUN:SEC 291 5 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 725

CLAERO = 0.9046 CDAERO = 0.8052 DCLC = 0.5439 DWLC = 1.0293 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
11.95	2134.25	2128.14	6.10	71.91	-0.04	2.72	0.591	1.083	1.674	88.8088	0.453373E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.04	2.48	0.24	-0.05	-0.02	0.00	-0.00	1.49	0.21	1.37	0.70

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-6.8032	-3.1433
2.5	-7.0042	-3.1834
5.0	-2.6607	-3.2237
10.0	0.8786	-3.4247
15.0	-1.3736	-3.7063
25.0	-1.2128	-4.3097
35.0	-1.1323	-3.4650
50.0	-1.2932	
56.0	-1.3335	-1.2932
65.0	-1.7758	-1.5345
76.0	-0.5291	-4.1889
79.0	-7.3259	-7.5673
80.5	-14.5041	-34.5369
81.0	-9.1041	-20.3981
82.0	-6.6073	-14.3008
84.0	-1.1492	
87.0	-1.9330	
89.0	-2.4766	-4.9226
93.0	-1.5325	
96.0	-0.5027	0.8275
100.0	1.2567	-1.1035
96.0	0.6559	0.4843
84.0	1.2567	-1.1035
73.0	0.6559	0.4843
50.0	0.7846	0.6988
35.0	0.8383	0.5970
25.0	0.5166	-1.6552
10.0	0.5166	0.3557
5.0	0.4362	0.3960
2.5	0.5970	0.4362

********** WING ******

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8612	-0.5793	-1.2558	-1.1995	-4.3567
2.5	-0.5793	0.3791	-1.1431	-1.0868	-3.2855
5.0	0.4918	0.3791	-1.3122	-0.9176	-2.2707
10.0	-0.6357	-0.0156	-1.4250	-0.8612	-1.3122
15.0	0.1535	-0.1847	-1.4814	-0.8049	-1.0304
24.0	-0.1283	-0.3539	-1.4250	-0.7485	-0.8049
33.0	-0.3047	-0.4677	-1.2827	-0.7937	-0.7937
54.0	-0.4677	-0.6307	-1.1197	-1.2827	-0.7937
65.0	-0.8889	-1.5754	-1.1035	-1.2322	-0.8889
78.5	-1.1464	-4.0644	-1.6183	-1.6183	-1.3610
79.5	-51.3457	-76.2274	-1.7002	-1.3815	-1.2364
80.5	-49.7207	-5.0976	-1.7298	-1.4105	-0.4524
81.3	-50.9984	220.6077	-1.3815	-1.3815	-1.2654
82.0	-51.0558	-6.3750	-1.5556	-1.2654	-1.2072
84.0	-49.5176	-3.7912	-1.2654	-1.3523	-1.3523
87.0	-21.7905	-2.9202	-1.4976	-1.2364	-0.3363
89.0	0.1958	-2.5635	-1.8300	-1.5856	-1.4808
93.0	4.2476	-1.5156	-1.5856	-0.6774	-0.5726
96.0	0.3705	-1.2712	-1.4458	-1.2712	-1.4458
100.0	-1.6554	-1.4110	-1.1664	-1.2013	-0.6075
96.0	0.1958	0.0561	0.0910	0.4054	0.3355
84.0	0.1958	0.1260	-0.6075	0.7547	0.6848
73.5	-0.1417	-0.4677	0.3473	0.6733	0.6733
54.0	-0.4677	-0.4677	-0.3047	0.0212	0.1843
33.0	-0.6307	-0.4677	-0.6307	-0.3047	0.1843
24.0	-0.5230	-0.6922	-0.6922	-0.2411	0.0408
10.0	-0.5230	-0.6922	-0.9176	-0.4102	0.5103
5.0	-0.6357	-0.6357	-0.8612	-0.4666	0.5103
2.5	-0.5230	-0.6922	-0.9176	-0.4102	0.5103

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.6314
43.4	-0.5456
50.4	-0.2882
56.1	-0.1165
62.5	-0.0307
69.7	-0.0736
76.9	-0.1594
83.4	-0.1594
89.4	-0.2452
95.4	-0.3311
101.1	-0.2882
---BOTTOM---	
38.2	0.2697
43.4	0.3126
50.4	0.0552
56.1	0.0122
62.5	-0.3311
69.7	-0.5027
76.9	-0.5027
83.4	-0.2882
89.4	0.0122
95.4	-0.0307
101.1	-0.7172

MISC.*

NO.	CP
1	-0.7937
2	-0.7937
3	-0.7937
4	0.0212
5	-0.7937
6	0.0212
7	-0.6075
8	-0.6075
9	-0.6075
10	-0.6075
11	-0.6425
12	-0.6425

TABLE A47. RUN 292, BC1W6V, DELF=45, CMU=2.0, (BN/B)W=0.25

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
292	1	CLAERO = -0.0390	CDAERO = 0.7616	DCLC = 0.8912	DWLC = 1.7183					BASEPR = 8.1904		
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.08		2134.46	2131.07	3.39	53.58	-0.05	2.67	1.089	1.986	3.075	87.2648	0.338019E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.05		2.57	-0.86	-0.47	-0.01	-0.00	-0.03	0.78	0.01	0.78	0.38	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.2061	-1.6037
2.5	-0.8797	-2.2550
5.0	-0.8073	-1.3864
10.0	1.0024	-1.1693
15.0	-0.7350	-1.0969
25.0	-0.7350	-0.9521
35.0	-0.8073	-1.0969
50.0	-1.0969	
56.0	-1.2416	-1.6037
65.0	-1.7484	-2.3276
76.0	-0.5177	-5.6573
79.0	-8.3357	-9.9284
80.5	-20.5266	-54.1780
81.0	-12.3226	-30.2454
82.0	-8.5081	-19.6383
84.0	-0.8265	
87.0	-1.4013	
89.0	-2.7050	-4.8677
93.0	-1.3919	
96.0	0.1528	0.6935
100.0	2.0838	-1.9326
96.0	0.4618	0.4618
84.0	2.0838	-1.9326
73.0	0.4618	0.4618
50.0	0.7707	0.8163
35.0	0.6404	0.4957
25.0	0.2061	-1.6037
10.0	0.1337	0.1337
5.0	0.0613	0.0613
2.5	0.1337	0.2785

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.4330	-1.2300	-0.6211	1.1039	0.2921
2.5	-0.4182	1.3069	0.1950	-0.4182	-0.9256
5.0	0.7995	0.9010	0.1906	-0.3167	-0.9256
10.0	-0.4182	0.3936	-0.4182	-0.3167	-0.7226
15.0	0.3936	0.1906	-0.3167	-0.4182	-0.7226
24.0	0.0892	-0.1138	-0.4182	-0.3167	-0.6211
33.0	0.1224	-0.1710	-0.4644	-0.4644	-0.7578
54.0	-0.4644	-0.4644	-0.7578	-2.5181	-0.7578
65.0	-0.8513	-2.2416	-0.7740	-0.7740	-0.9284
78.5	-3.3227	-4.1724	-1.3919	-1.3919	-2.8595
79.5	-94.9906	-112.1830	-1.0877	-1.2968	-3.5438
80.5	-92.0126	-3.2304	-1.1923	-1.4536	-0.3040
81.3	-93.5282	424.2244	-1.1400	-1.2968	-1.7671
82.0	-93.1095	-7.0970	-1.1400	-1.1923	-3.0736
84.0	-88.7192	-3.3870	-1.0877	-1.2445	-2.6032
87.0	-22.1987	-2.4464	-1.2968	-1.1923	-0.4085
89.0	8.5466	-2.3925	-1.5124	-1.7009	-2.[illegible]552
93.0	5.8435	-1.8267	-1.5124	-0.5065	-0.5065
96.0	0.3109	-1.5124	-1.4495	-1.6381	-1.8896
100.0	-3.7128	-1.6381	-1.3237	-1.7009	-0.4436
96.0	-0.2550	0.1851	0.2481	0.2481	0.3109
84.0	0.2481	0.5623	-0.5065	0.6881	0.5623
73.5	0.1224	-0.4644	0.4157	0.4157	0.4157
54.0	-0.4644	-0.4644	-0.1710	0.1224	0.1224
33.0	-1.3445	-0.4644	-0.7578	-0.4644	-0.1710
24.0	-1.3315	-1.5344	-1.0271	-0.4182	-0.1138
10.0	-1.1285	-1.2300	-1.4330	-0.4182	-0.1710
5.0	-1.1285	-1.3315	-1.7374	-0.3167	0.1224
2.5	-1.1285	-1.3315	-1.7374	-0.3167	0.1224

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.5423
43.4	-0.5423
50.4	-0.2334
56.1	-0.0789
62.5	-0.0017
69.7	-0.0017
76.9	-0.0789
83.4	-0.1561
89.4	-0.2334
95.4	-0.3878
101.1	-0.2334
---BOTTOM---	
38.2	0.1528
43.4	0.1528
50.4	-0.1561
56.1	-0.4651
62.5	-1.0829
69.7	-1.1602
76.9	-0.7740
83.4	-0.3106
89.4	-0.0789
95.4	-0.3106
101.1	-0.8513

MISC.*

NO.	CP
1	-0.1710
2	-1.3445
3	-0.4644
4	-0.1710
5	-0.4644
6	-0.1710
7	-0.5065
8	-0.5065
9	-0.3807
10	-0.5065
11	-0.4436
12	-0.5065

TABLE A47. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ	CLAERO = -0.0995	CDAERO = 0.7578	DCLC = 0.9651	DWLC = 1.8406	BASEPR = 8.1904
292 3					

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
4.16	2134.46	2131.07	3.39	53.57	-0.05	2.67	1.124	2.045	3.169	87.4664	0.338192E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	2.71	-0.71	-0.32	-0.03	0.01	-0.02	0.86	0.16	0.83	0.40

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-1.8932	-3.7031
2.5	-2.1828	-3.1962
5.0	-1.0969	-2.8342
10.0	1.1472	-2.8342
15.0	-1.0969	-1.8208
25.0	-0.9521	-1.1693
35.0	-0.8797	-1.0989
50.0	-1.1693	
56.0	-1.2416	-1.6037
65.0	-1.8208	-2.3276
76.0	-0.4454	-5.4403
79.0	-8.4082	-10.0007
80.5	-20.7357	-53.3938
81.0	-12.2182	-30.2454
82.0	-8.5603	-19.8473
84.0	-0.8265	
87.0	-1.5058	
89.0	-2.7822	-5.1767
93.0	-1.4692	
96.0	0.0756	0.6934
100.0	2.1610	-2.0097
96.0	0.4618	0.3845
84.0	2.1610	-2.0097
73.0	0.4618	0.3845
50.0	0.6934	0.5390
35.0	0.7129	0.4957
25.0	0.2785	-1.6760
10.0	0.3508	0.2061
5.0	0.2061	0.2061
2.5	0.4233	0.4957

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.3315	-0.9256	-1.4330	-0.0123	-1.7374
2.5	-0.5197	0.7995	-0.1138	-0.7226	-1.9403
5.0	0.6980	0.7995	-0.1138	-0.8241	-1.3315
10.0	-0.4182	0.4950	-0.4182	-0.6211	-1.1285
15.0	0.3936	0.2921	-0.5197	-0.6211	-0.9256
24.0	0.0892	-0.0123	-0.5197	-0.5197	-0.7226
33.0	-0.1710	-0.1710	-0.7578	-0.7578	-0.7578
54.0	-0.4644	-0.7578	-0.7578	-2.2247	-0.7578
65.0	-0.8513	-2.3960	-0.9285	-0.9285	-0.9285
78.5	-3.4002	-4.4042	-1.3147	-1.3920	-1.6236
79.5	-93.5783	-112.8612	-1.1400	-1.2968	-1.6626
80.5	-91.4377	-3.7529	-1.2445	-1.1400	-0.6175
81.3	-92.3252	400.4893	-1.2968	-1.4013	-1.6626
82.0	-92.2721	-7.2017	-1.1923	-1.2445	-1.5058
84.0	-88.7723	-3.7007	-1.1923	-1.4013	-1.6626
87.0	-24.6548	-3.0214	-1.2968	-1.2968	-0.4607
89.0	8.8612	-2.2040	-1.3237	-1.5752	-1.7639
93.0	6.5977	-1.7010	-1.2609	-0.4436	-0.3807
96.0	0.4995	-1.3866	-1.1980	-1.3866	-1.8896
100.0	-2.6440	-1.6381	-1.1980	-1.1980	-0.3807
96.0	-0.3179	-0.2550	0.0594	0.3737	0.1851
84.0	-0.0664	-0.1292	-0.3807	0.8139	0.6881
73.5	-0.4644	-0.4644	0.1224	0.4157	0.4157
54.0	-1.0512	-1.3445	-1.0512	-0.1710	0.1224
33.0	-1.3445	-0.4644	-1.3445	-0.7578	-0.1710
24.0	-1.0271	-1.2300	-1.1285	-0.7226	-0.3167
10.0	-1.0271	-1.0271	-1.2300	-0.6211	-0.1710
5.0	-0.9256	-1.0271	-1.2300	-0.7226	0.1224
2.5	-1.0271	-0.9256	-1.0271	-0.5197	0.1224

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.6196
43.4	-0.5423
50.4	-0.3106
56.1	-0.1561
62.5	-0.1561
69.7	-0.0789
76.9	-0.1561
83.4	-0.2334
89.4	-0.3106
95.4	-0.3878
101.1	-0.3878
---BOTTOM---	
38.2	0.1528
43.4	0.0756
50.4	-0.0789
56.1	-0.4651
62.5	-1.0058
69.7	-1.1602
76.9	-0.9285
83.4	-0.7740
89.4	-0.4651
95.4	-0.4651
101.1	-0.9285

MISC.*

NO.	CP
1	-0.4644
2	-1.3445
3	-0.7578
4	-0.4644
5	-0.7578
6	-0.1710
7	-0.3807
8	-0.4436
9	-0.3807
10	-0.4436
11	-0.5065
12	-0.4436

TABLE A47. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 292 5 CLAERO = 0.1906 CDAERO = 0.8547 DCLC = 1.0847 DWLC = 2.0372 BASEPR = 8.1904

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.13	2134.60	2131.32	3.28	52.67	-0.04	2.74	1.177	2.140	3.318	89.4884	0.332565E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.04	3.31	-0.26	-0.17	-0.02	0.00	-0.04	1.32	0.33	1.22	0.61

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-6.7051	-3.4108
2.5	-6.5554	-3.4108
5.0	-6.2559	-3.6356
10.0	1.2309	-3.7102
15.0	-1.3894	-4.1595
25.0	-1.3146	-4.7586
35.0	-1.1649	-4.1595
50.0	-1.4643	
56.0	-1.5392	-1.8387
65.0	-2.0633	-1.8387
76.0	-0.4911	-4.9081
79.0	-9.1758	-9.3255
80.5	-22.2867	-55.6857
81.0	-13.3154	-31.7446
82.0	-9.2081	-21.3140
84.0	-1.2635	
87.0	-1.9121	
89.0	-3.0677	-6.0232
93.0	-1.7096	
96.0	-0.1120	0.8466
100.0	2.2045	-2.0292
96.0	0.6068	0.3673
84.0	2.2045	-2.0292
73.0	0.6068	0.3673
50.0	0.8456	0.6068
35.0	0.8585	0.5571
25.0	0.4074	-1.9884
10.0	0.4074	0.4074
5.0	0.3325	0.3325
2.5	0.5571	0.2576

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-1.4075	-0.7778	-1.5125	-0.9877	-4.3460
2.5	-0.5680	0.4815	-1.5125	-0.8827	-3.1916
5.0	0.5865	0.4815	-1.5125	-0.8827	-1.8274
10.0	-0.5680	0.1667	-1.5125	-0.7778	-1.1976
15.0	0.2717	0.1667	-1.4075	-0.8827	-0.9877
24.0	-0.0432	-0.2531	-1.1976	-0.7778	-0.7778
33.0	-0.2073	-0.2073	-1.4210	-1.1176	-0.5108
54.0	-0.5108	-0.5108	-1.1176	-1.7244	-0.8142
65.0	-0.9109	-2.1889	-1.0707	-1.2304	-0.9908
78.5	-2.1889	-4.6653	-1.3902	-1.7096	-1.7096
79.5	-97.7861	-123.4592	-1.8581	-1.4257	-1.4797
80.5	-95.1392	-7.1543	-1.4797	-1.5338	-0.5069
81.3	-96.9231	404.7815	-1.6959	-1.3716	-1.3716
82.0	-96.0022	-8.6138	-1.4257	-1.4257	-1.4257
84.0	-92.8155	-4.8846	-1.3716	-1.5338	-1.5878
87.0	-23.8002	-3.5334	-1.6419	-1.4257	-0.3988
89.0	8.6790	-2.5698	-1.5946	-1.5946	-1.5946
93.0	6.4032	-1.8547	-1.5296	-0.6193	-0.4892
96.0	0.4212	-1.7246	-1.3995	-1.3345	-1.6596
100.0	-2.5048	-1.9197	-1.1394	-1.2044	-0.4892
96.0	-0.2941	-0.1641	0.2911	0.4862	0.3561
84.0	0.0310	0.0310	-0.5543	0.8113	0.8113
73.5	0.0961	-0.5108	0.0961	0.7029	0.7029
54.0	-0.8142	-0.8142	-0.5108	0.0961	0.0961
33.0	-1.4210	-0.5108	-1.1176	-0.5108	0.0961
24.0	-1.0927	-1.1976	-1.1976	-0.5680	-0.0432
10.0	-1.0927	-0.8827	-1.1976	-0.7778	0.3995
5.0	-0.8827	-1.0927	-1.1976	-0.9877	0.3995
2.5	-0.8827	-0.9877	-1.1976	-0.8827	0.7029

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.6712
43.4	-0.6712
50.4	-0.3517
56.1	-0.1120
62.5	-0.1120
69.7	-0.0322
76.9	-0.1120
83.4	-0.1120
89.4	-0.1919
95.4	-0.2718
101.1	-0.2718
---BOTTOM---	
38.2	0.2873
43.4	0.2873
50.4	-0.0322
56.1	-0.2718
62.5	-0.8310
69.7	-0.9908
76.9	-0.9109
83.4	-0.5913
89.4	-0.1919
95.4	-0.3517
101.1	-1.0707

MISC.*

NO.	CP
1	-0.5108
2	-1.4210
3	-1.1176
4	0.0961
5	-0.8142
6	0.0961
7	-0.4892
8	-0.5543
9	-0.5543
10	-0.4892
11	-0.4892
12	-0.5543

TABLE A48. RUN 296, BC1W6V, DELF=45, CMU=0.5, (BN/B)W=0.25, (BN/B)C=0.5

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 731

296 2 CLAERO = 0.4977 CDAERO = 0.4903 DCLC = 0.2721 DWLC = 0.3917 BASEPR = 8.1905

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.08	2142.17	2128.72	13.45	106.50	-0.05	2.65	0.333	0.453	0.785	86.7869	0.676155E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	1.16	0.07	-0.30	0.01	-0.01	0.00	0.69	-0.19	0.69	0.41

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.5295	0.1281
2.5	-0.4010	-0.8206
5.0	-0.3644	-0.5834
10.0	-0.1273	-0.5834
15.0	-0.4557	-0.6017
25.0	-0.4557	-0.5104
35.0	-0.4374	-0.4557
50.0	-0.4922	
56.0	-0.6381	-0.5286
65.0	-1.0212	-0.6199
76.0	-0.3462	10.8923
79.0	-3.7943	-0.5651
80.5	-27.2494	2.4001
81.0	-26.1957	0.7404
82.0	-25.6685	-1.6700
84.0	-14.1040	
87.0	-18.3715	
89.0	-5.6002	-2.0741
93.0	3.1857	
96.0	2.1727	0.5948
100.0	0.3220	-1.3339
96.0	0.4974	0.2051
84.0	0.3220	-1.3339
73.0	0.4974	0.2051
50.0	0.7896	0.5363
35.0	0.6207	0.5295
25.0	0.1829	-0.9118
10.0	0.1464	0.0917
5.0	-0.0178	0.0552
2.5	0.0187	0.2194

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.5828	-0.3525	0.1338	0.2361	-0.0966
2.5	-0.2757	0.5688	-0.1989	-0.5316	-0.7876
5.0	0.5432	0.4152	-0.2246	-0.3781	-0.7103
10.0	-0.3269	0.1849	-0.3525	-0.3269	-0.5316
15.0	0.1849	0.0570	-0.3781	-0.2501	-0.4549
24.0	-0.0198	-0.0710	-0.4293	-0.2757	-0.4037
33.0	-0.1801	-0.1801	-0.5501	-0.4022	-0.4761
54.0	-0.4022	-0.4022	-0.6981	-0.3282	-0.4761
65.0	-0.6715	-1.0222	-0.8079	-0.6715	-0.6520
78.5	0.0103	-2.4638	-1.0416	-1.1391	-1.1780
79.5	-25.6157	-24.0750	-0.9982	-1.0509	-1.1695
80.5	-24.0750	-1.9598	-0.9719	-1.0773	-0.3660
81.3	-26.6567	117.7275	-0.9851	-1.0773	-1.2485
82.0	-23.8772	-3.3296	-1.0245	-1.0509	-1.2221
84.0	-21.9408	-1.7885	-0.8929	-1.0509	-1.2353
87.0	-6.8992	-1.4461	-1.0245	-1.0641	-0.3528
89.0	0.6233	-1.5015	-1.0892	-1.1368	-1.2954
93.0	0.8770	-1.1843	-1.1527	-0.3915	-0.3757
96.0	-0.3123	-1.1051	-1.0099	-1.2002	-1.2478
100.0	-0.7563	-1.1209	-0.8038	-1.0575	-0.3281
96.0	0.3062	0.3696	0.3537	0.3379	0.1952
84.0	0.4489	0.5599	-0.3439	0.6392	0.5440
73.5	0.3379	-0.3282	0.5598	0.5598	0.4859
54.0	0.2639	0.1898	0.1898	0.1898	0.1158
33.0	-0.4022	-0.3282	0.0419	0.0419	0.0419
24.0	-0.4805	-0.4293	-0.0454	0.0825	0.0570
10.0	-0.4805	-0.6852	-0.1733	0.0825	0.1153
5.0	-0.5316	-0.6084	-0.1989	0.1081	0.1158
2.5	-0.5316	-0.5573	-0.1989	0.2105	0.1898

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.2819
43.4	-0.3014
50.4	-0.1650
56.1	-0.0676
62.5	-0.0481
69.7	-0.0676
76.9	-0.0871
83.4	-0.1260
89.4	-0.1845
95.4	-0.3014
101.1	-0.2624
---BOTTOM---	
38.2	0.0883
43.4	0.1272
50.4	-0.1066
56.1	-0.5352
62.5	-0.9442
69.7	-0.6130
76.9	-0.2040
83.4	0.0688
89.4	0.1272
95.4	-0.0286
101.1	-0.5352

MISC.*

NO.	CP
1	-0.3282
2	-0.0321
3	-0.4761
4	0.1158
5	-0.4022
6	-0.0321
7	-0.3757
8	-0.3598
9	-0.3439
10	-0.3598
11	-0.3757
12	-0.3439

TABLE A48. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ 296 4

CLAERO = 0.7493 CDAERO = 0.5916 DCLC = 0.2851 DWLC = 0.4074 BASEPR = 8.1905

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
4.02	2142.03	2128.69	13.34	106.12	-0.05	2.66	0.333	0.453	0.786	87.0501	0.672180E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	1.44	0.22	-0.30	0.00	-0.01	-0.00	0.98	-0.20	0.94	0.53

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.5114	-2.3512
2.5	-0.9713	-1.7257
5.0	-0.8057	-1.2472
10.0	-0.1056	-1.0081
15.0	-0.6953.	-0.8977
25.0	-0.5849	-0.7137
35.0	-0.5481	-0.6217
50.0	-0.6033	
56.0	-0.7689	-0.6585
65.0	-1.1001	-0.7873
76.0	-0.3825	9.4608
79.0	-3.9335	-0.6401
80.5	-27.7473	1.8209
81.0	-26.7776	0.7716
82.0	-26.2729	-1.8452
84.0	-15.1016	
87.0	-18.6217	
89.0	-4.7000	-1.9692
93.0	2.8244	
96.0	1.8814	0.7223
100.0	0.6241	-1.7924
96.0	0.5652	0.1722
84.0	0.6241	-1.7924
73.0	0.5652	0.1722
50.0	0.8206	0.6044
35.0	0.6294	0.5374
25.0	0.2614	-0.9897
10.0	0.2062	0.1510
5.0	0.0958	0.1327
2.5	0.3902	0.4638

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.6286	-0.4995	0.0682	-1.3254	-3.2353
2.5	-0.3447	0.3780	-0.6028	-1.1447	-1.6867
5.0	0.4812	0.2231	-0.4995	-0.8608	-1.1963
10.0	-0.3447	-0.0607	-0.4995	-0.7318	-0.9383
15.0	0.0682	-0.1640	-0.4995	-0.6286	-0.8092
24.0	-0.1124	-0.2931	-0.5253	-0.4995	-0.6544
33.0	-0.2713	-0.4206	-0.4952	-0.5698	-0.6444
54.0	-0.4952	-0.4952	-0.7190	-0.1966	-0.6444
65.0	-0.6725	-1.0655	-0.8100	-0.6922	-0.7315
78.5	0.3098	-2.6372	-1.1441	-1.1244	-1.3405
79.5	-25.7953	-28.5444	-0.9951	-1.0748	-1.3272
80.5	-24.5594	-2.4961	-0.9287	-1.0615	-0.3841
81.3	-24.4267	118.6727	-0.9951	-1.0581	-1.3537
82.0	-24.1215	-3.3728	-0.9154	-1.0482	-1.2607
84.0	-21.9962	-2.4430	-0.9420	-0.9819	-1.3139
87.0	-8.9516	-1.7555	-0.9686	-1.1412	-0.3708
89.0	-0.5697	-1.6731	-1.1454	-1.1614	-1.4013
93.0	0.6055	-1.5452	-1.1454	-0.4418	-0.4258
96.0	-0.4258	-1.3533	-1.0655	-1.2254	-1.4013
100.0	-0.8895	-1.2893	-0.9216	-1.1135	-0.4258
96.0	0.2778	0.3577	0.3577	0.3257	0.1658
84.0	0.5336	0.7735	-0.4418	0.7095	0.5496
73.5	0.3257	-0.3459	0.6242	0.6242	0.4750
54.0	0.2511	0.2511	0.2511	0.2511	0.1018
33.0	-0.1220	-0.3459	0.1018	0.1018	0.1018
24.0	-0.5511	-0.3189	0.0166	0.1973	0.0940
10.0	-0.6544	-0.8608	0.0682	0.2747	0.2511
5.0	-0.5511	-0.8867	0.1199	0.4038	0.3257
2.5	-0.5511	-0.8608	0.1457	0.4812	0.4750

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3386
43.4	-0.3386
50.4	-0.1814
56.1	-0.0635
62.5	-0.0242
69.7	-0.0632
76.9	-0.1225
83.4	-0.1421
89.4	-0.2011
95.4	-0.2993
101.1	-0.2207
---BOTTOM---	
38.2	0.1330
43.4	0.1919
50.4	-0.0242
56.1	-0.4565
62.5	-0.8690
69.7	-0.5547
76.9	-0.0635
83.4	0.1526
89.4	0.2312
95.4	0.0151
101.1	-0.6137

MISC.*

NO.	CP
1	-0.5698
2	0.1018
3	-0.5698
4	0.1018
5	-0.5698
6	0.0272
7	-0.4418
8	-0.4098
9	-0.4258
10	-0.4098
11	-0.4258
12	-0.4258

TABLE 48. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 735

296 6	CLAERO =	1.3521	CDAERO =	0.9399	DCLC =	0.3161	DWLC =	0.4451	BASEPR =	8.1905	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
12.08	2141.96	2128.73	13.22	105.76	-0.05	2.74	0.343	0.468	0.811	89.5966	0.867901E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.05	2.11	0.66	-0.37	0.02	0.01	0.02	1.72	-0.26	1.57	0.90	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-7.6076	-2.5196
2.5	-3.5957	-2.4824
5.0	-1.8516	-2.4024
10.0	-0.1446	-2.5567
15.0	-1.2208	-2.5937
25.0	-0.9424	-2.9463
35.0	-0.8867	-2.7607
50.0	-0.8125	
56.0	-0.8497	-0.9053
65.0	-1.0352	-0.5528
76.0	-0.6270	1.3955
79.0	-2.3711	-1.1651
80.5	-3.7363	1.7024
81.0	-3.3478	0.6843
82.0	-3.0263	-4.9553
84.0	-1.6868	
87.0	-3.0665	
89.0	-2.9765	-1.6689
93.0	-2.9170	
96.0	-3.3341	0.7483
100.0	0.7681	-0.4802
96.0	0.0945	0.3718
84.0	0.7681	-0.4802
73.0	0.0945	0.3718
50.0	0.7681	0.6492
35.0	0.7089	0.5791
25.0	0.3378	-1.0538
10.0	0.3935	0.3378
5.0	0.3378	0.3750
2.5	0.6162	0.5420

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	1.8907	0.6935	-2.8202	-5.0325	-2.2216
2.5	-0.5819	0.0688	-1.5188	-5.1626	-2.4818
5.0	0.3031	-0.2435	-1.2065	-3.8092	-2.4038
10.0	-0.5558	-0.5819	-1.1284	-1.1544	-2.5078
15.0	-0.5038	-0.6339	-1.0764	-1.0243	-2.5599
24.0	-0.4517	-0.7119	-1.0243	-0.8681	-2.5078
33.0	-0.4994	-0.6499	-0.9509	-0.8757	-2.2303
54.0	-0.6499	-0.8003	-1.0262	-0.1230	-1.1767
65.0	-0.9754	-1.1934	-1.2132	-1.0349	-1.0547
78.5	0.2926	-4.7002	-1.8472	-1.6689	-1.5897
79.5	-26.6566	-39.9720	-1.7939	-1.6198	-1.7537
80.5	-25.7324	-3.6023	-1.7805	-1.7403	-0.6552
81.3	-26.0002	119.6249	-1.7135	-1.7135	-1.7135
82.0	-25.7453	-5.2232	-1.7571	-1.6331	-1.6599
84.0	-24.9550	-3.9105	-1.6198	-1.6465	-1.7269
87.0	-16.3821	-2.9058	-1.8475	-1.7001	-0.6285
89.0	-6.9122	-2.8001	-1.9614	-1.7518	-1.7840
93.0	0.6833	-2.3646	-1.8808	-0.6713	-0.6390
96.0	-0.8326	-2.1388	-1.8163	-1.7840	-1.8002
100.0	-1.1390	-1.5260	-1.6389	-1.7195	-0.6068
96.0	1.1348	0.7478	0.3285	0.2479	0.0383
84.0	1.7798	1.5219	-0.6390	0.6994	0.3608
73.5	1.1563	-0.4994	0.7048	0.5543	0.3285
54.0	0.3285	0.3285	0.3285	0.3285	0.1779
33.0	0.1027	-0.5746	0.3285	0.3285	0.1027
24.0	-0.0613	0.0688	0.3291	0.4332	0.2510
10.0	-0.3218	-0.0092	0.3812	0.5113	0.4037
5.0	-0.6859	-0.2955	0.4072	0.5373	0.4790
2.5	-1.2063	-0.7380	0.3812	0.3291	0.2532

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.4802
43.4	-0.4405
50.4	-0.2820
56.1	-0.1829
62.5	-0.1631
69.7	-0.2226
76.9	-0.2622
83.4	-0.2424
89.4	-0.3415
95.4	-0.4207
101.1	-0.3415
---BOTTOM---	
38.2	0.1737
43.4	0.1935
50.4	-0.0442
56.1	0.0548
62.5	-0.0641
69.7	-0.0442
76.9	0.0548
83.4	0.1539
89.4	0.2529
95.4	-0.0442
101.1	-0.9556

MISC.*

NO.	CP
1	-0.8757
2	0.3285
3	-0.8757
4	0.3285
5	-2.2303
6	0.1027
7	-0.6390
8	-0.6390
9	-0.6229
10	-0.6552
11	-0.6390
12	-0.6068

TABLE A49. RUN 297, BC1W6V, DELF=45, CMU=1.0, (BN/B)W=0.25,(BN/B)C=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
297	1	CLAERO =	0.1632	CDAERO =	0.6370	DCLC =	0.6665	DWLC =	0.9592	BASEPR =	8.1905	
ALPHA		PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN
-0.05		2142.10	2136.56	5.54	68.42	-0.05	2.68	0.814	1.108	1.922	87.7528	0.431959E+06
BETA		CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR	
0.05		1.79	-0.39	-0.44	0.01	-0.01	-0.04	0.63	-0.18	0.64	0.45	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	0.4614	-0.0258
2.5	-0.4245	-0.9560
5.0	-0.4688	-0.7345
10.0	-0.1587	-0.7345
15.0	-0.4688	-0.7345
25.0	-0.6017	-0.6460
35.0	-0.6017	-0.6017
50.0	-0.5574	
56.0	-0.7788	-0.6460
65.0	-1.2219	-0.5574
76.0	-0.3359	5.3337
79.0	-5.2967	-3.9680
80.5	-53.6260	2.1122
81.0	-52.2829	0.8650
82.0	-51.7719	-2.5887
84.0	-26.1252	
87.0	-35.0147	
89.0	-14.7570	-3.3110
93.0	0.2362	
96.0	1.1821	0.6145
100.0	1.0875	-1.1354
96.0	0.3780	0.1416
84.0	1.0875	-1.1354
73.0	0.3780	0.1416
50.0	0.7564	0.5200
35.0	0.5500	0.5057
25.0	0.1513	-1.0888
10.0	0.1513	0.0627
5.0	-0.0702	0.0627
2.5	-0.0258	0.2399

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.8089	-0.4983	-0.0012	0.3716	0.3095
2.5	-0.2497	0.6823	-0.1876	-0.5604	-0.6847
5.0	0.6823	0.6201	-0.1876	-0.3740	-0.5604
10.0	-0.2497	0.4338	-0.2497	-0.4361	-0.4361
15.0	0.3095	0.1852	-0.3740	-0.1254	-0.4361
24.0	0.1852	-0.0633	-0.4361	-0.1254	-0.4361
33.0	-0.0850	-0.0850	-0.4444	-0.4444	-0.4444
54.0	-0.4444	-0.4444	-0.6240	-0.6240	-0.4444
65.0	-0.8044	-1.6084	-0.8044	-0.7570	-0.7098
78.5	-0.0949	-3.1691	-1.0881	-1.1354	-1.3246
79.5	-62.2610	-62.7088	-0.8298	-1.0217	-1.2455
80.5	-58.3272	-2.8126	-0.8939	-0.9258	-0.1263
81.3	-58.1033	284.7031	-0.9578	-1.0217	-1.2775
82.0	-57.6231	-5.1149	-0.9258	-0.8613	-1.3416
84.0	-50.6851	-2.7806	-0.8298	-0.8619	-1.1497
87.0	-6.6820	-2.0771	-0.8619	-0.8619	-0.0944
89.0	4.5731	-2.0868	-1.1629	-1.2399	-1.4324
93.0	1.6858	-1.5479	-1.2015	-0.4315	-0.3545
96.0	-0.4315	-1.3555	-1.0476	-1.2015	-1.6248
100.0	-1.8559	-1.3555	-0.8164	-1.0859	-0.3930
96.0	0.1845	0.3385	0.3385	0.2999	0.1845
84.0	0.3385	0.5695	-0.3930	0.6849	0.5695
73.5	0.4540	-0.2647	0.6336	0.6336	0.6336
54.0	0.2743	0.2743	0.2743	0.4540	0.2743
33.0	-0.4444	-0.2647	0.0946	0.0946	0.0946
24.0	-0.5604	-0.5604	-0.1876	0.0609	0.0609
10.0	-0.4983	-0.6225	-0.3740	0.1852	0.0946
5.0	-0.6225	-0.5604	-0.3740	0.1852	0.2743
2.5	-0.5604	-0.6225	-0.4361	0.2473	0.2743

FUSELAGE

-----TOP----

XLOC	CP
38.2	-0.3314
43.4	-0.3787
50.4	-0.2368
56.1	-0.0949
62.5	-0.0476
69.7	-0.0949
76.9	-0.1422
83.4	-0.1422
89.4	-0.2841
95.4	-0.2841
101.1	-0.2841

---BOTTOM---

38.2	0.0470
43.4	-0.0003
50.4	-0.2841
56.1	-1.3719
62.5	-1.6084
69.7	-0.9935
76.9	-0.3314
83.4	-0.0476
89.4	0.0470
95.4	-0.1895
101.1	-0.8044

MISC.*

NO.	CP
1	-0.2647
2	-0.2647
3	-0.4444
4	0.2743
5	-0.4444
6	0.0946
7	-0.4315
8	-0.3545
9	-0.3930
10	-0.3930
11	-0.3545
12	-0.3545

TABLE A49. (Continued)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
297 3	CLAERO =	0.4234	CDAERO =	0.6985	DCLC =	0.6448	DWLC =	0.9196	BASEPR =	8.1906		
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN	
3.99	2142.03	2135.92	6.10	71.84	-0.05	2.68	0.752	1.023	1.776	87.6917	0.453227E+06	
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.05	1.99	-0.14	-0.41	0.01	-0.01	-0.04	0.92	-0.18	0.88	0.54		

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	-0.7692	-3.0204
2.5	-1.1713	-2.0154
5.0	-0.7692	-1.4526
10.0	-0.0859	-1.1310
15.0	-0.6890	-1.0104
25.0	-0.6486	-0.8095
35.0	-0.6085	-0.6890
50.0	-0.6085	
56.0	-0.8497	-0.6486
65.0	-1.2919	-0.5683
76.0	-0.3271	1.6426
79.0	-5.1508	-5.4323
80.5	-52.0226	2.2474
81.0	-51.2099	0.9414
82.0	-46.8567	-2.3090
84.0	-26.3095	
87.0	-35.2486	
89.0	-15.7402	-3.9368
93.0	0.9565	
96.0	3.2314	0.7846
100.0	1.5145	-1.3184
96.0	0.5273	0.3556
84.0	1.5145	-1.3184
73.0	0.5273	0.3556
50.0	0.8277	0.6560
35.0	0.6778	0.5171
25.0	0.1954	-1.0908
10.0	0.2357	0.1553
5.0	0.1151	0.1151
2.5	0.3161	0.5572

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.8050	-0.8050	-0.1283	-0.8613	-2.1019
2.5	-0.4102	0.2664	-0.5794	-1.1996	-2.2710
5.0	0.5483	0.2100	-0.5794	-0.8050	-1.2560
10.0	-0.4102	-0.0155	-0.6922	-0.6358	-0.9177
15.0	0.0972	-0.1283	-0.8050	-0.5230	-0.7486
24.0	-0.0719	-0.2975	-0.9742	-0.4666	-0.6922
33.0	-0.1419	-0.3050	-0.7941	-0.4680	-0.4680
54.0	-0.3050	-0.3050	-0.7941	-0.1419	-0.4680
65.0	-0.7174	-1.4472	-0.9321	-0.6316	-0.6745
78.5	0.6990	-3.4216	-1.1896	-1.1467	-1.1467
79.5	-56.2595	-61.2812	-1.0610	-1.1192	-1.2932
80.5	-52.8057	-6.3719	-1.3224	-0.9740	-0.3065
81.3	-52.8647	246.5798	-0.9740	-1.0900	-1.1482
82.0	-52.7779	-6.1398	-1.0900	-1.0320	-1.1772
84.0	-47.6399	-3.0926	-1.0320	-1.0900	-1.2062
87.0	-8.9839	-2.2800	-1.0030	-0.9740	-0.2195
89.0	3.7246	-1.7606	-1.2365	-1.2365	-1.3065
93.0	1.6632	-1.3762	-1.2016	-0.4330	-0.3282
96.0	-0.4680	-1.2016	-1.0318	-1.1318	-1.3762
100.0	-1.5260	-1.3065	-0.6426	-0.8872	-0.3632
96.0	0.2658	0.4404	0.4055	0.4055	0.1609
84.0	0.4404	0.6500	-0.3632	0.6850	0.5802
73.5	0.5103	-0.3050	0.6733	0.6733	0.5103
54.0	0.3472	0.3472	0.3472	0.3472	0.1842
33.0	-0.3050	-0.3050	0.1842	0.1842	0.1842
24.0	-0.7486	-0.5794	-0.1847	0.2664	-0.0155
10.0	-0.6358	-0.7486	-0.1847	0.2664	0.3472
5.0	-0.6358	-0.8050	-0.0719	0.3792	0.5103
2.5	-0.6358	-0.6922	0.0972	0.4356	0.5103

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.3312
43.4	-0.3312
50.4	-0.1595
56.1	-0.0307
62.5	0.0122
69.7	-0.0307
76.9	-0.0736
83.4	-0.1166
89.4	-0.1595
95.4	-0.2024
101.1	-0.2024
---BOTTOM---	
38.2	0.1839
43.4	0.1839
50.4	-0.1166
56.1	-1.1896
62.5	-1.5330
69.7	-0.9321
76.9	-0.1595
83.4	0.1410
89.4	0.2268
95.4	-0.0736
101.1	-0.8462

MISC.*

NO.	CP
1	-0.4680
2	0.0212
3	-0.4680
4	0.1842
5	-0.4680
6	0.0212
7	-0.3981
8	-0.3632
9	-0.3981
10	-0.3981
11	-0.3981
12	-0.3981

TABLE A49. (Concluded)

RUN:SEQ PROPLULSIVE WING / CANARD PRESSURES PAGE 740

297 5 CLAERO = 0.9190 CDAERO = 0.9768 DCLC = 0.7383 DWLC = 1.0383 BASEPR = 8.1905

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
12.02	2141.89	2136.12	5.76	69.84	-0.05	2.73	0.802	1.092	1.894	89.2687	0.439947E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	2.70	0.33	-0.32	-0.00	-0.02	-0.02	1.58	-0.07	1.43	0.88

CANARD BP=3.375 %X/C	CP	CANARD BP=10.125 CP	WING BP = 2 %X/C	CP	WING BP = 6 CP	WING BP =12 CP	WING BP =16 CP	WING BP = 22 CP
0.0	-7.2933	-2.8670	0.0	-0.6656	-0.4865	-1.3819	-4.3074	-2.0984
2.5	-5.5484	-2.8670	2.5	-0.6059	-0.0089	-1.2029	-4.0685	-2.2179
5.0	-1.7604	-2.9520	5.0	0.3494	-0.1880	-0.9641	-3.2924	-2.1581
10.0	-0.2282	-2.8670	10.0	-0.7253	-0.4865	-0.9044	-0.7850	-2.2179
15.0	-1.4199	-3.0372	15.0	-0.0685	-0.4865	-0.9044	-0.7253	-2.3373
25.0	-1.2071	-3.2501	24.0	-0.3073	-0.6656	-0.9044	-0.6656	-2.2776
35.0	-1.1646	-3.2074	33.0	-0.3750	-0.7202	-0.8929	-0.7202	-1.9286
50.0	-1.0794		54.0	-0.7202	-0.7202	-1.0655	-0.7202	-0.8929
56.0	-1.2922	-1.3348	65.0	-0.8572	-1.6753	-1.0390	-0.9026	-0.9481
60.0	-1.5901	-0.5942	78.5	1.0061	-3.9930	-1.4026	-1.4026	-1.4480
76.0	-0.6538	2.1126	79.5	-59.9261	-73.8768	-1.3292	-1.2371	-1.4829
79.0	-5.8038	-3.2501	80.5	-56.4237	-8.4838	-1.3599	-1.2371	-0.7146
80.5	-56.0855	1.8059	81.3	-56.7308	245.2555	-1.4213	-1.2371	-1.5136
81.0	-54.4271	0.7603	82.0	-56.3006	-7.9971	-1.2678	-1.2985	-1.5136
82.0	-53.9344	-4.8935	84.0	-52.1216	-4.4326	-1.2062	-1.2371	-1.5443
84.0	-30.5815		87.0	-11.3771	-3.3880	-1.4213	-1.3906	-0.7455
87.0	-40.1988		89.0	3.1516	-2.9149	-1.6203	-1.3984	-1.1354
89.0	-13.3550	-2.1298	93.0	1.6719	-1.5464	-1.5464	-0.6585	-0.5845
93.0	3.6420		96.0	-0.8804	-1.0284	-1.2874	-1.4354	-1.5464
96.0	3.9147	0.7334	100.0	-1.7313	-0.7325	-1.0654	-1.3613	-0.6216
100.0	1.4605	-0.9481	96.0	0.3033	0.5622	0.4142	0.4142	0.2293
96.0	0.4607	0.1880	84.0	0.5992	0.8211	-0.6216	0.7471	0.5622
84.0	1.4605	-0.9481	73.5	0.4882	-0.7202	0.6608	0.4882	0.4882
73.0	0.4607	0.1880	54.0	0.3156	0.3156	0.3156	0.3156	0.1429
50.0	0.7334	0.5516	33.0	-0.0297	-0.7202	0.3156	0.3156	0.3156
35.0	0.6656	0.4528	24.0	-0.6656	-0.3073	0.1703	0.4091	0.1703
25.0	0.2400	-1.5051	10.0	-0.6059	-0.8447	0.2896	0.5285	0.4882
10.0	0.3677	0.2400	5.0	-0.5462	-0.7850	0.3494	0.6479	0.4882
5.0	0.2826	0.2826	2.5	-0.5462	-0.7253	0.3494	0.5285	0.4882
2.5	0.5379	0.4102						

FUSELAGE TOP XLOC	CP	MISC. NO.	CP
38.2	-0.5845	1	-1.0655
43.4	-0.5391	2	0.3156
50.4	-0.3573	3	-0.7202
56.1	-0.1755	4	0.3156
62.5	-0.1301	5	-1.9286
69.7	-0.1755	6	0.1429
76.9	-0.2209	7	-0.6216
83.4	-0.2209	8	-0.6585
89.4	-0.3119	9	-0.6216
95.4	-0.4028	10	-0.6216
101.1	-0.3573	11	-0.6216
BOTTOM		12	-0.5845
38.2	0.1880		
43.4	0.1880		
50.4	-0.3119		
56.1	-1.6298		
62.5	-1.8116		
69.7	-0.9481		
76.9	-0.1301		
83.4	0.0517		
89.4	0.1426		
95.4	-0.2664		
101.1	-1.4026		

TABLE A50. RUN 298, BC1W6V, DELF=45, CMU=2.0, (BN/B)W=0.25,(BN/B)C=0.5

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ												
298	1	CLAERO = -0.2258	CDAERO = 0.8233	DCLC = 1.2037	DWLC = 1.7253	BASEPR = 8.1906						
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HGT	RN	
0.10	2142.10	2138.93	3.16	51.77	-0.05	2.69	1.468	1.990	3.458	87.8940	0.325344E+06	
BETA	CL	CD	CK	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR		
0.05	2.70	-1.01	-0.66	0.02	-0.01	-0.09	0.65	-0.21	0.65	0.46		

********** CANARD ******

	BP=3.375	BP=10.125
%X/C	CP	CP
0.0	0.3135	-0.2293
2.5	-0.3068	-1.0822
5.0	-0.4619	-1.0046
10.0	-0.2293	-0.9271
15.0	-0.6170	-0.8496
25.0	-0.5395	-0.6945
35.0	-0.6170	-0.6170
50.0	-0.6170	
56.0	-0.9271	-0.6945
65.0	-1.3924	-0.6170
76.0	-0.3843	3.3376
78.0	-6.9754	-4.0289
80.5	-110.6728	3.3626
81.0	-107.9295	1.0114
82.0	-106.1944	-1.7877
84.0	-57.2106	
87.0	-59.6156	
89.0	-1.3898	-3.3769
93.0	15.1699	
96.0	8.9599	0.7630
100.0	3.2470	-1.3898
96.0	0.4318	0.1834
84.0	3.2470	-1.3898
73.0	0.4318	0.1834
50.0	0.9285	0.6802
35.0	0.6237	0.4686
25.0	0.0034	-1.2373
10.0	0.0809	0.0034
5.0	0.0034	-0.0742
2.5	-0.0742	0.3135

********** WING **********

	BP = 2	BP = 6	BP =12	BP =16	BP = 22
%X/C	CP	CP	CP	CP	CP
0.0	-0.7289	-0.5114	0.5763	0.5763	-0.1851
2.5	-0.2939	0.9026	-0.1851	-0.2939	-0.9454
5.0	0.6850	0.5763	-0.1851	-0.0763	-0.7289
10.0	-0.1851	0.3588	-0.2939	-0.1851	-0.6202
15.0	0.3588	0.1412	-0.2939	0.0325	-0.5114
24.0	0.1412	-0.0763	-0.1851	0.0325	-0.2939
33.0	0.0679	0.0679	-0.2467	-0.5612	-0.2467
54.0	-0.2467	-0.2467	-0.5612	-0.2467	-0.2467
65.0	-0.7274	-2.0521	-0.8102	-0.6446	-0.7274
78.5	0.5146	-4.1222	-1.1414	-0.9758	-1.4726
79.5	*********	-102.2755	-0.8920	-0.9480	-2.1794
80.5	*********	-6.4903	-1.0039	-0.9480	-0.2762
81.3	*********	467.8423	-0.8920	-0.7241	-1.7877
82.0	*********	-9.4573	-1.1719	-0.7241	-1.7877
84.0	-89.5673	-4.1949	-0.9480	-0.7800	-1.2278
87.0	-7.5538	-2.6835	-0.8920	-0.7241	-0.1063
89.0	9.7051	-2.0213	-1.1452	-1.2800	-1.6170
93.0	3.3027	-1.6170	-1.0778	-0.4039	-0.2691
96.0	-0.4713	-1.4148	-1.0104	-1.2800	-1.7518
100.0	-3.1671	-1.6170	-0.8082	-1.0778	-0.2691
96.0	0.2026	0.4722	0.3374	0.2701	0.3374
84.0	0.4722	0.6744	-0.3365	0.6744	0.6070
73.5	0.3823	-0.2467	0.6968	0.6968	0.6968
54.0	0.0679	0.3823	0.3823	0.0679	0.0679
33.0	-0.5612	0.0679	0.0679	0.0679	0.0679
24.0	-0.7289	-0.8377	-0.0763	0.2500	0.0325
10.0	-0.5114	-1.0552	-0.4026	0.3588	0.0679
5.0	-0.4026	-0.7289	-0.1851	0.2500	0.0679
2.5	-0.4026	-0.8377	-0.1851	0.2500	0.0679

FUSELAGE

-----TOP----	
XLOC	CP
38.2	-0.1478
43.4	-0.2306
50.4	-0.0650
56.1	0.1006
62.5	0.1006
69.7	0.0178
76.9	0.1006
83.4	-0.1478
89.4	-0.1478
95.4	-0.2306
101.1	-0.1478
---BOTTOM---	
38.2	0.1834
45.4	0.1006
50.4	-0.3134
56.1	-2.6489
62.5	-1.9694
69.7	-0.8930
76.9	-0.2306
83.4	0.1006
89.4	0.2662
95.4	-0.1478
101.1	-0.7274

MISC.*

NO.	CP
1	-0.2467
2	0.0679
3	-0.5612
4	0.0679
5	-0.2467
6	0.0679
7	-0.4039
8	-0.2691
9	-0.3365
10	-0.3365
11	-0.3365
12	-0.3365

C-5

TABLE A50. (Continued)

RUN:SEQ 298 3 — PROPLULSIVE WING / CANARD PRESSURES — PAGE 743

CLAERO = -0.0367 CDAERO = 0.6375 DCLC = 1.2544 DWLC = 1.7834 BASEPR = 8.1906

ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CMUC	CMUW	CMUT	HQT	RN
3.97	2142.03	2138.86	3.17	51.80	-0.05	2.37	1.464	1.985	3.449	87.4363	0.325082E+06

BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CMTR	CLTR	CDTR
0.05	3.00	-0.79	-0.63	0.02	-0.02	-0.11	0.69	-0.17	0.85	0.54

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-0.7259	-3.2066
2.5	-1.2656	-2.1213
5.0	-0.7259	-1.5786
10.0	0.0493	-1.2686
15.0	-0.6484	-1.2686
25.0	-0.5710	-0.9584
35.0	-0.5710	-0.6484
50.0	-0.4158	
56.0	-0.9584	-0.5710
65.0	-1.2686	-0.4934
76.0	-0.1834	-0.5710
79.0	-6.8499	-6.3075
80.5	-109.0489	4.7852
81.0	-106.4123	1.5391
82.0	-104.4044	-1.4830
84.0	-56.4424	
87.0	-60.3033	
89.0	-5.1460	-3.8215
93.0	14.5245	
96.0	9.2569	0.6483
100.0	3.9594	-1.5866
96.0	0.2345	0.2345
84.0	3.9594	-1.5866
73.0	0.2345	0.2345
50.0	0.6483	0.8140
35.0	0.8244	0.6694
25.0	0.2043	-1.1910
10.0	0.3593	0.2818
5.0	0.2818	-0.1058
2.5	0.5919	0.6694

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.5429	-0.5429	0.0000	-0.4341	-1.1963
2.5	-0.1079	0.5445	-0.3254	-1.0886	-1.5215
5.0	0.7620	0.4358	-0.1079	-0.2167	-0.6691
10.0	-0.1079	0.0008	-0.2167	-0.1079	-0.5429
15.0	0.3271	-0.3254	-0.2167	0.0008	-0.4341
24.0	0.2183	-0.5429	-0.3254	0.0008	-0.3254
33.0	0.0363	-0.2782	-0.5926	-0.9071	-0.5926
54.0	-0.2782	-0.5926	-0.9071	0.3506	-0.5926
67.0	-0.6760	-1.9176	-0.9243	-0.6760	-0.6760
78.5	1.3106	-3.9870	-1.0899	-1.0071	-1.1726
79.5	*********	-109.5545	-0.4157	-0.8114	-1.2032
80.5	*********	-8.7586	-0.5876	-0.4757	0.0840
81.3	*********	443.1594	-0.5316	-0.6998	-0.7555
82.0	*********	-9.9899	-0.5316	-0.5876	-1.0013
84.0	-90.0758	-4.0675	-0.4757	-0.6436	-0.9793
87.0	-7.6951	-2.6035	-0.8674	-0.9234	-0.0279
89.0	9.9403	-2.0523	-1.0417	-1.1092	-1.3787
93.0	3.3376	-1.4460	-1.1092	-0.3006	-0.1659
96.0	-0.4354	-1.2439	-1.1766	-0.3744	-1.4460
100.0	-3.0629	-1.5806	-0.6396	-0.9070	-0.3006
96.0	0.3730	0.5078	0.3057	0.3730	0.3057
84.0	0.5752	0.8447	-0.5028	0.7773	0.7099
73.5	0.3506	-0.2782	0.3506	0.6650	0.6650
54.0	0.0363	0.0363	0.0363	0.3506	0.0363
33.0	-0.5926	-0.2782	-0.2782	0.0363	0.0363
24.0	-0.6517	-0.9778	0.2183	0.5445	0.2183
10.0	-0.4341	-1.3040	0.1095	0.6532	0.0363
5.0	-0.2137	-0.7603	0.2183	0.6532	0.3506
2.5	-0.2167	-0.5429	0.2183	0.7620	0.6650

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3449
43.4	-0.3449
50.4	-0.0138
56.1	0.1517
62.5	0.0690
69.7	0.1517
76.9	-0.0138
83.4	-0.0138
89.4	-0.0966
95.4	-0.2621
101.1	-0.1793
---BOTTOM---	
38.2	0.2345
43.4	0.2345
50.4	-0.3410
56.1	-2.4143
62.5	-2.0832
69.7	-0.9243
76.9	-0.0966
83.4	0.3173
89.4	0.4000
95.4	-0.1793
101.1	-0.9243

**MISC.

NO.	CP
1	-0.5926
2	0.0363
3	-0.5926
4	0.0363
5	-0.5926
6	0.0363
7	-0.3006
8	-0.2333
9	-0.1659
10	-0.2333
11	-0.3006
12	-0.3006

TABLE A50. (Concluded)

PROPLULSIVE WING / CANARD PRESSURES

RUN:SEQ											
298 5	CLAERO =	0.4507	CDAERO =	1.0036	DCLC =	1.3027	DWLC =	1.8319	BASEPR =	8.1906	
ALPHA	PTOT	PSTAT	Q	VEL	YAW	H/C	CHUC	CHUW	CHUT	HGT	RN
12.04	2141.96	2138.68	3.28	82.74	-0.05	2.73	1.415	1.928	3.341	89.4116	0.330638E+06
BETA	CL	CD	CM	CROLL	CN	CY	CNTR	CNTR	CLTR	CDTR	
0.05	3.59	-0.14	-0.55	0.02	-0.01	-0.11	1.49	-0.10	1.36	0.83	

********** CANARD ******

%X/C	BP=3.375 CP	BP=10.125 CP
0.0	-6.2510	-2.5100
2.5	-5.6525	-2.3606
5.0	-2.5850	-2.4353
10.0	-0.1908	-2.2856
15.0	-1.3978	-2.2856
25.0	-1.2382	-2.3606
35.0	-1.1634	-2.1362
50.0	-1.0886	
54.0	-1.3131	-1.8367
65.0	-1.6872	-1.6123
76.9	-0.4901	2.2033
79.0	-7.2982	-8.7946
80.5	-110.9857	3.9771
81.0	-107.4010	1.2224
82.0	-106.7269	-1.9105
84.0	-57.3293	
87.0	-57.0610	
89.0	1.4856	-2.0296
93.0	15.1468	
96.0	8.2761	0.7667
100.0	3.5628	-1.7099
99.0	0.3672	0.0476
84.0	3.5628	-1.7099
73.0	0.3672	0.0476
50.0	0.8466	0.5867
35.0	0.7818	0.6322
25.0	0.2581	-1.4627
10.0	0.4826	0.4078
5.0	0.1833	0.1833
2.5	0.5574	0.3329

********** WING **********

%X/C	BP = 2 CP	BP = 6 CP	BP =12 CP	BP =16 CP	BP = 22 CP
0.0	-0.5680	-0.2531	-2.8671	-4.0313	-1.9322
2.5	-0.5680	0.1867	-1.5125	-3.9261	-2.0372
5.0	0.3766	-0.1482	-1.3026	-4.2411	-1.7224
10.0	-0.5680	-0.4630	-1.1978	-0.5680	-1.8274
15.0	-0.1482	-0.4630	-0.7778	-0.3580	-1.7224
24.0	-0.2531	-0.7778	-0.5680	-0.3580	-2.0372
33.0	-0.5110	-0.8145	-0.8145	-1.1180	-2.0283
54.0	-0.5110	-1.1180	-1.1180	-0.2076	-1.7248
65.0	-0.7513	-2.3490	-1.0709	-0.5110	-1.7898
78.5	3.0834	-4.4283	-1.3903	-1.3903	-2.0685
79.6	*********	-111.2920	-1.2083	-0.8842	-1.5324
80.5	-98.2740	-9.6386	-1.2624	-1.1003	-0.1820
81.3	-98.4906	410.2706	-1.2083	-0.8302	-2.1267
82.0	-98.1642	-9.8509	-1.2083	-1.0463	-1.8565
84.0	-88.6043	-5.0974	-0.8842	-0.8842	-1.6406
87.0	-10.6068	-3.9091	-0.9382	-1.0463	-0.2361
89.0	9.3297	-3.4165	-1.3996	-1.5297	-1.7248
93.0	2.7620	-1.9849	-1.4647	-0.6193	-0.4243
96.0	-0.9445	-2.0490	-1.3996	-1.5297	-1.3996
100.0	-3.3504	-2.1801	-1.1396	-1.3996	-0.3592
96.0	0.0309	0.4210	0.3560	0.2910	0.2910
84.0	0.5511	0.7461	-0.6153	0.8112	0.7461
73.5	0.3993	-0.5110	0.7028	0.7028	0.7028
54.0	0.3993	0.3993	0.3993	0.0960	0.3993
33.0	0.0960	-0.5110	0.0960	0.3993	0.3993
24.0	-0.0432	0.0618	0.4815	0.9014	0.3766
10.0	-0.8927	-0.3580	0.5865	1.0063	0.3993
5.0	-0.4630	-0.5680	0.9014	1.1112	0.7028
2.5	-0.4630	-0.6729	1.0063	1.1112	0.7028

FUSELAGE

-----TOP---- XLOC	CP
38.2	-0.3515
43.4	-0.3618
50.4	-0.1121
56.1	0.0476
62.5	0.0476
69.7	0.0476
76.9	-0.0322
83.4	0.0476
89.4	-0.0322
96.4	-0.1121
101.1	-0.1121
---BOTTOM---	
38.2	0.2074
43.4	0.2074
50.4	-0.4317
56.1	-2.3490
62.5	-2.2691
69.7	-0.8311
76.9	0.2074
83.4	0.3672
89.4	0.2873
96.4	-0.3518
101.1	-1.3903

MISC.

NO.	CP
1	-0.8145
2	0.0960
3	-1.4215
4	0.0960
5	-1.4215
6	0.0960
7	-0.4243
8	-0.3592
9	-0.4243
10	-0.4893
11	-0.3592
12	-0.4243

A4.0 TABULATED DOWNWASH DATA

Table A51 presents the tabulation of downwash measurements behind the wing of the propulsive wing/canard. Downwash data are presented for the full span and 1/2 span blowing nozzles. The downwash of the short (1/4) span was not measured.

TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.*	δ_F	VARIABLE	C_{μ_w}	C_L	$C_{L_{TR}}$
57	11	157.16	97.52	1.92	BW6V	0	α=0	0	0.142	
57	12	157.86	99.79	1.95			2	0	0.232	
57	13	158.52	100.07	2.95			4	0	0.322	
57	15	158.95	96.66	4.67			8	0	0.532	
57	17	159.05	98.66	9.52			12	0	0.752	
59	2	164.34	102.56	4.04	BW6V	0	α=0	0.5	0.369	0.265
59	3	163.83	103.37	5.14			2	0.5	0.495	0.377
59	6	163.33	103.37	5.89			4	0.5	0.617	0.483
59	8	163.68	106.57	9.08			8	0.5	0.862	0.704
59	10	164.63	107.09	14.76			12	0.5	1.143	0.962
61	4	101.174	66.06	5.50	BW6V	0	α=0	1.0	0.580	0.296
61	5	100.682	69.36	6.22			2	1.0	0.732	0.412
61	6	101.144	68.14	7.45			4	1.0	0.887	0.533
61	9	101.103	68.55	11.32			8	1.0	1.144	0.723
61	12	100.587	74.05	16.35			12	1.0	1.501	0.100
62	2	70.399	46.98	8.89	BW6V	0	α=0	2.0	0.898	0.320
62	3	70.389	49.51	8.64			2	2.0	1.075	0.427
62	4	70.377	48.73	10.44			4	2.0	1.266	0.551
62	6	70.358	49.61	12.94			8	2.0	1.650	0.800
62	8	71.666	53.94	18.28			12	2.0	2.029	1.098
64	3	159.158	111.19	4.11	BW6V	15	α=0	0	0.470	
64	4	159.230	111.44	5.20			2	0	0.560	
64	5	159.581	112.44	6.59			4	0	0.653	
64	7	157.348	114.17	9.60			8	0	0.845	
64	9	159.087	117.55	15.60			12	0	1.083	

*CONFIG. code on Table 3

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
65	4	152.700	114.11	7.98	BW6V	15	α=0	0.5	0.923	0.687
65	5	152.700	115.72	9.93			2	0.5	0.988	0.740
65	6	153.322	116.33	11.45			4	0.5	1.053	0.792
65	8	153.308	119.76	15.62			8	0.5	1.186	0.902
65	10	152.031	123.60	21.09			12	0.5	1.355	1.042
66	2	105.836	83.27	10.88	BW6V	15	α=2	1.0	1.433	0.910
66	3	105.357	81.80	8.82			0	1.0	1.308	0.806
66	5	105.756	83.32	14.52			4	1.0	1.584	1.034
66	7	105.716	85.93	17.00			8	1.0	1.892	1.294
66	10	104.784	87.75	22.70			12	1.0	2.279	1.622
67	2	76.092	63.51	9.91	BW6V	15	α=0	2.0	1.860	0.901
67	3	76.076	62.95	11.48			2	2.0	2.041	1.029
67	4	76.675	63.04	13.25			4	2.0	2.179	1.136
67	6	76.655	63.77	17.01			8	2.0	2.255	1.417
67	8	76.022	66.61	21.80			12	2.0	3.018	1.768
69	3	157.198	110.18	2.57	BC1W6V	15	α=0	0	0.489	
69	4	157.884	109.93	1.25			2	0	0.592	
69	6	158.280	115.66	- 0.31			4	0	0.700	
69	8	158.358	143.29	27.10			8	0	0.938	
69	10	157.539	139.48	30.16			12	0	1.221	
70	2	151.288	115.73	6.20	BC1W6V	15	α=0	0.5	1.099	0.792
70	3	151.608	114.91	6.90			2	0.5	1.246	0.922
70	4	151.906	125.83	8.31			4	0.5	1.381	1.042
70	6	156.534	134.22	8.59			8	0.5	1.656	
70	9	155.363	136.49	14.52			12	0.5	2.038	1.649

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
71	2	110.406	85.05	8.12	BC1W6V	15	α=0	1.0	1.420	0.833
71	3	110.384	87.34	10.07			2	1.0	1.594	0.972
71	5	110.775	85.71	11.48			4	1.0	1.769	1.116
71	7	109.895	102.83	9.20			8	1.0	2.183	1.455
71	9	109.027	114.91	14.29			12	1.0	2.638	1.838
72	3	78.716	62.31	9.61	BC1W6V	15	α=0	2.0	2.062	0.883
72	5	78.669	61.64	11.72			2	2.0	2.229	1.045
72	6	78.665	62.22	11.32			4	2.0	2.471	1.221
72	9	78.623	70.24	12.99			8	2.0	1.577	2.948
72	12	77.443	76.28	13.09			12	2.0	2.019	3.547
75	10	149.540	113.19	4.23	BC1W6V	45	α=0	0	2.371	
75	11	148.960	117.31	2.33			2	0	2.535	
75	12	148.311	112.20	2.47			4	0	2.688	
76	2	148.624	116.45	8.06	BC1W6V	45	α=0	0.5	1.578	1.049
76	3	150.850	123.54	9.55			2	0.5	1.641	1.118
76	4	150.576	123.08	12.15			4	0.5	1.756	1.221
76	7	149.156	132.45	15.70			8	0.5	2.079	1.523
76	10	149.540	141.47	18.59			12	0.5	2.371	1.805
77	2	105.958	85.68	8.35	BC1W6V	45	α=0	1.0	2.330	1.273
77	3	105.509	84.81	10.09			2	1.0	2.469	1.386
77	4	105.962	85.04	13.51			4	1.0	2.585	1.495
77	6	104.656	92.92	17.84			8	1.0	2.960	1.822
77	8	102.856	100.85	19.22			12	1.0	3.459	2.256
79	3	74.745	59.82	9.13	BC1W6V	45	α=0	1.0	3.456	1.298
79	9	74.750	60.62	14.47			8	1.0	4.110	1.870
79	11	74.139	71.08	18.58			12	1.0	4.614	2.304

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
85	2	158.756	114.66	5.13	BC2W6V	45	α=0	0	0.514	
85	4	158.864	113.93	6.36			2	0	0.601	
85	7	159.301	118.97	10.30			8	0	0.917	
85	9	159.381	117.55	14.50			12	0	1.139	
86	1	150.652	120.53	7.48	BC2W6V	45	α=0	0.5	1.543	1.037
86	2	150.023	119.40	8.39			2	0.5	1.615	1.084
86	3	151.610	124.14	11.63			4	0.5	1.675	1.147
86	4	153.514	121.23	10.24			4	0.5	1.657	1.149
86	5	152.911	125.90	15.44			8	0.5	1.945	1.436
86	6	150.731	143.52	20.68			12	0.5	2.222	1.727
86	7	151.082	144.70	21.91			12	0.5	2.218	1.677
87	1	104.471	87.42	8.74	BC2W6V	45	α=0	1.0	2.649	1.608
87	2	104.412	84.46	9.64			2	1.0	2.761	1.702
87	4	103.928	84.77	10.80			4	1.0	2.857	1.774
87	5	103.467	85.80	12.51			8	1.0	2.982	1.863
87	6	101.585	95.25	18.98			12	1.0	3.367	2.185
88	1	74.305	62.35	6.65	BC2W6V	45	α=0	2.0	3.950	1.918
88	2	72.376	59.71	8.43			2	2.0	4.217	2.029
88	4	74.898	60.09	9.39			4	2.0	4.140	2.037
88	5	74.229	58.58	13.68			8	2.0	4.301	2.106
88	7	73.583	63.99	17.54			12	2.0	4.617	2.337
89	2	149.841	122.26	11.31	BC2W6V	45	h=14.3	0.5	1.881	1.365
89	3	149.817	124.14	11.71			16.4	0.5	1.827	1.310
89	5	151.079	123.17	9.25			32.7	0.5	1.607	1.102
89	6	150.171	115.52	9.14			65.6	0.5	1.529	1.018
89	7	150.609	117.29	9.05			87.0	0.5	1.504	0.995

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
90	1	75.349	60.62	11.70	BC2W6V	45	h=14.5	2.0	4.157	2.149
90	2	75.318	59.75	12.98			16.4	2.0	4.111	2.102
90	3	76.441	66.96	8.30			32.7	2.0	3.812	1.863
90	4	79.283	57.25	8.05			65.4	2.0	3.573	1.757
90	5	74.694	54.92	8.87			87.0	2.0	3.852	1.804
92	2	157.717	113.93	4.71	BC2W6V	45	h=14.2	0	0.741	
92	3	158.016	115.63	4.42			16.4	0	0.702	
92	4	156.860	108.91	6.25			32.7	0	0.602	
92	5	158.314	113.75	6.83			65.4	0	0.543	
92	6	158.591	111.70	6.52			87.0	0	0.524	
97	2	159.243	113.73	8.32	BC2W5V	45	α =0	0	0.577	
97	3	159.412	111.22	8.98			0	0	0.576	
97	4	159.822	111.86	10.71			2	0	0.663	
97	5	159.596	112.61	11.95			4	0	0.750	
97	6	159.982	115.65	14.55			8	0	0.891	
97	7	160.046	113.36	19.59			12	0	1.076	
98	1	143.023	124.91	8.24	BC2W5V	45	α =0	0.5	1.834	1.242
98	3	143.766	126.32	11.04			2	0.5	1.905	1.304
98	4	144.800	137.69	11.24			4	0.5	1.987	1.385
98	5	144.834	148.34	17.71			8	0.5	2.217	1.599
98	6	144.523	157.68	23.03			12	0.5	2.393	1.758
99	2	99.412	90.60	8.01	BC2W5V	45	α =0	1.0	3.088	1.855
99	4	101.778	92.31	10.50			2	1.0	3.111	1.914
99	6	99.834	93.63	11.33			4	1.0	3.295	2.034
99	7	100.766	104.14	13.56			8	1.0	3.485	2.212
99	8	98.838	108.76	18.55			12	1.0	3.761	2.413

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
100	2	73.145	60.78	6.89	BC2W5V	45	α=0	2.0	4.332	2.042
100	3	71.161	59.04	8.42			2	2.0	4.587	2.127
100	4	72.457	59.41	7.27			4	2.0	4.581	2.174
100	5	72.481	63.81	11.02			6	2.0	4.720	2.281
100	6	71.798	66.82	9.69			8	2.0	4.885	2.370
100	7	73.106	74.45	16.28			12	2.0	5.062	2.586
108	3	159.033	109.16	3.64	BC3W6V	45	α=0	0	0.544	
108	4	159.396	109.93	3.59			2	0	0.647	
108	6	160.087	94.61	8.76			8	0	0.962	
108	9	159.846	110.55	28.23			12	0	1.199	
109	4	152.576	118.25	8.19	BC3W6V	45	α=0	0.5	1.408	0.935
109	5	152.565	120.10	9.96			2	0.5	1.496	1.019
109	7	152.273	120.74	12.35			4	0.5	1.586	1.094
109	12	148.460	136.04	16.83			12	0.5	2.128	1.576
110	2	103.301	82.34	8.79	BC3W6V	45	α=0	1.0	2.368	1.364
110	3	103.727	81.30	11.06			2	1.0	2.523	1.512
110	5	105.510	86.35	13.44			4	1.0	2.646	1.657
110	6	104.616	93.96	13.20			8	1.0	2.990	1.950
110	7	105.019	100.78	21.72			12	1.0	3.286	2.202
111	2	75.707	58.21	10.64	BC3W6V	45	α=0	2.0	3.374	1.466
111	3	75.061	57.32	10.18			2	2.0	3.575	1.599
111	5	75.036	58.32	10.97			4	2.0	3.786	1.778
111	8	75.015	65.95	14.93			8	2.0	4.345	2.282
111	10	75.630	76.21	19.86			12	2.0	4.704	2.625

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
142	2	160.726	112.19	6.28	BW6V	45	α=0	0	0.528	
142	3	161.723	113.92	6.50			2	0	0.615	
142	4	162.090	113.92	7.89			4	0	0.704	
142	5	161.542	112.19	12.00			8	0	0.883	
142	6	161.597	119.95	19.34			12	0	1.088	
143	1	129.940	121.41	8.88	BW6V	45	α=0	0.5	1.462	1.080
143	2	129.570	122.13	11.28			2	0.5	1.490	1.100
143	3	131.750	124.91	11.69			4	0.5	1.498	1.116
143	4	131.710	131.11	16.01			8	0.5	1.570	1.179
143	5	133.756	136.07	20.08			12	0.5	1.629	1.242
144	1	88.261	88.95	9.64	BW6V	45	α=0	1.0	2.564	1.751
144	2	87.153	87.79	13.27			2	1.0	2.763	1.915
144	3	88.759	91.27	12.53			4	1.0	2.851	2.021
144	4	87.540	91.12	17.89			8	1.0	3.243	2.372
144	5	87.569	92.75	20.71			12	1.0	3.405	2.512
145	1	65.763	61.99	12.76	BW6V	45	α=0	2.0	3.439	1.991
145	2	64.344	63.04	10.53			2	2.0	3.652	2.105
145	3	63.569	63.77	12.30			4	2.0	3.834	2.225
145	4	65.081	66.08	15.28			8	2.0	4.122	2.538
145	5	64.329	67.74	19.23			12	2.0	4.423	2.777
146	1	63.788	53.70	9.76	BW6V	45	α=0	3.0	4.499	2.276
146	2	64.528	55.10	11.01			2	3.0	4.588	2.380
146	3	64.522	56.92	11.43			4	3.0	4.773	2.533
146	4	63.755	56.73	16.58			8	3.0	5.238	2.891
146	5	63.762	57.72	19.26			12	3.0	5.477	3.088

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_w}	C_L	$C_{L_{TR}}$
147	1	54.752	45.49	9.02	BW6V	45	α =0	4.0	5.497	2.465
147	2	54.749	46.32	10.53			2	4.0	5.672	2.595
147	3	55.620	46.72	10.97			4	4.0	5.689	2.669
147	4	56.480	49.45	14.12			8	4.0	5.935	2.939
147	5	55.616	50.38	17.89			12	4.0	6.420	3.271
148	1	67.401	57.52	9.48	BW6V	45	α =0	2.0	3.496	2.083
148	2	67.391	58.29	12.74			2	2.0	3.663	2.227
148	3	67.383	54.24	14.57			4	2.0	3.800	2.343
148	4	68.104	58.47	16.12			8	2.0	4.197	2.659
148	5	67.376	61.29	18.26			12	2.0	4.534	2.932
149	1	67.389	54.24	10.68	BW6V	45	α =0	1.0	2.490	1.912
149	2	67.383	54.93	12.19			2	1.0	2.624	2.034
149	3	68.093	55.75	13.92			4	1.0	2.685	2.101
149	4	67.370	57.87	18.10			8	1.0	3.031	2.419
149	5	67.370	59.96	21.27			12	1.0	3.358	2.670
153	1	76.777	46.24	13.54	BW6V	45	h=14.7	2.0	2.154	0.638
153	2	76.154	50.32	8.79			16.8	2.0	2.513	0.973
153	3	76.095	58.95	12.36			32.7	2.0	3.530	1.989
153	4	76.729	58.21	7.73			65.5	2.0	3.465	1.946
153	5	76.123	58.99	7.50			87.7	2.0	3.468	1.923
154	1	155.226	115.49	9.92	BW6V	45	h=15.3	0.5	0.984	0.600
154	2	154.628	115.90	10.02			16.5	0.5	0.961	0.581
154	3	153.611	122.99	9.27			32.7	0.5	0.831	0.455
154	4	151.573	119.20	7.96			65.4	0.5	0.815	0.429
154	5	151.350	118.52	8.23			86.9	0.5	0.816	0.425

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
155	1	160.556	110.94	4.49	BW6V	45	h=14.9	0	0.723	
155	2	160.883	110.44	4.86			16.5	0	0.701	
155	3	160.013	109.42	4.84			33.3	0	0.606	
155	4	160.313	108.91	6.11			65.7	0	0.552	
155	5	160.362	109.16	5.35			87.1	0	0.530	
156	1	161.623	110.69	5.37	BW6V	45	h=14.9	0	0.727	
156	2	161.651	111.45	4.94			16.5	0	0.706	
156	3	161.067	109.42	5.86			32.6	0	0.609	
156	4	161.779	113.61	7.22			65.5	0	0.550	
156	5	160.204	110.69	6.42			87.3	0	0.526	
157	1	154.846	113.29	11.23	BW6V	45	h=15.5	0.5	0.998	0.626
157	2	154.213	114.61	12.22			16.4	0.5	0.982	0.607
157	4	154.248	124.48	10.06			32.6	0.5	0.828	0.447
157	5	152.819	121.66	8.83			65.6	0.5	0.789	0.395
157	6	154.085	121.11	8.89			87.3	0.5	0.764	0.377
158	1	106.073	78.39	11.69	BW6V	45	h=15.6	1.0	2.078	1.269
158	2	106.456	79.66	12.92			16.4	1.0	2.052	1.248
158	3	107.147	89.87	10.34			32.6	1.0	1.883	1.088
158	4	107.154	85.91	8.45			65.6	1.0	1.814	1.018
158	6	106.308	85.54	10.89			87.0	1.0	1.810	0.999
159	1	78.740	56.33	8.93	BW6V	45	h=14.9	2.0	2.428	0.949
159	2	77.613	50.67	11.09			16.4	2.0	2.536	1.007
159	3	76.241	64.61	11.38			32.9	2.0	3.668	2.095
159	4	77.458	62.63	10.30			65.5	2.0	3.559	2.029
159	5	76.856	61.93	11.53			86.9	2.0	3.558	2.002

TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
160	1	76.226	64.45	16.38	BW6V	45	h=30.4	2.0	3.899	2.269
160	2	76.817	62.80	15.63			32.6	2.0	3.850	2.245
160	3	76.817	64.63	12.61			65.5	2.0	3.575	2.039
160	4	78.000	64.45	14.27			87.2	2.0	3.600	2.040
161	1	104.795	86.85	16.69	BW6V	45	h=30.4	1.0	2.140	1.269
161	2	107.436	90.46	15.88			32.4	1.0	2.024	1.192
161	3	108.308	88.48	12.67			65.5	1.0	1.894	1.078
161	4	107.033	87.48	13.92			87.1	1.0	1.900	1.065
162	1	151.796	124.64	16.64	BW6V	45	h=30.4	0.5	0.916	0.520
162	2	152.435	126.17	15.18			32.8	0.5	0.892	0.499
162	3	153.404	122.49	13.17			65.8	0.5	0.814	0.425
162	4	153.291	123.13	12.45			87.0	0.5	0.806	0.417
163	1	169.088	111.45	8.22	BW6V	45	h=29.4	0	0.808	
163	2	160.708	111.20	7.87			32.9	0	0.797	
163	3	160.120	110.44	8.24			65.3	0	0.729	
163	4	161.011	111.20	8.20			87.1	0	0.702	
								C_{μ_C}		
171	2	157.605	118.97	- 2.67	BC1V	45	α=0	0	0.201	
171	3	157.701	116.84	- 2.54			2	0	0.239	
171	4	157.488	110.94	3.41			4	0	0.279	
171	5	157.872	105.00	23.62			8	0	0.348	
171	6	157.985	115.87	13.51			12	0	0.374	
172	1	150.558	108.39	- 2.14	BC1V	45	α=0	0.25	0.487	
172	2	149.074	111.70	- 2.00			2	0.25	0.531	
172	3	149.454	119.68	- 3.09			4	0.25	0.562	
172	5	149.817	142.64	23.99			8	0.25	0.663	
172	6	149.540	139.61	17.36			12	0.25	0.837	

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_C}	C_L	$C_{L_{TR}}$
173	1	102.061	72.13	- 0.59	BC1V	45	α=0	0.5	0.724	0.435
173	2	102.521	73.29	- 1.58			2	0.5	0.769	0.491
173	3	102.890	75.54	- 1.37			4	0.5	0.827	0.558
173	4	102.964	92.69	- 1.06			8	0.5	0.926	0.674
173	5	102.506	81.93	4.54			12	0.5	1.025	0.788
174	1	74.926	52.36	1.99	BC1V	45	α=0	1.0	0.971	0.430
174	2	70.393	50.73	- 0.42			2	1.0	1.083	0.486
174	3	70.370	50.17	- 0.31			4	1.0	1.141	0.560
174	4	72.956	53.42	1.85			8	1.0	1.182	0.692
176	2	158.081	112.20	- 2.23	BC1V	45	α=0	0	1.115	1.092
176	3	158.123	107.61	- 2.45			2	0	1.466	1.444
176	4	158.495	107.61	8.77			4	0	1.778	1.755
176	5	158.571	111.70	14.37			8	0	0.252	0.249
176	6	158.355	114.66	9.13			12	0	0.343	0.341
177	1	143.431	106.04	- 2.24	BC1V	45	α=0	0.25	0.427	0.303
177	2	146.725	110.69	- 3.44			2	0.25	0.465	0.344
177	3	146.081	123.86	- 3.53			4	0.25	0.519	0.394
177	4	148.665	147.57	- 0.85			8	0.25	0.576	0.452
177	5	148.378	137.84	30.28			12	0.25	0.754	0.622
178	1	102.817	73.29	- 0.62	BC1V	45	α=0	0.5	0.596	0.355
178	2	102.788	73.67	- 1.15			2	0.5	0.640	0.392
178	3	102.308	73.29	- 0.78			2	0.5	0.642	0.391
178	4	102.746	77.01	- 0.38			4	0.5	0.694	0.440
178	5	103.651	99.16	2.11			8	0.5	0.810	0.548
178	6	106.320	77.01	11.82			12	0.5	0.938	0.679

TABLE A51. TABULATED DOWNWASH DATA

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_c}	C_L	$C_{L_{TR}}$
179	1	74.097	50.73	1.17	BC1V	45	α=0	1.0	0.819	0.352
179	2	72.814	48.47	2.49			2	1.0	0.898	0.401
179	3	69.533	50.73	1.58			4	1.0	1.014	0.456
179	4	70.187	56.96	3.18			8	1.0	1.114	0.542
179	5	70.181	71.35	5.59			12	1.0	1.280	0.687
181	3	157.599	111.70	- 0.15	BV	0	α=0	0	-0.000	-0.038
181	4	158.035	115.63	2.39			2	0	-0.000	-0.033
181	5	157.780	118.97	- 0.46			4	0	0.104	-0.027
181	6	157.831	121.99	- 0.55			8	0	0.022	-0.015
181	7	157.583	125.85	- 2.04			12	0	0.039	0.002
								C_{μ_w}		
187	2	156.237	128.68	6.93	BW6V	45	α=0	0	0.546	
187	3	156.939	128.33	7.93			2	0	0.636	
187	4	157.007	127.43	9.66			4	0	0.720	
187	5	157.097	125.92	12.41			8	0	0.911	
187	6	156.262	124.39	20.65			12	0		
								b_N=1/2		
188	1	155.868	130.67	7.01	BW6V	45	α=0	0.5	0.885	0.627
188	2	153.483	129.06	8.02			2	0.5	0.964	0.692
188	3	154.131	130.34	9.23			4	0.5	1.013	0.738
188	4	158.090	133.78	14.00			8	0.5	1.080	0.810
188	5	155.132	149.57	21.90			12	0.5	1.200	0.912
								b_N=1/2		
189	1	107.023	99.72	7.41	BW6V	45	α=0	1.0	1.527	0.977
189	2	109.171	101.22	9.55			2	1.0	1.587	1.047
189	3	107.427	96.56	15.39			8	1.0	1.872	1.244
189	4	110.382	108.82	20.90			12	1.0	1.970	1.400

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
								b_N=1/2		
190	1	77.350	73.37	6.26	BW6V	45	α =0	2.0	2.142	1.088
190	2	77.745	72.82	9.14			2	2.0	2.279	1.186
190	3	77.927	73.85	8.95			4	2.0	2.330	1.250
190	4	77.337	77.30	15.95			8	2.0	2.627	1.495
190	5	76.732	77.76	17.96			12	2.0	2.800	1.619
216	2	128.442	104.65	7.59	BC1W6V	45	α =0	0	0.526	0.433
216	3	128.477	107.87	5.19			2	0	0.650	0.556
216	4	128.864	113.07	5.50			4	0	0.759	0.665
216	5	128.888	89.89	19.02			8	0	0.996	0.901
216	7	128.566	113.07	34.54			12	0	1.226	1.131
								b_N=1/2		
217	1	80.186	73.71	9.82	BC1W6V	45	α =0	1.0	2.023	0.932
217	3	80.167	73.72	8.94			2	1.0	2.003	0.892
217	4	80.155	73.08	9.27			4	1.0	2.076	0.943
217	5	79.563	79.67	11.81			8	1.0	2.434	1.246
217	6	78.405	93.10	14.33			12	1.0	2.893	1.632
								b_N=1/2		
218	1	57.454	39.95	5.14	BC1W6V	45	α =0	2.0	3.292	1.160
218	2	55.827	38.61	9.49			2	2.0	3.510	1.200
218	3	56.622	36.68	10.17			4	2.0	3.412	1.122
218	4	58.215	36.10	11.88			8	2.0	3.434	1.188
218	5	56.610	40.97	12.82			12	2.0	4.018	1.571

TABLE A51. TABULATED DOWNWASH DATA (Continued)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
								b_N=1/2		
227	1	60.963	52.89	5.46	BW6V	45	h=14.3	2.0	2.104	0.655
227	2	61.674	56.64	7.13			16.4	2.0	2.770	1.321
227	3	65.085	48.35	9.85			32.6	2.0	2.668	1.369
227	4	60.253	43.83	5.70			65.6	2.0	2.765	1.250
227	5	59.489	40.03	10.87			86.7	2.0	2.817	1.266
								b_N=1/2		
228	1	83.702	68.75	9.39	BW6V	45	h=14.4	1.0	1.913	1.176
228	2	84.223	68.79	11.78			16.3	1.0	2.034	1.261
228	3	90.182	68.27	8.28			32.6	1.0	1.824	1.144
228	5	87.728	63.91	8.15			65.5	1.0	1.801	1.084
								b_N=1/2		
229	1	116.119	94.92	9.84	BW6V	45	h=14.8	0.5	1.582	1.239
229	2	111.035	84.76	9.21			16.3	0.5	1.659	1.281
229	3	114.876	89.20	6.46			32.7	0.5	1.478	1.126
229	5	112.610	87.10	6.46			64.7	0.5	1.428	1.062
229	6	115.743	87.65	7.00			87.2	0.5	1.385	1.039
230	1	128.458	93.75	8.65	BW6V	45	h=14.8	0	0.762	0.816
230	2	128.446	94.15	8.69			16.3	0	0.742	0.796
230	3	128.089	94.56	7.23			32.7	0	0.641	0.696
230	4	126.983	95.67	7.80			65.3	0	0.582	0.638
230	5	126.668	94.74	6.73			86.9	0	0.565	0.622
							α=8			
231	1	127.604	92.52	14.25	BW6V	45	h=43.7	0	1.006	1.063
231	2	127.545	93.42	13.30			64.8	0	0.949	1.006
231	3	127.024	95.50	12.64			86.8	0	0.928	0.986

TABLE A51. TABULATED DOWNWASH DATA (Concluded)

RUN	POINT	V_∞	V_{PROBE}	ϵ	CONFIG.	δ_F	VARIABLE	C_{μ_W}	C_L	$C_{L_{TR}}$
							α =8			
232	2	114.951	97.45	17.04	BW6V	45	h=44.5	0.5	1.785	1.405
232	4	114.968	91.63	14.78			65.4	0.5	1.726	1.344
232	5	115.076	88.92	14.92			87.1	0.5	1.686	1.302
							α =8			
233	1	77.974	67.83	15.47	BW6V	45	h=44.5	1.0	2.618	1.791
233	2	78.993	63.06	14.52			65.5	1.0	2.546	1.745
233	3	78.434	61.75	17.31			87.0	1.0	2.532	1.721
233	4	57.364	51.41	15.53			44.5	2.0	3.462	1.953
233	7	58.113	45.47	14.96			65.2	2.0	3.298	1.833
233	9	58.091	45.13	16.78			86.7	2.0	3.251	1.792

A5.0 TABULATED RUN INDEX

Table A52 presents the Run Index for NASA-LaRC Test 290.

TABLE A52. RUN INDEX

PREPARED BY	
CHECKED BY	
DATE	26 JAN 84

NASA - LaRC
TEST 230
14.5 x 21.75 VGTS
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV DRAG	DOWN WASH		H/C	NPR	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	Time Started	Time Finished	DATE
40	BW	1	–	6	–	–	0°	–					∞	A	∞	–	0	0°	0°	Noz Calib & Static Thrust	5:50		1/26
41	BC	–	1	–	1	0°	–	0°					∞	B	–	∞	0	0°	0°	" " " "	6:30		1/26
42	BWC	1	1	6	1	0°	0°	0°					∞	C	∞	∞	0	0°	0°	" " " "			1/27
43																				Alpha Calib Run ??			1/27
44	BWC	1	1	6	1	0°	0°	0°					∞	1.0	–	–	0		0°	Wt Tare			1/27
45	BWC	1	1	6	1	0°	0°	0°		Y			∞	C	∞	∞	0	0°	0°	Static Run	10:10		1/27
46	BWC	1	1	6	1	0°	0°	0°		Y			∞	1.0	0	0	30	A	0°		10:16		1/27
47	BWC	1	1	6	1	0°	0°	0°		Y			∞	1.0	0	0	30	A	0°				
48										Y										?			
49										Y										?			
50	BWC	1	1	6	1	0°	0°	0°		Y			∞	2.5	.5	.2	28	C	0°	Includes Wt Tare Run	5:00		1/30
51																				XDUCER CALIB RUN			1/30
52	BWC	1	1	6	1	0°	0°	0°		Y			∞	2.5	.5	.2	28	D	0°	Cont'd from run 50			1/30
53	BWC	1	1	6	1	0°	0°	0°		Y			∞	2.5	1.0	.4	13.6	A	0°				1/30
54	BWC	1	1	6	1	0°	0°	0°		Y			∞	2.5	2.0	.8	7.1	A	0°				1/30
55	BWC	1	1	6	1	0°	0°	0°		Y			∞	1.0	0	0	30	A	0°				1/30
56	BW	1	–	6	–	–	0°	–		Y			∞	D	∞	–	0	0°	0°	STATIC RUN	9:50		1/31
57	BW	1	–	6	–	–	0°	–		Y			∞	1.0	0	0	30	A	0°	Includes Wt Tare	9:35		1/31
58	BW	1	–	6	–	–	0°	–					A	1.0	0	0	30	0°	0°	Gnd Hi Speed	10:05		1/31
59	BW	1	–	6	–	–	0°	–		Y			∞	2.5	.5	0	32	A	0°		10:20		1/31

NOTES

	NPR	α	H/C	Q
A	1.6, 2.4, 3.1, 3.6	-2° → 22° Δ = 2°	.4, .5, 1, 2, ∞	30, 24, 15, 8, 3
B	1.6, 2.3, 3.0, 3.6, 4.3	0, 2, 4, 8, 12, 16, 20	.65, 1, 2, ∞	
C	1.5, 2.1, 2.5	-2°, 0, 4°	.9, 1, 2, ∞	
D	1.5, 2, 2.5, 3, 3.5	4° → 22° Δ = 2°		
E	1.5, 2, 2.5, 3	0 → 12°		

ORIGINAL PAGE IS OF POOR QUALITY

TABLE A52. (Continued)

PREPARED BY
CHECKED BY
DATE

NASA - LaRC
TEST 290
14.5 x 21.75 VSTT
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV DRIVE	DOWN WASH		H/C	NPR	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TIME		DATE
60																							
61	BW	1	-	6	-	-	0°	-		Y										RUN ABORTED - Tunnel failure			
62	BW	1	-	6	-	-	0°	-		Y			∞	2.5	1.0	0	12	A	0°		2:00		1/31
63	BW	1	-	6	-	-	15°	-					∞	2.5	2.0	0	6.1	A	0°			2:40	1/31
64	BW	1	-	6	-	-	15°	-		Y			∞	D	∞	-	0	0°	0°	STATIC RUN			1/31
65	BW	1	-	6	-	-	15°	-		Y			∞	1.0	0	-	30	A	0°				1/31
66	BW	1	-	6	-	-	15°	-		Y			∞	2.5	.5	0	28	A	0°				1/31
67	BW	1	-	6	-	-	15°	-		Y			∞	2.5	1.0	0	13.3	A	0°				1/31
68	BWC	1	1	6	1	0°	15°	15°					∞	2.5	2.0	0	7	A	0°				1/31
69	BWC	1	1	6	1	0°	15°	15°		Y			∞	C	∞	∞	0	0°	0°	WT TARE & STATIC RUN			1/31
70	BWC	1	1	6	1	0°	15°	15°		Y			∞	1.0	0	0	30	A	0°				1/31
71	BWC	1	1	6	1	0°	15°	15°		Y			∞	2.5	.53	.22	29	A	0°				1/31
72	BWC	1	1	6	1	0°	15°	15°		Y			∞	2.5	1.0	.43	14.5	A	0°				1/31
73	BWC	1	1	6	1	0°	45°	45°					∞	2.5	2.0	.35	7.5	A	0°				1/31
74	BWC	1	1	6	1	0°	45°	45°					∞	C	∞	∞	0	0°	0°	STATIC RUN	9:00		2/1
75	BWC	1	1	6	1	0°	45°	45°		Y			∞	1.0	0	0	30		0°	RUN ABORTED - No WT TARE			2/1
76	BWC	1	1	6	1	0°	45°	45°		Y			∞	1.0	0	0	30	A	0°		10:10		2/1
77	BWC	1	1	6	1	0°	45°	45°		Y			∞	2.5	.5	.23	27	A	0°		10:46		2/1
78	BWC	1	1	6	1	0°	45°	45°					∞	2.5	1.0	.5	13.3	A	0°		11:15		2/1
79	BWC	1	1	6	1	0°	45°	45°		Y			A	2.5	1.0	.5	13	0°	0°	Gnd Hgt Sweep			
													∞	2.5	2.0	1.0	7	A	0°				

NOTES

A
B
C 1.5, 2.0, 2.5
D
E

TABLE A52. (Continued)

PREPARED BY	
CHECKED BY	
DATE	

NASA - LaRC
TEST 290
14.5 x 21.75 VSTL
FULL-SPAN PROPULSIVE WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_w}	δ_{F_c}		SCV DATA	DOWN WASH		H/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
80	BWC	1	1	6	1	0°	45°	45°					A	2.5	2.0	1.0	6.7	0°	0°				2/1
81	BWC	1	1	6	1	0°	45°	45°					A	2.5	.5	.23	28.5	0°	0°				2/1
82	BWC	1	1	6	1	0°	45°	45°					A	1.0	0	0	30	0°	0°				2/1
83	BWC	1	1	6	2	0°	45°	45°					∞	C	∞	∞	0	0°	0°	STATIC THRUST RUN			2/1
84	BWC	1	1	6	2	0°	45°	45°					∞	1.0	–	–	0		0°	WT TARE	3:30		2/1
85	BWC	1	1	6	2	0°	45°	45°		Y			∞	1.0	0	0	30	B	0°				2/1
86	BWC	1	1	6	2	0°	45°	45°		Y			∞	2.5	.5	.23	28	B	0°				2/1
87	BWC	1	1	6	2	0°	45°	45°		Y			∞	2.4	1.0	.5	13	B	0°				2/1
88	BWC	1	1	6	2	0°	45°	45°		Y			∞	2.5	2.0	1.0	6.7	B	0°				2/1
89	BWC	1	1	6	2	0°	45°	45°		Y			A	2.5	.5	.24	27	0°	0°				2/1
89	BWC	1	1	6	2	0°	45°	45°		Y			A	2.5	1.0	.5	13.8	0°	0°	Pt. 2, 5, 10, 11, 12 Will fix run # 2 recommends			2/1
90	BWC	1	1	6	2	0°	45°	45°		Y			A	2.5	2.0	1.0	7	0°	0°				2/1
91																							
92	BWC	1	1	6	2	0°	45°	45°		Y			A	1.0	0	0	30	0°	0°	SPARE RUN NO-USE FOR 89 # 2			2/1
93	BWC	1	1	5	2	0°	45°	45°					∞	1.0	–	–	0		0°	WT TARE N.G.	8:30		2/1
94	BWC	1	1	5	2	0°	45°	45°					∞	1.0	–	–	0		0°	WT TARE			2/1
95	BWC	1	1	5	2	0°	45°	45°					∞	E	∞	∞	0	0°	0°	STATIC RUN (N.G. Fitting leak)			2/1
96	BWC	1	1	5	2	0°	45°	45°					∞	E	∞	∞	0	0°	0°	STATIC & WT Tare			2/2
97	BWC	1	1	5	2	0°	45°	45°		Y			∞	1.0	0	0	30	B	0°				2/2
98	BWC	1	1	5	2	0°	45°	45°		Y			∞	2.3	.5	.29	24.6	B	0°				2/2

NOTES

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E 1.5, 2.0, 2.5

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TABLE A52. (Continued)

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NASA - LaRC
TEST 250
14.5 x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV PRES	DOWN WASH		M/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TAPE STATUS	TAPE PROCESSED	PAGE
99	BWC	1	1	5	2	0°	45°	45°		Y			∞	2.3	1.0	.6	12	B	0°				1/2
100	BWC	1	1	5	2	0°	45°	45°		Y			∞	2.3	2.0	1.2	6.2	B	0°				2/2
101	BWC	1	1	5	2	0°	45°	45°		Y			A	2.3	.5	.28	25	0°	0°				1/2
102	BWC	1	1	5	2	0°	45°	45°					A	2.3	1.0	.6	12.5	0°	0°				2/2
103	BWC	1	1	5	2	0°	45°	45°					A	2.3	2.0	1.1	6.3	0°	0°				2/2
104	BWC	1	1	5	2	0°	45°	45°					A	1.0	0	0	30	0°	0°				2/2
105	BWC	1	1	6	3	0°	45°	45°					∞	1.0	–	–	0		0°	WT TARE N.G.			1/2
106	BWC	1	1	6	3	0°	45°	45°					∞	1.0	–	–	0		0°	WT TARE OK			2/2
107	BWC	1	1	6	3	0°	45°	45°					∞	E	∞	∞	0	0°	0°	STATIC RUN			2/2
108	BWC	1	1	6	3	0°	45°	45°		Y			∞	1.0	0	0	30	B	0°				2/2
109	BWC	1	1	6	3	0°	45°	45°		Y			∞	2.5	.5	.21	28	B	0°				2/2
110	BWC	1	1	6	3	0°	45°	45°		Y			∞	2.5	1.0	.42	13.3	B	0°				1/2
111	BWC	1	1	6	3	0°	45°	45°		Y			∞	2.5	2.0	.86	6.9	B	0°				2/2
112	BWC	1	1	6	3	0°	45°	45°					A	2.4	.5	.21	27	0°	0°				2/2
113	BWC	1	1	6	3	0°	45°	45°					A	2.5	1.0	.44	13	0°	0°				2/2
113	BWC	1	1	6	3	0°	45°	45°					A	2.5	2.0	.85	7.1	0°	0°	Pc. 6, 7, 8, 9, 10			2/2
114	BWC	1	1	6	3	0°	45°	45°					A	1.0	0	0	30	0°	0°				
115																				SPARE RUN No - Use for 113 #2			2/2
116	BWC	1	1	6	1	10°	45°	45°					∞	1.0	–	–	0		0°	WT TARE			2/2
117	BWC	1	1	6	1	10°	45°	45°					∞	E	∞	∞	0	0°	0°	STATIC RUN			

NOTES

A
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TABLE A52. (Continued)

PREPARED BY	
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NASA - LaRC
TEST 290
14.5 x 21.75 VSTL
FULL-SPAN PROPULSIVE WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_W}	δ_{F_C}		SCV DATA	DOWN WASH		H/C	N_{P_R}	C_{μ_W}	C_{μ_C}	Q	α	β	REMARKS	[illegible]	[illegible]	[illegible]
118	BWC	1	1	6	1	10°	45°	45°		Y			∞	1.0	0	0	30	B	0°				2/2
119	BWC	1	1	6	1	10°	45°	45°		Y			∞	2.5	.5	.21	27.5	B	0°				2/2
120	BWC	1	1	6	1	10°	45°	45°		Y			∞	2.5	1.0	.43	13.9	B	0°				2/2
121	BWC																			RUN ABORTED - 1 Pt only BAD Pt			2/2
122	BWC	1	1	6	1	10°	45°	45°					∞	1.0	-	-	0		0°	WT TARE (New Tare)			2/3
123	BWC									Y										N.G.			
124	BWC	1	1	6	1	10°	45°	45°		Y			∞	2.5	2.0	.88	6.7	B	0°				2/3
125	BWC	1	1	6	1	10°	45°	45°					A	2.5	2.0	.87	7.0	0°	0°				2/3
126	BWC	1	1	6	1	10°	45°	45°					A	2.5	1.0	.44	14	0°	0°				2/3
127	BWC	1	1	6	1	10°	45°	45°					A	2.5	.5	.22	28.5	0°	0°				2/3
128	BWC	1	1	6	1	10°	45°	45°		Y			A	1.0	0	0	30	0°	0°				2/3
129										Y										RUN N.G.			
130	BWC	1	1	6	1	0°	45°	45°					∞	2.4	1.0	.44	12.8	B	0°				2/3
131	BWC	1	1	6	1	0°	45°	45°					∞	2.5	3.0	1.28	4.5	B	0°				2/3
132	BWC	1	1	6	1	0°	45°	45°					∞	2.5	4.0	1.74	3.4	B	0°				2/3
133	BWC	1	1	6	1	0°	45°	45°					∞	2.5	2.0	.87	6.9	2°	A				2/3
134	BWC	1	1	6	1	0°	45°	45°					∞	2.5	.5	.21	28	2°	A				2/3
135	BWC	1	1	6	1	0°	45°	45°					∞	1.0	0	0	30	2°	A				2/3
136	BWC	1	1	6	1	0°	45°	45°					B	1.0	0	0	30	2°	0.8				2/3
137	BWC	1	1	6	1	0°	45°	45°					B	2.5	.5	.22	28	2°	0.8				2/3

NOTES

A	-0.8 → +0.8
B	-8.0 → +8.0
C	
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TABLE A52. (Continued)

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NASA - LaRC
TEST 230
14.5 x 21.75 WTM
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV DEFL	DOWN WASH		H/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS			
138	BWC	1	1	6	1	0°	45°	45°					B	2.5	1.0	.44	13.7	2°	0.8				2/3
139	BWC	1	1	6	1	0°	45°	45°					B	2.5	2.0	.87	6.9	2°	0.8				2/3
140	BW	1	–	6	–	–	45°	–					∞	1.0	–	–	0		0°	WT TARE			2/3
141	BW	1	–	6	–	–	45°	–					∞	E	∞	∞	0	0°	0.8	STATIC RUN			2/3
142	BW	1	–	6	–	–	45°	–		Y			∞	1.0	0	0	30	B	0°				2/3
143	BW	1	–	6	–	–	45°	–		Y			∞	2.5	.5	0	28.6	B	0°	10000+ IN-LB PM			2/3
144	BW	1	–	6	–	–	45°	–		Y			∞	2.5	1.0	0	13.6	B	0°				2/3
145	BW	1	–	6	–	–	45°	–		Y			∞	2.5	2.0	0	6.9	B	0°				2/3
146	BW	1	–	6	–	–	45°	–		Y			∞	2.5	3.0	0	4.9	B	0°	HIGH C_M			2/3
147	BW	1	–	6	–	–	45°	–		Y			∞	2.5	4.0	0	3.6	B	0°	" "			2/3
148	BW	1	–	6	–	–	45°	–		Y			∞	2.1	2.0	0	5.3	B	0°	Constant Q, VARY NPR			2/3
149	BW	1	–	6	–	–	45°	–		Y			∞	1.43	1.0	0	5.3	B	0°	" " " "			2/3
150	BW	1	–	6	–	–	45°	–		Y			∞	1.0	0	0	30	0°	A				2/3
151	BW	1	–	6	–	–	45°	–		Y			∞	2.5	.5	0	29.5	0°	A	PM > 10K IN-LB			2/3
152	BW	1	–	6	–	–	45°	–		Y			∞	2.5	2.0	0	6.9	0°	A				2/4
153	BW	1	–	6	–	–	45°	–		Y			A	2.5	2.0	0	6.8	0°	0.8				2/4
154	BW	1	–	6	–	–	45°	–		Y			A	2.5	.5	0	28	0°	0.8	PM > 10K IN-LB			2/4
155	BW	1	–	6	–	–	45°	–		Y			A	1.0	0	0	30	0°	0.8				2/4
156	BW	1	–	6	–	–	45°	–		Y			A	1.0	0	0	30	0°	0°				2/4
157	BW	1	–	6	–	–	45°	–		Y			A	2.5	.5	0	27.8	0°	0°	PM > 10K IN-LB			2/4

NOTES

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TABLE A52. (Continued)

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NASA - LaRC
TEST 290
14.5 x 21.75 VSTT
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV DRAG	DOWN WASH		H/C	N_{PR}	C_{M_w}	C_{M_c}	Q	α	β	REMARKS	TIME BEGIN	TIME END	DATE
158	BW	1	-	6	-	-	45°	-		Y			A	2.5	1.0	0	13.4	0°	0°		8:25		2/4
159	BW	1	-	6	-	-	45°	-		Y			A	2.5	2.0	0	6.8	0°	0°		8:35		2/4
160	BW	1	-	6	-	-	45°	-		Y			C	2.5	2.0	0	6.9	4°	0°				2/4
161	BW	1	-	6	-	-	45°	-		Y			C	2.5	1.0	0	13.6	4°	0°				2/4
162	BW	1	-	6	-	-	45°	-		Y			C	2.5	.5	0	27.4	4°	0°	PM > 10K INLB			2/4
163	BW	1	-	6	-	-	45°	-		Y			C	1.0	0	0	30	4°	0°				2/4
164	BW	1	-	6	-	-	45°	-		Y			∞	2.5	.5	0	28.4	8	0°	Repeat 143 - 10000+ in lb PM			2/4
165	BW	1	-	6	-	-	45°	-		Y			∞	1.9	.5	0	17.6	8	0°	" 164 - lower NPR - 9400 in lb PM			2/4
166	BW	1	-	6	-	-	45°	-		Y			∞	1.0	0	0	A	8°	0°				2/4
167	BW	1	-	6	-	-	45°	-		Y			∞	C	∞	0	0	0	0	WT TARE & STATIC			2/4
168	BW	1	-	6	-	-	45°	-		Y			∞	VARY	.5	0	VARY	8°	0°	10000+ INLB PM			2/4
169	BC	-	1	-	1	10°	-	45°					∞	1.0	-	-	0		0°	WT TARE			2/6
170	BC	-	1	-	1	10°	-	45°					∞	E	0	∞	0	0°	0°	STATIC RUN			2/6
171	BC	-	1	-	1	10°	-	45°		Y			∞	1.0	0	0	30	B	0°		9:05		2/6
172	BC	-	1	-	1	10°	-	45°		Y			∞	2.6	0	.25	26.7	B	0°		9:50		2/6
173	BC	-	1	-	1	10°	-	45°		Y			∞	2.6	0	.5	12.7	B	0°		10:25		2/6
174	BC	-	1	-	1	10°	-	45°		Y			∞	2.6	0	1.0	6.8	B	0°			11:00	2/6
175	BC	-	1	-	1	0°	-	45°					∞	E	0	∞	0	0°	0°	STATIC RUN	11:35		2/6
176	BC	-	1	-	1	0°	-	45°		Y			∞	1.0	0	0	30	B	0°		12:30		2/6
177	BC	-	1	-	1	0°	-	45°		Y			∞	2.5	0	.25	25.6	B	0°		1:26		2/6

NOTES
A
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TABLE A52. (Continued)

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NASA - LaRC
TEST 290
14.5 x 21.75 VSTT
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_w}	δ_{F_c}		SCV DATA	DOWN WASH		H/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TIME BEGIN	TIME END	DATE
178	BC	-	1	-	1	0°	-	45°		Y			∞	2.5	0	.5	12.8	B	0°		1:50		2/6
179	BC	-	1	-	1	0°	-	45°		Y			∞	2.5	0	1.0	6.0	B	0°		2:10		2/6
180	B	-	-	-	-	-	-	-					∞	1.0	-	-	0		0°	WT TARE	2:50		2/6
181	B	-	-	-	-	-	-	-		Y			∞	1.0	0	0	30	B	0°		3:15		2/6
182	BW	1/2	-	6	-	-	45°	-					∞	VARY	∞	0	0	0°	0°	$P_{T_{NOZ}}$ VS P_{PL} calib			2/6
183	BW	1/2	-	6	-	-	45°	-					∞	E	∞	0	0	0°	0°	WT TARE + STATIC RUN			2/6
184	BW	1/2	-	6	-	-	45°	-					∞	E	∞	0	0	0°	0°	STATIC RUN			2/6
185	BW	1/2	-	6	-	-	45°	-					∞	VARY	∞	0	0	0°	0°	$P_{T_{NOZ}}$ VS P_{PLEN} CALIB			2/6
186	BW	1/2	-	6	-	-	45°	-					∞	F	∞	0	0	0°	0°	STATIC RUN			2/6
187	BW	1/2	-	6	-	-	45°	-		Y			∞	1.0	0	0	30	B	0°		11:00		2/6
188	BW	1/2	-	6	-	-	45°	-		Y			∞	2.5	.5	0	28.7	B	0°		11:15		2/6
189	BW	1/2	-	6	-	-	45°	-		Y			∞	2.5	1	0	14.9	B	0°		11:25		2/6
190	BW	1/2	-	6	-	-	45°	-		Y			∞	2.5	2	0	7.5	B	0°		11:40		2/6
191	BW	1/2	-	6	-	-	45°	-		Y			∞	1.0	0	0	30	0°	B		10:15		2/7
192	BW	1/2	-	6	-	-	45°	-		Y			∞	2.5	.5	0	30.5	0°	B		11:00		2/7
193	BW	1/2	-	6	-	-	45°	-		Y			∞	2.5	2	0	7.6	0°	B				
194	BW	1/2	-	6	-	-	45°	-					∞	1.0	-	-	0		0°	WT TARE (w/ 40 lb WTs)			2/7
195	BW	1/2	-	6	-	-	45°	-		Y			∞	G	∞	0	0	0°	0°	STATIC RUN			2/7
196	BW	1/2	-	6	-	-	45°	-					∞	1.0	-	-	0		0°	WT TARE (w/o 40 lb WTs)	2:45		2/7
197	BW	1/2	-	6	-	-	45°	-					∞	G	∞	0	0	0°	0°	STATIC RUN			2/7

NOTES

A
B -8.0 → +8.0
C
D
E

TABLE A52. (Continued)

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NASA - LaRC
TEST 290
14.5 x 21.75 VSTL
FULL-SPAN PROPULSIVE WING

SUB NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_W}	δ_{F_C}		SCV DATA	DOWN WASH		H/C	N_{PR}	C_{μ_W}	C_{μ_C}	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
198	BW	1/2	-	6	-	-	45°	-												BAL INTERFERENCE CHK RUN			2/7
199	BW	1/2	-	6	-	-	45°	-												" " " "			2/7
200	BW	1/2	-	6	-	-	45°	-					∞	1.0	0	0	30	B	0°				2/7
201	BWC	1/2	1	6	1	0°	45°	45°					∞	1.0	-	-	0		0°	WT TARE	6:45		2/7
202	BWC	1/2	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	STATIC RUN	7:20		2/7
203	BWC	1/2	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	" "			2/7
204																				No Data wt Tare ??			2/7
205	B																			STING Press Tare Run			2/7
206																				Model loading run w/ Jack			2/8
207																				" " " "			2/8
208																				" " " "			2/8
209																				" " " "			2/8
210																				" " " "			2/8
211	B																			STING PRESS TARE RUN			2/8
212	BWC	1/2	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	STATIC THRUST RUN N.G.			2/8
213	BWC	1/2	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	" " "			2/8
214	BWC	1/2	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	20°	0°	" " " α = 20°			2/8
215	BWC	1/2	1	6	1	0°	45°	45°					∞	1.97	.5	.31	17	B	0°		4:35		2/8
216	BWC	1/2	1	6	1	0°	45°	45°		Y			∞	1.0	0	0	20	B	0°		5:15		2/8
217	BWC	1/2	1	6	1	0°	45°	45°		Y			∞	1.92	1.0	.63	7.9	B	0°		5:40		2/8

NOTES

A
B
C
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TABLE A52. (Continued)

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NASA - LaRC
TEST 290
14.5 x 21.75 VSTT
FULL-SPAN PROPULSIVE WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV PARA	DOWN WASH		H/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TIME STARTED	TIME FINISHED	DATE
218	BWC	1/2	1	6	1	0°	45°	45°		Y			∞	1.93	2.0	1.22	4.1	B	0°				2/8
219	BW	1/2	–	6	–	–	45°	–					∞	1.0	–	–	0		0°	WT TARE			2/8
220	BW	1/2	–	6	–	–	45°	–					∞	1.6 2.5	∞	0	0	0°	0°	STATIC RUN			2/8
221	BW	1/2	–	6	–	–	45°	–		Y			∞	1.0	0	0	20	B	0°				2/8
222	BW	1/2	–	6	–	–	45°	–					∞	1.7	.5	0	14.4	E	0°		7:10		2/8
223	BW	1/2	–	6	–	–	45°	–					∞	1.7	1.0	0	8.1		0°				2/8
224	BW	1/2	–	6	–	–	45°	–					∞	1.9	1.0	0	9		0°				2/8
225	BW	1/2	–	6	–	–	45°	–					∞	1.9	2.0	0	5.1		0°				2/8
226	BW	1/2	–	6	–	–	45°	–		Y			∞	1.95	2.0	0	5		B				2/8
227	BW	1/2	–	6	–	–	45°	–		Y			A	1.9	2.0	0	4.7	0°	0°				2/8
228	BW	1/2	–	6	–	–	45°	–		Y			A	1.9	1.0	0	9	0°	0°				2/8
229	BW	1/2	–	6	–	–	45°	–		Y			A	1.7	.5	0	16.7	0°	0°				2/8
230	BW	1/2	–	6	–	–	45°	–		Y			A	1.0	0	0	20	0°	0°				2/8
231	BW	1/2	–	6	–	–	45°	–		Y			D	1.0	0	0	20	8°	0°				2/8
232	BW	1/2	–	6	–	–	45°	–		Y			D	1.7	.5	0	16.5	8°	0°				2/8
233	BW	1/2	–	6	–	–	45°	–		Y			D	1.7	1.0	0	7.8	8°	0°	PT 1,2,3			2/8
233	BW	1/2	–	6	–	–	45°	–		Y			D	1.7	2.0	0	4.2	8°	0°	PT 4,7,9			2/8
234	BW	1/2	–	6	–	–	45°	–					D	1.7	.74	0	11.6	8°	0°		11:06		2/8
235	BW	1/2	–	6	–	–	45°	–		Y			∞	1.7	.5		16	0°	B				2/8
236	BW	1/2	–	6	–	–	45°	–		Y			∞	1.0	0		20	0°	B				2/8

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TEST 290
14.5 x 21.75 VSTT
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_w}	δ_{F_c}		SCV DRIP	DOWN WASH		H/C	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	TIME BEGIN	TIME END	DATE
237	BWC	1/2	1	6	1	0°	0°	0°					∞	1.0	–	–	0		0°	WT TARE	9:30		2/9
238	BWC	1/2	1	6	1	0°	0°	0°					∞	1.0	–	–	0		0°	WT TARE	9:45		2/9
239	BWC	1/2	1	6	1	0°	0°	0°					∞	VARY	∞	∞	0		0°	STATIC RUN	10:00		2/9
240	BWC	1/2	1	6	1	0°	0°	0°		Y			∞	1.0	0	0	30	A	0°		10:40		2/9
241	BWC	1/2	1	6	1	0°	0°	0°		Y			∞	2.0	.5	.31	21.1	B	0°		11:15		2/9
242	BWC	1/2	1	6	1	0°	0°	0°					∞	2.0	1.0	.66	10.6	B	0°		11:51		2/9
																				CHANGE TO BENT STING			
243	B																			STING-PRESS TARE RUN			2/9
244																				? POSSIBLY XDUCER CALIB			2/10
245																				? Possibly Hot & α Calib			2/10
246																				? Plen Press "			2/10
247	B																			STING-PRESS TARE RUN			2/10
248	BWC	1/2	1	6	1	0°	0°	0°					∞	VARY	∞	∞	0	0°	0°	STATIC RUN	11:05		2/10
249	BWC	1/2	1	6	1	0°	0°	0°					∞	1.0	–	–	0		0°	WT TARE	11:35		2/10
250	BWC	1/2	1	6	1	0°	0°	0°					∞	2.0	.5	.28	17.5	B	0°		12:50		2/10
251	BWC	1/2	1	6	1	0°	0°	0°					∞	2.4	1.0	.54	12	B	0°				2/10
252	BWC	1/2	1	6	1	0°	0°	0°					∞	2.4	2.0	1.1	6	B	0°				2/10
253	BW	1/2	–	6	–	–	0°	–					∞	VARY	∞	–	0	0°	0°	Wing P_{TUBE} vs P_{PLEN} Calib			2/10
254	BW	1/2	–	6	–	–	0°	–					∞	1.0	–	–	0		0°	WT TARE			2/10
255	BW	1/2	–	6	–	–	0°	–					∞	1.5 & 2.4	∞	0	0	0	0°	STATIC RUN			2/10

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TEST 290
14.5 x 21.75 VSTM
FULL-SPAN PROPULSIVE WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_w}	δ_{F_c}		SCV DRAG	DOWN WASH		H/C	N_{P_R}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS	[illegible]	[illegible]	DATE
256	BW	1/2	–	6	–	–	0°	–		Y			∞	1.0	0	0	30	B	0°				2/10
257	BW	1/2	–	6	–	–	0°	–		Y			∞	2.4	.5	0	28	B	0°		5:45		2/10
258	BW	1/2	–	6	–	–	0°	–		Y			∞	2.4	1.0	0	14	B	0°				2/10
259	BW	1/2	–	6	–	–	0°	–		Y			∞	2.4	2.0	0	8	B	0°				2/10
260	BW	1/4	–	6	–	–	0°	–					∞	VARY	∞	0	0	0°	0°	1/4 SPAN NOZZ/PLEN CALIB	[illegible]		2/10
261	BW	1/4	–	6	–	–	0°	–					∞	VAR	∞	0	0	0°	0°	STATIC RUN (USE 263)	[illegible]		2/10
262	BW	1/4	–	6	–	–	0°	–		Y			∞	1.0	0	0	30	B	0°		[illegible]		2/10
263	BW	1/4	–	6	–	–	0°	–					∞	1.0	–	–	0		0°	WT TARE (NEW TAPE)			2/11
264	BW																			Vent AP Xducer Calib	8:45		2/11
265	BW	1/4	–	6	–	–	0°	–		Y			∞	2.4	.5	0	17.4	B	0°		9:30		2/11
266	BW	1/4	–	6	–	–	0°	–		Y			∞	2.4	1.0	0	7.9	B	0°				2/11
267	BW	1/4	–	6	–	–	0°	–		Y			∞	2.4	2.0	0	4.1	B	0°				2/11
268	BW	1/4	–	6	–	–	0°	–					∞	1.5→2.6	∞	0	0	0°	0°	STATIC RUN (C_μ calib)	10:20		2/11
269	BWC	1/4	1	6	1	0°	0°	0°					∞	1.5→2.3	∞	∞	0	0°	0°	STATIC RUN	11:15		2/11
270	BWC	1/4	1	6	1	0°	0°	0°					∞	1.0	–	–	0		0°	WT TARE	11:25		2/11
271	BWC	1/4	1	6	1	0°	0°	0°		Y			∞	1.0	0	0	30	B	0°		11:35		2/11
272	BWC	1/4	1	6	1	0°	0°	0°		Y			∞	2.4	.5	.3	18	B	0°		11:55		2/11
273	BWC	1/4	1	6	1	0°	0°	0°		Y			∞	2.4	1.0	.6	9	B	0°		12:15		2/11
274	BWC	1/4	1	6	1	0°	0°	0°		Y			∞	2.4	2.0	1.2	4.5	B	0°		12:30		2/11
275	BW	1/4	–	6	–	–	45°	–					∞	1.5→2.3	∞	∞	0	0°	0°	Static run NG	2:40		2/11

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TEST 290
14.5 x 21.75 VSTM
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	δ_c	δ_{F_W}	δ_{F_C}		SCV DRAG	DOWN WASH		H/C	N_{PR}	$C_{\mu W}$	$C_{\mu C}$	Q	α	β	REMARKS	[illegible]	[illegible]	DATE
276	BW	1/4	-	6	-	-	45°	-					∞	1.0	-	-	0		0°	WT TARE	[illegible]		2/11
277	BW	1/4	-	6	-	-	45°	-		Y			∞	1.0	0	0	30	B	0		[illegible]		2/11
278	BW	1/4	-	6	-	-	45°	-					∞	VARY	∞	0	0	0°	0°	STATIC RUN N.G.	[illegible]		2/11
279	BW	1/4	-	6	-	-	45°	-					∞	VARY	∞	0	0	0	0	STATIC RUN			2/11
280	BW	1/4	-	6	-	-	45°	-		Y			∞	1.9	.5	0	10.4	B	0°				2/11
281	BW	1/4	-	6	-	-	45°	-		Y			∞	1.9	1.0	0	4.7	B	0°				2/11
282	BW	1/4	-	6	-	-	45°	-		Y			∞	1.9	2.0	0	2.7	B	0°				2/11
283	BW	1/4	-	6	-	-	45°	-		Y			∞	1.0	0.	0	30	0°	B		[illegible]		2/11
284	BW	1/4	-	6	-	-	45°	-		Y			∞	1.9	.5	0	10	0°	B				2/11
285	BW	1/4	-	6	-	-	45°	-		Y			∞	1.9	2	0	2.6	0°	B				2/11
286	BWC	1/4	1	6	1	0°	45°	45°					∞	1.0	-	-	0		0	WT TARE N.G.	[illegible]		2/11
287	BWC	1/4	1	6	1	0°	45°	45°					∞	1.0	-	-	0		0	WT TARE			2/11
288	BWC	1/4	1	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	STATIC RUN			2/11
289	BWC	1/4	1	6	1	0°	45°	45°		Y			∞	1.0	0	0	30	B	0°				2/11
290	BWC	1/4	1	6	1	0°	45°	45°		Y			∞	1.9	.5	.27	13.5	B	0°				2/11
291	BWC	1/4	1	6	1	0°	45°	45°		Y			∞	1.95	1.0	.55	6.4	B	0°				2/11
292	BWC	1/4	1	6	1	0°	45°	45°		Y			∞	1.96	2.0	1.1	3.4	B	0°				2/11
293	BWC	1/4	1/2	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	Nozz Calib			
294	BWC	1/4	1/2	6	1	0°	45°	45°					∞	1.0	-	-	0		0°	WT TARE			
295	BWC	1/4	1/2	6	1	0°	45°	45°					∞	VARY	∞	∞	0	0°	0°	STATIC RUN			

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TEST 290
14.5 x 21.75 VSTL
FULL-SPAN
PROPULSIVE
WING

RUN NO	CONFIGURATION	W-NOZ SPAN	C-NOZ SPAN	WING POS	CAN POS	S_c	S_{F_w}	S_{F_c}		SCV DATA	DOWN WASH		H/c	N_{PR}	C_{μ_w}	C_{μ_c}	Q	α	β	REMARKS			
296	BWC	1/4	1/2	6	1	0°	45°	45°		Y			∞	1.95	.5	.3	12.4	B	0°				2/12
297	BWC	1/4	1/2	6	1	0°	45°	45°		Y			∞	1.95	1.0	.66	6.1	B	0°				2/12
298	BWC	1/4	1/2	6	1	0°	45°	45°		Y			∞	1.96	2.1	1.27	3.2	B	0°				2/12
299	BC	–	1/2	–	1	0°	–	45°					∞	1.0	–	–	0		0°	WT TARE			2/12
300	BC	–	1/2	–	1	0°	–	45°					∞	VARY	0	∞	0	0°	0°	STATIC RUN			2/12
301	BC	–	1/2	–	1	0°	–	45°					∞	1.96	0	.28	18.7	B	0°	PTS 1-8			2/12
301	BC	–	1/2	–	1	0°	–	45°					∞	1.96	0	.59	10.0	B	0°	PTS 9-15 (1 PT @ C_{μ_c} = .92)			2/12
303	BC	–	1/2	–	1	0°	–	45°					∞	1.94	0	1.09	4.5	B	0°				2/12
304	BW	1	–	6	–	–	45°	–					∞	1.8 + 2.4	∞	0	0	0	0°	PTOT/PSTN Calib			2/12
305	BW	1	–	6	–	–	45°	–					∞	1.0	–	–	0		0°	WT TARE N.G.			2/12
306	BW	1	–	6	–	–	45°	–					∞	1.0	–	–	0		0°	WT TARE OK			2/12
307	BW	1	–	6	–	–	45°	–					∞	1.5 + 2.2	∞	0	0	0	0°	STATIC RUN			2/12
308	BW	1	–	6	–	–	45°	–		151			∞	1.8	.5	0	20	0	B				2/12
309	BW	1	–	6	–	–	45°	–		–			∞	1.8	1.0	0	9.7	0	B				2/12
310	BW	1	–	6	–	–	45°	–		152			∞	1.8	2.0	0	5	0	B				2/12
311	BW	1	–	6	–	–	45°	–		147 148			∞	1.9	2.0	0	5	B	0°	See 145, 148			2/12
312	BW	1	–	6	–	–	45°	–		144 149			∞	1.8	1.1	0	9	B	0°	See 144, 149. for SCV			2/12
313	BW	1	–	6	–	–	45°	–		165			∞	1.8	.5	0	20	C	0°				

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15. Supplementary Notes

LANGLEY TECHNICAL MONITOR: J. W. PAULSON, JR.
Interim Report

16. Abstract

A full span model of a wing/canard concept representing a fighter configuration has been tested at STOL conditions in the NASA Langley 4 x 7 meter tunnel. The results of this test are presented, and comparisons are made to previous data of the same configuration tested as a semispan model. The potential of the propulsive wing/canard to develop very high lift coefficients was investigated with several nozzle spans (nozzle aspect ratios). Although longitudinal trim was not accomplished with the blowing distributions and configurations tested, the propulsive wing/canard appears to offer an approach to managing the large negative pitching moments associated with trailing edge flap blowing. Also presented are data showing the effects of large flap deflections and relative wing/canard positions. Presented in the appendix to the report are limited lateral-directional and ground effects data, as well as wing downwash measurements.

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